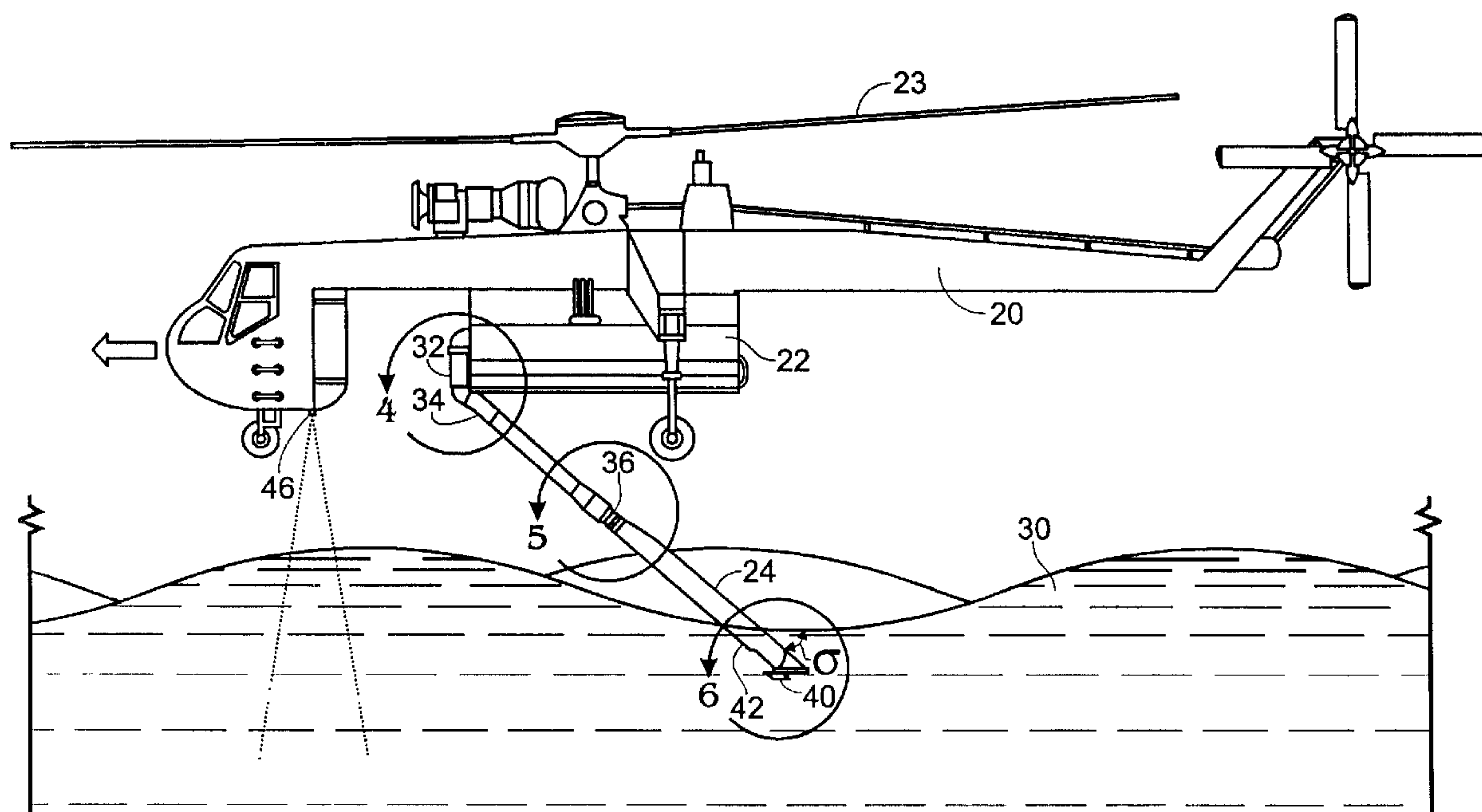




(86) Date de dépôt PCT/PCT Filing Date: 2001/03/09
(87) Date publication PCT/PCT Publication Date: 2001/09/20
(45) Date de délivrance/Issue Date: 2009/11/10
(85) Entrée phase nationale/National Entry: 2002/09/10
(86) N° demande PCT/PCT Application No.: US 2001/007657
(87) N° publication PCT/PCT Publication No.: 2001/069058
(30) Priorité/Priority: 2000/03/10 (US60/188,547)

(51) Cl.Int./Int.Cl. *B64D 1/22* (2006.01),
A62C 3/02 (2006.01), *B64D 1/00* (2006.01)
(72) Inventeurs/Inventors:
RAMAGE, LEE, US;
MACDONALD, ALAN B., US;
PARKER, RORY D., US
(73) Propriétaire/Owner:
ERICKSON AIR-CRANE INCORPORATED, US
(74) Agent: FETHERSTONHAUGH & CO.

(54) Titre : SYSTÈME DE CHARGEMENT DE FLUIDE
(54) Title: FLUID LOADING SYSTEM



(57) Abrégé/Abstract:

A pump system for loading fluid into a tank onboard a translating aircraft includes a substantially rigid tube member having a diving device on the distal end of the tube member. A strut structure may also be provided on a front face of the tube member. In use, the tube member is directed rearward relative to the translating aircraft so that splashing of fluid onto the underside of the aircraft is substantially avoided.



(12) INTERNATIONAL APPLICATION PUBLISHED UNDER THE PATENT COOPERATION TREATY (PCT)

(19) World Intellectual Property Organization
International Bureau



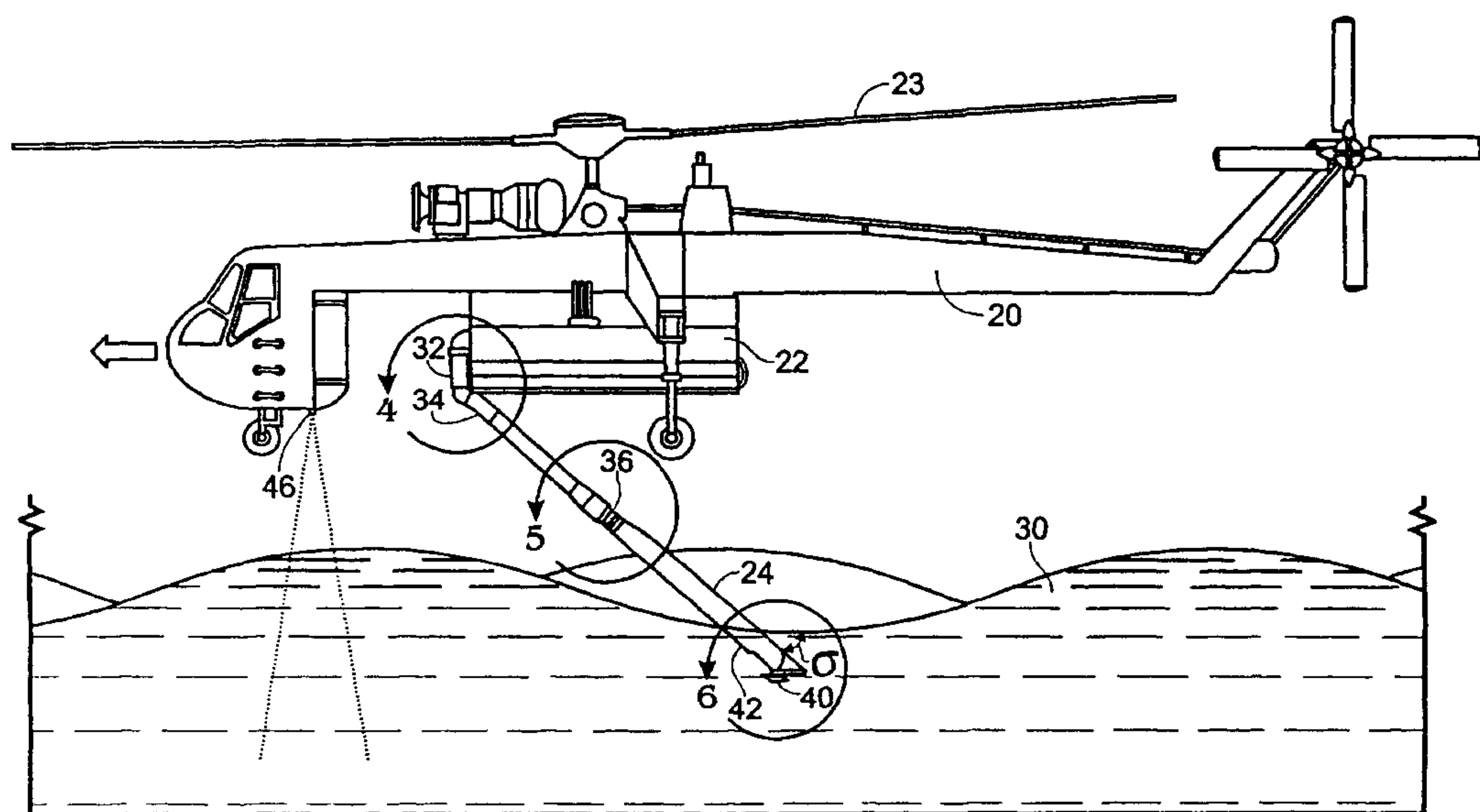
(43) International Publication Date
20 September 2001 (20.09.2001)

PCT

(10) International Publication Number
WO 01/69058 A2

- (51) International Patent Classification⁷: **F02B**
- (21) International Application Number: PCT/US01/07657
- (22) International Filing Date: 9 March 2001 (09.03.2001)
- (25) Filing Language: English
- (26) Publication Language: English
- (30) Priority Data:
60/188,547 10 March 2000 (10.03.2000) US
- (71) Applicant (for all designated States except US): **ERICKSON AIR-CRANE INCORPORATED** [US/US]; 3100 Willow Springs Road, Central Point, OR 97502 (US).
- (72) Inventors; and
- (75) Inventors/Applicants (for US only): **RAMAGE, Lee** [US/US]; 271 Black Oak Drive, Medford, OR 97504 (US). **MACDONALD, Alan, B.** [US/US]; 17230 Ford Road, Rogue River, OR 97537 (US). **PARKER, Rory, D.** [US/US]; 2055 Bradbury Street, Medford, OR 97504 (US).
- (74) Agents: **KOLISCH, J., Pierre et al.**; Kolisch, Hartwell, Dickinson, McCormack & Heuser, Suite 200, 520 S.W. Yamhill Street, Portland, OR 97204 (US).
- (81) Designated States (*national*): AE, AG, AL, AM, AT, AU, AZ, BA, BB, BG, BR, BY, BZ, CA, CH, CN, CO, CR, CU, CZ, DE, DK, DM, DZ, EE, ES, FI, GB, GD, GE, GH, GM, HR, HU, ID, IL, IN, IS, JP, KE, KG, KP, KR, KZ, LC, LK, LR, LS, LT, LU, LV, MA, MD, MG, MK, MN, MW, MX, MZ, NO, NZ, PL, PT, RO, RU, SD, SE, SG, SI, SK, SL, TJ, TM, TR, TT, TZ, UA, UG, US, UZ, VN, YU, ZA, ZW.
- (84) Designated States (*regional*): ARIPO patent (GH, GM, KE, LS, MW, MZ, SD, SL, SZ, TZ, UG, ZW), Eurasian patent (AM, AZ, BY, KG, KZ, MD, RU, TJ, TM), European patent (AT, BE, CH, CY, DE, DK, ES, FI, FR, GB, GR, IE, IT, LU, MC, NL, PT, SE, TR), OAPI patent (BF, BJ, CF, CG, CI, CM, GA, GN, GW, ML, MR, NE, SN, TD, TG).
- Published:**
— without international search report and to be republished upon receipt of that report
- For two-letter codes and other abbreviations, refer to the "Guidance Notes on Codes and Abbreviations" appearing at the beginning of each regular issue of the PCT Gazette.

(54) Title: FLUID LOADING SYSTEM



(57) Abstract: A pump system for loading fluid into a tank onboard a translating aircraft includes a substantially rigid tube member having a diving device on the distal end of the tube member. A strut structure may also be provided on a front face of the tube member. In use, the tube member is directed rearward relative to the translating aircraft so that splashing of fluid onto the underside of the aircraft is substantially avoided.

WO 01/69058 A2

FLUID LOADING SYSTEM

The invention relates to pump devices for loading fluid into a tank. In particular, the invention involves conduit devices that can load fluid onto an aircraft, either fixed wing or rotocraft, from a fluid source such as a lake, river, reservoir, or ocean while remaining airborne.

For many years, special aircraft have been designed and used to carry and deliver water to a fire. Helicopters or “helitankers” may be equipped with fire suppression systems that are capable of loading water from a source such as a lake or an ocean, into an onboard tank, and then transporting and dispensing the water onto a fire. Recently, the world-wide demand for helitanker fire fighting services has expanded. Helitanker fire fighting operations are now often required in areas where fresh water sources are scarce and salt water sources are abundant. Therefore, helitankers must be able to utilize sea water as a retardant without damaging aircraft components or compromising safety.

Accordingly, an object of the invention is to provide a system for loading fluid onto an aircraft from a salt water source without ingesting fluid into the aircraft engines.

Another object of the invention is to provide a conduit device for loading fluid into a tank on board an aircraft, in which the conduit device can remain partially stably submerged in the fluid source while the aircraft flies over the source at a ground speed sufficient to stay ahead of the spray generated by the rotor wash.

Another object of the invention is to provide a fluid loading system that may be versatily employed to load fluid from different types of sources, for example, oceans, lakes, rivers, and reservoirs, in varying shapes, depths, and degrees of salinity.

Summary of the Invention

The invention provides a fluid loading and transport system for an aircraft. A substantially rigid tube member is connected to the aircraft in communication with the tank. The tube member is movable into a downward orientation for accessing a fluid source while the aircraft flies over the source and may be retracted in times of non-use.

One aspect of the invention relates to a fluid conduit device having a distal end equipped with a diving device, preferably in the form of an inverted hydrofoil, that substantially maintains the distal end of the conduit device below a surface of the water source while the aircraft flies over the water source.

Another aspect of the invention provides a fluid conduit device in which a tube member has a strut running along at least a portion of the length of the tube member. The strut is configured to stabilize a submerged distal end of the tube member while the aircraft moves over a fluid source.

Still another aspect of the invention involves a fluid conduit device in which a tube member is connected to an aircraft in communication with an onboard tank so that the tube member maintains a substantially rearward angular orientation when the aircraft translates over the fluid source and the distal end of the tube member is submerged.

The invention also provides a method of loading water onto a moving aircraft. Fluid is transferred from a source through a tube member into a tank on an aircraft by a pump or ram pressure generated by the forward velocity of the aircraft as it moves over the source. In a preferred procedure, the tube member is directed rearward relative to the

aircraft's direction of travel. A distal portion of the tube member is maintained below the surface of the fluid source by providing a diving device near the distal portion.

Various embodiments of this invention provide a fluid conduit device comprising
a tube member having a proximal end and a distal end with a fluid inlet portion, the
5 proximal end being configured for connection to an aircraft in communication with an
onboard tank, the distal end having a diving device positioned below the fluid inlet portion
that substantially maintains the distal end of the tube member below a surface of a fluid
source while the aircraft translates over the fluid source.

Other embodiments of this invention provide a pump system comprising an aircraft
10 having an underside, a tank onboard the aircraft for containing fluid, a substantially rigid
tube member, connected to the aircraft in communication with the tank, wherein the tube
member is movable to a downward orientation rearward relative to the aircraft for accessing
a fluid source below the aircraft, and a flexible tube member connected to the tank
separately, from the rigid tube member, for loading fresh water while the aircraft hovers
15 over a source.

Other embodiments of this invention provide a fluid conduit device comprising a
substantially rigid tube member having a proximal end configured for connection to an
aircraft in communication with an onboard tank, wherein the tube member is movable into a
downward orientation for accessing a fluid source spaced away from the aircraft,
20 wherein the tube member has a distal end with a fluid inlet portion equipped with a diving
device positioned below the fluid inlet portion that substantially maintains the fluid inlet
portion of the tube member below a surface of a water source while the aircraft translates
over the water source. The diving device may be configured to substantially maintain a
rearward linear orientation of the tube member relative to the aircraft when the aircraft
25 translates over the fluid source.

Other embodiments of this invention provide a method of loading water onto an
aircraft comprising pumping fluid from a source through a tube member onto a tank on an
aircraft, while the aircraft moves over the source, and maintaining a fluid inlet portion on a
distal portion of the tube member below the surface of a fluid source by providing a diving
30 device positioned below the fluid inlet portion.

Other embodiments of this invention provide a pump system comprising an aircraft
having an underside, a tank onboard the aircraft for containing fluid, a substantially rigid

3a

tube member, connected to the aircraft in communication with the tank, wherein the substantially rigid tube member is movable to a downward orientation rearward relative to the aircraft for accessing a fluid source below the aircraft, and a flexible tube member connected to the tank in parallel with the rigid tube member, wherein the flexible tube member is movable to a downward orientation relative to the aircraft for accessing a fluid source below the aircraft.

Other embodiments of this invention provide a fluid conduit device for loading water onto an aircraft comprising a substantially rigid tube member having a proximal end and a distal end, the proximal end being configured for connection to an aircraft in communication with an onboard tank, the distal end having an inlet and a diving device with substantially planar shoulder members which extend outward and down on opposite sides of the inlet for substantially maintaining the distal end inlet below a surface of a fluid source while the aircraft translates over the fluid source.

Other embodiments of this invention provide a pump system comprising an aircraft having an underside, a tank onboard the aircraft for containing fluid, a substantially rigid tube member, connected to the aircraft in communication with the tank, wherein the substantially rigid tube member is movable to a downward orientation rearward relative to the aircraft for accessing a fluid source below the aircraft, and further wherein the substantially rigid tube member is rotatable around an axis that is substantially perpendicular to the surface of the fluid source.

3b

Brief Description of the Figures

Figure 1 is a side view of a helicopter equipped with a tank and associated fluid
5 conduit device for loading water in to the tank.

Figure 2 is a side view of a helicopter equipped with a tank and a fluid conduit
device consisting of the upper portion of the tube member shown in the system of Figure
1.

Figure 3 is a side view of a helicopter equipped with a tank and fluid conduit
10 device retracted against the underside of the helicopter in its non-use position.

Figure 4 is a partial, side view of the junction between the tube member and the
tank of the system shown in Figure 1.

Figure 5 is a partial, cut-away view of a central portion of the tube member
shown in the system of Figure 1.

Figure 6 is a partial, side view of the distal portion of the tube member in the
15 system of Figure 1.

Figure 7 is a partial, front view of a preferred embodiment of the invention
suspended from a tank on the underside of a helicopter.

Figure 8 is a perspective view of the diving device that is mounted on the distal
20 end of the conduit device shown in Figure 7.

Figure 9 is a partial, side view of an alternative embodiment of the invention
showing a modified intake design.

Figure 10 is a cross-sectional view through the conduit device of Figure 9.

Figure 11 is a partial front view of the distal end of the conduit device shown in
25 Figure 9.

Description of the Invention

The invention provides a system for loading fluid into a tank on an aircraft while
the aircraft translates over the surface of the fluid source. Tests have demonstrated that a
helicopter can fly at a loading level, for example, within about 10 feet of the water
30 surface, at a speed of approximately 25 knots or greater, and thereby substantially avoid
the ingestion of water into the engines. The in-flight loading system can be operated

stably and safely in a turbulent or wavy salt water source such as an ocean without splashing significant amounts of salt water onto the underside of the aircraft. In a preferred application of the invention, a diving device such as a hydrofoil structure is used to stabilize a submerged distal end of a conduit while loading.

5 Figures 1-6 illustrate details of a first embodiment of the invention. Figure 1 shows helicopter 20 which is a conventional helicopter in many respects, except that it has been equipped with a fluid loading and containing system according to the invention. Tank 22 is connected to a central portion of the helicopter substantially directly below the main rotor 23. A front view of a similar tank is shown in Figure 7. Substantially rigid
10 tube member 24, also referred to as conduit structure or snorkel, is connected to the front end of tank 22. A distal end of tube member 24 is submerged in fluid source 30. Tube member 24 is connected to tank 22 by horizontal rotating union 32 and a vertical swivel joint 34, further details of which are shown in Figure 4 and discussed below. Upper and lower portions of tube member 24 are connected in a central portion 36. The distal end of
15 tube member 24 has a diving device in the form of an inverted hydrofoil structure or other diving device 40 which helps to maintain tube member 24 in the fluid source. A streamlined strut 42 is formed in the front face of tube member 24 for decreasing drag when helicopter 20 is moving forward, and also to at least partially stabilize lateral movement of tube member 24. Helicopter 20 is equipped with an altimeter device 46 for
20 monitoring the relative height of helicopter 20 over the surface of fluid source 30. Information from the monitoring device is reported to the pilot/copilot in an appropriate instrument.

As shown in Figure 1, tube member 24 is directed rearward at a preferred angle θ of approximately 40° . Diving device 40 is configured to stabilize the distal end of tube
25 member 24 in the fluid at angle θ while helicopter 20 translates forward at approximately 40 knots. The system shown in Figure 1 substantially avoids splashing of water on the underside of the helicopter by moving forward at a sufficient velocity, preferably about 25 knots or higher, so that water that is splashed up by rotor downwash occurs primarily behind the moving helicopter.

30 Figure 2 shows a different application of the invention. Helicopter 20 and tank 22 are the same as illustrated and described in relation to Figure 1. Tube member 24 has

been shortened by detaching the bottom portion so that upper portion 48 of tube member 24 directly connects tank 22 to fluid source 50. Fluid source 50 may be a lake or reservoir that has a substantially calm or flat surface. The application shown in Figure 2 may be appropriate when fluid source 50 is relatively fresh water (negligible salt content) so that fluid loading may be performed while helicopter 20 hovers.

Figure 3 shows the same system of Figure 1 in its retracted position. Tube member 24 is pulled up and secured against the underside of tank 22 by a winch (not shown) onboard helicopter 20.

An enlarged view of the joinder mechanism between tube member 24 and tank 22 is shown in Figure 4. Horizontal rotating union 32 allows tube member 24 to rotate around an axis that is substantially perpendicular to the surface of the fluid source, and perpendicular to the direction of travel. Vertical swivel joint 34 permits tube member 24 to rotate around axis AA which is perpendicular to the figure. Vertical swivel joint 34 permits movement of tube member 24 between the use position, as shown in Figure 1, and the non-use position, as shown in Figure 3. Outer skin 60 of vertical swivel joint 34 is flexible to permit movement of the joint.

In Figure 5 the drawing is cut-away in central portion 36 of tube member 24 to show an optional pump device 64 that is contained inside tube member 24. When the helicopter translates forward with tube member 24 submerged in a water source, even without any assistance from pump device 64, a ram effect is observed which naturally pressures water from the fluid source up through tube member 24 into tank 22. Pump device 64 may be used to increase the rate of fluid loading.

Figure 6 shows the distal end of tube member 24. Diving device 40 is configured to retain and stabilize the desired orientation of tube member 24 in the fluid source as the helicopter translates forward. Water inlet 66 is provided on the front face of tube member 24 for receiving and channeling water into tube member 24. Internal volutes 68 are shown in dashed lines, and is provided to smoothly redirect in-flowing fluid up into tube member 24 toward tank 22.

A preferred embodiment of the invention is illustrated in Figure 7. Helicopter 100 has a tank 102 for carrying water from a source such as an ocean, to a fire. Two different conduit devices are connected to the front end of tank 102. Rigid conduit 104 is

connected to tank 102 on one side of the front end of tank 102. Conduit 104 is designed primarily for accessing fluid from a salt water or a fresh water source while in forward motion. Flexible hose conduit 106 is connected to the other side of the front end of tank 102, and is used primarily for accessing fluid from a fresh water source while helicopter 100 hovers. Rigid conduit 104 is substantially the same as the one described in relation to Figure 1, except that conduit 104 has no internal pump device located in mid-portion 110. When the distal end of conduit 104 is submerged and moving forward in the water at a velocity of, for example, 25 knots, a substantial ram effect pressures water up through conduit 104 into tank 102 without any additional pump mechanism. Experiments have shown that conduit 104 can load water into tank 102 at a rate of approximately 2,000 gallons in 28 seconds while translating forward at a velocity of approximately 35 knots, without any additional pumping mechanism.

The front view of conduit 104 in Figure 7 also illustrates the narrow front profile of lower portion 112 of conduit 104. The narrow front profile streamlines the portion of conduit 104 that is submerged in the water and minimizes drag or water resistance during loading. Intake 114 channels fluid into two internal tubes which extend on up through conduit 104. Diving device 116 is connected to the distal tip of conduit 104 in an orientation that maintains submersion of conduit 104 at a desired angle and depth while loading water into tank 102.

It is advantageous to provide two types of conduits or snorkels for maximum versatility. The rigid conduit is most desirable when loading fluid from a salt water source when it is particularly important to avoid ingestion of salt water into the helicopter engines. Other times it may be desirable to use the flexible conduit, for example, when loading from a fresh water source or a source that is not large enough to fly over while loading.

Figure 8 shows a perspective view of diving device 116. Diving device 116 has a pair of substantially planar shoulder members 120 which taper as they extend down and generally outward on opposite sides of conduit 104. Shoulder members 120 are then connected to lateral wing members 122. Three fastening flanges 124 may be provided on the top side of diving device 116 for connecting diving device 116 to conduit 104.

Figures 9-11 illustrate another embodiment of the invention in which the outer shape of the intake portion of the conduit is modified. Lower conduit portion 140 has a distal intake portion 142. A diving device 144 is connected to conduit 140 substantially as previously described. Internal tubes 146a and 146b present openings for receiving oncoming fluid as lower conduit portion 140 is submerged and translating forward. Figure 10 shows a cross-section through lower conduit portion 140 and internal tubes 146a and 146b. Figure 11 shows a front view of intake portion 142 where the openings of tubes 146a and 146b can be seen, in addition to the front profile of diving device 144.

Although the invention has been disclosed in its preferred forms, the specific embodiments thereof as disclosed and illustrated herein are not to be considered in a limiting sense, because numerous variations are possible. Applicants regard the subject matter of their invention to include all novel and nonobvious combinations and subcombinations of the various elements, features, functions, and/or properties disclosed herein. No single feature, function, element or property of the disclosed embodiments is essential. The following claims define certain combinations and subcombinations of features, functions, elements, and/or properties that are regarded as novel and nonobvious. Other combinations and subcombinations may be claimed through amendment of the present claims or presentation of new claims in this or a related application. Such claims, whether they are broader, narrower, equal, or different in scope to the original claims, also are regarded as included within the subject matter of applicants' invention.

WE CLAIM:

1. A fluid conduit device comprising
a tube member having a proximal end and a distal end with a fluid inlet portion, the
5 proximal end being configured for connection to an aircraft in communication with an
onboard tank, the distal end having a diving device positioned below the fluid inlet portion
that substantially maintains the distal end of the tube member below a surface of a fluid
source while the aircraft translates over the fluid source.
- 10 2. The device of claim 1, wherein the tube member is substantially rigid.
3. The device of claim 1 or 2, wherein the tube member has an upper portion
and a detachable lower portion.
- 15 4. The device of claim 1, 2 or 3, wherein the tube member has a front face, and
a strut structure formed on the front face for reducing drag when the aircraft translates over
the fluid source.
- 20 5. A pump system comprising
an aircraft having an underside,
a tank onboard the aircraft for containing fluid,
a substantially rigid tube member, connected to the aircraft in communication with
the tank, wherein the tube member is movable to a downward orientation rearward relative
to the aircraft for accessing a fluid source below the aircraft, and
25 a flexible tube member connected to the tank separately, from the rigid tube
member, for loading fresh water while the aircraft hovers over a source.
6. The pump system of claim 5, wherein the aircraft is a helicopter.
- 30 7. The pump system of claim 5, wherein the aircraft is an airplane.

8. The pump system of claim 5, 6 or 7, wherein the substantially rigid tube member forms an angle of approximately 40° with the underside of the aircraft when the substantially rigid tube member is in the downward orientation.

5 9. The pump system of any one of claims 5 to 8, wherein the substantially rigid tube member has a distal end and a diving device near the distal end for maintaining the distal end of the substantially rigid tube member in the fluid source while the aircraft moves over the source.

10 10. The pump system of any one of claims 5 to 9, wherein the substantially rigid tube member has a front face and a strut structure formed on the front face of the substantially rigid tube member for reducing drag when the aircraft moves over the source.

11. The pump system of any one of claims 5 to 10, further comprising an
15 altimeter device onboard the aircraft for monitoring the height of the aircraft over the water source.

12. The pump system of any one of claims 5 to 11, further comprising a pump
20 device substantially contained in the substantially rigid tube member for increasing the loading capacity of the pump system.

13. A fluid conduit device comprising
a substantially rigid tube member having a proximal end configured for connection to an aircraft in communication with an onboard tank, wherein the tube member is movable
25 into a downward orientation for accessing a fluid source spaced away from the aircraft,
wherein the tube member has a distal end with a fluid inlet portion equipped with a diving device positioned below the fluid inlet portion that substantially maintains the fluid inlet portion of the tube member below a surface of a water source while the aircraft translates over the water source.

14. The device of claim 13, wherein the diving device is configured to substantially maintain a rearward linear orientation of the tube member relative to the aircraft when the aircraft translates over the fluid source.

5 15. A method of loading water onto an aircraft comprising
pumping fluid from a source through a tube member onto a tank on an aircraft, while
the aircraft moves over the source, and
maintaining a fluid inlet portion on a distal portion of the tube member below the
surface of a fluid source by providing a diving device positioned below the fluid inlet
10 portion.

16. The method of claim 15, further comprising the step of providing a pump
device in a middle portion of the tube member for increasing the pumping rate through the
tube member.

15 17. A pump system comprising
an aircraft having an underside,
a tank onboard the aircraft for containing fluid,
a substantially rigid tube member, connected to the aircraft in communication with
20 the tank, wherein the substantially rigid tube member is movable to a downward orientation
rearward relative to the aircraft for accessing a fluid source below the aircraft, and
a flexible tube member connected to the tank in parallel with the rigid tube member,
wherein the flexible tube member is movable to a downward orientation relative to the
aircraft for accessing a fluid source below the aircraft.

25 18. The pump system of claim 17, wherein the flexible tube member is
configured to load fluid while the aircraft hovers over the fluid source.

30 19. The pump system of claim 17 or 18, wherein the substantially rigid tube
member does not access the fluid source while the aircraft hovers over the fluid source.

20. The pump system of claim 17, 18 or 19, wherein the substantially rigid tube member is configured to load fluid while the aircraft traverses over the fluid source.

5 21. The pump system of any one of claims 17 to 20, wherein the parallel flexible tube member does not access the fluid source while the aircraft traverses over the fluid source.

10 22. A fluid conduit device for loading water onto an aircraft comprising
a substantially rigid tube member having a proximal end and a distal end, the proximal end being configured for connection to an aircraft in communication with an onboard tank, the distal end having an inlet and a diving device with substantially planar shoulder members which extend outward and down on opposite sides of the inlet for substantially maintaining the distal end inlet below a surface of a fluid source while the
15 aircraft translates over the fluid source.

23. The device of claim 22, wherein the tube member has an upper portion, and a detachable lower portion.

20 24. A pump system comprising
an aircraft having an underside,
a tank onboard the aircraft for containing fluid,
a substantially rigid tube member, connected to the aircraft in communication with the tank, wherein the substantially rigid tube member is movable to a downward orientation
25 rearward relative to the aircraft for accessing a fluid source below the aircraft, and further wherein the substantially rigid tube member is rotatable around an axis that is substantially perpendicular to the surface of the fluid source.

Fig. 1

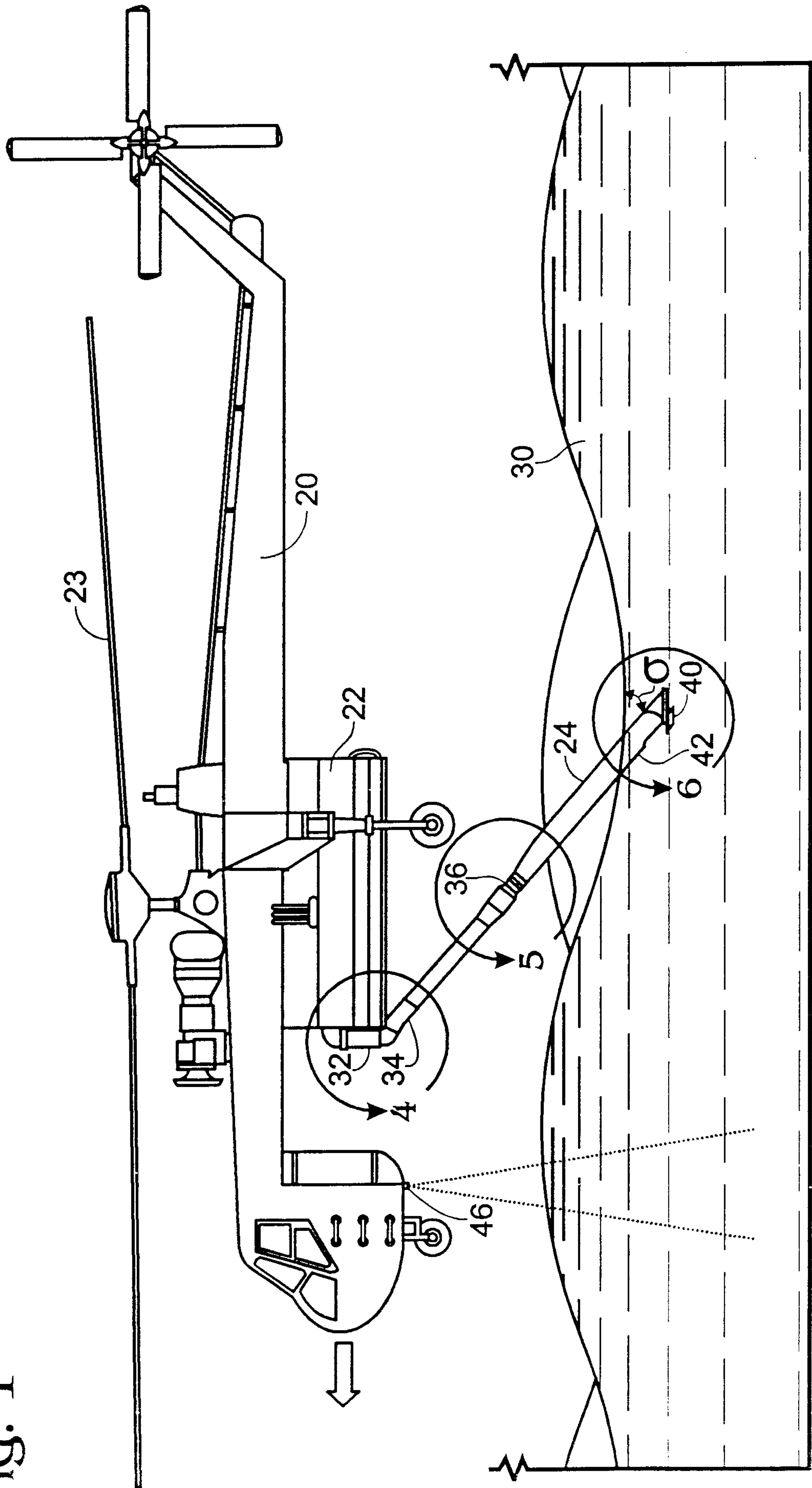


Fig. 2

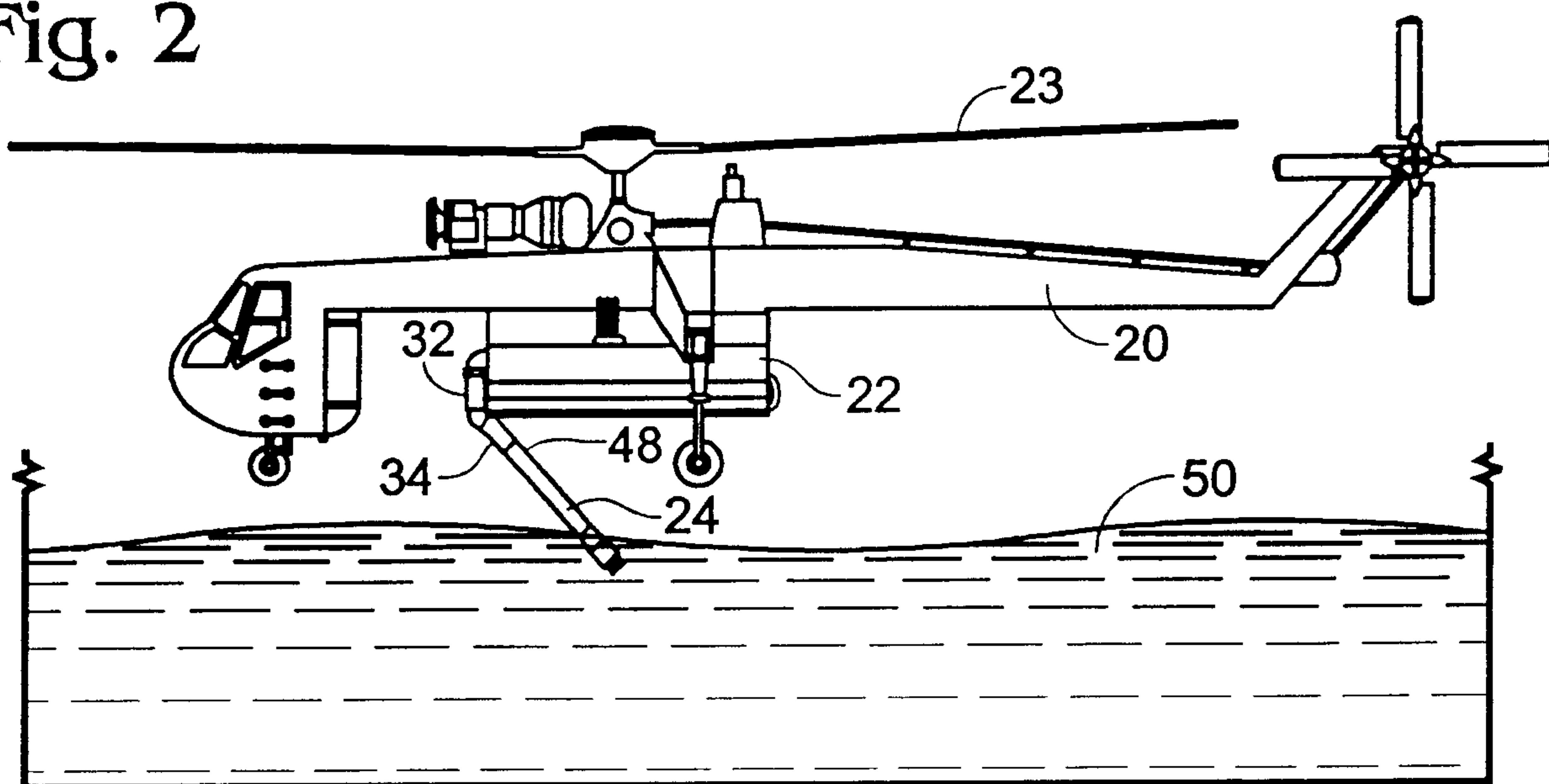


Fig. 3

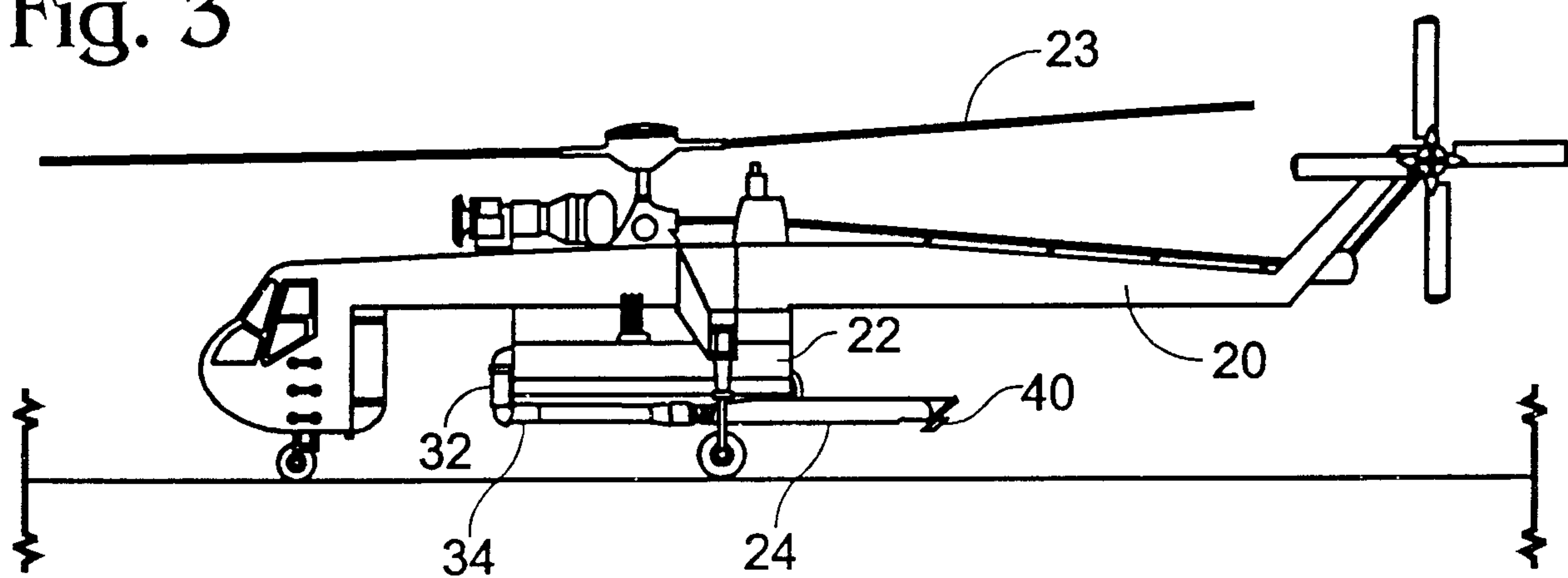


Fig. 4

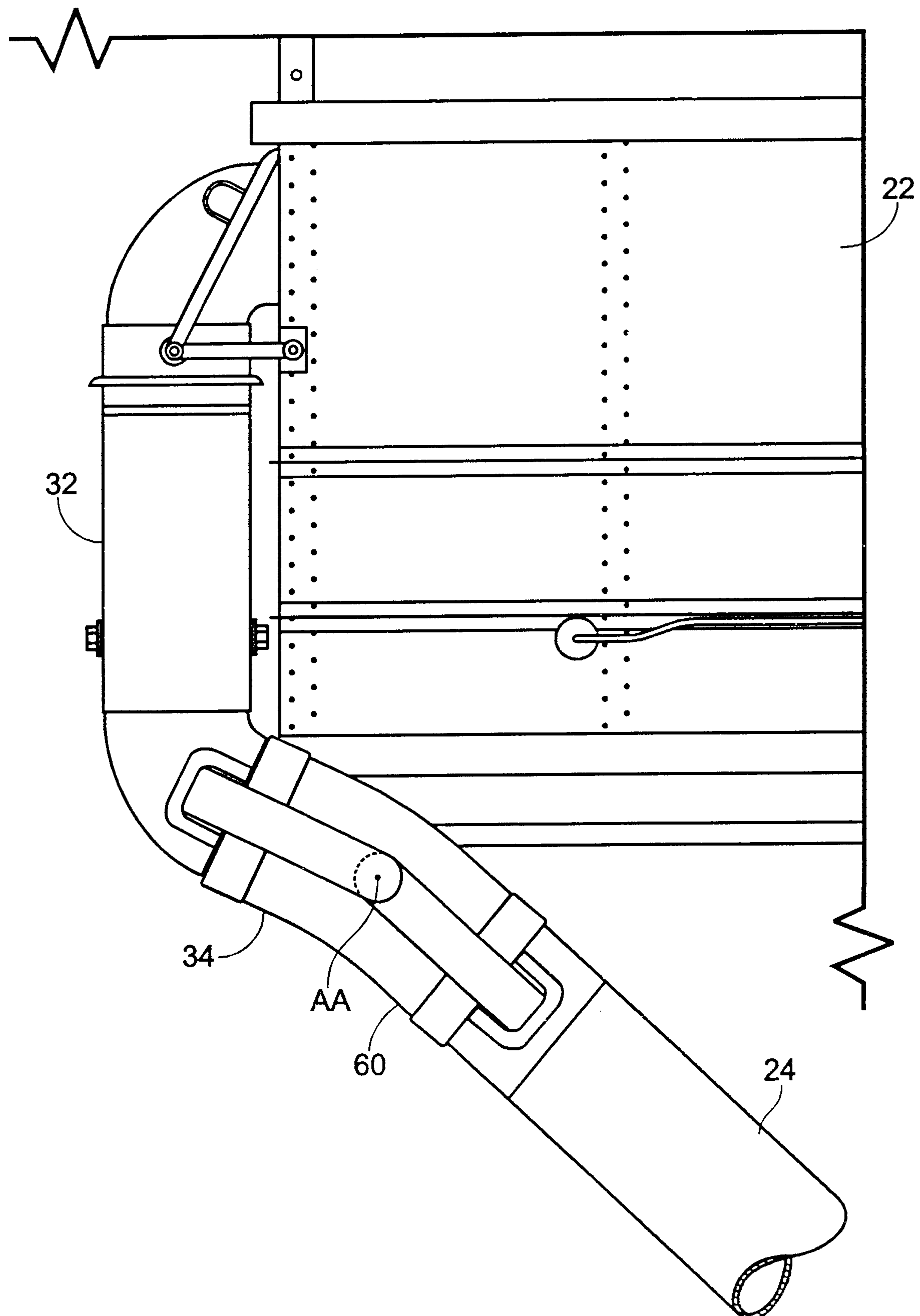


Fig. 5

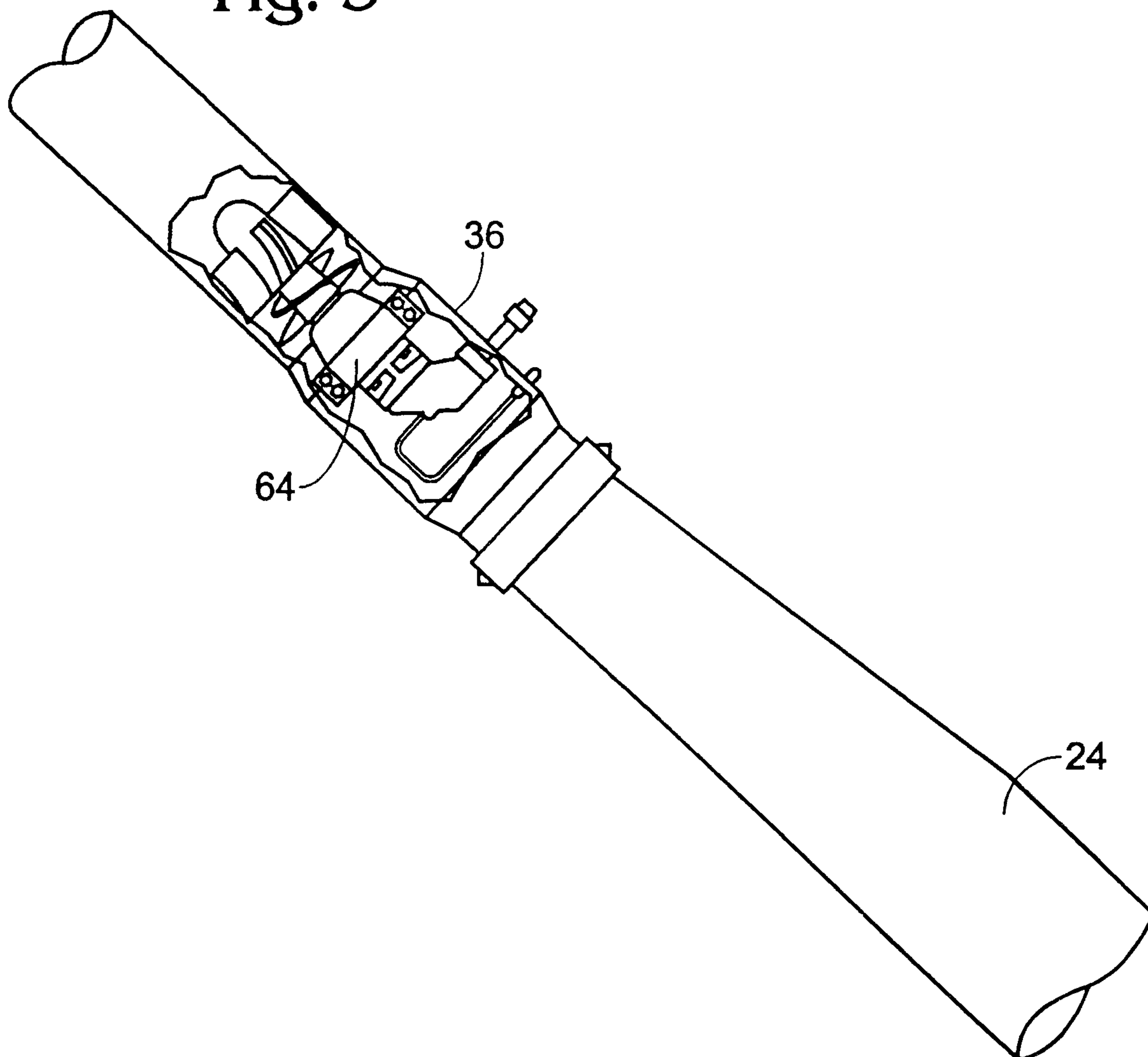


Fig. 6

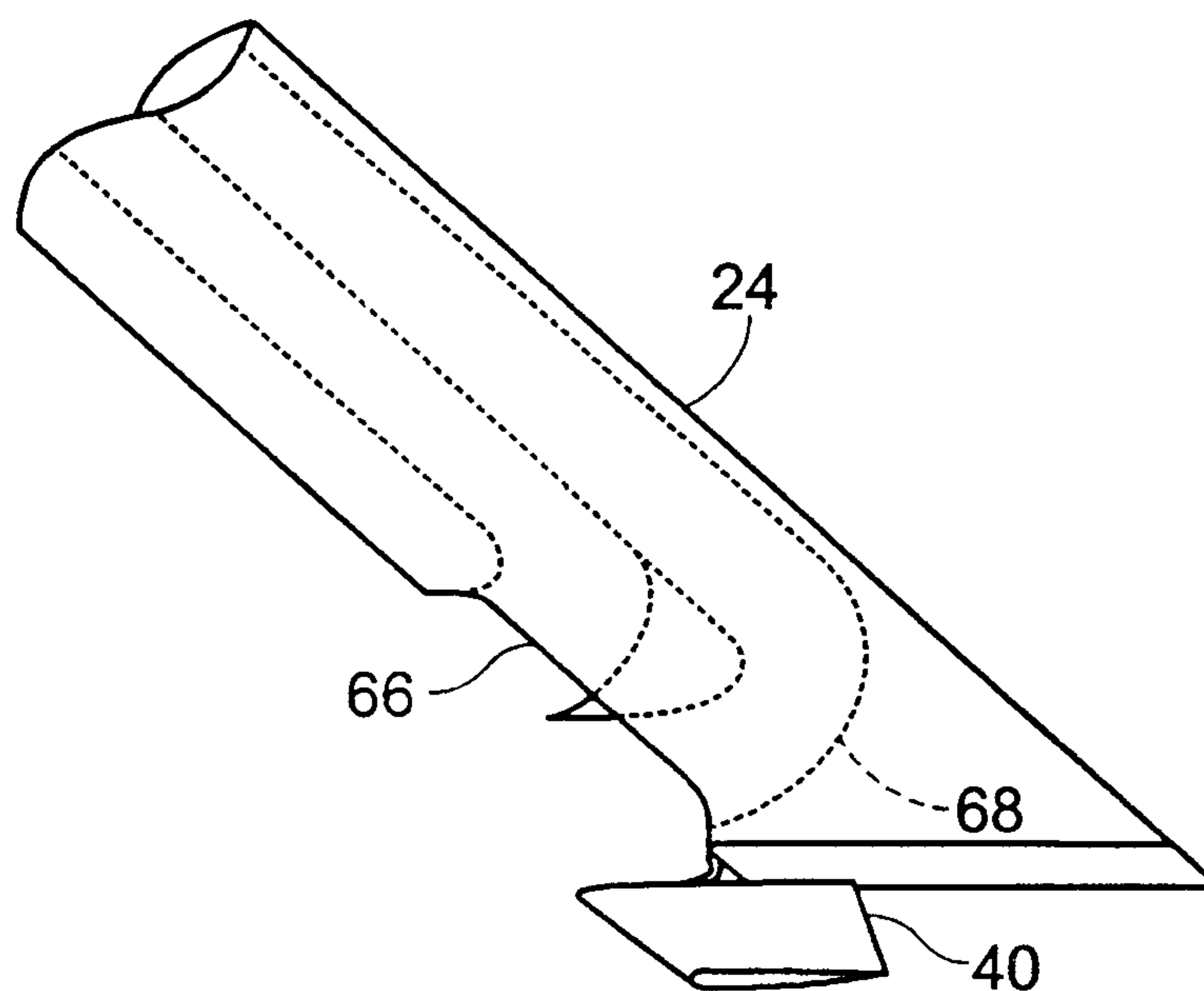


Fig. 7

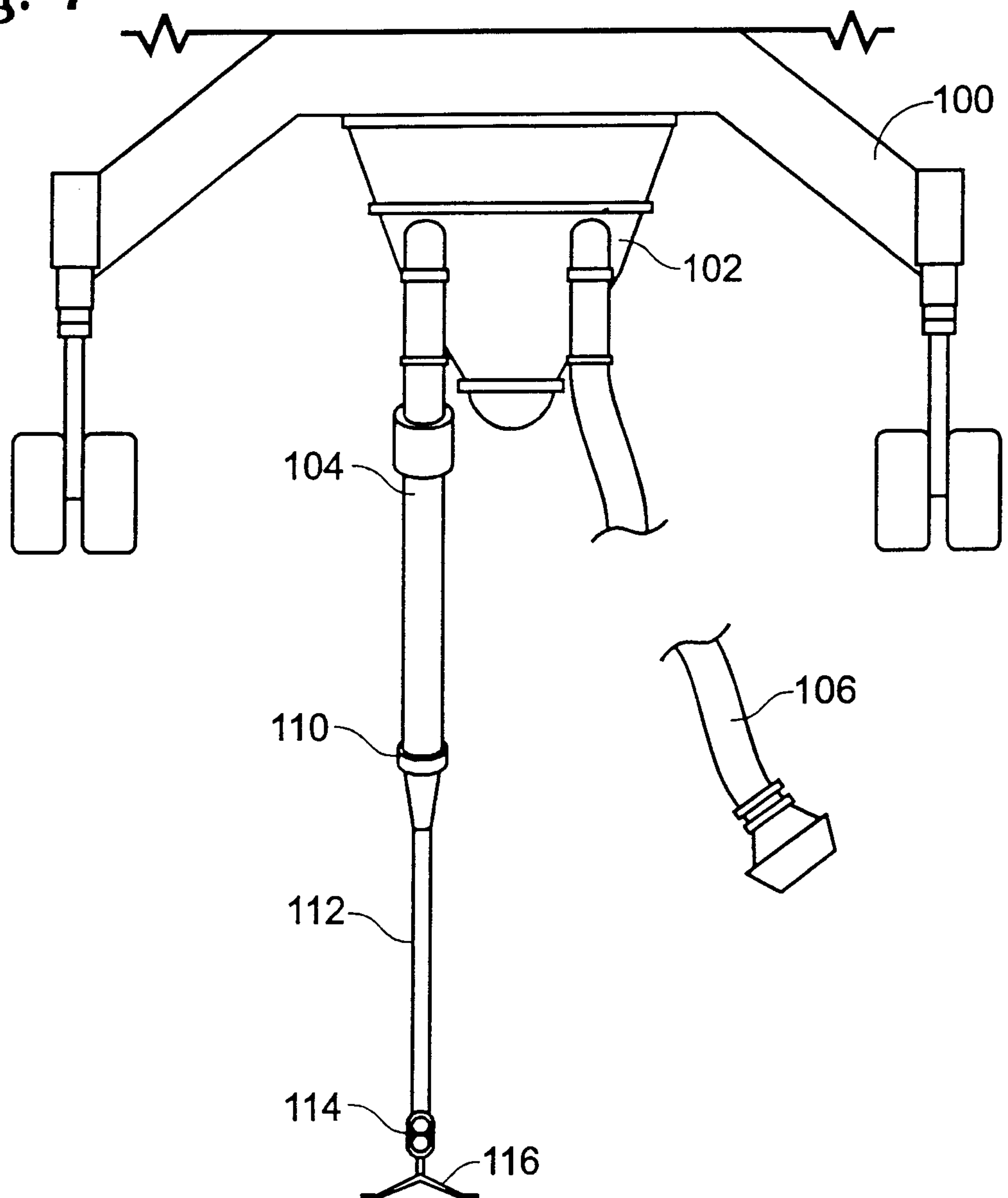


Fig. 8

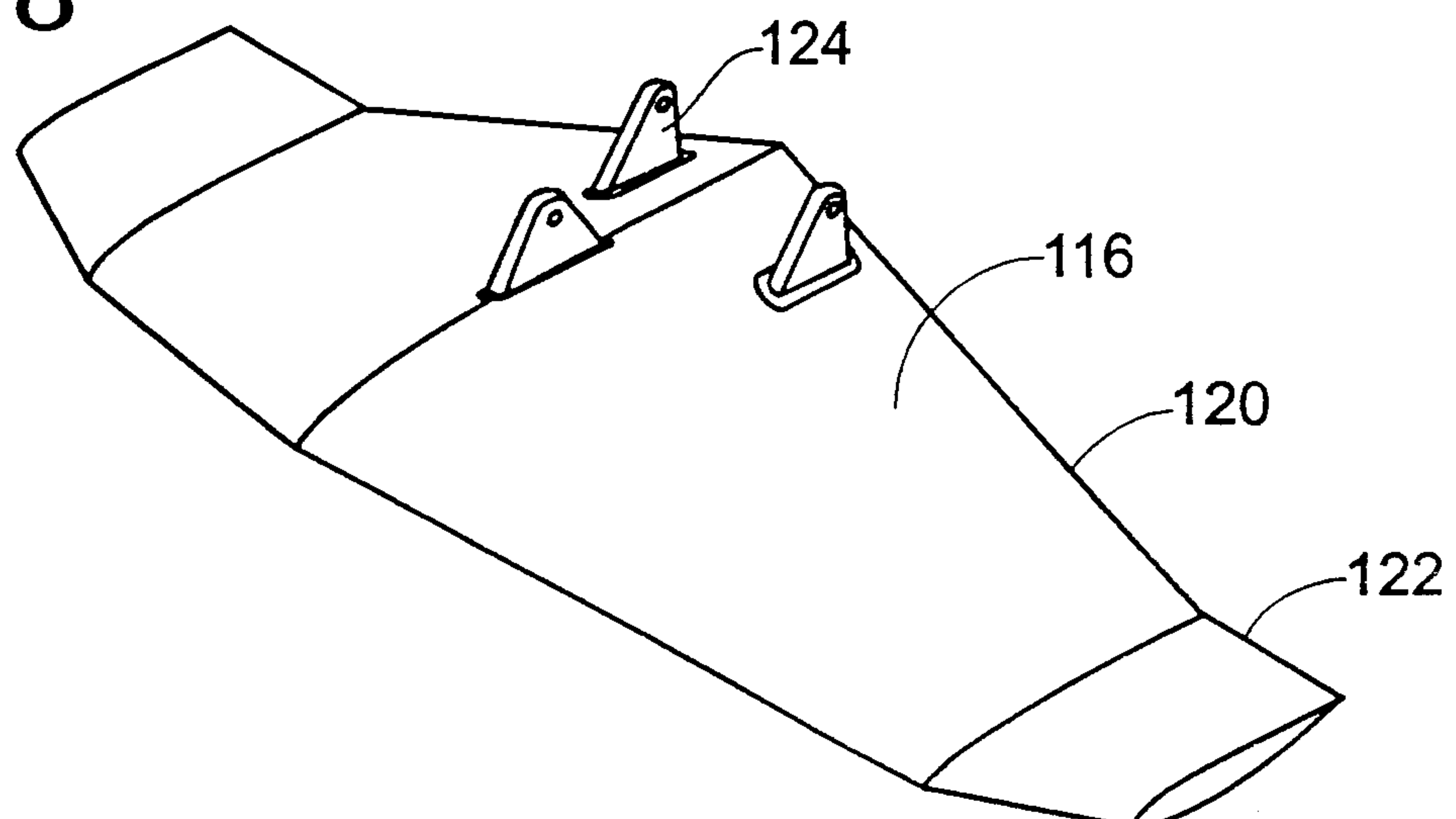


Fig. 9

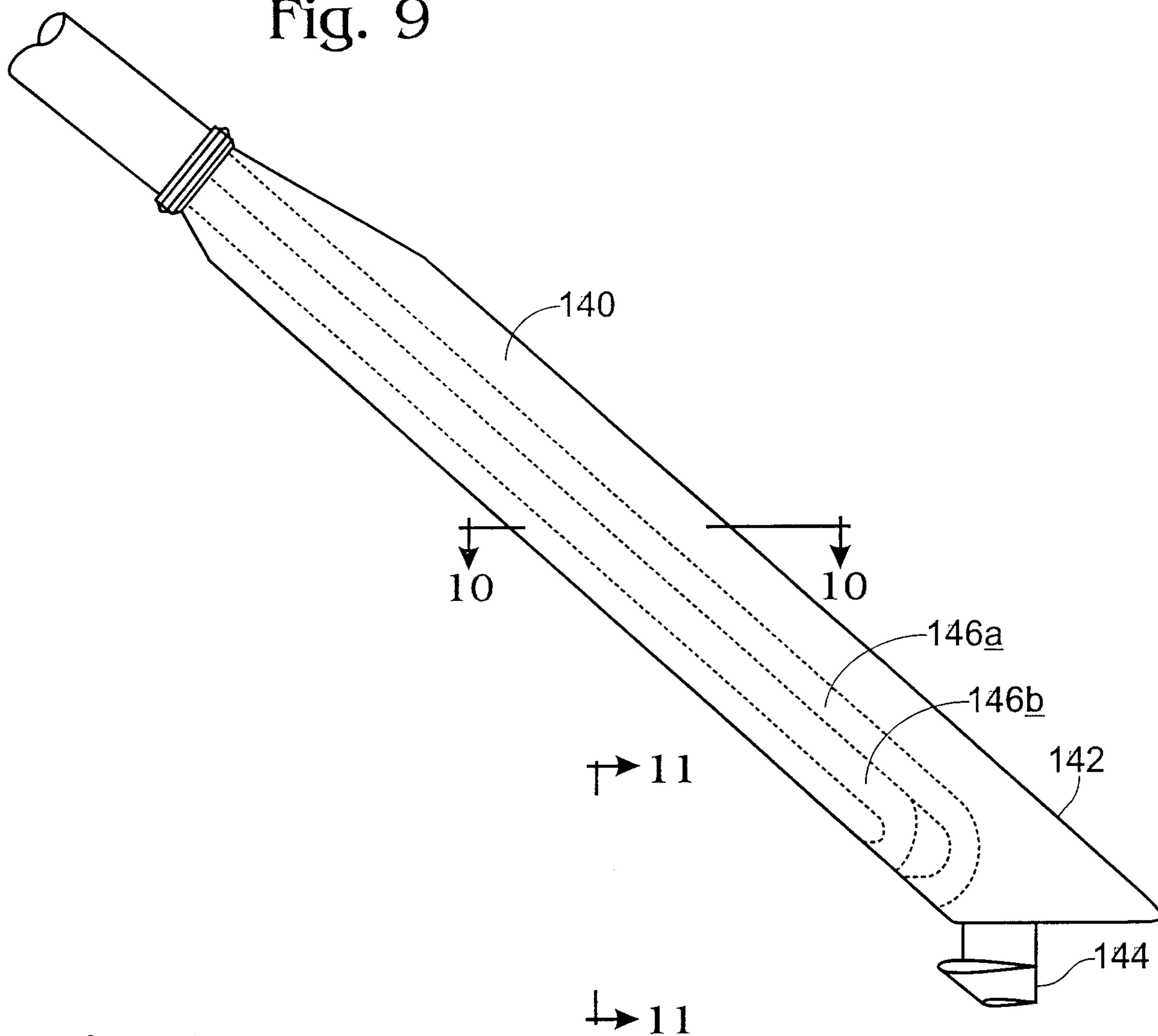


Fig. 10

