



(51) International Patent Classification:

F01D 5/28 (2006.01) F02C 3/16 (2006.01)
F01D 5/02 (2006.01) F23R 3/42 (2006.01)
F02C 3/04 (2006.01) F23R 3/56 (2006.01)

(21) International Application Number:

PCT/CA2017/051423

(22) International Filing Date:

27 November 2017 (27.11.2017)

(25) Filing Language:

English

(26) Publication Language:

English

(30) Priority Data:

15/361,351 25 November 2016 (25.11.2016) US

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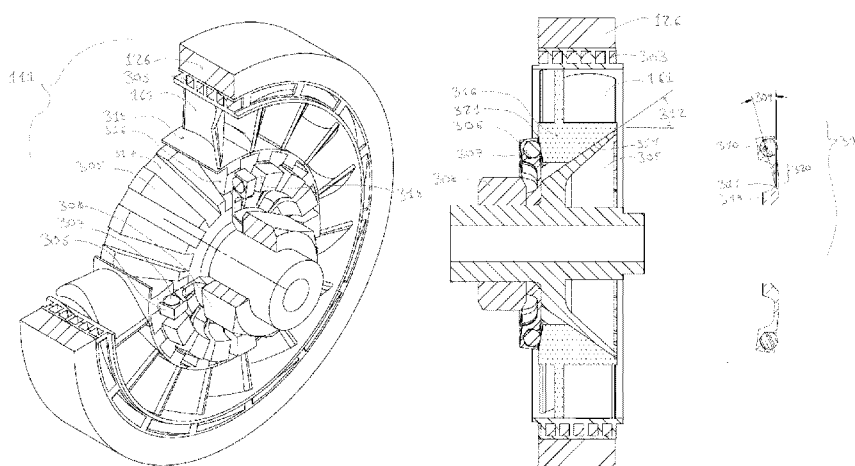
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(81) Designated States (unless otherwise indicated, for every kind of national protection available): AE, AG, AL, AM, AO, AT, AU, AZ, BA, BB, BG, BH, BN, BR, BW, BY, BZ, CA, CH, CL, CN, CO, CR, CU, CZ, DE, DJ, DK, DM, DO, DZ, EC, EE, EG, ES, FI, GB, GD, GE, GH, GM, GT, HN, HR, HU, ID, IL, IN, IR, IS, JO, JP, KE, KG, KH, KN, KP, KR, KW, KZ, LA, LC, LK, LR, LS, LU, LY, MA, MD, ME, MG, MK, MN, MW, MX, MY, MZ, NA, NG, NI, NO, NZ, OM, PA, PE, PG, PH, PL, PT, QA, RO, RS, RU, RW, SA, SC, SD, SE, SG, SK, SL, SM, ST, SV, SY, TH, TJ, TM, TN, TR, TT, TZ, UA, UG, US, UZ, VC, VN, ZA, ZM, ZW.

(84) Designated States (unless otherwise indicated, for every kind of regional protection available): ARIPO (BW, GH, GM, KE, LR, LS, MW, MZ, NA, RW, SD, SL, ST, SZ, TZ, UG, ZM, ZW), Eurasian (AM, AZ, BY, KG, KZ, RU, TJ, TM), European (AL, AT, BE, BG, CH, CY, CZ, DE, DK,

(54) Title: HIGH TEMPERATURE CERAMIC ROTARY TURBOMACHINERY

FIG. 19



(57) Abstract: The present disclosure generally relates to rotary turbomachinery methods and integrated processes requiring high-energy efficiency. In one embodiment, the present invention relates to rim-rotor configurations enabling long-term survival under conditions of either high temperature or oxidation resistance or saturated fluid abrasion.



EE, ES, FI, FR, GB, GR, HR, HU, IE, IS, IT, LT, LU, LV,
MC, MK, MT, NL, NO, PL, PT, RO, RS, SE, SI, SK, SM,
TR), OAPI (BF, BJ, CF, CG, CI, CM, GA, GN, GQ, GW,
KM, ML, MR, NE, SN, TD, TG).

Published:

— *with international search report (Art. 21(3))*

HIGH TEMPERATURE CERAMIC ROTARY TURBOMACHINERY

CROSS-REFERENCE TO RELATED APPLICATIONS

This patent application claims priority from U.S. Provisional Patent Application No. 5 62/166124 also titled "High G-field Combustion" on May 25th, 2015, and U.S. Patent Application No. 15/164642 also titled "High G-field Combustion" on May 25th, 2016.

FIELD OF THE INVENTION

The present disclosure relates to rim-rotor turbomachinery where the combustion 10 process predominantly takes place at high g-field forces and the turbine is radially supported by a composite, including carbon, reinforced rim-rotor that empowers the use of ceramics. Both technologies enable the increase of temperature including in a recuperated Brayton cycle to achieve high efficiency while maintaining low NOx levels. The rotary turbomachinery is perfectly suited for both compression and/or expansion in which severe conditions of high temperature, 15 high oxygen or other flammable fluids, and/or highly saturated fluids creating abrasion are present independent of any integral combustion process/methods, notably the inventive high g-field combustion.

BACKGROUND OF THE INVENTION

20 Due to a variety of factors including global warming issues, fossil fuel availability and environmental impacts, crude oil price and availability issues, alternative combustors with or without power generation methods must be developed to reduce carbon dioxide (CO2) and nitrogen oxides (NOx) emissions.

When considering power generation cycles such as the recuperated Brayton cycle, it is 25 recognized in the art that increasing cycle efficiency requires increasing combustion

temperature, yet it is also known that increasing combustion temperature is accompanied by an increasing challenge of maintaining NO_x emissions below environmental requirements. Typical gas turbines use lean premixed combustion to minimize the maximum flame temperature within the combustor and hence reduce NO_x emissions. However, for recuperated cycles these combustors are limited to air preheat temperatures below the autoignition temperature of the fuel-air mixture to avoid instabilities which can ultimately lead to catastrophic failure of the combustor. Lean premixed combustors are thus restricted to lower recuperated cycle temperatures, lower cycle efficiencies and higher carbon dioxide emissions.

Mobile applications require compactness, minimal weight and volume.

Furthermore, another challenge with increasing the temperature of a recuperated Brayton cycle lies in the turbine itself, where typical alloys require large amounts of cooling to be able to withstand high gas temperatures. This is even more challenging for small scale turbines (< 1 MW) where film cooling is very hard to implement and significantly reduces cycle efficiency. Attempts have been made to use ceramics, such as Silicon Nitride and Silicon Carbide, for gas turbines since these materials can withstand very high temperature, but due to their brittleness they show reliability issues. Prior attempts have been made to build ceramic turbines contained in a rim-rotor, such as U.S. Patent No. 4017209, but do not propose a viable cooling solution for the composite rim-rotor, which is limited by glass transition for carbon-polymer composites, or oxidation for carbon-carbon composites. In this specific case, cooling air goes through long slender blades operating beyond 1200 C, meaning the air is inevitably pre-heated, and thus, unless massive mass flows are used, cannot perform any meaningful cooling to a composite rim-rotor having a maximum operating temperature in the 250-350C range, making the approach useless for high-efficiency applications. These attempts have also been limited to purely axial turbine designs, which do not take full advantage of the rim-rotor that could be used for hub-less designs allowing inversed radial, axial or mixed flow configurations that optimize the temperature distribution of the engine packaging by keeping the hot gases on one single side of the turbine wheel, therefore separating structural and thermal loops.

Furthermore, when considering rim-rotor machinery, there is a significant challenge in matching the large displacement of the rim-rotor to the small displacement of a rigid hub. The rim-rotor also needs to be thermally insulated from the hot combustion gases, with ceramics

being a choice candidate due to their low conductivity and high temperature resistance. Prior art exists showing attempts to design and build flexible, compliant hubs for rim-rotor machinery as well as thermal protection layers for the rim-rotor. Some of this prior art has been limited to conceptual designs with no experimental validation (GE, Stoffer 1979), or component failure
5 during experimental validation (R. Kochendorfer 1980). These designs failed due to tensile loading of ceramics components under circumferential stress, and hence an improper use of the rim-rotor design to reduce, or even eliminate, the tensile stresses.

Accordingly, there is a need for a compact, low NOx combustor that can operate at high air preheat temperatures without the risk of instabilities or failures, that could be used in
10 industrial (furnaces, heaters) and power applications such as distributed CHP, aerospace and automotive applications. For maximum efficiency and emissions benefits in power applications, this combustor would need to be used with rim-rotor ceramic turbomachinery allowing high combustion temperatures, and hence high cycle efficiency.

15 **SUMMARY OF THE INVENTION**

In a first aspect, the present disclosure provides a high g-field combustor whose embodiment can be in a static, rotating or otherwise accelerating reference frame. The combustor comprises fuel injection sites, flame-holding (or flame-stabilizing) devices, means of igniting the fuel-air mixture and means of generating a high g-field.

20 In a second aspect, the present disclosure provides a gas turbine configuration that uses a rim-rotor configuration to allow the use of ceramics under compression. The rim-rotor turbine comprises a high-strength composite rim-rotor, ceramic or high temperature insulating layer, ceramic or high temperature alloy aerodynamic blades, and a radially flexible hub.

25 In accordance with an embodiment, there is provided a rim-rotor turbine rotating assembly comprising: a hub adapted to be mounted or connected to a rotating shaft; a rim-rotor; and a plurality of blades with each of the plurality of blades contacting the rim-rotor, each of the plurality of blades operatively connected to the hub by a sliding contact with the hub allowing at

least a radial motion between the blades and the hub to compensate for rim-rotor radial deformation under centrifugal loads; wherein the blades are ceramic blades.

Further in accordance with the embodiment, the blades are for instance axial-flow blades with the fluid flow entering generally axially and leaving generally axially.

5 Still further in accordance with the embodiment, the blades are for instance radial-flow blades, in which the fluid flow enters generally radially and leaves generally axially on a same side as the fluid flow has entered.

10 Still further in accordance with the embodiment, the fluid flow enters for instance generally radially at an angle ranging between 5 and 45 degrees inclusively from a radial plane of the rim-rotor.

Still further in accordance with the embodiment, the sliding contact is for instance defined by complementary sliding surfaces between the hub and the blades, wherein the complementary sliding surfaces are for instance at an angle ranging between 15 to 75 degrees inclusively from a rotational axis in a radial – axial plane.

15 Still further in accordance with the embodiment, the complementary sliding surfaces are for instance at an angle of 35 degrees from the rotational axis in the radial – axial plane.

Still further in accordance with the embodiment, at least two blades are joined for instance together into a single part having a common inner shroud and a common root for at least two airfoils.

20 Still further in accordance with the embodiment, a shape of the blade roots complementarily matches for instance a shape of hub surfaces to create a complementary male-female engagement creating the sliding contact.

Still further in accordance with the embodiment, the shape of the blade roots shape is for instance one of a rectangle, an ellipse, a dovetail, and a fir-tree.

25 Still further in accordance with the embodiment, at least one spring biases for instance the plurality of the blades in sliding contact with the hub against the rim-rotor.

Still further in accordance with the embodiment, the at least one spring applies for instance at least one contact force on blades at their roots.

Still further in accordance with the embodiment, the at least one spring applies for instance a force on the blades that is generally perpendicular to the rotational axis, the at least one spring being radially under each of the blades.

Still further in accordance with the embodiment, contact forces are for instance applied on blades at their tips by the at least one spring.

Still further in accordance with the embodiment, the at least one spring includes for instance a plurality springs each applying an independent biasing force on a respective one of the blades.

Still further in accordance with the embodiment, the at least one spring is for instance at least one of a leaf spring, a disc spring, a coil spring, a wounded spring, an air spring, and a fingered dome spring.

Still further in accordance with the embodiment, an insulation material is for instance between the at least one spring and the blades.

Still further in accordance with the embodiment, the insulation material is for instance at one of a ceramic coating, a low conductivity ceramic pad or a ceramic ball.

Still further in accordance with the embodiment, axial contact forces on the blades are for instance provided by using a radial centrifugal force on a spring head converted to an axial force by locating the spring head further away from the hub than a spring attachment point on the shaft.

In accordance with another embodiment of the present disclosure, there is provided a rotary turbomachine comprised of a rim-rotor having a rim-rotor radius, a blade mounting assembly and a shaft in the radial-axial plane, whereby the rim-rotor has a radial deformation, whereby the blade mounting assembly is comprised of at least two blades, at least two blade roots and a filled hub, whereby the blade mounting assembly enables relative motion between the at least two blade roots and the filled hub during operation with a radial displacement of the at least two blades by an amount from 1 micron to the radial deformation of the rim-rotor.

BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 is a schematic view illustrating the principal of operation of a high g-field combustor.

5 **FIG. 2** is a cutaway view of an embodiment of a rotating high g-field combustor using a rim-rotor to support the various flame-holders required.

FIG. 3 is a schematic view illustrating the various fuel injection points possible within a high g-field combustor either in a static or rotating configuration.

10 **FIG. 4** is a schematic view illustrating how combustion can be fully completed within the high g-field combustor or only partially within the high g-field combustor and the remainder of the combustion reaction fully completed in a pressure expansion device to achieve isothermal expansion.

FIG. 5 is a schematic view of a static high g-field combustor.

15 **FIG. 6** is a schematic view of a possible embodiment of a high g-field combustor used for a gas turbine application in which the curvature radius of the high g-field combustor can be different than the radius of the turbine, allowing optimum configuration of flow velocity and g-field.

FIG. 7 is a schematic cut view of a high g-field combustor used with a turbine.

20 **FIG. 8** is a CFD result showing NO_x concentration contours in ppm in a channel submitted to a large g-field of 100,000 g's in which fuel is injected from the top into a hot air at 1000 K stream and combustion fully takes place within the channel.

FIG. 9 is a CFD result showing Temperature contours in degrees K in a channel submitted to a large g-field of 100,000 g's in which fuel is injected from the top into a hot air at 1000 K stream and combustion fully takes place within the channel.

FIG. 10 is a CFD result showing NO_x concentration contours in ppm in a channel submitted to a small g-field of 10,000 g's in which fuel is injected from the top into a hot air at 1000 K stream and combustion only partially takes place within the channel.

FIG. 11 is a CFD result showing Temperature contours in degrees K in a channel submitted to a small g-field of 10,000 g's in which fuel is injected from the top into a hot air at 1000 K stream and combustion only partially takes place within the channel.

FIG. 12 is an embodiment of a high g-combustor used for high radiant applications.

FIG. 13 is a schematic view of a rotating high g-field combustor in which stator guide vanes can be individually closed to control the mass flow rate to the rotating combustor.

FIG. 14 is a schematic view of a rotating high g-field combustor in which stator guide vanes can be individually oriented to control tangential velocity of the flow to the rotating combustor.

FIG. 15 is an embodiment of combining a static high g-field combustor with a ceramic rim-rotor turbomachine.

FIG. 16 is an embodiment of a rim-rotor turbomachine using a rim-rotor, counter-flux insulation substrate, ceramic blades and a compliant hub.

FIG. 17 is a schematic view of a counter-flux insulation substrate consisting of different cooling channel configuration.

FIG. 18 is an embodiment of different counter-flux insulation substrates which can be manufactured using additive manufacturing methods.

FIG. 19 is an embodiment of a rim-rotor turbomachine using a rim-rotor, insulation substrate, ceramic blades with a sliding plane below the root, a matching sliding plane hub and a spring-mass retainer.

FIG. 20 is an embodiment of a rim-rotor turbomachine using a rim-rotor, insulation substrate, ceramic blades with a sliding plane below the root, a matching sliding plane hub and a spring loaded retainer.

FIG. 21 is a variation of the blade root geometry for the sliding plane hub, including dovetail, rectangle, ellipse, fir-tree shapes.

FIG. 22 is an embodiment of a rim-rotor turbomachine using a rim-rotor, insulation substrate, ceramic blades with a pin-and-socket relative motion below the male root (pin) and a female receptacle (socket) in the hub .

FIG. 23 is a schematic view of various rim-rotor turbomachine configurations that isolate the hot combustion gas from critical turbomachine components (shaft, bearings), essentially allowing “hubless” turbines.

FIG. 24 is an embodiment of a rim-rotor turbomachine in a reversed mixed flow configuration using an outer section of split blades and an inner section in a single part, where the rim-rotor is in structural communication with the shaft by a hub with an array of radially compliant springs.

FIG. 25 is an embodiment of a rim-rotor turbomachine in a reversed mixed flow configuration using an outer section of split blades and an inner section in a single part, where the rim-rotor is in structural communication with the shaft by a thin hub made of orthotropic material.

FIG. 26 is an embodiment of a rim-rotor turbomachine in a reversed mixed flow configuration using an outer section of split blades and an inner section in a single part, where the rim-rotor is in structural communication with the inner section through the blades that are maintained in contact with a sliding plane by a rear spring mechanism.

FIG. 27 is a variation in the outer section blade where it is split into an insulation substrate, a rear wall flow guide and a blade.

FIG. 28 is a mixed flow turbine inlet velocity triangles that can be used with reversed mixed flow turbines.

FIG. 29 is a highly compact turbomachine design where a centrifugal compressor is mounted back-to-back with the reversed mixed flow turbine.

FIG. 30 is a schematic illustration of a concept of the rim-rotor of **FIG. 19**.

FIG. 31 is a schematic illustration of a concept of the rim-rotor of **FIG. 26**.

FIG. 32 is a schematic illustration of a concept of the rim-rotor of **FIGS. 24-25**.

FIG. 33 is a schematic illustration of a concept of the rim-rotor of **FIG. 22**.

FIG. 34 is a schematic illustration of a concept of the rim-rotor of **FIG. 24-25**.

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DETAILED DESCRIPTION OF THE INVENTION

Traditional flame propagation mechanism in combustion reactions is driven by turbulent mixing, buoyancy forces between reactants and products, and species diffusion. Under normal low g-field conditions, the buoyancy forces are very small and do not significantly contribute to the flame propagation. However, at high g-fields (at a minimal embodiment of g-field greater than 10,000 g's in which buoyant forces obtain meaningful mixing, and at the preferred embodiment of g-field greater than 100,000 g's in which buoyant forces are dominant), buoyancy forces between combustion products and reactants (or fuel and air) dominate the flame propagation mechanism by greatly increasing the Rayleigh-Taylor instabilities between the fluids, improving mixing between products and reactants and hence increasing the heat release rates. High g-field is the key element for fast mixing and thus short reaction distances and residence times. Furthermore, it is expected that a high g-field rotating combustor would be most beneficial for small scale turbomachinery (< 1MW) because for a given turbine tip speed the g-field is inversely proportional to the machine radius. This results in g-fields in the 100,000 g's for turbines in the 10cm scale, and in over 1,000,000 g's for turbines in the 10mm scale.

Conventional turbines normally use internally supported blades, i.e. blades that are supported at their root connected to a hub whose diameter is smaller than the root radius of the blade. Such configurations result in the blades being loaded under tensile stress due to the centrifugal forces occurring during rotation, which limits the blades to being made of materials having high tensile strength. These materials are typically metallic alloys that are limited to relatively low temperatures. High temperature materials such as ceramics cannot be used in conventional turbines due to their low tensile strength and high brittleness: any small crack present in the blade will rapidly grow and lead to failure of the turbine due to the tensile loading

of the blades. To increase the efficiency of a recuperated Brayton cycle, it is desirable to increase the turbine inlet temperature to levels significantly higher than the maximum allowable blade temperature for metallic alloys. Conventional turbines can achieve this by using blade cooling strategies, but the manufacturing difficulties and efficiency penalties limits its use to large-scale turbines (> 1MW). For small scale turbines, the only viable approach to increase the turbine inlet temperature is to use ceramic blades, which is only possible using a rim-rotor turbine by holding the blades on their outer radius, the blades are loaded in compression which inhibits crack growth in ceramics. A rim-rotor turbine thus greatly increases the reliability of ceramic blades, which allows an increase in turbine inlet temperatures without the added complexity and cost of blade cooling.

Example embodiments will now be described more fully with reference to the accompanying figures.

FIG. 1 depicts a high g-field combustor where the combustion reaction is driven by buoyant forces caused by a centrifugal acceleration field and the density difference between combustion products **101** and premixed reactants **102**. The g-field can also be used for the mixing of air **158** and fuel **157** to achieve a homogeneous reactants mixture **102** before ignition: fuels heavier than air could be injected on the inner radii and lighter fuels on the outer radii to provoke mixing. High g-field is the key element for fast mixing, short reaction distances and short residence time. In the various contexts of use of a combustor, both cold and highly pre-heated air inlet situations exist. For cold air inlet (below fuel auto-ignition temperature), both non-premixed and premixed combustion need a flame holding device **103** and an ignition device that can be within the g-field **159** or external to the g-field **195**. Prior art (US 2014/0290259 A1) has shown that ignition devices such as glow plugs, spark plugs, sparkles or pilot flames can be used. For hot air inlet (above fuel auto-ignition temperature), only non-premixed combustion is possible and has no need for a flame holding device **103** nor an internal ignition device **159** within the combustor. A flame holding device **103** and internal ignition device **159** may optionally be used for start-up modes until the temperature exceeds the fuel auto-ignition temperature when using a heat-exchanger to pre-heat the combustion air from the exhaust gases.

FIG. 2 is an embodiment of a high g-field combustor in a rotating configuration **108**. The g-field is imposed by rotating the combustor around an axis, driving the heavier reactants **102** or air **158** outwards of the rotating axis and the hot combustion products **101** or fuel **157** inwards of the rotating axis. Multiple flame-holding devices such as an upper flame-holder **104**, a vertical flame-holder **105** and a lower flame-holder **106** in reference to the g-field direction can be used to stabilize the flame within the combustor. The rotating combustor **108** can be operable with either premixed cold air inlet, non-premixed cold air inlet or non-premixed hot air inlet, with the fuel injection for non-premixed configuration occurring either just before or within the rotating frame.

The location of the chemical reaction within the combustor can be shaped by carefully placing multiple injection points in each flow channel of the main combustor. **FIG. 3** shows the multiple location scenarios including: A) injection points within the rotating combustor (e.g. from the blades surface **112**, from the shroud **113** inner surface **114**, from the flame-holders **103** or from the hub **115** outer surface **116**, with fuel being transferred to the rotating frame either from the hub or the shroud), B) injection points within the interface or gap **117** between the static inlet and rotating combustor, or C) injection points within the static inlet **118** or ahead of the static inlet from holes **119** in the channel walls. The inventive high g-field rim-rotor is the only configuration that enables injecting fuel from the shroud side, which is good for high g-field combustion since the combustion products **101** will naturally flow across the main flow. Preferably, most of the combustion is done in the rotating frame **108** to limit Rayleigh pressure losses associated with high speed flows. The hub and shaft may be an integral piece.

FIG. 4 shows two embodiments that can be used to design a rotating combustor **108**: A) combustion can take place only in the combustor **108** at a constant pressure or B) combustion can be initiated and stabilized in the combustor **108** and can “continuously” take place during a pressure expansion in an expansion device such as a turbine **120** in which work is extracted by means of shaft power. It is understood that “continuous” combustion is when at least 1% of the combustion occurs concurrently with expansion, though preferably continuous combustion is at least 15% of the combustion occurs concurrently with expansion, particularly preferred is continuous combustion is at least 50% occurs concurrently with expansion, specifically preferred is continuous combustion is at least 90% occurs concurrently with expansion. And the absolute best configuration is in which 100% of the combustion takes place concurrently with

expansion to achieve full/complete isothermal expansion for the highest thermodynamic efficiency of a shaft power device.

FIG. 5 depicts a high g-field combustor in a static **107** configuration. Reactants **102** or air **159** enter the combustion chamber and are submitted to centrifugal acceleration due to the curvature of the combustion chamber. An outward positioned flame-holder **103** located in the curvature, hence in the g-field, is used to stabilize the flame in the case of premixed combustion. For non-premixed combustion, fuel injection **157** is done within the g-field without a flame-holder (or optionally with a flame-holder solely as a pilot flame-holder or for enhanced flame stabilization). The colder, denser reactants **102** are driven towards the outer radius of the channel whereas the hotter, less dense combustion products **101** are driven inwards. The static configuration can be operable with either cold air or premix inlet, or hot air inlet. For cold intake, an internal ignition device **159** such as a spark plug can be used to ignite the mixture. For hot intake, the ignition device **159** may optionally be used during start-up modes.

FIG. 6 is an embodiment of a static high g-field combustor **128**, where the g-field is created in a non-rotating reference frame by turning the flow around an axis **129** that is perpendicular to the inlet/outlet flow axis **130**, is the latter also being the rotation axis of turbomachinery at the combustor outlet. Previous attempts of similar combustors have been limited to swirling type combustors using channels wrapped around the turbine axis of rotation in a way such that the channels and the turbine basically have the same radii and are concentric. This limits the radius of curvature that can be used for the combustor, and results in a sub-optimal combination of velocity, pressure losses, and number of g's. A first order model can be drawn to illustrate this sub-optimal operation. The radius of curvature of a static g-field combustor r_c , depends on the flow Mach number M , inlet temperature T_{in} , and desired acceleration a , such that

$$r_c = \frac{M^2 \gamma R T_{in}}{a}$$

where R and γ are gas properties. When designing for ideal design parameters, for example preferred design values of $M < 0.3$ and $a > 100,000$ g's, and a fixed inlet temperature T_{in} , the radius of curvature is found to be unrelated to the turbine radius such as in prior art. Hence prior art designs imply sub-optimal solutions by constraining the radius to be equal to the

radius of the turbine. The embodiment shown in **FIG. 6** allows the use of different flow radius for the high g-combustor and the turbine. The preferred ratio of turbine radius to combustor radius is greater than 2, the particularly preferred ratio of turbine radius to combustor radius is greater than 5, and the specifically preferred ratio is greater than 10. The larger power rating of the
5 turbomachinery yields a higher turbine radius to combustor radius ratio.

FIG. 7 shows a cutaway of a preferred embodiment of a static g-field combustor **107**. To generate high g-fields in static combustors, preference is to have a small radius rather than increasing the velocity to limit Fanno and Rayleigh losses. Preference is for the radius of combustion chamber **109** to always be equal or smaller than the radius **110** of the turbine **111**
10 preferably by at least 10%.

FIG. 8 and **FIG. 9** show the results of CFD analysis of methane non-premixed combustion at a g-field of 100,000 g's. NOx concentration in ppm is shown in **FIG. 8** and temperature contour in degrees Kelvin is shown in **FIG. 9**. In the simulations, the g-field is imposed on the reference frame as gravity and fuel injection is done in hot air at 1000 K to
15 achieve non-premixed combustion at overall lean conditions. The results show that very low NOx concentration below 10 ppm is achieved even for combustion temperatures above 1600 K due to very good mixing resulting in low residence time at near-stoichiometric conditions.

FIG. 10 and **FIG. 11** show the CFD analysis under the same conditions but at a lower g-field of 10,000 g's. The results show that the temperature is higher than 2000 K, resulting in high
20 NOx concentration above 50 ppm, which is due to poor mixing and long reaction lengths at lower g-fields resulting in higher residence time at near-stoichiometric conditions. High g-field clearly enables a reduction in NOx concentration as well as combustor compactness by at least 50%. The preferred embodiment of the invention has g-fields of at least 100,000 g's in order to have a short combustion distance and corresponding low NOx emissions. The preferred
25 embodiment further maintains a flow Mach number below 0.3 in order to limit Fanno and Rayleigh losses.

A high radiant combustion process would emit combustion energy in the form of emitted radiation outside of the combustion chamber. High emissivity is preferred (preferably with emissivity greater than 30%, particularly preferred with emissivity greater than 70%, and
30 specifically preferred with emissivity greater than 85%) when the high g-field combustor is

utilized for industrial applications including: top cycle or bottom cycle for high radiant processes or industrial processes that can achieve higher production throughput by high radiant heat transfer (e.g., steel, glass, cement, etc.) or higher efficiency in the combination of solid-state energy conversion (e.g., thermophotovoltaic, thermoelectric, photovoltaic, etc.). A standalone
5 high radiant and high g-field combustor with solid-state energy conversion significantly increases emitted energy while uniquely limiting the production of NO_x formation. It is understood that the radiated/emitted energy is enabled by using designs with radiation transparent materials and/or by providing an obstacle free path. Such a design is shown in **FIG. 12** where a static high g-field combustor **107** has a radiation transparent container **131** that
10 allows high radiant combustion.

FIG. 13 shows a method of maintaining combustor performance, both static **107** and rotating **108**, during load and/or rpm variations by blocking the number of stator **118** channels, with the objective being to keep pre-heated air temperature above fuel ignition temperature under all load condition when using a hot air combustor. Flow can be blocked with a slide valve
15 or a butterfly valve **121** for at least one of the combustor channel passage, preferably all channel passages could be turned ON/OFF individually or collectively. An advantage here is that combustor performance can remain independent of load since each individual (or group of) combustor passage(s) can see a relatively constant mass flow rate. This is a key inventive feature of the highly compact high g-field combustor.

FIG. 14 shows variable stator guide vanes **122** that can be used to change the flow's angular momentum at the stator **118** exit. The fuel injection system consists of multiple injection points **119** that can be turned ON/OFF individually or collectively as a subset of the total fuel injection points for flow modulation depending on operating conditions. Preferred injection point locations are: after the compressor and optional recuperator but before the main combustion
25 chamber, one in each combustor channel of the main combustion chamber, and finally, after the turbine but before the recuperator. In all cases, injectors can be placed in curved channels and benefit from the piping's natural curvature (e.g. recuperator) to generate high g-fields. Another key feature of the invention is to place injectors to provide precise control of system ramp-up and ramp-downs for temperature control to mitigate thermal shock resistance issues on critical
30 components (e.g. notably components made with ceramic materials). It is understood that the

injectors will include an igniter such as when the autoignition temperature of the fuel is not reached or to alter the temperature profile within the high g-field combustor.

FIG. 15 shows the system combining the high g-field combustor **128** with a ceramic rim-rotor turbomachinery turbine **111**, providing a very compact system with high thermodynamic efficiency capability. The short combustion length of the high g-field combustor together with a high temperature rim-rotor turbomachinery turbine consisting of at least one or more composite rings, at least two blades, a counter-flux thermal insulation substrate and whereby the at least two blades are retained under compressive loading, create a very short gas path between pre-combustion and post-expansion, therefore reducing thermal loss by at least 1% when compared to traditional combustor.

FIG. 16 shows the proposed construction of the rim-rotor turbomachinery turbine **111**, which consists of an external rim-rotor **126** with its annular body, supportive shield **160**, counter-flux thermal insulation substrate **165** and at least two aerodynamic blades **161**, connected to a shaft through opposing arrays of radially compliant springs in the radial-axial plane **163 164**. The radial-axial plane is also known as the meridional plane, and is the plane in which lies the rotational axial and a radial axis of a point, object, etc. The external rim-rotor **126** can be made from an arrangement of one or more composite rings in the axial and/or radial axis and has the primary function of maintaining the at least two blades under compression loading. The radially compliant hub **163 164** requires at least one array of such compliant springs in order to provide adequate stiffness in all axis in order to be dynamically stable. The at least two blade **161** are in physical communication to the shaft through the at least one rotating array of radially compliant springs comprised of an at least one cantilevered beam in the radial – axial plane, whereby the at least two radially compliant springs are in physical communication with the at least two blades at the first axial position, and to the hub at a second axial position, and whereby the first axial position is different from the second axial position and the second axial position is located at a distance from the shaft greater by at least 0.01 inches from the first axial position. The radial compliance is adjusted by the length of the cantilever beams in the radial-axial plane. The stiffness can be selected by adjusted the thickness and profile of the beams, and if required a second set of beams can be added to largely increase the stiffness of the assembly. If required, the second set of beams can be joined by friction, permanently joined or made in a single piece. In the event where the spring in contact with the blade exceeds its material maximum

temperature, cooling features that provide cooling flow directly under the blade **162** is the most effective method to maintain the hub structural integrity.

The rim-rotor **126** is insulated by a counter-flux thermal insulation substrate **165**, physically located between the rim-rotor and the at least two blades in order to maintain its temperature below its maximum operating temperature. The counter-flux thermal insulation substrate consist of at least two cooling channels, whereby the cooling fluid circulates, having a channel inlet and a channel outlet, whereby the center of the channel inlet **166** is located at a channel inlet distance " D_i " **173** from the rim-rotor inner surface to the channel inlet, whereby the channel outlet distance is located at a channel outlet distance " $D_o + D_i$ " **172** from the rim-rotor inner surface, and whereby the channel inlet distance is at least 0.010 inches greater than the channel outlet distance. The cooling channels further have at least a portion of the channels that have a segment **186** that is at an angle between +45 and -45 from the radial axis. This configuration provides that at least a portion of the cooling flow is being directed radially toward the rotating axis (inward), which is against the dominant temperature gradient, therefore against the dominant conductive heat flux in the channel walls (i.e. counter-flux). Means of producing such radially inward flows are illustrated in **FIG. 17**. The insulation substrate **165** with at least a portion of the channels that have Z-shape channels **174**, U-shape channels **177**, micro-channel or micro-holes **176**, porous material construction **178**, or an arrangement of such. The substrate can be made from metals (e.g. titanium, superalloys), ceramics (e.g. Si₃N₄, SiC, Mullite, Al₂O₃, SiO₂, ZrO₂), blend of ceramics, ceramic coatings or a mix of those options. Superalloys are known in the art as high-performance alloys, being an alloy that exhibits several key characteristics: excellent mechanical strength, resistance to thermal creep deformation, good surface stability and resistance to corrosion or oxidation. In the case where porous material is used, open porosity is required to allow the cooling flow to circulate towards the main gas path. Low porosity, between 50% and 80% by mass, is preferred for material between the rim-rotor and the blade that are subject to compression loading. Higher porosity, between 20 and 50% by mass, is preferred where the insulation substrate is not in contact with the blade in order to reduce the substrate mass. Intricate small sizes required by the substrate optimal features consist of cooling channels of characteristic dimension between 0.004 and 0.040 inches. Preferred micro-holes dimensions are 0.001 to 0.020 inches.

FIG. 18 shows example of design implementing the advantages of additive manufacturing. By removing material at selected section(s), the insulation substrate consisting of a continuous monolithic material ring with intricate radially inwards flow channels **180** provides the low circumferential stiffness required to follow the rim-rotor circumferential expansion without breaking. This configuration is especially adapted to the additive manufacturing (also known as 3d printing manufacturing and used interchangeably throughout), but can also be fabricated using subtractive manufacturing methods. This configuration also introduces radial micro-holes at selected position **185**. A insulation substrate made of at least one layer of individual bricks **181** with complex cooling channels, which can also be produced with additive manufacturing or other manufacturing method, prevent the counter-flux thermal insulation layer from breaking due to rim-rotor circumferential expansion while undergoing centrifugal loading. In specific configurations, the optimal insulation substrate can be made with the same material as the at least two blades, and therefore the insulation substrate bricks or ring can be joined or directly fabricated together with at least one blade **182** by different method including soldering, casting, machining, additive manufacturing, diffusion bonding or laser welding. In given specific configuration, it is important to introduce a supportive shield **160** between the rim-rotor and the insulation substrate to distribute uniformly the centrifugal forces between substrate features and the rim-rotor, reducing the stress concentration at the contact of surfaces. The supportive shield **160** may provide axial constraints for the insulation substrate by incorporating at least one side with positive locking features. The supportive shield **160**, may also provide an effective mean of integrating different type of seals **179** having therefore a sealing function between the main flow and the cooling flow inlet to avoid hot gas being directed to the cooling channels and to reduce by at least 5% the amount of cooling fluid leaking in a non-cooling channel region (i.e., the main gas path), especially in configuration(s) where the insulating substrate is built from bricks and has tangential breaks or gaps. In addition, the supportive shield **160** is used in given embodiment(s) to provide a tightly controlled axial gap in front of one or more of the exit of the insulation substrate **171**, which provides an effective mean of controlling the flow rates of small cooling flows. Gaps between the insulation substrate cooling channel outlet and the supportive shield ranging between 0.002 and 0.020 inches creates a flow area that provides adequate regulation of the cooling fluid flow rate. The supportive shield **160** can consist of one or more part to complete its function.

In order to reduce the amount of cooling flow, which directly impacts the efficiency of the turbomachinery, the counter-flux thermal insulation substrate is preferred to the prior art where only a mix of axial and tangential cooling flow is used. In a configuration where the rim-rotor is wound from carbon fiber in a polymer matrix, the maximum operating temperature of the insulation substrate **165** is much greater than the maximum operating temperature of the rim-rotor **126**, therefore the cooling flow exiting the insulation substrate which has inward radial feature allows the cooling flow to get up to the maximum temperature of the substrate and extract considerably higher amount of heat, having effectively a higher calorific capacity for a given flow rate. The axial and tangential cooling flow of the prior art is in direct or indirect thermal contact with the rim-rotor **126** until the channel exit, which limits the temperature of cooling flow to the maximum temperature of the rim-rotor itself, extracting less heat, and therefore requiring considerably higher amount of cooling flow. Furthermore, the cooling channel air flow is also self-stabilized due to the high-g field where the density difference between hot and cold fluid is such that the coldest fluid is sent toward the cooling channels inlet, and therefore protects the rim-rotor, while the hottest fluid is sent toward the cooling channel exit. The counter-flux insulation substrate would reduce the cooling flow required from approximately 5+% to less than 1% of the main flow, therefore by at least a 20% reduction in cooling flow up to the optimal and specifically preferred cooling flow reduction by at least 80%. Based on approximately 0.5 point of cycle efficiency loss per 1% of main flow used for cooling, the counter-flux insulation substrate results in efficiency gain of 2+ point on the cycle, therefore by at least a 0.5% of cycle efficiency up to the optimal of at least a 2.0% of cycle efficiency in the specifically preferred embodiment.

In addition to the benefit of the increase calorific capacity, the radial inward features allow at least a portion of the cooling flow to be used as transpiration cooling (preferably 50%, up to the optimal of 100% in the specifically preferred embodiment), effectively injecting cold air between the main gas path and the insulation substrate. This creates the film-cooling effect, defined by an insulating layer of cold air between the surface of the insulating substrate and the hot main flow to reduce wall temperatures and thus heat flux. This effect is typically used on turbine blades or static shroud surfaces, which are not radially inward in the rotating frame like in this embodiment. The high centrifugal forces, combined with the density difference between the injected cooling flow that is relatively cold and the main gas flow that is hot, provides a

highly beneficial stabilizing effect on the film-cooling layer by generating stratification between the hot and cold gases therefore limiting mixing to a lower level. This lower level of mixing keeps a colder gas temperature near the insulating substrate inner wall, therefore reducing the heat transfer to the insulating substrate and the subsequent radially outward component (i.e. the rim-rotor). The film cooling itself, and the stabilization of the film cooling due to high centrifugal forces, are essential distinctions from rim-rotor prior art to insulate the thermal flux from the primary gas flow to the rim-rotor and provides about 50% of the benefit claimed by the insulating substrate. Furthermore, the inventive transpiration cooling utilizing a cooling flow inlet at a radius closer to the rim-rotor eliminates above 80% of the thermal stress gradient within the turbine blades compared to a configuration where the cooling flow inlet is further away from the rim-rotor, being from the hub and through the blades in the prior art, which is particularly important in ceramic blades that are subject to thermal shock.

Another advantage of this embodiment is the ability to design the cooling channels to direct the cooling flow to the higher heat flux region, which is typically the contact surface between the blade and the insulation substrate **184**. It is preferred that the ratio between the cooling flow directed to the channels thermally connected to the contact surface of the blade and the total cooling flow is sized to match the heat flux ratio between the conduction from the blade and the convection under the insulation substrate. The heat flux ratio for this configuration is typically 50%, therefore the recommended cooling flow ratio directed over the blades is between 35 and 65%. All cooling flow can then be directed through the at least one main channel the front (or inlet side) **170**, to the back **169** (opposite from the front side), or radially into the main flow through at least two orifices **168**. Specific configurations requires that at least 5% of the totality of cooling fluid exits the at least one main channel through the at least two orifices located on the insulation substrate inner wall.

Another embodiment of a rim-rotor turbomachine further comprised of a rim-rotor cavity **152**, a rotating boy and a static housing, utilizes fuels or inert gases to protect components from oxidative degradation at high temperature. In particular when oxidation sensitive rim-rotor materials such as carbon-fiber polyimide or carbon-carbon composites are used. The rim-rotor cavity **152** is filled with non-oxidative gases such as inert gases (nitrogen, helium) or non-oxidative fuels (e.g. hydrogen, methane, propane) to limit/prevent oxidation of the material. The fuels or inert gases will also concurrently reduce windage drag of rotating components. In

particular, the rim-rotor **126** surfaces are moving at high relative velocities and generate frictional drag with fluids from surrounding environment. Drag is a function of gas density, hence filling the rim-rotor cavity **154** with gases having molecular weight preferably 40% lower than air, and specifically preferably 90% lower than air (e.g., methane, helium, hydrogen) minimizes drag. In order to minimize drag, whether the cavity contains air or other gases, the optimal radial and axial gap(s) **155** that minimizes windage losses is between 0.020 to 0.200 inches. The actual gap is a balance between viscous losses at a small gap and turbulence induced losses at large gaps.

A major obstacle in a functional rim-rotor turbine is due to the larger radial deformation between an annular rim-rotor and a classical prior art filled hub submitted to high rotational speed. The radial elongation of the rim-rotor is from 1 micron up to 2% of the rim-rotor inner radius and between 10 to 80 times larger depending on materials and exact geometries than the radial elongation of a filled hub. An innovative measure is to allow relative motion at the interface between the blades and the hub in a controlled manner in order to allow blades to follow the radial displacement of the rim-rotor while maintaining a sufficiently rigid connection between the blades and the hub to withstand dynamic effort during operation. The rigidity of the hub assembly in the radial-axial plane being at least 5 times larger than the shaft bending rigidity in the case of supercritical rotor to ensure the modal vibration due to the shaft rigidity do not excite the hub components. This has the advantage of decoupling the flexibility and the strength requirement of the hub. Decoupling the flexibility and the strength into two components also provides better material options for the hub to resist the high temperature at the contact point with the blade roots, and the potential to insulate the flexible component from the blade root.

The present disclosure also describes a plurality of rim-rotor turbine rotating assembly configurations in which the plurality of blades are operatively connected to the hub by a sliding contact with the hub. The sliding contact allows a radial motion and may allow axial motion as well, between the blades and the hub to compensate for rim-rotor radial deformation under centrifugal loads. In these embodiments, the blades may be ceramic blades, and may have their centers-of-mass at a smaller radius than the rim-rotor radius of gyration, to cause the blades to move radially into contact with the rim-rotor due to centrifugal forces, while the blades remain in operative contact with the rim-rotor (i.e., whether the contact is direct or not, it is the

rim-rotor that blocks the blades from further radially outward movement. These configurations are described with reference to **FIGS. 19-34**.

FIG. 19 depicts an axial rim-rotor turbine rotating assembly **111**, which includes a rim-rotor **126**, a set of at least 2 blades **161** with inner shroud **315** and a blade root **316**, an insulation substrate **303** interfacing the blades **161** to a remainder of the rim-rotor **126**, a hub **305**, and a spring-mass retainer **306**. The substrate **303** may be present in the rim-rotor **126** to release heat of the blades **161**, but it is contemplated to have the rim-rotor **126** without the substrate **303**. **FIG. 30** schematically illustrates the concept of the rim-rotor turbine rotating assembly **111** of **FIG. 19**. The individual blades **161** are not retained by the hub like a classical gas turbine blade, and they are free to move radially to follow the radial expansion **401** of the rim-rotor under rotation and maintain radial contact at their tips with the rim-rotor or intermediate component (i.e. insulation component). To ensure the turbine maintains its integrity under rotation and transfers the power to the turbine drive shaft connected to the hub, it is necessary that the blades stay in contact with the hub with high rigidity achieved by forcing the blades to move following a sliding plane **314** between the blade root **316** and the hub slots bottom surface **317**. The sliding plane is angled from the turbine axial axis which results in a preselected axial blade movement based on its radial displacement **401**. The minimum angle **312** is based on the static coefficient of friction between the blade and hub material to ensure the blade will not jam on speed ramp down. A high angle **312** reduces the static friction necessary at maintaining a high rigidity under dynamic vibration. Therefore, angles between 15 and 75 degree are possible depending on materials and rigidity needs, with a preferred value of 35 degrees that provides sufficient margin on the static friction to avoid ceasing on rotational speed ramp down while ensuring the static friction is sufficient to avoid sliding due to unbalance forces. To maintain the blade root in contact with the sliding plane, an axial force may be applied on the blade, preferably on the blade root backface **321**. This axial force is created by a spring-mass component **306** in contact with the blade root and with a retainer part **308**, depicted here by a nut, creating a spring effect resulting in an axial force on the blades root. To allow an axial relative position and motion of each blades while maintaining contacts, the spring-mass component has partial radial cuts **138** creating an individual beam spring **320** per blade attached to a spring-mass center ring **319** by each beam feet **311**. As the turbine starts spinning, the blades start moving axially reducing the spring preload. To ensure sufficient force

is maintained on the blade root during high rotational speed, an offset mass **310** may be added at the end of the spring beam causing an angle **309** between the spring head center of mass **310** and the radial axis, resulting in an increase of axial force on the blade root when rotational speed increases (shown on the sliced cut view of the spring-mass **313**). To ensure sufficient force is applied by the spring-mass during rotation, the mass head is between 10 and 50 % of the blade assembly mass it makes contacts and where the angle **309** is between 3° and 30°, with a preferred angle of 15° which limits the radial forces being transferred while the creating sufficient axial forces for the assembly rigidity criteria. This configuration uses oxidation-resistant material at gas temperature above 900°C such as ceramics for the blades with compressive loading where maximal tensile stresses are at 40% or less than the maximal compressive stresses. Those blades will conduct a large heat flux to the components in thermal communication. Due to the low centrifugal forces experienced by this hub, the hub can be made of oxidation-resistant material at gas temperature above 900°C with tensile strength above 100 MPa at temperature above 900°C (e.g., ceramics and/or ceramic-metal matrix composites and/or ceramic-matrix composites, as compared to metallic alloys). In cases where small axial displacement are made possible (for example when high sliding plane angle are used), a low flexibility of the spring-mass would allow this component to be made of high temperature materials like ceramic. In the depicted case where higher flexibility of the spring component is required, a metallic superalloy can be used with reduced amount of air cooling by introducing an insulation piece between the spring-mass and the blade root, namely a ceramic coating, or a low conductivity ceramic pad or ball **307**, reducing the spring-mass air cooling by 50%. The insulation has a secondary benefit of increasing the resistance to lateral friction seen at the contact point between the spring-mass and the blade by increasing the hardness by at least 50%.

FIG. 20 depicts a rim-rotor turbine rotating assembly **111** using the aforementioned sliding plane **314** principle, using a disc spring assembly of **331** alone or of both **331** and **333** maintaining the axial force on the blade root **316**. The primary disc spring **331** is in structural communication with the blade root, applying axial force based on the disc spring initial preload, given at assembly and maintained by a retaining component/feature **308**, depicted here with the component being a nut. A stack of additional disc springs in series **333** can be added for added axial displacement, and/or in parallel for added rigidity. The disc spring can be partially cut

radially **335** to allow relative axial position between 0.001 and 0.010 inches of one blade to the next one, without losing contact between the spring and any of the blades, which is required for the manufacturing and assembly tolerances. The primary disc spring can be further insulated from the blade heat by having an insulation ceramic pad **332** or coating applied between the spring and the blade. The spring type used is not limited to disc spring, and other types could be used to produce similar axial forces and displacement.

FIG. 21 depicts another embodiment that proposes a different blade root shape option for an airfoil **340** (or multiple airfoils for a single root, and a single inner shroud) to generate a different sliding pattern, potentially beneficial in durability due to friction pattern, accumulation of residual particles away from the sliding plane and/or providing easier captive assembly features. A dovetail root **342**, a rectangular root **343**, a rounded root **344**, a fir-tree **345** or other root shape perpendicular to the sliding plane can provide additional benefit in the sliding plane **347**, by forming a complementary joint of complementary surfaces with the hub. As shown with only the blades and hub assembly **349**, a female corresponding hub **346** would have the selected pattern blade root imprinted at the selected sliding plane **347** angle, depicted with the single dovetail root pattern **348**. Additionally, the airfoil **340** can be manufactured in one part with an inner shroud **341** and a root system comprised of **342 343 344** or **345**, or all in individual parts assembled together in the turbine, or a hybrid comprised of multiple individual combination parts. In the case of the turbine with high blade density, it could be beneficial for packaging to use a single root attachment **353** for a series of airfoils **350**. This arrangement can also reduce the number of inner shroud segments, therefore reducing the number of joints on the inner shroud. Those joints are leading to hot gas flow leaking inwards to the hub cavity, therefore overlapping feature like depicted on **351** and **352** might be required for sealing purposes. Some of arrangements shown may be used in the assembly 111 in such a way that no spring or like biasing member is required to displace the blades 161 axially and radially. The airfoils may be an integrally cast piece with shroud and root (e.g., cast ceramic), but the airfoils (e.g., ceramic) may be in a material with the other parts are in another material.

FIG. 22 depicts an embodiment of a rim-rotor turbine **111** using a pin-and-socket relative motion assembly, while **FIG. 33** shows the concept schematically. The blade root is a male extrusion (pins) **363** are inserted in the hub's **361** inside matching shaped female receptacle (socket) **362** providing support and friction to the blade root preventing angular movement while

allowing radial movement, purposefully maintaining rotor structural integrity under rotation. To ensure consistent contact between the blade root **363** and the hub socket, a radial spring system is introduced in the socket cavity. The radial springs are designed to have sufficient spring back to stay in contact to both the hub **361** and the blade root **363** in all conditions of operation. An assembly of leaf springs **365** is depicted, but it is not limited to this option. For instance, springs may be absent. Disc springs, coil springs, wounded springs, air springs, fingered dome springs are potential design that would provide the same function. Due to the low tensile stresses experienced by the hub **361** in this configuration, high temperature materials with low tensile strength such as ceramics are possible, allowing higher operating temperatures at the structural contact point between the blades and the hub. The springs can be made of ceramic or metallic superalloys. In the latter, an insulation pad or coating **364** is applied on the blade root at the structural contact location with the spring, where a heat flux is caused by heat conduction through the hub towards the shaft, or by air cooling the springs. The blade root is a planar extruded profile/shape that can be rectangular, rounded **362**, one or more circular pins **366**, the blade root profile itself **367**. The matching socket is shaped to ensure sufficient sliding plane contacts **402** to block all rotations and translation in all axes except the radial axis while allowing a slidefit throughout the operating envelope including all operating temperatures and rotational speed conditions.

In order to improve the thermal management of the turbomachinery, it is beneficial to isolate the main gas path from the shaft, bearings and other turbomachinery components. **FIG. 23** depicts the general configurations that provides the isolated “gas” side **189** from the “shaft” side **156**. Those configurations all provide the advantage of not having a structural component in contact with the bottom of the blade (void from the hub or “hubless”), providing easier thermal implementation with ceramic blade (e.g. Si₃N₄, SiC), especially in configuration where hot combustion exhaust products gases are acting on the at least two blades in the “gas” side **189**, and the advantage of a lighter structure, therefore able to achieve higher tangential speed of the hot gas path meanline in excess of 400 m/s. The connection between the blade and the shaft is made through the rim-rotor **126** or through the counter-flux thermal insulating substrate **165**. The main flow inlet **141** can either be located at the inner radius or the outer radius, and the main flow exit **142** is at the opposite. Both pure axial flow, and partially radial flow (also known as mixed flow) configuration are possible. In the mixed flow configuration, the inlet angle **193**

can be at a range of angles between 0 and 90 degrees relative to the radial plane, with a preferred 45 degree to maximize the ratio of inlet radius over exit radius while maintaining structural integrity. The exit angle **194** can also be at a range of 0 to 90 degrees. The gas side **189** circumscribes the rim-rotor's inner surface **190** and a side face **191** of the connecting element **192**.

A rim-rotor axial turbine configuration requires the hot gas path to surround the rim-rotor, which is especially problematic for recuperated cycles where the exhaust is re-circulated into the heat-exchanger, coupling the exhaust gas with the fresh air path. Due to the limited temperature resistance of the rim-rotor, including rim-rotors comprised of a fiber (e.g., carbon) polymer rim-rotor, the surroundings of the rim-rotor must be maintained at low temperatures. Therefore, our inventive configuration keeps the hot gas flow path on only one side of the rim-rotor and a cooler section on the other side of the rim-rotor providing considerable cooling and thermal protection advantages, effectively increasing the cycle efficiency and rim-rotor reliability. The configuration provides this thermal separation having similarities to radial and mixed flow turbines, but including a change in flow direction in the meridional plane greater than prior art mixed flow and radial turbines, therefore effectively creating a reversed mixed flow turbine (RMFT). The meridional plane is defined as a plane cutting a turbomachine through a diametral line and the rotating axis. The embodiments are not limited to a single rim-rotor on the outside of the turbine, it can utilize an assembly of multiple (i.e., at least two) rim-rotors providing higher radial preloading as necessary or varying material compositions providing different functions of each material. It is understood that the depictions in the figures are for turbines/expanders but in fact any turbo-machinery including compressors.

FIG. 24 depicts a rim-rotor reversed mixed flow turbine assembly **201**, while **FIG. 32** shows the concept schematically. Hot gas flow meridional inlet **141** forms an angle **211** smaller than 90° from the hot gas flow meridional outlet **142**. In this configuration the hot gas flow makes a change in direction in the meridional plane between 120° and 180° , preferably 150° (calculated as 180° minus the inlet to outlet angle **211**). This is larger than the 90° change in direction of radial turbines, much larger than regular mixed flow turbine which are between 0° and 90° , and naturally larger than axial flow turbine where there is no change in direction (0°). Such change in direction provides a definitive separation between the "gas side" **189** and the "shaft side" **156**, enabling a more effective cooling feature **204** and cooler environment around

the rim-rotor **126**. This large change in direction also generates a pressure gradient **216** as the flow turns reducing pressure at the turbine blade tip **217**, effectively reducing tip leakage and associated efficiency losses. The turbine blades are split into two sections, preferably at the point **214** where the meridional blade curvature **219** is tangent with a radial line **218**, to form an

5 outer section that is mostly in compressive loading by being forced outward during rotation into the rim-rotor **126**, and an inner section **206** that is in tensile loading by being retained at its center. This separation generates an outer section maximal tensile stresses that are at 40% or less than the maximal compressive stresses, and an inner section maximal compressive stresses that are at 40% or less than the maximal tensile stresses. The outer section is

10 subsequently split into at least two separated blade sections **203** to avoid any circumferential stresses in those parts, allowing oxidation-resistant material at gas temperature above 900°C such as ceramics to be used. The inner section **206** is a single part containing the exducer section of the blades, and makes a structural connection to the turbine shaft **208**. The exducer is the exit blade section that provides the rotor exit angle at the rotor exit for the flow to minimize

15 energy losses at the turbine exit and following diffuser. The lower tangential speed of the exducer section, relative to the turbine tip speed, considerably reduces the stresses in the component compared to traditional single part radial turbines, allowing oxidation-resistant material at gas temperature above 900°C with tensile strength material above 100 MPa at temperature above 900°C such as monolithic ceramics, especially Si₃N₄ or SiC, and ceramic

20 matrix composites (CMCs), to be used for this component. Since the rim-rotor **126** radial elongation under rotation from 1 micron up to 2% of the rim-rotor inner radius and between 10 to 80 times larger depending on materials and exact geometries than the radial elongation of the exducer section **206** backface, a flexible connection must maintain contact between the blades, the rim-rotor and the exducer or directly between the rim-rotor, turbine blades and the turbine

25 shaft to ensure the turbine structural integrity is maintained under rotation. A flexible hub **205** that connects the rim-rotor **126** and the blade outer section **203** to the turbine shaft **208** requires advance design features to ensure flexibility, rigidity and strength are sufficient to operate at high rotational speed. One exemplary design feature is using an array of radially compliant springs **207** comprised of a cantilevered beam in the radial-axial plane, such as for a functional

30 isotropic material, namely most metallic alloys and specific ceramics. Components in the "shaft side" **156**, which includes the hub, rim-rotor and not depicted turbomachine component like bearing(s), electric generator, compressor(s), can be further insulated from the hot components

by having a thermal barrier coating or insulation component(s) **215** applied onto the backfaces of the blades outer section **203** and inner section **206**.

Another exemplary embodiment for the turbine connection is further detailed in **FIG. 25**, with a radially deformable thin dome hub **240** based on orthotropic material properties to control deformation and strength respectively in the radial and axial axis. The thickness **261** of the dome is between 2.5 and 25 % of the rim-rotor thickness, preferably 10%, and its curvature in the axial-radial plane **260** is between 15 and 50% of its outer radius. A dome hub provides an improved and easier sealing capability of the inner cavity **245** than an array of compliant springs, therefore reducing drag, particularly preferred drag reduction of 66%, by avoiding air flow exiting and entering the cavity **245**. A fiber-polymer composite, including a carbon fiber, is the preferred high strength orthotropic material where the fiber placement adjusts the rigidity in all axes, i.e. radially, axially and tangentially. An exemplary method is a placement of the composite layers on a domed mold with layers made at angles between 0 and ± 90 degrees between the axial axis and the fibers, preferably at repeated layers at +45 then -45 degree, notably by using the preferred filament winding technique. This fiber placement allows above 1.5% elastic deformation in the radial-axial plane, and specific strength above 200 MPa/(g/cm³) in the radial-axial plane to support its own mass. The dome hub **240** is depicted to be inserted between the rim-rotor **126** and the outer blade sections **203**, but attachment with the dome hub on the outside of the rim-rotor or on the side, or even between the blade section and the cooling features **204** are also options, depending on the hub material composition.

As shown on **FIG. 26**, and schematically in **FIG. 31**, the integrity of the turbine can be also maintained by introducing a sliding planes **244** between the base of each turbine blade outer section **203** and the inner section **206**. This angled sliding plan between 15° and 75°, preferably 35°, from the axial axis allows the blade to move freely radially while maintaining contact with the inner section. A spring system, which includes a spring **241** and a preloading and locking mechanism **242**, ensures an axial force is applied at the back of the blade so the contact on the sliding plane is maintained with sufficient friction that the turbine keeps its structural integrity under rotation. The depicted spring **241** is a disc spring with partial radial cuts **243** to allow a relative axial position and motion of each blades. Exemplary features to provide the axial force include raw disc springs, coil springs, wounded springs, air springs, fingered dome springs, magnetic or hydraulic mechanism are potential design that would perform the

same functions. An insulation component or coatings could be applied at the contact point **245** in the instance where the blade outer section is made of ceramic and the springs are metallic.

The individual outer blade sections are further detailed on **FIG. 27**. An integrated outer section **224** provides simplicity during the assembly. Flexibility in material selection for each function is better achieved with a separated outer section **226**, which includes the insulation substrate **220** with insulation and cooling features **204**, the rear wall flow guide **222**, and the blade **221** inserted in a blade holder **223**. The outer blade section **203** has at most one rear wall flow guide per blade, but can also include multiple blade per rear wall flow guide portion. The rear flow guide **222** can be made of a metallic superalloy with oxidation-resistant thermal barrier coatings while the blade **221** is made from oxidation-resistant material at gas temperature above 900°C such as ceramic. Reducing mass of the outer blade section **203**, which leads to lower stresses in the rim-rotor **126**, is achieved by making the rear wall flow guide partially hollow by removing non-loaded material sections.

FIG. 28 shows a typical mixed flow turbine inlet velocity triangles. The advantage of the mixed flow turbine as compared to a purely radial turbine is the capability to create a positive inlet blade angle **232** while using purely radial blades, meaning that all sections of the blade are attached radially to the supportive section. This advantage is carried over to the rim-rotor reversed mixed flow turbine configuration since the inlet flow has an axial velocity component. The outer section of the reverse mixed flow turbine blade **237** is shown in the inlet tangential plane, where the positive blade angle **232** allows the inlet flow tangential speed **234** to be larger by at least 5%, preferably 22%, than the inlet blade tip velocity **236**, resulting in an increased specific power for the same stage efficiency, ultimately resulting in a more compact and efficient engine.

FIG. 29 illustrates a highly compact turbomachine design where a centrifugal compressor **250** is mounted back-to-back with the reversed mixed flow turbine **201**, where the distance between the compressor and the radial turbine backfaces is less than 50% of the turbine rim-rotor radius. Such a compact configuration can be connected to a load or an electric generator on an extension of the turbine shaft **208** where both the compressor and turbine mass are overhung from the bearings. The high compactness provides a reduction in mass of the turbomachine of a least 5% by reducing thermal insulation needs. Installing the compressor and

turbine back faces **253** back-to-back removes the drag loss induced between the back faces and a static component preferably reducing the drag by up to 100% versus the configuration where the compressor and turbine are installed separately (i.e., not back to back). In this configuration, the hot section "gas side" **189** is kept on one side of the machine, and the cooler
5 section **156** is on the other side, reducing thermal heat losses and importantly reducing cooling air requirements notably on the fiber-polymer rim, therefore increasing system efficiency.

FIGS. 32 and **34** depicts the rim-rotor reversed mixed flow turbine assembly **201** in which there is a radial sliding surface **403** of contact between each outer blade section **203** and the inner section **206**. Under rotational speed, the rim-rotor **126** is loaded by the centrifugation
10 of the outer blade sections **203** which results in a radial displacement **401**. The rim-rotor and outer blade assembly is attached to the shaft through radially compliant springs **207** or dome hub **240**. To avoid dynamic or sealing issues during operation, the outer blade sections **203** and the inner section **206** must remain in contact, effectively made on the radial sliding surfaces **403**. Those surfaces can be incorporated at the rear wall flow guide **222**, and additionally on the
15 side of the blade in the meridional plane **404**, creating overlapping surfaces.

The high g-field combustor is a key inventive component with the embodiment of a rim-rotor turbomachinery, most notably when the rim-rotor turbomachinery is a ceramic turbomachinery with high tip speed (e.g. compressor, turbine, rotating ramjet, or rotating combustors). Although the invention has been described in detail with particular reference to
20 certain embodiments detailed herein, other embodiments can achieve the same results. Variations and modifications of the present invention will be obvious to those skilled in the art and the present invention is intended to cover in the appended claims all such modifications and equivalents.

CLAIMS

1. A rim-rotor turbine rotating assembly comprising:
 - a hub adapted to be mounted or connected to a rotating shaft;
 - a rim-rotor; and
 - 5 a plurality of blades with each of the plurality of blades contacting the rim-rotor, each of the plurality of blades operatively connected to the hub by a sliding contact with the hub allowing at least a radial motion between the blades and the hub to compensate for rim-rotor radial deformation under centrifugal loads;
 - wherein the blades are ceramic blades.
- 10 2. The rim-rotor turbine rotating assembly of claim 1, wherein the blades are axial-flow blades with the fluid flow entering generally axially and leaving generally axially.
3. The rim-rotor turbine rotating assembly of claim 1, wherein the blades are radial-flow blades, in which the fluid flow enters generally radially and leaves generally axially on a same side as the fluid flow has entered.
- 15 4. The rim-rotor turbine rotating assembly of claim 3, wherein the fluid flow enters generally radially at an angle ranging between 5 and 45 degrees inclusively from a radial plane of the rim-rotor.
5. The rim-rotor turbine rotating assembly of any one of claims 1 to 4, wherein the sliding contact is defined by complementary sliding surfaces between the hub and the blades, wherein
20 the complementary sliding surfaces are at an angle ranging between 15 to 75 degrees inclusively from a rotational axis in a radial – axial plane.
6. The rim-rotor turbine rotating assembly of claim 5, wherein the complementary sliding surfaces are at an angle of 35 degrees from the rotational axis in the radial – axial plane.
7. The rim-rotor turbine rotating assembly of any one of claims 1 to 6, wherein at least two
25 blades are joined together into a single part having a common inner shroud and a common root for at least two airfoils.

8. The rim-rotor turbine rotating assembly of any one of claims 1 to 7, wherein a shape of the blade roots complementarily matches a shape of hub surfaces to create a complementary male-female engagement creating the sliding contact.
9. The rim-rotor turbine rotating assembly of claim 8 wherein the shape of the blade roots
5 shape is one of a rectangle, an ellipse, a dovetail, and a fir-tree.
10. The rim-rotor turbine rotating assembly of any one of claims 1 to 9, further comprising at least one spring to bias the plurality of the blades in sliding contact with the hub against the rim-rotor.
11. The rim-rotor turbine rotating assembly of claim 10, wherein the at least one spring
10 applies at least one contact force on blades at their roots.
12. The rim-rotor turbine rotating assembly of claim 10, wherein the at least one spring applies a force on the blades that is generally perpendicular to the rotational axis, the at least one spring being radially under each of the blades.
13. The rim-rotor turbine rotating assembly of claim 10, wherein contact forces are applied
15 on blades at their tips by the at least one spring.
14. The rim-rotor turbine rotating assembly of any one of claims 10 to 13, wherein the at least one spring includes a plurality springs each applying an independent biasing force on a respective one of the blades.
15. The rim-rotor turbine rotating assembly of any one of claims 10 to 14 wherein the at least
20 one spring is at least one of a leaf spring, a disc spring, a coil spring, a wounded spring, an air spring, and a fingered dome spring.
16. The rim-rotor turbine rotating assembly of any one of claims 10 to 15, further comprising an insulation material between the at least one spring and the blades.
17. The rim-rotor turbine rotating assembly of claim 16, wherein the insulation material is at
25 one of a ceramic coating, a low conductivity ceramic pad or a ceramic ball.

18. The rim-rotor turbine rotating assembly of any one of claims 10 to 12, wherein axial contact forces on the blades are provided by using a radial centrifugal force on a spring head converted to an axial force by locating the spring head further away from the hub than a spring attachment point on the shaft.

5 19. A rotary turbomachine comprised of a rim-rotor having a rim-rotor radius, a blade mounting assembly and a shaft in the radial-axial plane, whereby the rim-rotor has a radial deformation, whereby the blade mounting assembly is comprised of at least two blades, at least two blade roots and a filled hub, whereby the blade mounting assembly enables relative motion
10 between the at least two blade roots and the filled hub during operation with a radial displacement of the at least two blades by an amount from 1 micron to the radial deformation of the rim-rotor.

20. The rotary turbomachine according to claim 19 whereby the radial displacement of the at least two blades is from 1 micron to 2% of an inner radius of the rim-rotor and from 10 to 80 times larger than the radial displacement of the filled hub, and utilizes a force on the at least two
15 blade roots that creates a friction to form the blade mounting assembly with at least 5 times more rigidity than the rigidity of the shaft in the radial-axial plane.

21. The rotary turbomachine according to claim 19 whereby the at least two blades are in compression loading, whereby the at least two blades are made of an oxidation-resistant material having a maximal tensile stress at least 40% less than a maximal compressive stress
20 at a gas temperature above 900°C.

22. The rotary turbomachine according to claim 19 whereby the filled hub has a tensile strength greater than 100 MPa at a temperature above 900°C and whereby the filled hub is comprised of an oxidation-resistant material.

23. The rotary turbomachine according to claim 19 whereby the at least two blades are
25 made into a single part having a common inner shroud and a common root attachment.

24. The rotary turbomachine according to claim 19 wherein the filled hub has at least two matching hub slots and whereby the at least two blade roots and the at least two matching hub slots have an angular plane in an axial direction whereby the angular plane is at a minimum of 15 degrees up to 75 degrees from the axial direction.

25. The rotary turbomachine according to claim 24 whereby the angular plane is at 35 degrees.
26. The rotary turbomachine according to claim 19 whereby the filled hub has an axial force created by at least one spring including a disc spring or mass or spring system, and wherein the
5 at least one spring has a spring head, a spring head offset, a spring beam, a spring head mass center and a spring feet.
27. The rotary turbomachine according to claim 25 whereby the at least one spring enables an individual axial movement per the at least two blades between 0.001 and 0.010 inch by having partial at least one radial cut in the at least one spring.
- 10 28. The rotary turbomachine according to claim 25 whereby the blade mounting assembly has a mass, whereby the axial force is created by an array of compliant springs having a mass from 10 to 50% of the blade mounting assembly mass, whereby the blade mounting assembly has a physical contact point at an extremity of the spring head, with the spring mass center or at
15 the spring head offset from the spring beam creating an increasing axial force for an increasing rotational speed, and whereby an angle made by a line passing by the spring head mass center and the spring feet and a radial axis is between 3 and 30 degrees.
29. The rotary turbomachine according to claim 28 further comprised of an insulation piece between the spring-mass and the blade root, whereby the insulation piece includes a ceramic coating, or a low conductivity ceramic pad or ball, and whereby the at least one spring with the
20 insulation piece reduces an air cooling by at least 50% compared to the at least one spring without the insulation piece and whereby the insulation piece increases a lateral friction resistance at the physical contact point between the spring mass and the blade by at least 50%.
30. The rotary turbomachine according to claim 19 further comprised of a counterpart hub having a counterpart hub shape, wherein the blade root has a blade root shape, and wherein
25 the counterpart hub shape and the blade root shape have an identical shape and are at an identical angle.
31. The rotary turbomachine according to claim 12 whereby the identical shape includes a rectangle, an ellipse, a dovetail, and a fir-tree.

32. The rotary turbomachine according to claim 25 wherein the at least one spring under each at least one blade is in a pin-and-socket assembly and whereby the pin-and-socket assembly has a pin shape including round, rectangular, elliptical, or of a blade profile of the at least one blade.

5 33. The rotary turbomachine according to claim 19 further comprising a hot gas path exit, wherein the hot gas path has a meridional plane, whereby the hot gas path exit is axial and where the hot gas path exit has a change in direction within the meridional plane.

10 34. The rotary turbomachine according to claim 19 further comprised of an outer section and an inner section, whereby the at least two blades are split between the outer section and inner section, whereby a cut is between the outer section and the inner section wherein the cut is made to maintain the outer section in compressive loading, whereby the outer section has a maximal tensile stress at least 40% less than a maximal compressive stress, and whereby the inner section has a maximal compressive stress that is at least 40% less than a maximal tensile stress of the inner section.

15 35. The rotary turbomachine according to claim 34 whereby the outer section splits into an at least two blade sections comprising the at least one blade.

36. The rotary turbomachine according to claim 16 whereby the at least two blade sections is made of a single piece that is further comprised of a one rear wall flow guide section for each of the at least one blades.

20 37. The rotary turbomachine according to claim 36 wherein the at least one blades is a separate material than the one rear wall flow guide section.

25 38. The rotary turbomachine according to claim 33 whereby the outer section has an inlet flow blade angle, an inlet flow relative angle, an inlet blade tip velocity and an inlet flow tangential speed, and whereby the inlet flow relative angle is not zero and whereby the inlet flow tangential speed is greater by at least 5% than the inlet blade tip velocity.

39. The rotary turbomachine according to claim 19 further comprised of a shaft, a hub, and a connection between the rim-rotor and the shaft, wherein the rim-rotor has an inner radius,

whereby the connection is a flexible connection having up to 2% of the rim-rotor inner radius in a radial deformation directly between the rim-rotor and the shaft.

40. The rotary turbomachine according to claim 39 whereby the flexible connection is further comprised of at least one rotating array having an at least two radially compliant springs having
5 an at least one cantilevered beam in a radial-axial plane.

41. The rotary turbomachine according to claim 39 whereby the flexible connection is further comprised of a compliant material that has greater than a 1.5% elastic deformation in the radial-axial plane.

42. The rotary turbomachine according to claim 39 whereby the flexible connection is
10 comprised of a material having a specific strength above 200 MPa/(g/cm³) in the radial-axial plane.

43. The rotary turbomachine according to claim 33 whereby the outer section is in structural communication with both the inner section and the rim-rotor by having a root that slides on an angular sliding plane at a hot gas path exit, where an angle between the sliding plane and an
15 axial axis is between 15° and 75°.

44. The rotary turbomachine according to claim 19 further comprised of a rim-rotor having a rim-rotor radius, a back-to-back centrifugal compressor with a radial turbine having a backface drag, wherein a distance between the compressor and turbine is less than 50% of the rim-rotor radius, wherein the compressor and turbine have a system efficiency, and whereby the system
20 efficiency is at least 5% greater than a centrifugal compressor and a radial turbine that is not back-to-back by reducing the backface drag from 0.01% up to 100%.

FIG. 1

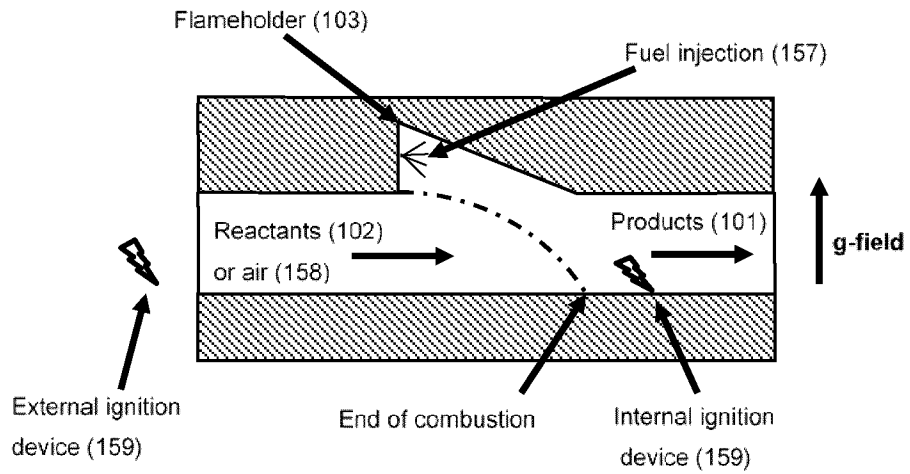


FIG. 2

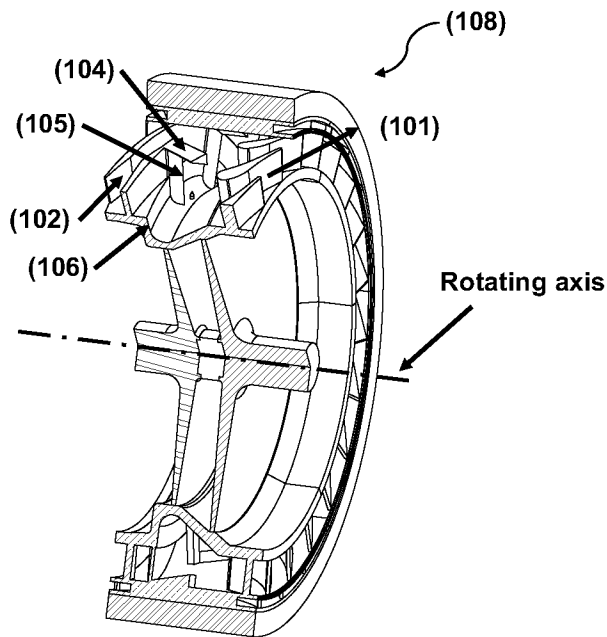


FIG. 3

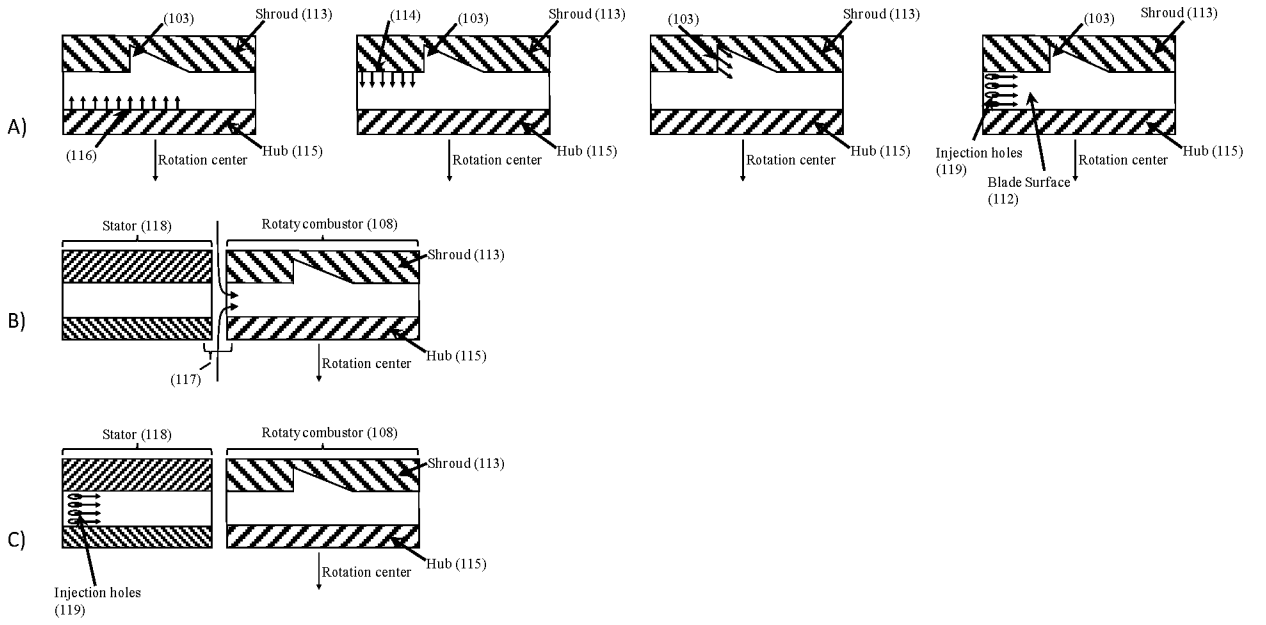


FIG. 4

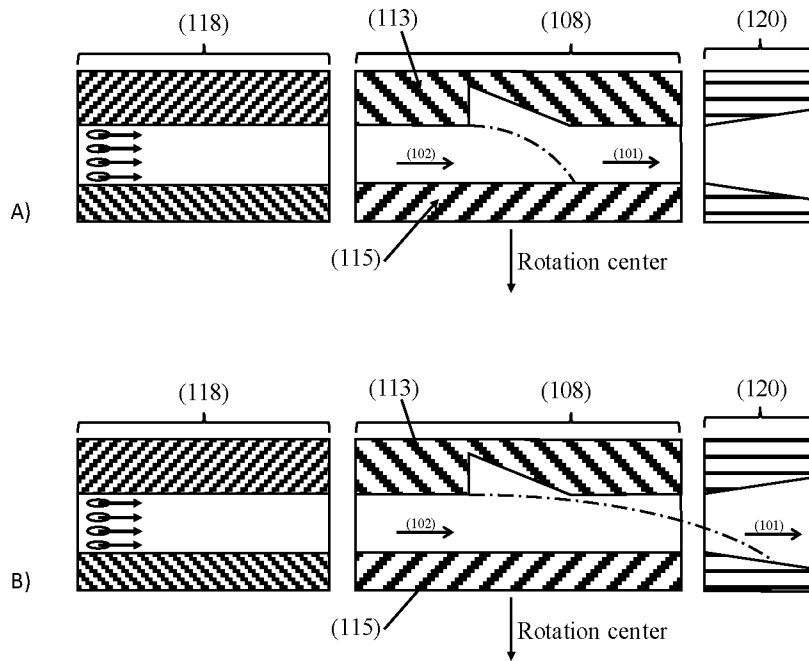


FIG. 5

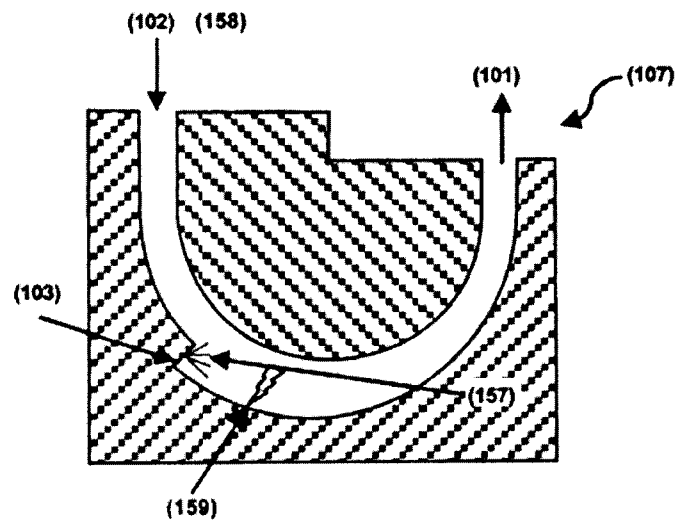


FIG. 6

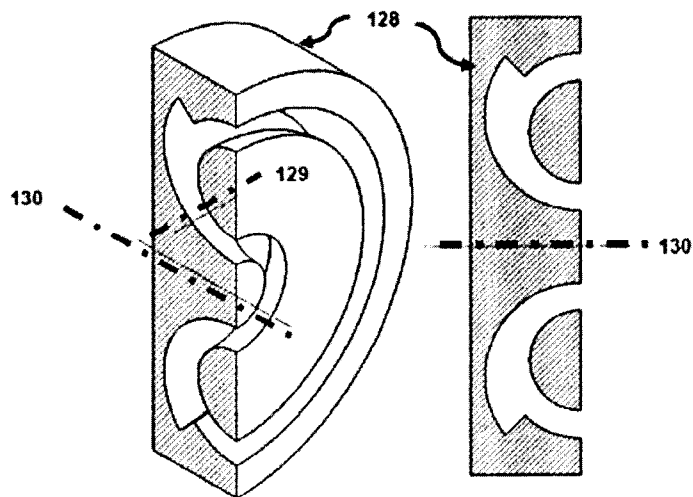


FIG. 7

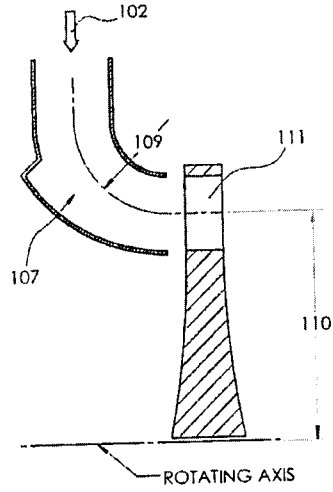


FIG. 8

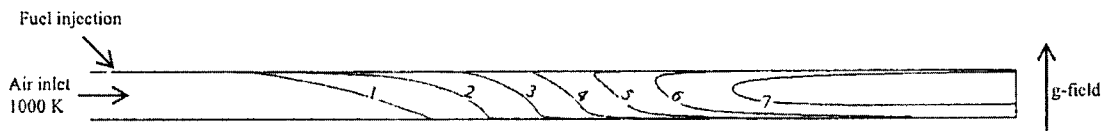


FIG. 9

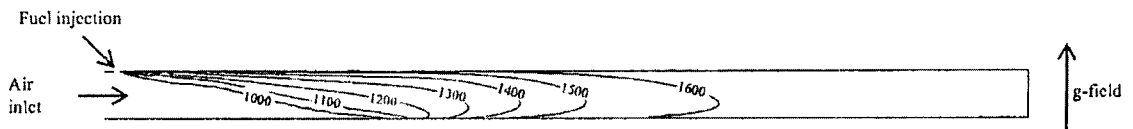


FIG. 10

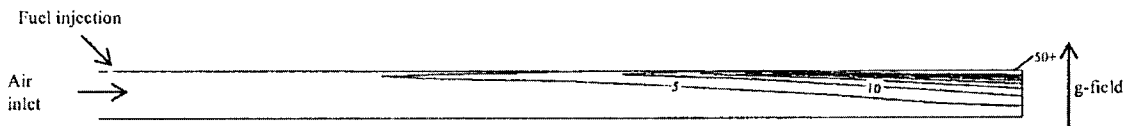


FIG. 11

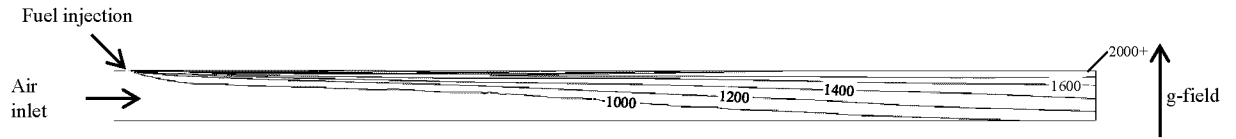


FIG. 12

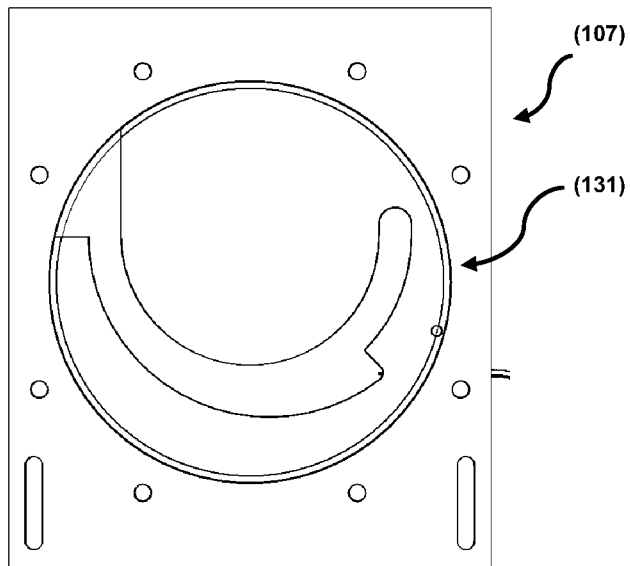


FIG. 13

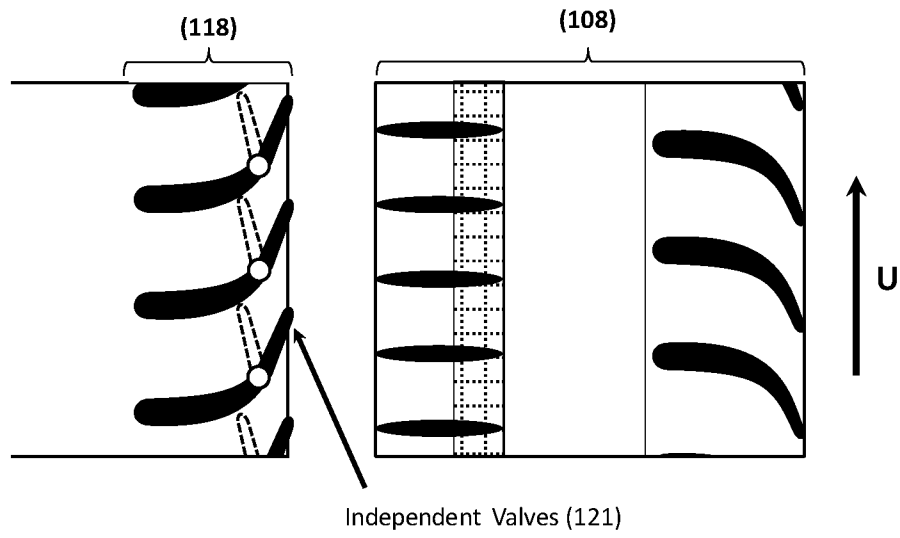


FIG. 14

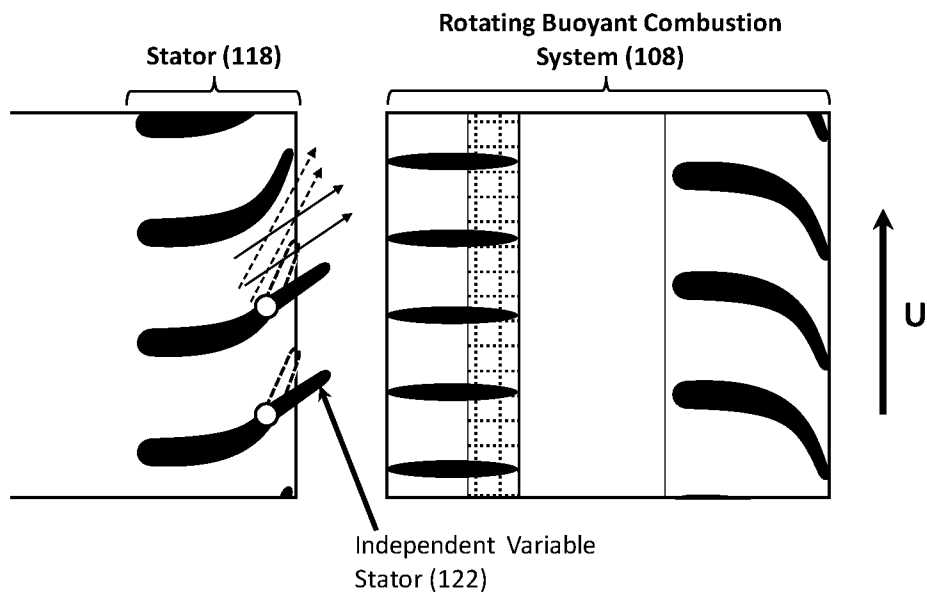


FIG. 15

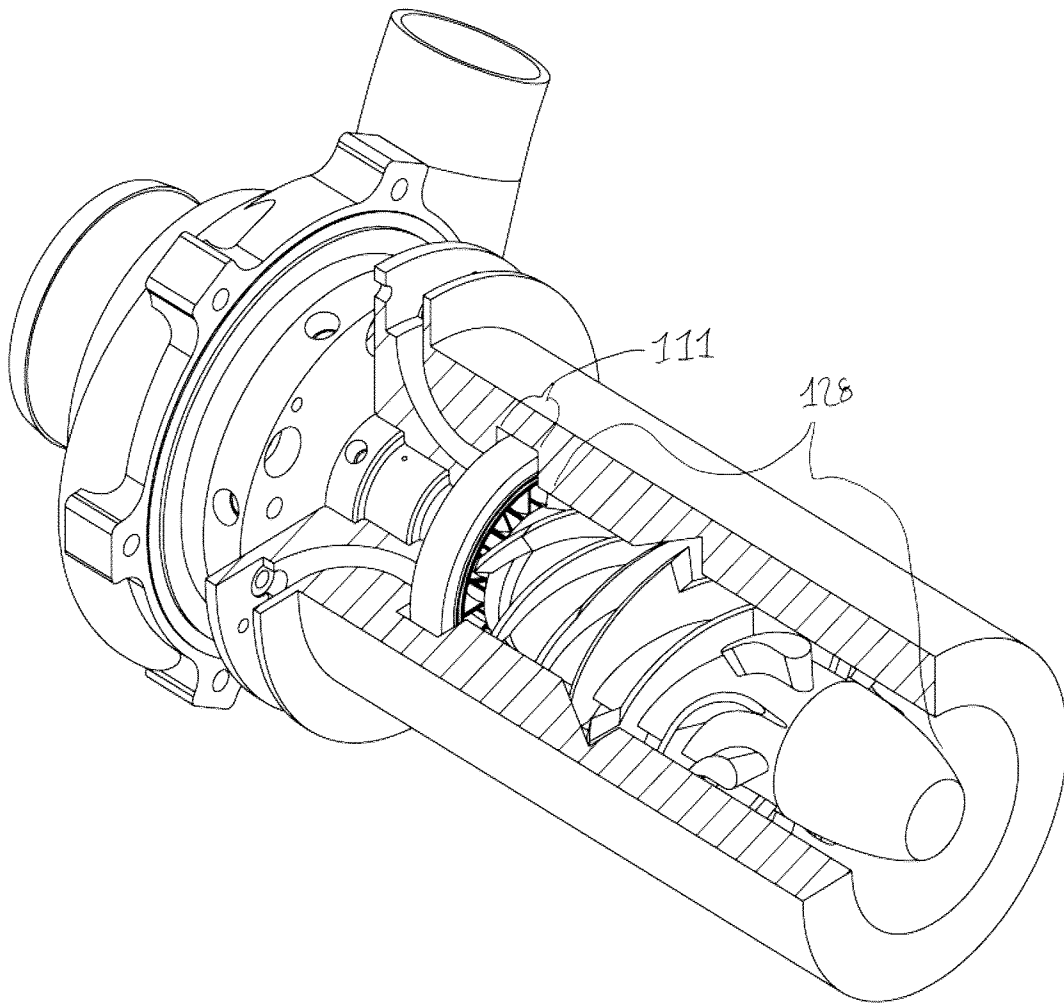


FIG. 16

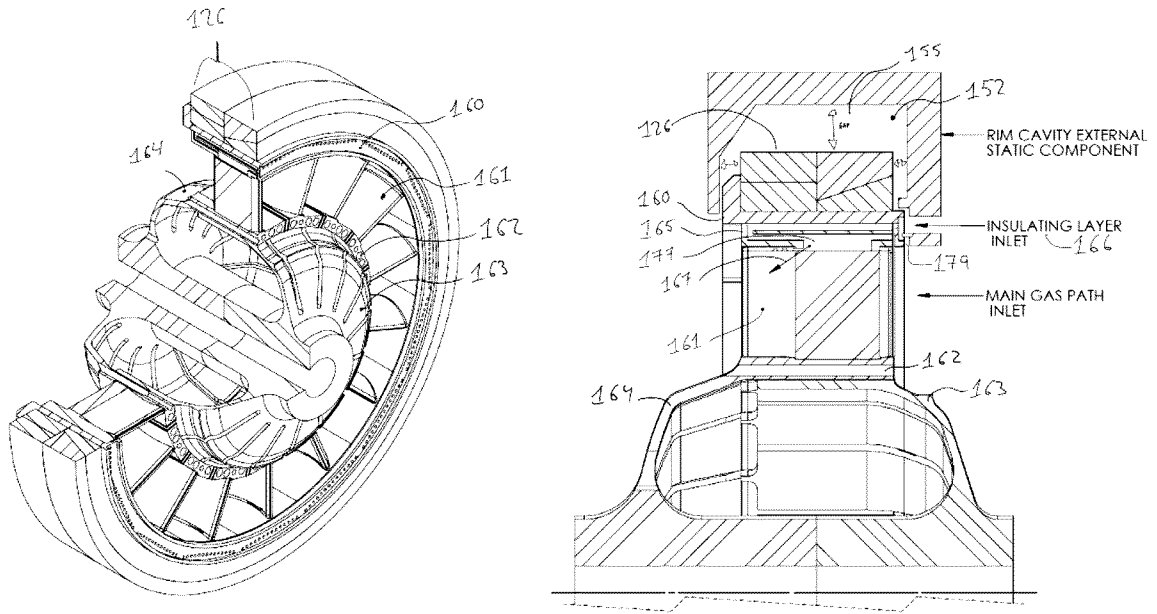


FIG. 17

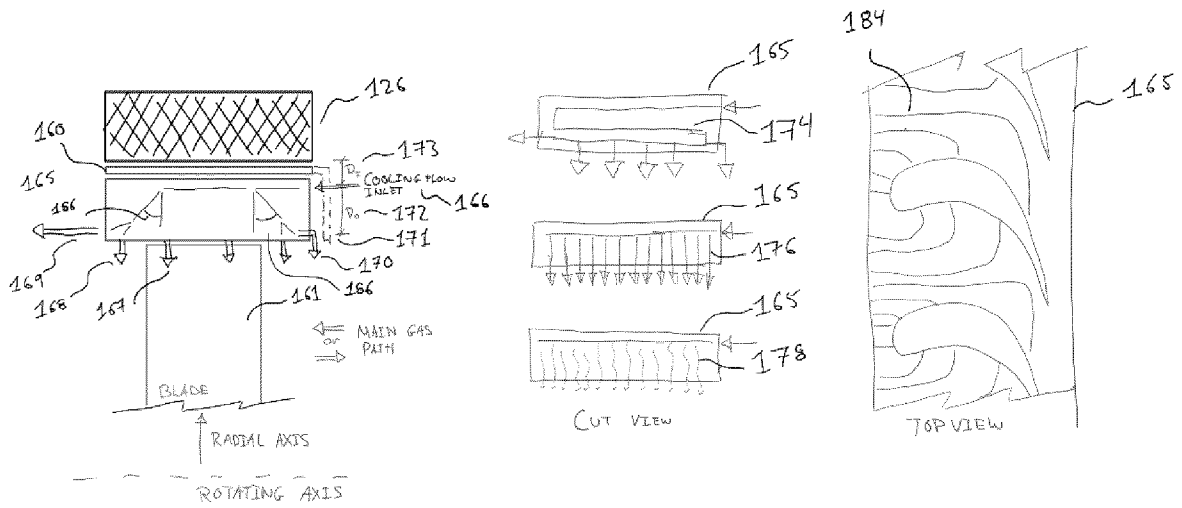


FIG. 18

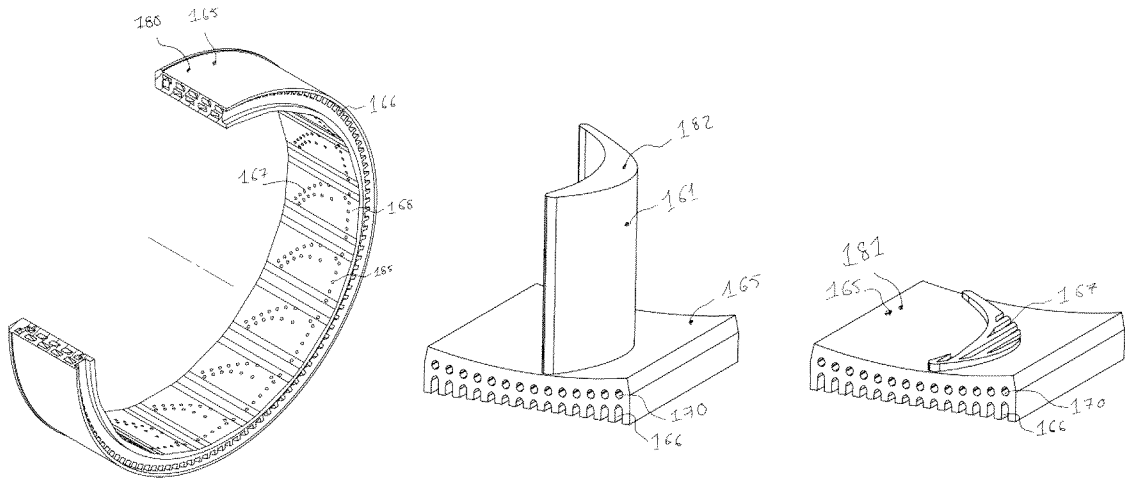


FIG. 19

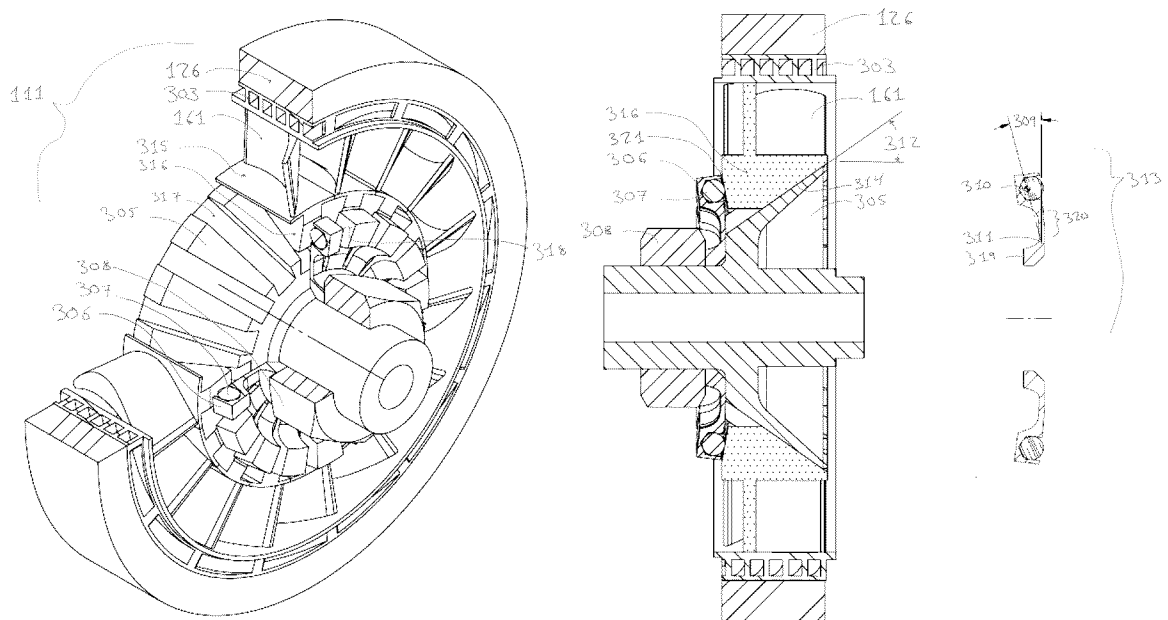


FIG. 20

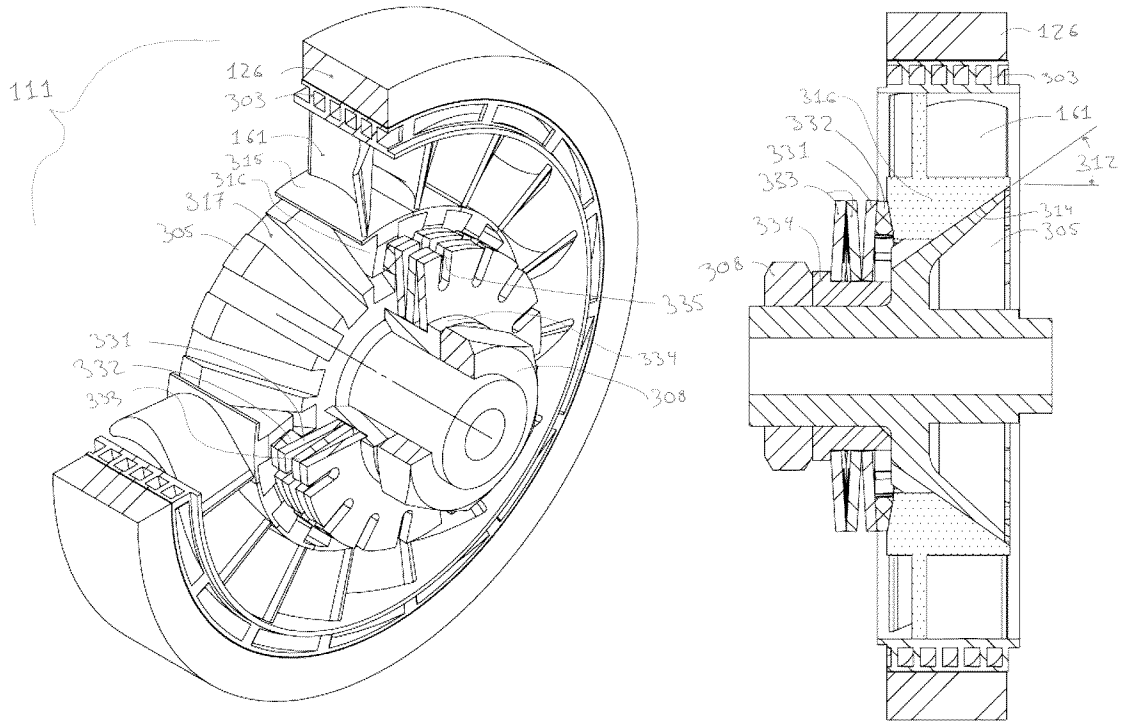


FIG. 21

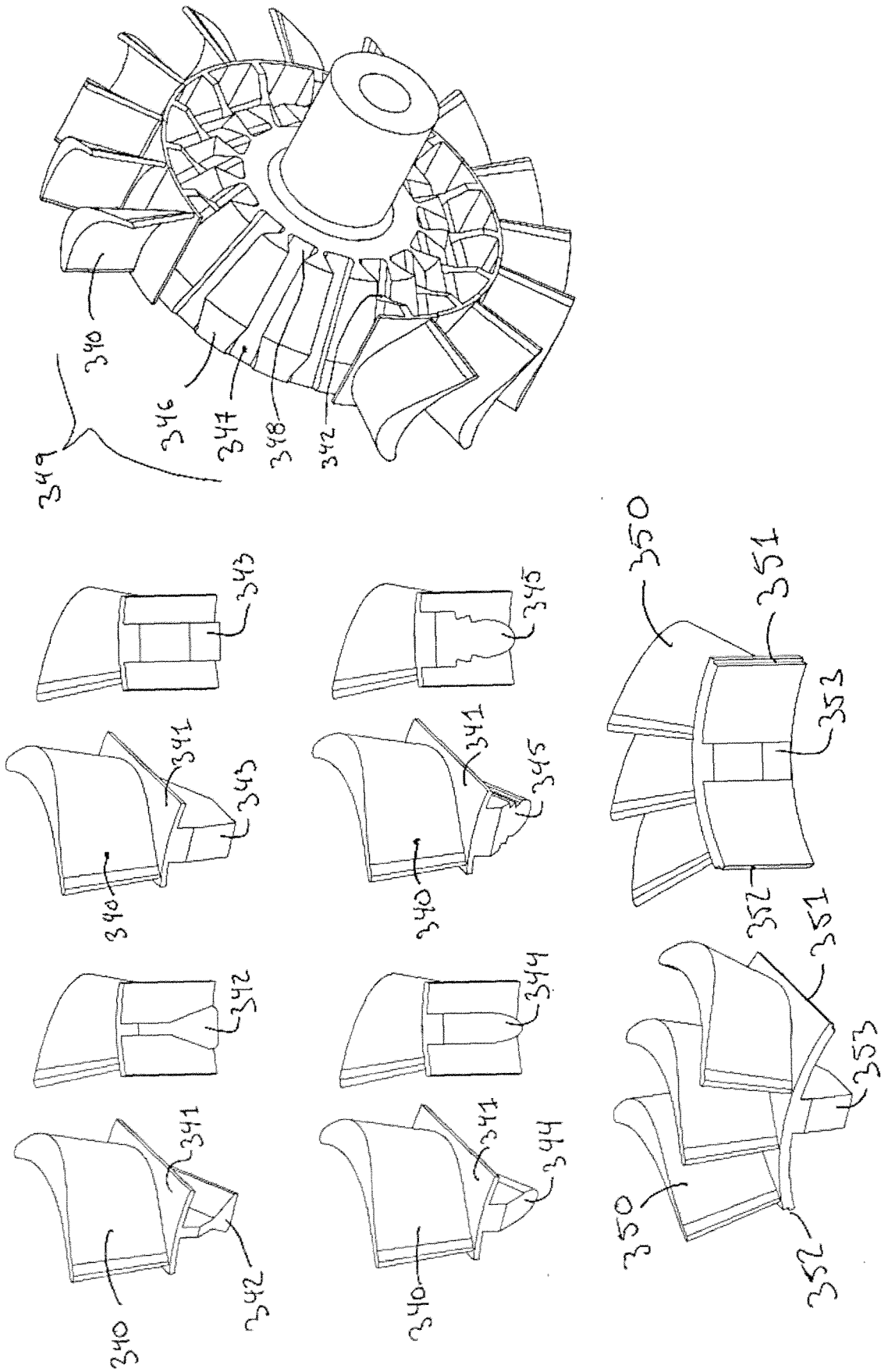


FIG. 22

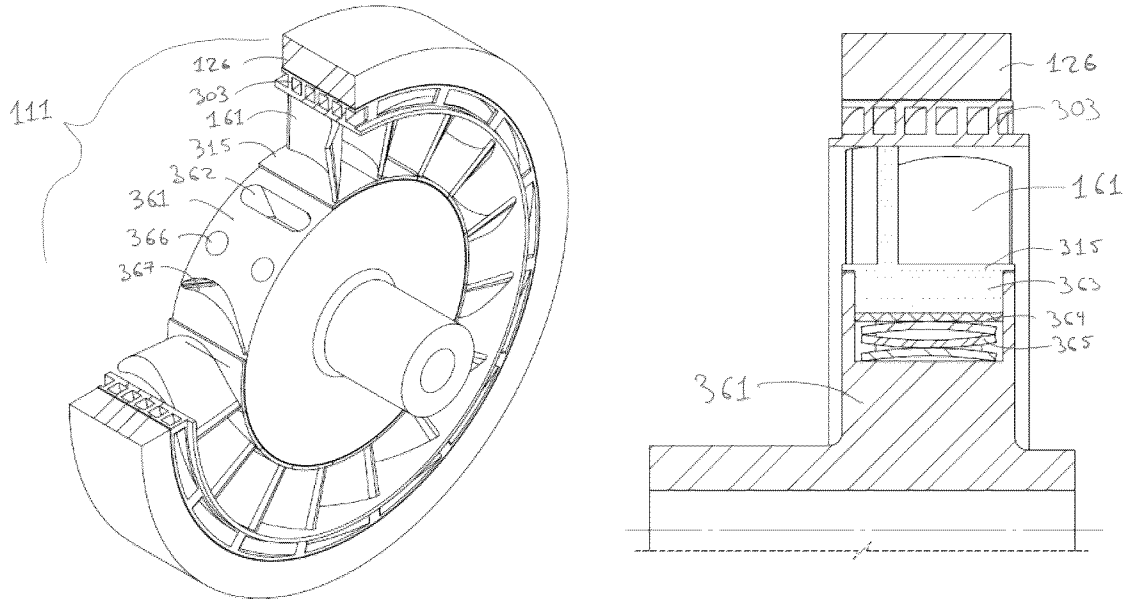


FIG. 23

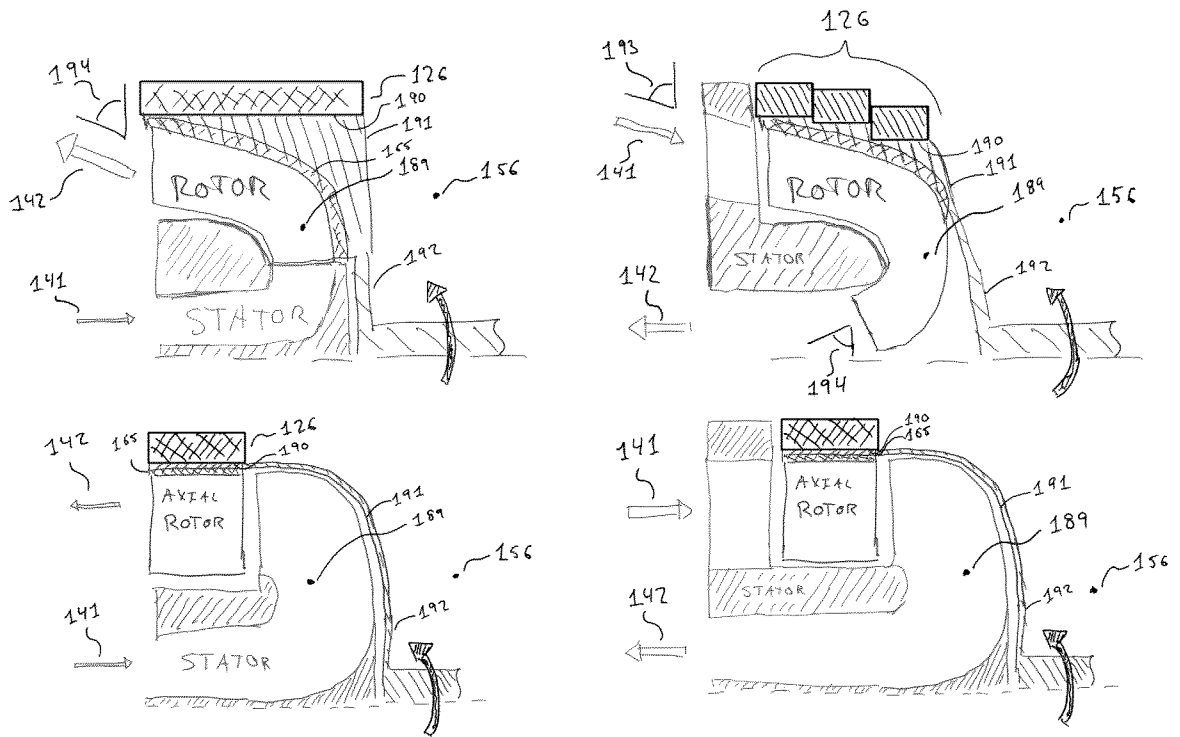


FIG. 24

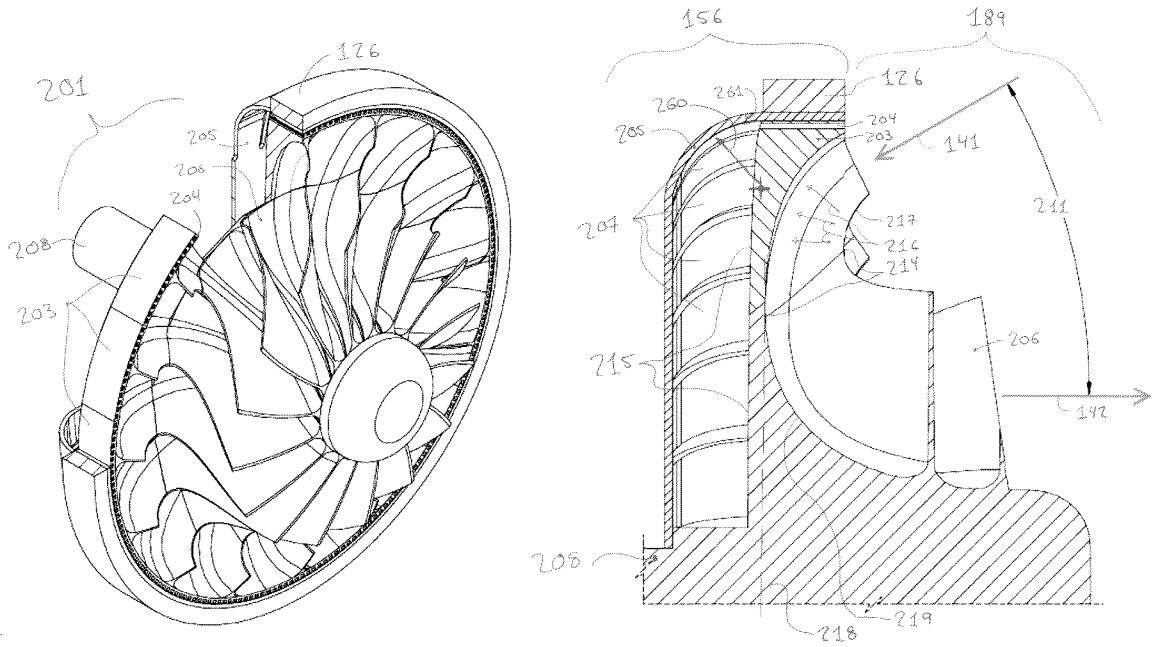


FIG. 25

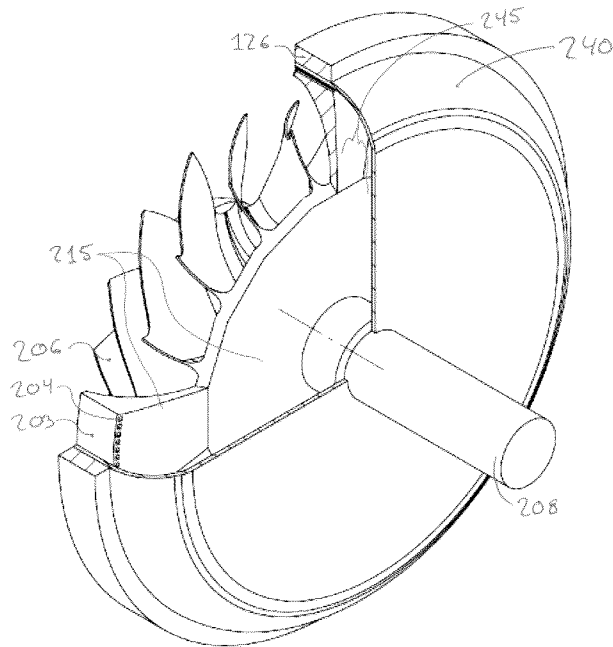


FIG. 26

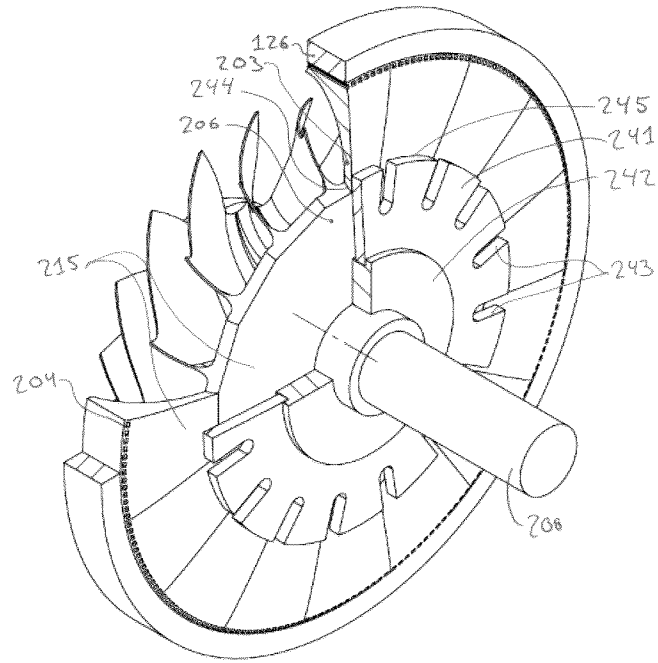


FIG. 27

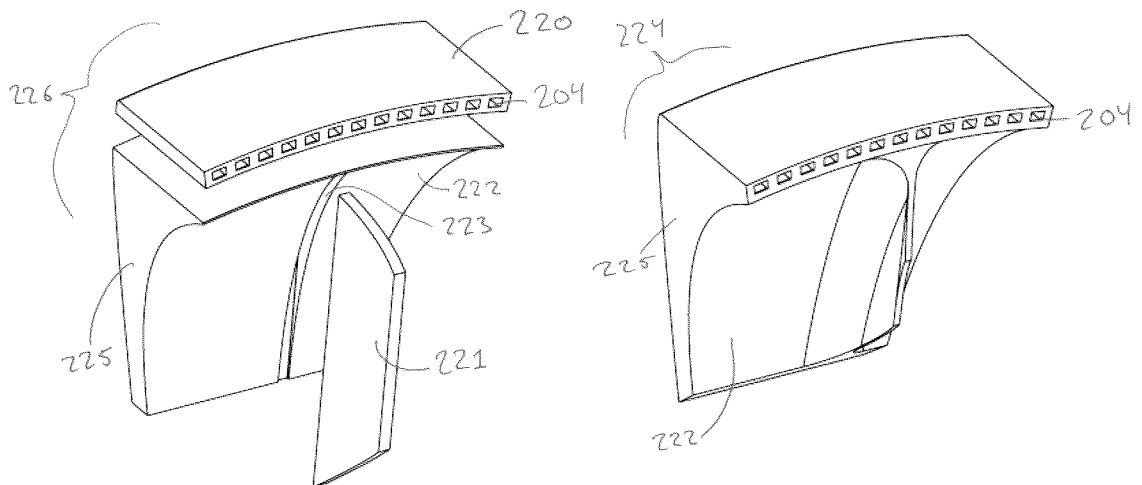


FIG. 28

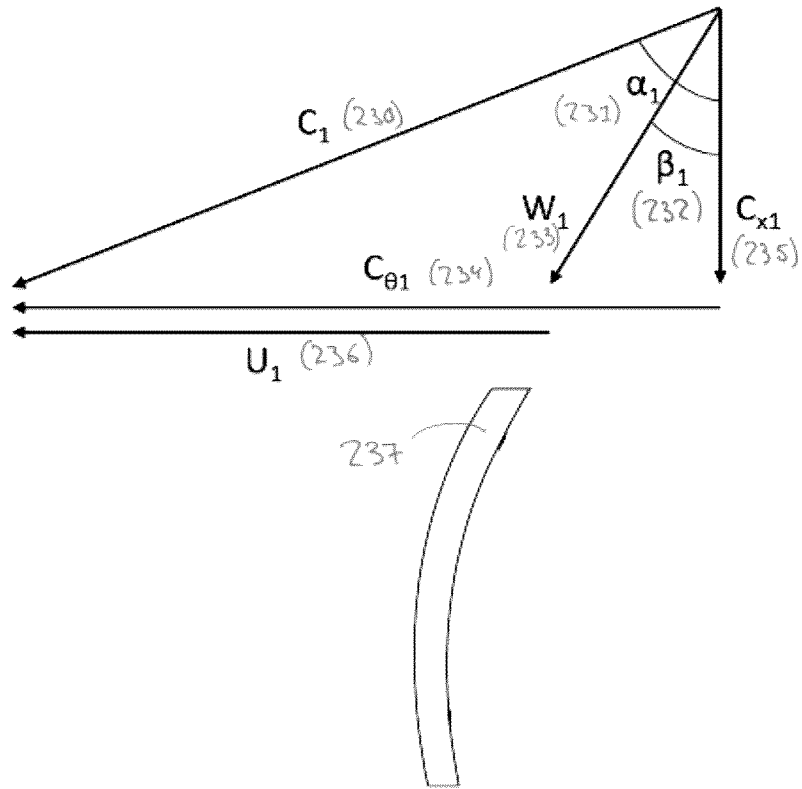
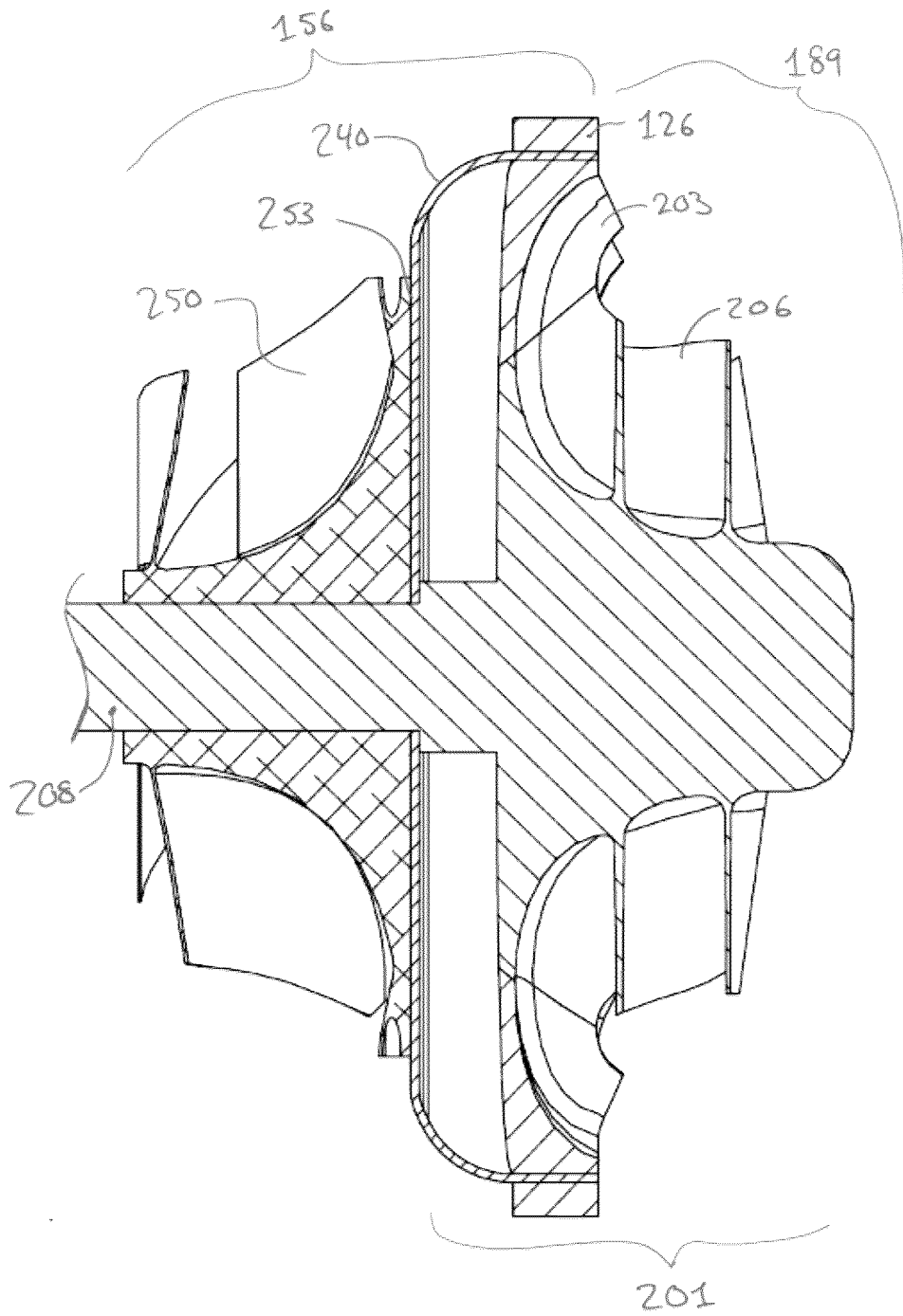


FIG. 29



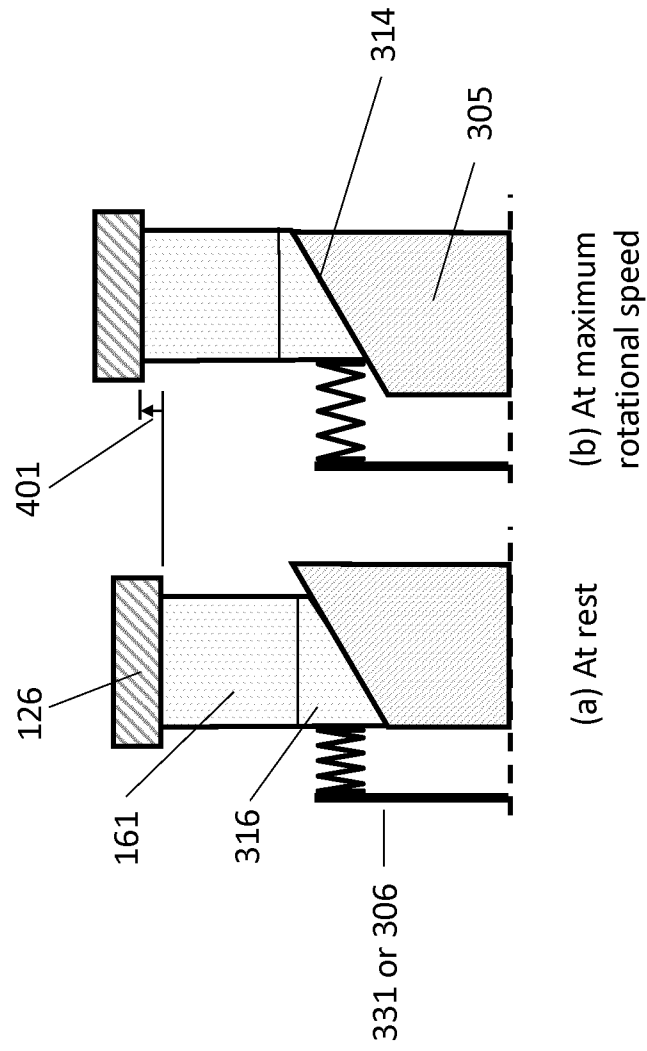


Fig. 30

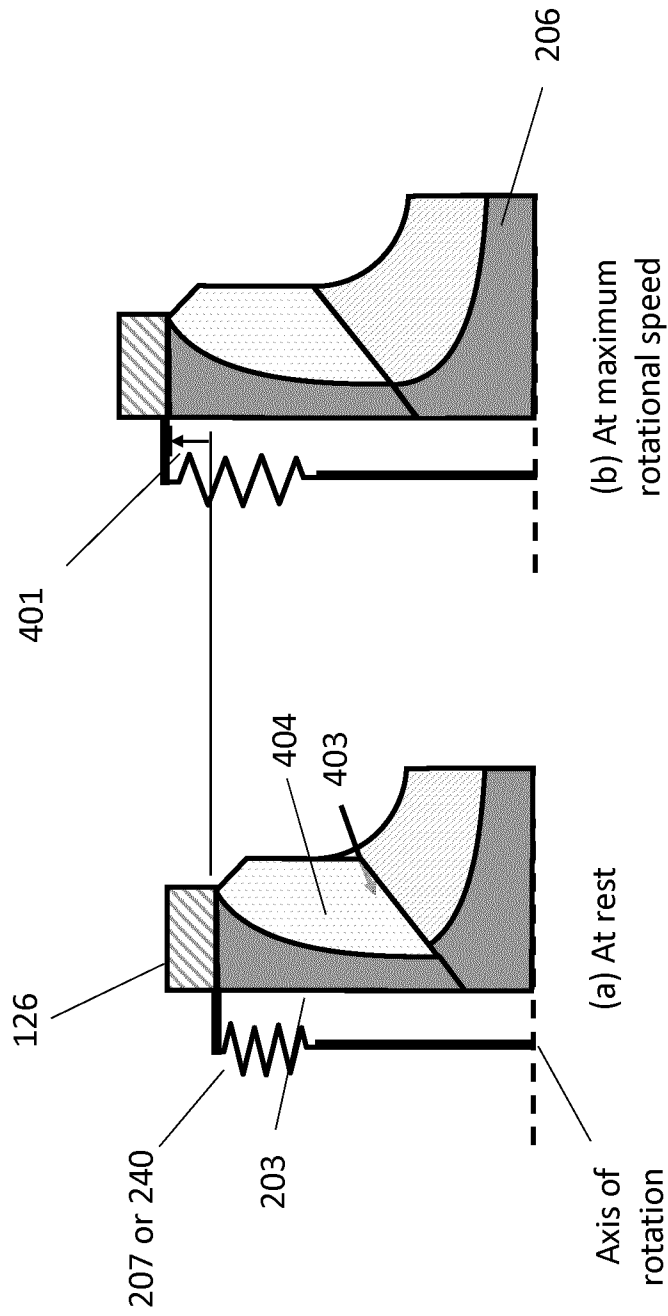


Fig. 32

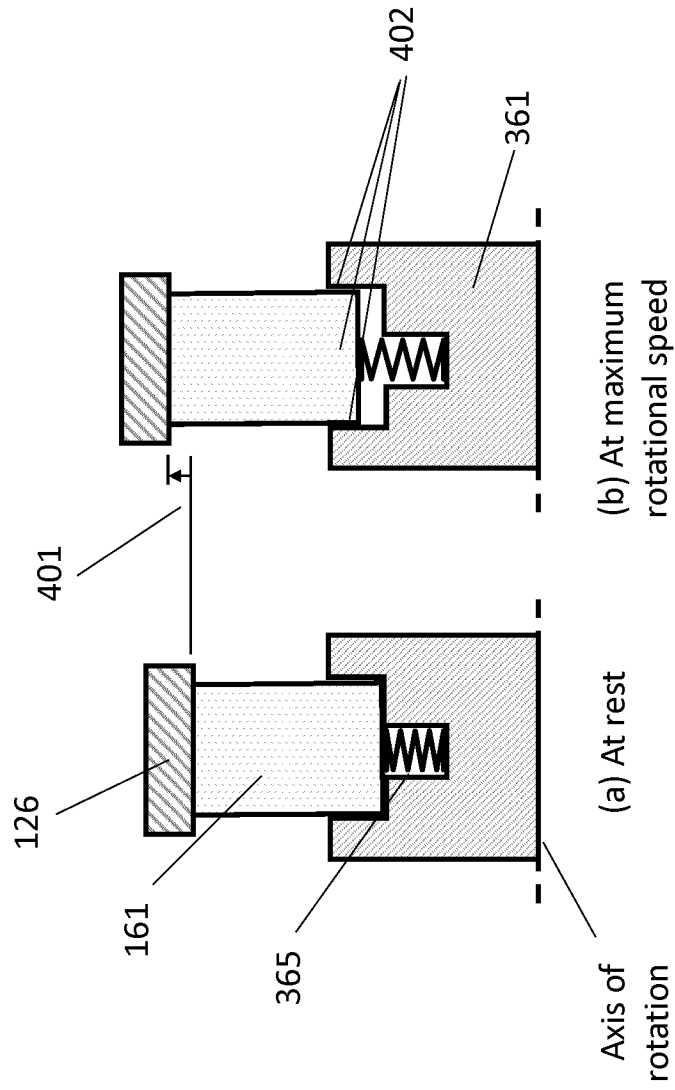


Fig. 33

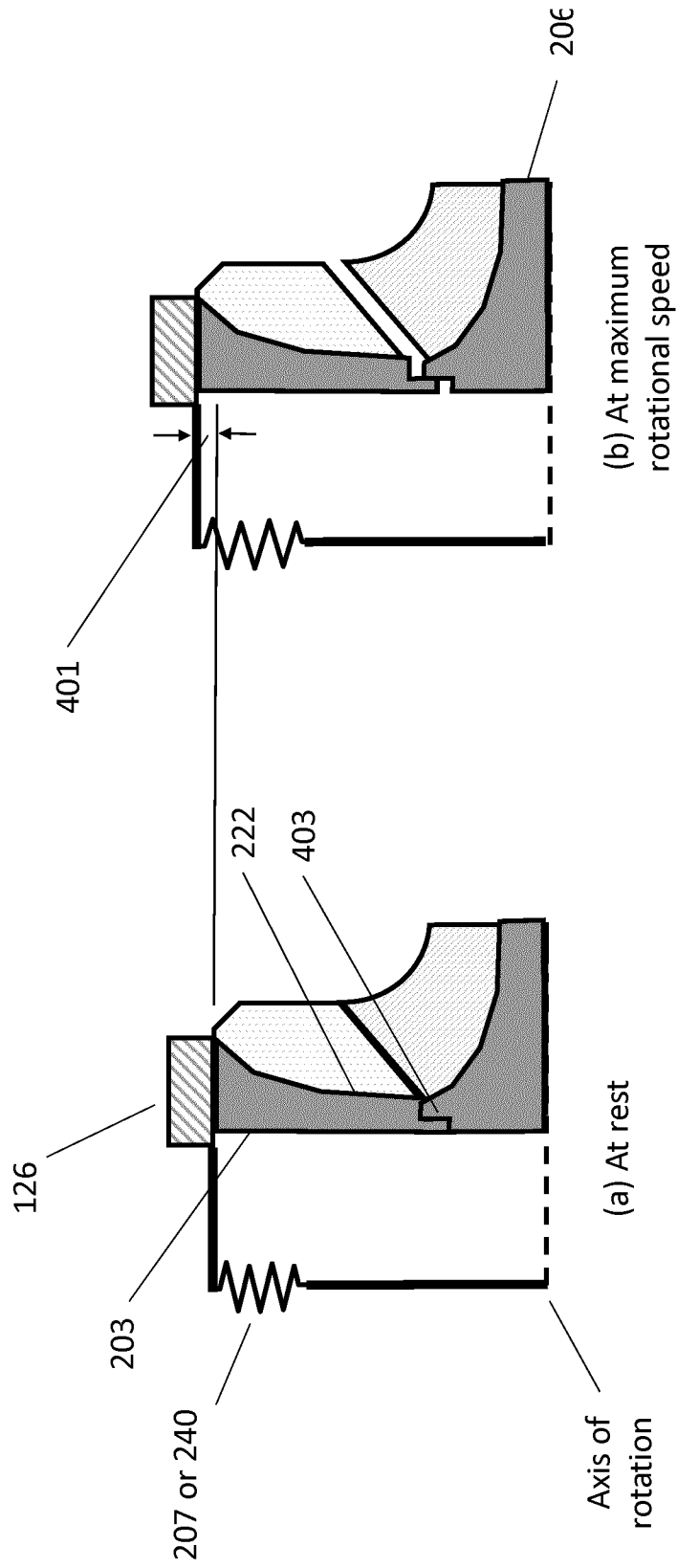


Fig. 34

INTERNATIONAL SEARCH REPORT

International application No.
PCT/CA2017/051423

<p>A. CLASSIFICATION OF SUBJECT MATTER IPC: F01D 5/28 (2006.01), F01D 5/02 (2006.01), F02C 3/04 (2006.01), F02C 3/16 (2006.01), F23R 3/42 (2006.01), F23R 3/56 (2006.01)</p> <p>According to International Patent Classification (IPC) or to both national classification and IPC</p>																				
<p>B. FIELDS SEARCHED</p> <p>Minimum documentation searched (classification system followed by classification symbols) IPC: F01D 5/28 (2006.01), F01D 5/02 (2006.01), F02C 3/04 (2006.01), F02C 3/16 (2006.01), F23R 3/42 (2006.01), F23R 3/56 (2006.01)</p> <p>Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched None</p> <p>Electronic database(s) consulted during the international search (name of database(s) and, where practicable, search terms used) Canadian Patents Database QUESTEL-ORBIT (FamPat database) Keywords: turbine; blade; rotor; r4e; rim rotor; centrifugal; displace+; deform+; radial; ceramic; thermal expan+</p>																				
<p>C. DOCUMENTS CONSIDERED TO BE RELEVANT</p> <table border="1"> <thead> <tr> <th>Category*</th> <th>Citation of document, with indication, where appropriate, of the relevant passages</th> <th>Relevant to claim No.</th> </tr> </thead> <tbody> <tr> <td>X</td> <td>GB 2 069 065 A (CAIN) 19 August 1981 (19-08-1981) * Whole document *</td> <td>1 - 4, 7 - 9, 19 - 23, 30, 31, 33, 38, 39, 41 & 42</td> </tr> <tr> <td>X</td> <td>US 4,111,603 A (STAHL) 05 September 1978 (05-09-1978) * Whole document *</td> <td>1, 2, 8, 9, 19 - 21, 30, 31, 39</td> </tr> <tr> <td>A</td> <td>US 7,393,182 B2 (MATHENY) 01 July 2008 (01-07-2008) * Whole document *</td> <td></td> </tr> <tr> <td>A</td> <td>WO 2013/040679 A1 (PLANTE ET AL.) 28 March 2013 (28-03-2013) * Same Applicant / Whole document *</td> <td></td> </tr> </tbody> </table>			Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.	X	GB 2 069 065 A (CAIN) 19 August 1981 (19-08-1981) * Whole document *	1 - 4, 7 - 9, 19 - 23, 30, 31, 33, 38, 39, 41 & 42	X	US 4,111,603 A (STAHL) 05 September 1978 (05-09-1978) * Whole document *	1, 2, 8, 9, 19 - 21, 30, 31, 39	A	US 7,393,182 B2 (MATHENY) 01 July 2008 (01-07-2008) * Whole document *		A	WO 2013/040679 A1 (PLANTE ET AL.) 28 March 2013 (28-03-2013) * Same Applicant / Whole document *				
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A	WO 2013/040679 A1 (PLANTE ET AL.) 28 March 2013 (28-03-2013) * Same Applicant / Whole document *																			
<p><input checked="" type="checkbox"/> Further documents are listed in the continuation of Box C. <input checked="" type="checkbox"/> See patent family annex.</p>																				
<table border="1"> <tr> <td>* Special categories of cited documents:</td> <td>"T"</td> <td>later document published after the international filing date or priority date and not in conflict with the application but cited to understand the principle or theory underlying the invention</td> </tr> <tr> <td>"A" document defining the general state of the art which is not considered to be of particular relevance</td> <td>"X"</td> <td>document of particular relevance; the claimed invention cannot be considered novel or cannot be considered to involve an inventive step when the document is taken alone</td> </tr> <tr> <td>"E" earlier application or patent but published on or after the international filing date</td> <td>"Y"</td> <td>document of particular relevance; the claimed invention cannot be considered to involve an inventive step when the document is combined with one or more other such documents, such combination being obvious to a person skilled in the art</td> </tr> <tr> <td>"L" document which may throw doubts on priority claim(s) or which is cited to establish the publication date of another citation or other special reason (as specified)</td> <td>"&"</td> <td>document member of the same patent family</td> </tr> <tr> <td>"O" document referring to an oral disclosure, use, exhibition or other means</td> <td></td> <td></td> </tr> <tr> <td>"P" document published prior to the international filing date but later than the priority date claimed</td> <td></td> <td></td> </tr> </table>			* Special categories of cited documents:	"T"	later document published after the international filing date or priority date and not in conflict with the application but cited to understand the principle or theory underlying the invention	"A" document defining the general state of the art which is not considered to be of particular relevance	"X"	document of particular relevance; the claimed invention cannot be considered novel or cannot be considered to involve an inventive step when the document is taken alone	"E" earlier application or patent but published on or after the international filing date	"Y"	document of particular relevance; the claimed invention cannot be considered to involve an inventive step when the document is combined with one or more other such documents, such combination being obvious to a person skilled in the art	"L" document which may throw doubts on priority claim(s) or which is cited to establish the publication date of another citation or other special reason (as specified)	"&"	document member of the same patent family	"O" document referring to an oral disclosure, use, exhibition or other means			"P" document published prior to the international filing date but later than the priority date claimed		
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Date of the actual completion of the international search 21 February 2018 (21-02-2018)		Date of mailing of the international search report 27 February 2018 (27-02-2018)																		
Name and mailing address of the ISA/CA Canadian Intellectual Property Office Place du Portage I, C114 - 1st Floor, Box PCT 50 Victoria Street Gatineau, Quebec K1A 0C9 Facsimile No.: 819-953-2476		Authorized officer Jason Fisher (819) 639-7805																		

INTERNATIONAL SEARCH REPORT

International application No.

PCT/CA2017/051423

C (Continuation). DOCUMENTS CONSIDERED TO BE RELEVANT

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