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(54) Title: **COBALT-BASE COMPOSITION AND METHOD FOR DIFFUSION BRAZE REPAIR OF SUPERALLOY ARTICLES**

(57) Abstract: An improved cobalt-base braze alloy composition and method for diffusion brazing are provided for use in repairing superalloy articles, such as gas turbine engines, power generation turbines, refinery equipment, and heat exchangers. The improved cobalt-base braze alloy composition includes nickel; at least one element selected from the group of rhenium, palladium, platinum, ruthenium, and iridium; at least one element selected from the group of boron and silicon; and the remaining balance consists of cobalt. This composition may also include aluminum and/or one or more rare earth/lanthanide series elements, and the composition may be combined with one or more powdered base metal superalloy compositions to form an improved diffusion braze alloy mixture. In the improved method for repairing superalloy articles, the foregoing mixture is applied to a region of the superalloy article to be repaired. The mixture is then heated to melt the cobalt-base braze alloy, thereby joining the base metal superalloy powder particles together, and joining the entire mixture to the region being repaired. The molten mixture is next subjected to a diffusion braze heat treatment cycle in order to break down undesirable boride and silicide phases and to diffuse the melting point depressants into the mixture. In a preferred embodiment, the long term diffusion heat treatment cycle consists of heating the repaired article to 2000 °F (1093 °C), holding that temperature for 2 hours, heating the repaired article to 2100 °F (1149 °C), holding that temperature for 22 hours, and cooling the article to 250 °F (121 °C). After cooling, an environmental coating is applied to the final repair composite, and this composite significantly improves the cyclic oxidation resistance of the coating compared to the properties of the superalloy base metal.

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## COBALT-BASE COMPOSITION AND METHOD FOR DIFFUSION BRAZE REPAIR OF SUPERALLOY ARTICLES

### REFERENCE TO RELATED APPLICATIONS

5           The present application claims priority to U.S. Patent Application filed April 8, 1997 and assigned Serial No. 08/827,723; U.S. Patent Application filed May 7, 1999 and assigned Serial No. 09/307,616; and U.S. Patent Application filed May 7, 1999 and assigned Serial No. 09/306,968, which are hereby incorporated by reference in their entirety.

10

### FIELD OF THE INVENTION

          This invention relates generally to diffusion braze repair of superalloy articles and more particularly to cobalt-base braze alloy compositions containing at  
15   least one of the following elements: rhenium, palladium, platinum, ruthenium, iridium; and to long term diffusion heat treatment of repaired superalloy articles.

### BACKGROUND OF THE INVENTION

          High temperature operating environments such as those present in gas turbine  
20   engines, power generation turbines, refinery equipment, and heat exchangers demand parts composed of a variety of cobalt-, iron-, and nickel-base metals known as superalloys. These superalloys are capable of withstanding extremely high temperatures for extended periods of time, but the extremely stressful temperature conditions to which superalloy articles are subjected eventually take their toll upon the metal in a number of ways.

25           The main types of damage to a superalloy article are cracks from thermal fatigue, wide gap cracks, foreign object impact damage, and dimensional reduction from mechanical wear. Because the cost of these superalloy components is quite high, there is considerable incentive to repair these types of defects rather than to scrap the part and replace it with a new one. The high cost of these components, as well as the fact that  
30   superalloy components, once damaged, tend to fail repeatedly in the same region, also makes it critical that any repairs made have mechanical, environmental, and processing properties equivalent to or better than the original superalloy base metal.

          Traditional methods for repairing damaged superalloy articles involve choosing or  
35   creating an alloyed combination of elements that will melt at a temperature below the melting temperature of the superalloy substrate. These compositions are known in the

industry as braze alloys, and the most useful prior art braze alloys are characterized as either nickel-base or cobalt-base alloys. Historically, the most popular braze alloys contain a melting point depressant such as silicon or boron; a complex of some of the same alloying elements used in the superalloy article to be repaired such as chromium,  
5 aluminum, titanium, tungsten, etc.; and either nickel or cobalt as the base. In fact, one braze alloy, sometimes known as B-28, is simply the combination of an alloy frequently used to manufacture cast turbine airfoils, named Rene'80, with about 2% boron.

Advances in the braze alloy composition art have introduced multi-constituent alloy compositions that are mixtures of at least one braze alloy and at least one base metal  
10 alloy, the base metal alloy differing from the braze alloy in that it melts at a higher temperature than the braze alloy and contains no melting point depressants that can weaken the repair site. These multi-constituent compositions result in stronger repairs because the low-melting brazing alloy liquefies first, wetting the base metal constituent and joining the entire mixture to the superalloy article.

15 Once a braze alloy or alloy mixture has been chosen, the damaged superalloy article is cleaned to remove any environmental coating that may be over the base metal and any oxides that may have developed inside the damaged regions. The braze alloy composition is then applied to the region to be repaired, and the article subjected to a high temperature brazing cycle to melt and join the braze alloy to the superalloy article. Upon  
20 the completion of this cycle, typical braze alloys will have formed undesirable large blocky or script-like brittle phases composed of chromium, titanium, and the family of refractory elements (e.g., tungsten, tantalum) combined with the melting point depressants. These brittle phases weaken the repair composite and cannot be removed from conventional braze alloys.

25 However, certain braze alloy compositions, known as diffusion braze alloys, are capable of withstanding higher temperatures than conventional braze alloys. Diffusion braze alloys form the same bad phases during brazing as conventional alloys, but diffusion braze alloys can be subjected to a second, long-term high temperature heat cycle known as a diffusion cycle. This diffusion cycle allows the brittle borides, carbides, and silicides to  
30 break down into fine, discrete blocky phases. The diffusion cycle also diffuses the elemental melting point depressants into the braze alloy matrix. These actions result in a stronger repair that is less susceptible to incipient melting when the part is returned to service.

Unfortunately, the diffusion braze alloys of the prior art have failed to attain the  
35 crucial part-like mechanical and environmental properties demanded by the increased stresses to which today's superalloy articles are subjected. The main reason for this failure

is that prior high temperature braze alloys and alloy powder mixtures tend to use only those elements present in the superalloy article being repaired.

This lack of flexibility in the compositions of the prior art has caused a stagnation in the development of truly new braze alloy compositions which employ elements and elemental combinations without regard to the composition of the superalloy substrate. As well, previous multi-constituent alloy compositions were so precisely matched to the particular superalloy to be repaired that it was considered unthinkable to select base metal powders for the mixture based solely on their mechanical and environmental properties.

For these reasons, prior art compositions cannot provide a flexible diffusion braze alloy system capable of accommodating various new elements and base metal powders to increase the strength, flow characteristics, and oxidation resistance of the braze alloy system. Prior art heat treatment cycles are similarly incapable of effectively breaking down brittle phases and allowing the elemental melting point depressants to diffuse both into the superalloy substrate and the base metal matrix. As well, prior art diffusion braze alloy compositions frequently rely upon intentional carbon additions for strength, and these prior art compositions do not effectively impart improved environmental resistance to the superalloy substrate and/or any environmental coating which may be applied to the substrate.

A need therefore exists for a new diffusion braze alloy system that desirably employs the elements rhenium, platinum, palladium, ruthenium, iridium, and/or aluminum in order to improve significantly over the hot corrosion and oxidation resistance properties provided by prior art braze alloys. Additionally, such an improved braze alloy composition preferably uses boron and silicon concurrently as melting point depressants in order to reduce the undesirable mechanical and environmental properties associated with the use of either boron or silicon alone. The present invention addresses these needs.

## SUMMARY OF THE INVENTION

5 Briefly describing one aspect of the present invention, there is provided an improved cobalt-base braze alloy composition and method for diffusion braze repair of superalloy articles that achieves mechanical, processing, and environmental properties equivalent to and, in many cases, better than those properties possessed by the superalloy articles. The present cobalt-base braze alloy composition comprises nickel; at least one  
10 element selected from the following group: rhenium, palladium, platinum, ruthenium, iridium; boron; silicon; and cobalt. This composition may also include one or more of the rare earth elements such as yttrium, cerium, lanthanum, and other lanthanide series elements; aluminum; chromium; titanium; tungsten; molybdenum; niobium; hafnium; tantalum; iron; manganese; and/or zirconium, which elements appear in many advanced  
15 superalloy base metal compositions. This cobalt-base braze alloy composition may be combined with one or more powdered base metal superalloy compositions to form an improved diffusion braze alloy mixture having enhanced mechanical, environmental, and processing properties compared to prior art braze alloy mixtures. The present invention also provides new cobalt-base base metal alloy compositions for use in such improved  
20 diffusion braze alloy mixtures, which base metal alloy compositions do not include melting point depressants but which are otherwise similar to those of the braze alloy compositions.

In the case of non-eutectic alloys according to the present invention, the instant invention employs melting point depressants such as boron, silicon, and aluminum to  
25 reduce the melting point of the braze alloy. Although the present braze alloy compositions contain relatively low amounts of melting point depressants, these depressants nonetheless adversely affect the mechanical and/or environmental properties of a repaired article unless they are subjected to a long-term diffusion heat treatment cycle.

The present invention therefore also describes an improved diffusion heat  
30 treatment method to break down the undesirable phases formed by the melting point depressant(s) and diffuse the depressant(s) into the base metal alloy matrix. Use of this long-term diffusion heat treatment method minimizes the negative properties associated with the use of conventional melting point depressants.

In the brazing method of the present invention, a damaged region of a superalloy  
35 article is repaired by first cleaning the article by any conventional means; preparing a braze alloy composition mixture according to the present invention, wherein the

mechanical and environmental properties of that mixture are chosen to equal and preferably improve upon those properties of the superalloy article to be repaired; depositing this mixture on the region to be repaired; and placing the superalloy article in a furnace under an inert gas atmosphere or under a vacuum. Once in such a furnace, the pressure in the furnace chamber should be reduced to approximately  $1 \times 10^{-3}$  torr or a lower pressure and the brazing cycle initiated by heating the repaired region to a temperature of about 800°F (427°C). The 800°F (427°C) temperature is maintained for approximately 15 minutes, whereafter the temperature is increased to about 1800°F (982°C) and that temperature maintained for approximately 15 minutes. Next, the temperature is again raised to a temperature less than the incipient melting temperature of the article being repaired, which incipient melting temperature typically exceeds 2350°F (1288°C), and that less than incipient melting temperature maintained for between 15 and 45 minutes. Finally, the furnace is vacuum cooled from the less than incipient melting temperature to about 1800°F (982°C). This step completes the conventional brazing cycle which causes the formation of undesirable brittle phases. The next steps in the present method constitute the diffusion heat treatment cycle that will break down these brittle phases.

Upon completion of the high temperature brazing cycle, the superalloy article is subjected to a pressure higher than the pressure used in the brazing cycle and reheated to a temperature of between 1 and 400°F (1°-225°C) below the chosen brazing temperature for the article. This temperature is maintained for at least 20 hours, whereafter the temperature is lowered to about 250°F (121°C). At this point, the superalloy article is fully repaired and ready for machining.

The superalloy article is then usually coated with a metal or ceramic, diffusion or overlay coating according to any known application method. This coating protects the superalloy base metal from oxidation and hot corrosion attack, and, if the superalloy article is given a multi-layer coating of which at least one layer is a cobalt-base braze alloy according to the present invention, the coating remains resistant to environmental attack much longer than a traditional coating.

These and other objects, advantages, and features are accomplished according to the compositions and methods of the following description of the preferred embodiment of the present invention.

## DESCRIPTION OF THE PREFERRED EMBODIMENT

For the purposes of promoting an understanding of the principles of the invention, reference will now be made to the preferred embodiments thereof, and specific language  
5 will be used to describe the same. It will nevertheless be understood that no limitation of the scope of the invention is thereby intended, such alterations, modifications, and further applications of the principles of the invention being contemplated as would normally occur to one skilled in the art to which the invention relates.

The principal objective of the present invention is to achieve mechanical,  
10 processing, and environmental resistance properties in a braze alloy repair composite that equal, if not exceed, the levels of these properties enjoyed by the superalloy substrate, or base metal. Prior art braze alloy compositions have failed to achieve this objective for several reasons. First, prior art braze alloy systems are unable to reduce the melting point of the brazing alloy without embrittling either the repair composite or the superalloy  
15 substrate. Second, skilled artisans have for so long considered it crucial that the elements of the braze alloy system match those of the superalloy article to be repaired that it was thought impossible or unworkable to repair a superalloy article using any other elements. And lastly, this old way of thinking about braze alloy compositions prevented artisans from investigating what base metal alloys could be added to a braze alloy system to  
20 improve a repair's solid solution strengthening and oxidation resistance properties.

It is clear that braze alloy systems having the foregoing problems cannot effectively repair today's higher temperature and higher strength superalloys which undergo greater mechanical and thermal stresses than ever before, and which cost more to manufacture than ever before. Therefore, the compositions and method of the present  
25 invention introduce new elements and elemental combinations not previously considered for use in the field of diffusion braze repair in order to overcome the disadvantages of the prior art and to give the superalloy component a longer useful life than was previously considered attainable.

The cobalt-base diffusion braze alloy composition of the present invention has the  
30 following general composition range, listed in Table 1 in percent by weight based on the total weight of the alloy composition:

Table 1

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	0.001-<Co
Chromium	0-40
Aluminum	0-12
Titanium	0-6
Tungsten	0-15
Molybdenum	0-15
Niobium	0-12
Ruthenium	0.001-12
Rhenium	0.001-15
Iridium	0.001-12
Hafnium	0-6
Tantalum	0-15
Platinum	0.001-40
Palladium	0.001-40
Iron	0-3
Manganese	0-1
Rare Earth/Lanthanide	0-5
Series Element(s)	
Carbon	0-2
Boron	0.001-6
Silicon	0.001-10
Zirconium	0-2

- 5 While the foregoing constitutes a general description of the cobalt-base braze alloy composition of the present invention, the following are specific examples of preferred compositions according to the present invention. These specific examples are provided for purposes of illustrating the invention, and no limitations on the invention are intended thereby. Before proceeding further, it should also be noted that when zeroes are used in
- 10 the composition tables of the present invention, those zeroes indicate no intentional addition of the element, not that the element is absent from the composition. It is usually not economically feasible to use 100% pure elemental additions, and therefore some impurities may be introduced into the composition.

- 15 A first preferred embodiment of the cobalt-base braze alloy composition of the present invention is listed in Table 2 and is known as "RCA-C1", which has the following composition in percent by weight:



Table 2

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10.5
Chromium	23
Aluminum	1.5
Titanium	1.75
Tungsten	3
Rhenium	1
Tantalum	6
Platinum	0-40
Palladium	0-40
Carbon	0-0.55
Boron	1.5
Silicon	5

- 5 A second preferred embodiment of the cobalt-base braze alloy composition of the present invention is listed in Table 3 and is known as "RCA-C2", which has the following composition in percent by weight:

Table 3

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10
Chromium	22.5
Titanium	0.1
Tungsten	7
Rhenium	0.001-15
Tantalum	3.5
Platinum	0-40
Palladium	0-40
Carbon	0-0.6
Boron	1.5
Silicon	5
Zirconium	0.5

10

A third preferred embodiment of the cobalt-base braze alloy composition of the present invention is listed in Table 4 and is known as "RCA-C3", which has the following composition in percent by weight:

Table 4

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10.5
Chromium	20.5
Aluminum	2.25
Tungsten	1.25
Rhenium	1
Tantalum	7.75
Platinum	0-40
Palladium	0-40
Carbon	0-0.28
Boron	3

- 5           A fourth preferred embodiment of the cobalt-base braze alloy composition of the present invention is listed in Table 5 and is known as "RCA-C6", which has the following composition in percent by weight:

Table 5

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	30.75
Chromium	14.75
Aluminum	3.38
Tungsten	1.38
Rhenium	0.5
Tantalum	8.8
Hafnium	0.5
Platinum	0-40
Palladium	3
Carbon	0-0.3
Boron	2.33
Silicon	3.38

10

A most preferred embodiment of the cobalt-base braze alloy composition of the present invention is listed in Table 6 and is known as "RCA-C4", which has the following composition in percent by weight:

Table 6

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10.5
Chromium	23
Aluminum	1.75
Tungsten	1.25
Rhenium	1
Platinum	0-40
Palladium	0-40
Tantalum	6.5
Carbon	0-0.55
Boron	2.15
Silicon	3.25

5           Turning now to discuss the novelty of the foregoing compositions, it will be obvious to one of ordinary skill that certain preferred embodiments of the instant diffusion  
 braze alloy compositions are formulated with concurrent boron and silicon additions as  
 melting point depressants. Prior art braze alloys, in contrast, have traditionally used boron  
 alone as the melting point depressant for two major reasons: (1) boron diffuses  
 10 exceptionally well into the base metal matrix of a braze alloy mixture, and (2) this boron  
 diffusion results in a higher remelt temperature of the final repair composite. Boron in a  
 braze alloy thus ensures that the repair composite will be able to withstand the same high  
 temperatures withstood by the superalloy substrate itself.

15           However, exceptionally high boron concentrations in a braze alloy promote embrittlement of the superalloy base metal and incipient melting. These deleterious  
 effects reduce the number of repairs that can be performed upon any one region of a  
 superalloy article and thereby shorten the operating life of the part since superalloy  
 components tend to fail repeatedly in the same area.

20           Silicon alone is typically used in conventional (non-diffusion) brazing alloy compositions to speed the alloy's rate of flow into a damaged region. Unfortunately,  
 silicon-only braze alloys do not typically have a high degree of diffusivity into the base  
 metal matrix, and they tend to form very stable silicides. These silicides form large,  
 brittle, script-like phases in the microstructure of the repair composite, which phases can  
 degrade the mechanical properties of both the repair composite and the superalloy base  
 25 metal.

Embodiments of the present invention combine the two elements to minimize the  
 undesirable effects of using either boron or silicon alone and maximize the beneficial

properties imparted by each element. When boron and silicon are combined, the amount of boron necessary to reduce the melting temperature of the alloy is decreased, which reduces the high concentrations of boron in the superalloy substrate. The instant braze alloy system thus enjoys the strength and high temperature melting properties imparted by  
5 boron without having to sacrifice the superalloy base metal in the process.

Similarly, the silicon additions in the present braze alloy compositions improve the flow characteristics of the braze alloy without embrittling the repair composite with large amounts of script-like silicide phases. This latter benefit is assured when the long-term diffusion heat treatment cycle of the present invention is used to homogenize the  
10 braze alloy/base metal mixture and diffuse the elemental boron and silicon into the base metal matrix. Silicon also has the unexpected benefit of improving the performance of any environmental coating placed over the repaired region. This feature helps assure long life of the repaired area and gives it improved environmental resistance properties over the original superalloy substrate.

15 It should be understood, however, that the use of either boron or silicon alone as a melting point depressant is also considered and intended to come within the scope of the present invention. As will be discussed in greater detail below, the use of iridium, ruthenium, palladium, platinum, and especially rhenium in the preferred compositions of the present invention significantly reduce the deleterious brittle phases associated with the  
20 use of boron alone and thereby help to increase the re-melt temperature of the final repair composite. The present compositions, therefore, achieve unexpected results over traditional boron- or silicon-only diffusion braze alloys.

Cobalt-base base metal alloy compositions are also intended to come within the scope of the present invention. As discussed previously, braze alloy compositions may be  
25 used alone to repair part damage, but significant benefits in mechanical strength and processing properties can be achieved when a part is repaired using a mixture of one or more braze alloys and one or more base metal components. The main reason for these improvements over single-component braze alloy systems is that the amount of melting point depressants used can be significantly reduced. To achieve such property  
30 improvements, then, the present invention has described braze alloy compositions which may be combined with any known superalloy base metal to create an improved repair composite. The following discussion describes new base metal alloy compositions that can be combined with the instant and/or any other known braze alloy compositions to also create an improved repair composite.

35 The base metal alloy compositions described herein possess the same general composition range as the braze alloy compositions of the present invention, but obviously

do not include boron or silicon. Therefore, the instant base metal alloy compositions listed in Table 7 and comprise generally, in percent by weight:

Table 7

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	0.001-<Co
Chromium	0-40
Aluminum	0-12
Titanium	0-6
Tungsten	0-15
Molybdenum	0-15
Niobium	0-12
Ruthenium	0.001-12
Rhenium	0.001-15
Iridium	0.001-12
Hafnium	0-6
Tantalum	0-15
Platinum	0.001-40
Palladium	0.001-40
Iron	0-3
Manganese	0-1
Rare Earth/Lanthanide Series Element(s)	0-5
Carbon	0-2
Zirconium	0-2

5

While the foregoing constitutes a general description of the cobalt-base base metal alloy composition of the present invention, the following are specific examples of preferred compositions according to the present invention. These specific examples are provided for purposes of illustrating the invention, and no limitations on the invention are intended thereby.

10

A first preferred embodiment of the cobalt-base base metal alloy composition of the present invention listed in Table 8 is known as "RCA-B1", which has the following composition in percent by weight:

Table 8

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10
Chromium	22.5
Aluminum	2
Tungsten	5
Rhenium	0.5
Tantalum	6
Platinum	0-40
Palladium	0-40
Carbon	0-0.55
Zirconium	0.5

- 5        A most preferred embodiment of the cobalt-base base metal alloy composition of the present invention is listed in Table 9 and known as "RCA-B2", which has the following composition in percent by weight:

Table 9

<u>Elements</u>	<u>Weight Percent</u>
Cobalt	Balance
Nickel	10.5
Chromium	22
Aluminum	1.75
Tungsten	4
Tantalum	6.5
Rhenium	0-15
Palladium	0-40
Platinum	0.001-40
Carbon	0-0.55

10

- As can be seen in all the foregoing diffusion alloy compositions, braze alloys and base metal alloys alike, the instant alloy compositions contemplate use of one or more elements from the following group: rhenium, palladium, platinum, ruthenium, and iridium. The use of rhenium, palladium, platinum, ruthenium, and iridium in cobalt-base diffusion alloys represents a significant advance in the art of diffusion braze repair of superalloy articles because it departs radically from the traditional diffusion braze alloy composition: a powder of the same superalloy as the damaged component with a measure of melting point depressants added to lower the brazing temperature. These new alloy compositions are formulated to not only repair, but also to improve, the mechanical, processing, and environmental properties possessed by the superalloy base metal.
- 15
- 20

It is well known that failures in superalloy components regularly occur in the same or an immediately adjacent location. It is therefore extremely important that these areas of fatigue be repaired to be even stronger than the original superalloy base metal. The compositions of the present invention achieve this objective by successfully combining  
5 certain elements such as rhenium, platinum group elements, aluminum, and one or more rare earth/lanthanide series elements, and by removing carbon from the compositions.

The first of these new preferred elements, rhenium, is preferably added to the cobalt-base alloy compositions of the present invention in an amount from 0 to 15 weight percent. Rhenium additions give the present compositions significantly  
10 improved mechanical and environmental properties over other, more traditional, solid solution strengthening elements such as tungsten, molybdenum, or hafnium. The mechanical properties associated with rhenium compositions are similar to those achievable by using tungsten and molybdenum; however, rhenium has significant oxidation resistance properties that the tungsten and molybdenum-type  
15 elements do not have. Therefore, the inclusion of rhenium in the compositions of the present invention permits a skilled artisan to reduce or completely remove other solid solution strengthening elements that are undesirable for use in oxidizing environments. Of additional benefit to the preferred composition embodiments, rhenium does not promote sigma phase formation in the repair composite or the  
20 adjacent superalloy base metal.

Another benefit of rhenium-containing compositions according to the present invention relates to rhenium's effect on melting point depressants in the alloy matrix. Unexpectedly, the addition of rhenium to the present preferred braze alloy compositions works so well to bind up significant amounts of melting point  
25 depressants that the elements which traditionally form brittle phases (e.g., chromium, tungsten) are left in solid solution to strengthen the repair composite and improve environmental resistance. As well, the instant preferred compositions eliminate the diffusion of excess melting point depressants into the adjacent base metal of the superalloy article. This is true even when silicon is not used  
30 concurrently with boron, and the amount of melting point depressants that can be successfully incorporated in the alloy matrix increases with the length of the long-term heat treatment diffusion cycle. The present compositions can therefore use boron alone to lower the melting temperature of the braze alloy and achieve the

benefit of a higher re-melt temperature for the repair composite without experiencing the weak and destructive brittle phases or the excess boron diffusion experienced with the prior art boron-containing braze alloys.

Platinum may be added to the present compositions in a range of from 0 to 40 weight percent. The addition of platinum and/or other platinum group elements, such as ruthenium, osmium, rhodium, iridium, and palladium, improves the hot corrosion and oxidation resistance properties of the repair composite. As well, platinum and other platinum group metals added to the present invention in sufficiently high concentrations improve the ductility, or plasticity, of the repair composite.

The addition of palladium is contemplated by the present invention because it achieves improvements in the repair composite similar to those achieved by platinum. For example, palladium enhances the oxidation resistance of the repair site and improves the ductility of the repair composite. Palladium also enhances the flow characteristics of the instant braze alloy compositions, and nickel and palladium are 100% soluble when combined in a braze alloy mixture. Further, palladium additions have been shown to retard the formation of undesirable borides and silicides in the alloy matrix.

The addition of ruthenium in the compositions of the present invention improves the repair composite in ways similar to those discussed above for the other platinum group additions. Ruthenium additions are also beneficial in that they reduce alloy density while simultaneously providing strength equivalent to or better than that achieved by the foregoing elemental additions. This strength characteristic is especially beneficial given that ruthenium's atomic weight is 30% less than other similar refractory elements commonly used in cobalt superalloys. An increase in the strength to weight ratio presents a significant benefit to the aerospace industry because a lighter structure having strength equal to or greater than a structure formed from more traditional, heavy materials may be formed using the compositions of the present invention.

Another element contemplated by and intended to come within the scope of the present invention is aluminum. Conventional high temperature braze alloys such as AMS 4783 do not have aluminum in them. This is because aluminum reduces the flowability of the braze by the rapid formation of aluminum oxide, a material commonly used for the prevention of braze flow. Additionally, different surface tensions and viscosities occur that change the braze flow characteristics when aluminum is used. Because diffusion braze alloys do not have the same flow requirements as conventional braze alloys, diffusion braze alloys allow the use of aluminum. Nonetheless, aluminum is not normally used in cobalt-base superalloy repair because prior art repair systems typically use



powdered cobalt superalloys combined with a braze alloy to repair a cobalt superalloy substrate, and cobalt superalloys do not contain aluminum.

Cobalt superalloys are typically used in the temperature range at which the superalloy base metal is subject to hot corrosion attack and damage. Certain turbine  
5 manufacturers have recently begun to push the operating temperatures for cobalt superalloys above this temperature range and into the oxidation mode of base metal attack and damage. It is for this reason that the present invention includes aluminum in a cobalt-base diffusion braze alloy composition. By including aluminum in the instant compositions, the final repair composite receives additional protection from preferential  
10 oxidation at the repaired areas of the superalloy components; the gamma prime phase of the alloy matrix is strengthened over non-aluminum containing cobalt-base braze alloys; and the introduction of aluminum helps reduce the melting point of the braze alloy composition. These benefits outweigh any previously encountered difficulties with braze flow characteristics, and the inclusion of aluminum represents a significant advance in the  
15 diffusion braze alloy art.

The present invention also contemplates and intends that the preferred embodiments of the present invention incorporate one or more rare earth elements such as yttrium, cerium, lanthanum, and other lanthanide series elements. This addition so significantly improves the unexpected and novel oxidation resistance enjoyed by the braze  
20 alloy and the base metal alloy compositions of the present invention and the brazeability of the instant braze alloy compositions that the amounts of aluminum, boron, and silicon used in these compositions may be reduced. By reducing the aluminum content, any problems that might arise from alumina formation during brazing can be minimized. The reduction of boron and silicon additions permits the properties of the repair composite to more  
25 closely resemble the properties of the base metal substrate, and the reduction of boron yields additional oxidation resistance in the present braze alloy compositions.

It is well known in the art that, other than using solid solution strengthening elements, carbides are the primary strengthening mechanism for cobalt-base alloys. Because the compositions of the present invention include such effective solid solution  
30 strengthening elements as rhenium, and because the present compositions contemplate the use of silicides and/or borides to strengthen the alloy matrix as effectively as carbides, carbon may effectively be removed from the present compositions without suffering any loss in mechanical properties.

It is particularly beneficial to remove carbon from diffusion alloy compositions  
35 because carbon prefers to agglomerate and precipitate out of the alloy matrix at lower temperatures. Carbides therefore exhibit poor ductility and have poor oxidation resistance.

Carbide particles in a cobalt-base alloy system also tend to go into solution in the alloy matrix and disappear at high temperatures. However, as soon as the superalloy cools, the carbides precipitate out of the matrix and form a carbide line at the interface of the repair composite and the superalloy substrate. This carbide line allows the repair composite to  
5 break away from the superalloy substrate in a zipper-like fashion. The mere possibility of such a significant repair failure makes removing carbon from the present invention a significant improvement in the art.

Of importance, the most preferred embodiments of the present compositions are prealloyed powders. The prealloying is accomplished using well-known methods  
10 according to the following procedure: the basic elements are first mixed in the required weight percentages in a container; this mixture is then melted at high temperature; and the molten mixture is atomized by spraying the metal through a high pressure nozzle and cooling it with argon gas. This technique solidifies the once discrete elements into uniform powder particles. Skilled artisans will recognize that the properties of a  
15 prealloyed mixture are significantly different from those of a simple mixture of elements, and the improvements achieved by the present invention rely in part upon the fact that these compositions are prealloyed.

The present alloy compositions contemplate the inclusion of a number of other elements typically used in advanced superalloy compositions, including solid solution  
20 strengtheners such as cobalt, molybdenum, and tungsten; gamma-prime formers such as nickel, hafnium, niobium, titanium, and tantalum; sacrificial oxide formers such as chromium; carbide formers such as zirconium; elements to improve ductility such as manganese; and other elements such as iron. Because these elements are commonly used in superalloy base metals and braze alloys and because the properties they impart to those  
25 systems are well known in the art, those of ordinary skill will understand which elements to choose to customize the instant compositions to their specifications.

Having now described the preferred composition formulations of the present invention, it is necessary to discuss the preferred mixtures for use in repairing a damaged superalloy component. It is known in the art of superalloy repair that combining in a braze  
30 alloy mixture a high temperature melting composition and one or more compositions which melt at a lower temperature will improve the strength of the repair composite while still providing adequate flow characteristics to facilitate placement and insertion of the braze alloy system into the damaged region. However, the high temperature component used in prior mixtures was nothing more than a powder of the same superalloy as the  
35 article being repaired.

The present invention, in contrast, describes a diffusion braze alloy system that employs base metal powders chosen without regard to the composition of the superalloy substrate. Instead, the present invention chooses which base metal powders to incorporate based on the properties those base metals will impart to the braze alloy system or, the  
5 repair composite. In certain preferred embodiments of the present invention, the use of multiple base metal components, whether iron-, cobalt-, or nickel-base, enhances the mechanical, environmental, and processing properties of the instant braze alloy system.

As an example, one base metal powder may be chosen for its strength and another base metal powder chosen for its improved braze flow characteristics. One preferred  
10 embodiment of the mixture of the present invention uses a base metal alloy powder known in the industry as Mar-M509. Mar-M509 is known to provide a very strong repair composite, but it is not preferred for use in diffusion braze repair because it slows the flow of molten braze mixture during the high temperature braze cycle. This slow flow characteristic is especially undesirable when the damage to the superalloy article is in the  
15 form of a crack or a wide gap. It is therefore desirable when repairing cracks and gaps to include a second base metal powder known in the industry as X40. When used alone, X40 makes for a relatively weak repair composite, but when combined with Mar-M509, it improves the flow characteristics of the braze alloy system and permits cracks and gaps to be filled with a stronger repair composite. Certain other preferred embodiments of the  
20 present invention choose the high temperature base metal alloy compositions of the present invention in order to impart the improved properties associated with those base metal powders to the braze alloy mixture.

Although the following may generally be known in the industry, it is instructive for practicing the present invention that in the embodiments of the present braze alloy  
25 mixtures preferred for repairing cracks, the braze alloy composition or compositions comprise no more than 50% by weight of the total braze alloy mixture. Wide cracks and gaps may be repaired with the present mixtures if the percentage by weight of the braze alloy composition or compositions is kept to about 40%. Similarly, dimensional repairs, or build-ups, are most effectively performed when the total weight of braze alloy in the  
30 mixture does not exceed 40%.

It will be obvious to those of ordinary skill which mixture percentages should be applied to which types of structural damage. Accordingly, one preferred embodiment of the braze alloy mixture of the present invention comprises a powder metal slurry. Another preferred embodiment of the present mixture invention takes the form of a plasticized  
35 powdered metal alloy tape. Another preferred embodiment of this mixture comprises a

pre-sintered alloy tape. Alternatively, one preferred embodiment of the present invention especially useful for dimensional repair comprises a pre-sintered alloy preform.

In practice, after the damage has been assessed, the preferred braze alloy composition or compositions of the present invention chosen, the base metal alloy  
5 composition or compositions chosen, and the braze alloy and base metal compositions combined in the appropriate ratio corresponding to the damage to be repaired, the superalloy article is cleaned of all coatings and oxides using techniques known in the art for such cleaning. The chosen braze alloy mixture in the embodiment appropriate to repair the damage, e.g., powder metal slurry, tape,  
10 etc., is then applied to the damaged region and the superalloy article subjected to a high temperature brazing cycle in a vacuum or in an inert gas atmosphere. This high temperature brazing cycle melts the braze alloy portion of the mixture, thereby creating a base metal powder matrix within the braze alloy composition, and joining the entire mixture to the now-repaired superalloy substrate.

15 One preferred inventive method for repairing damaged superalloy components involves a high temperature brazing cycle having the following steps: placing the mixture-coated superalloy article in an inert gas atmosphere or under vacuum in a brazing furnace; obtaining a pressure of  $1 \times 10^{-3}$  torr or lower pressure in the inert gas atmosphere or under the vacuum; heating the braze alloy mixture to  
20 a temperature of about 800°F (427°C) and holding that temperature for approximately 15 minutes; thereafter increasing the temperature to about 1800°F (982°C) and holding that temperature for approximately 15 minutes; then increasing the temperature again to a temperature less than the incipient melting temperature of the article being repaired, which incipient melting temperature  
25 typically exceeds 2350°F (1288°C), and holding that less than incipient melting temperature for between 15 and 45 minutes; whereafter the furnace is vacuum cooled from the less than incipient melting temperature to about 1800°F (982°C).

While the foregoing high temperature braze cycle has been described, it will be understood by skilled artisans that any series of temperatures and brazing  
30 times capable of melting only the braze alloy composition and permitting that braze alloy composition sufficient time to flow and effect the repair while forming a solid solution matrix and precipitating gamma-prime phase particles are considered and intended to be encompassed herein. Those of ordinary skill in the

art will also understand that the lower the pressure in the brazing furnace during this brazing cycle, the lower the vapor pressure of the sacrificial oxide forming elements, and thus the better the flow of the braze alloy during the braze cycle.

The next series of steps in the preferred repair method of the present invention comprise the long term diffusion heat treatment cycle. The diffusion cycle facilitates homogenizing the remaining solidified braze alloy system microstructure and diffusing the elemental melting point depressants into the alloy matrix. The particular steps used in this diffusion heat treatment cycle comprise the following: obtaining a pressure in the furnace higher than the pressure used in the high temperature braze cycle, preferably in the range of about 250 torr; heating the mixture deposited on the repaired region to a temperature of about 2000°F (1093°C); holding the temperature at about 2000°F (1093°C) for approximately 2 hours; increasing the temperature to about 2100°F (1149°C); holding the temperature at about 2100°F (1149°C) for approximately 22 hours; and lowering the temperature from about 2100°F (1149°C) to about 250°F (121°C).

While this diffusion cycle may be altered slightly in terms of the temperatures employed, the range of preferred temperatures for the diffusion braze cycle of the present invention are between 1° and 400°F (1°-225°C) less than the highest temperature achieved during the high temperature brazing cycle. The range of preferred pressures includes any pressure higher than the pressure used in the braze cycle but lower than atmospheric pressure. Those of ordinary skill will recognize that the higher the pressure, the less chromium and other elemental vaporization from the repair composite and the superalloy article there will be, and therefore the less elemental loss there will be.

Additionally, the diffusion braze holding times may vary slightly from the holding times described above, but preferred holding times are in the range of at least 20 hours to about 32 hours in order to permit the repair composite sufficient time to break down the script-like silicide phases into fine discrete particles. Preferred diffusion cycle times are also adequate both to reduce the size and quantity of brittle boride phases in the repair matrix caused by chromium, titanium, and members of the refractory family of elements (tungsten, tantalum, etc.)

combining with boron, and to diffuse the elemental boron and silicon into the repair composite matrix.

Upon completion of the long term diffusion heat treatment cycle, the repaired part is usually given a new metal or ceramic, diffusion or overlay coating by means of any known coating method. Such coatings protect both the superalloy article and/or the repaired area from oxidation, hot corrosion, and extreme thermal gradients. Examples of typical environmental coatings are simple aluminides, platinum aluminides, MCrAl(X)-type overlays, and ceramics. Typical metal coatings such as these may be used alone as a single layer coating, as the final layer of a multilayer coating, or as a bonding coat for a ceramic top coat; and the ceramic coatings may be used alone directly atop the superalloy article surface, or as the final coating atop a bonding coat. However, it is also contemplated by and intended to come within the scope of the present invention to use the present cobalt-base braze alloy compositions as a new type of metal coating that may be used to coat a superalloy article by means of any coating method. The instant compositions may also form part of a multilayer coating system in which the present compositions are applied to the surface of a superalloy article either before or after another environmental coating has been applied.

It has been discovered through the course of high temperature cyclic oxidation testing of superalloy parts coated and/or repaired according to the present invention that the combination of the present braze alloy composition(s) with one or more environmental coatings yields unexpected, inventive, and beneficial improvements in oxidation resistance. Specifically, the instant cobalt-base braze alloy compositions significantly improve the adhesion of an environmental coating to the repair composite.

By way of example and not of limitation, the cyclic oxidation testing was performed at both 2075°F (1135°C) and 2000°F (1093°C) on repaired cobalt base metal coupon specimens, and the specimens of both test conditions exhibited similar results. The test performed at 2075°F (1135°C) indicated that the coating spalled off of the cobalt base metal specimens after 40 cycles. The coating did not spall off the braze repaired areas of the coupons, but it was consumed after 300 cycles. The coating around the brazed areas started to spall after approximately 100 cycles. At 2000°F (1093°C), the test results were identical, except the coating

over the repair composite lasted over 500 cycles with no loss of coating. It is believed that these surprising achievements in oxidation resistance are a result of the careful balance struck between the oxidation properties and the mechanical properties of the elements used in the present preferred compositions.

- 5           For the purpose of promoting further understanding and appreciation of the present invention and its advantages, the following Examples are provided. It will be understood, however, that these Examples are illustrative and not limiting in any fashion.

#### EXAMPLES

- 10           The following examples in Tables 10 and 11 illustrate the preferred composition of cobalt-base alloys for use in accordance with the present invention. Unless otherwise stated, they are listed in percent by weight based on the total weight of the cobalt base alloy.

Table 10

	Example 1	Example 2	Example 3	Example 4	Example 5	Example 6
Nickel	0.001 < Co	0.001 < Co	0.001 < Co	9.5-11.5	9-11	29-32
Chromium	-	0.001- 30**	0.001- 30**	22-24	21.5-23.5	13.75- 15.75
Aluminum	0.001-12‡	0.001-12	0.001-12	0.5-2.5	-	2.3-4.4
Titanium	-	-	0.001-6	0.75-2.75	0.001- 0.25	-
Tungsten	-	0-15**	0.001- 15**	2-4	6-8	0.3-2.4
Molybdenum	-	0-15**	0.001- 15**	-	-	-
Niobium	-	0-12**	0.001- 12**	-	-	-
Ruthenium	0.001-12*	0.001-12*	0.001-12*	-	-	-
Iridium	0.001-12*	0.001-12*	0.001-12*	-	-	-
Rhodium	0.001- 15*‡	0.001-15*	0.001-15*	0.001-2	0.001-15	0.001- 1.5
Hafnium	-	-	0.001-3	-	-	0.001- 1.5
Tantalum	-	0-15**	0-15**	5-7	2.5-4.5	7.8-9.8
Platinum	16-40†	0-40	0-40	0-40	0-40	0-40
Palladium	16-40†	0-40	0-40	0-40	0-40	2-4
Iron	-	-	0.001-3	-	-	-
Manganese	-	-	0.001-1	-	-	-
Rare Earth/ Lanthanide	0.001-5†	0.001-5†	0.001-5†	0.001-5	0.001-5	0.001-5
Carbon	0-2‡	0-2‡	0.001-2‡	0.05- 1.05‡	0.1-1.1‡	0.001- 0.8‡
Boron	0.001-6†	0.001-6†	0.001-6†	0.5-2.5	0.5-2.5	1.3-3.4
Silicon	0.001-10†	0.001-10†	0.001-10†	4-6	4-6	2.3-4.4
Zirconium	-	-	0.001-2	-	0-1.5	-
Cobalt	Balance	Balance	Balance	Balance	Balance	Balance

\*At least one of the selected metals

\*\*At least one of the selected metals

†At least one of the selected metals

‡These elements are optional in selected alloys.



Table 11

	Example 7	Example 8	Example 9	Example 10	Example 11
Nickel	9.0-11.0	9.5-11.5	29-31	9.0-11.0	0.001 < Co
Chromium	21.5-23.5	21-23	13.75- 15.75	21.5-23.5	0-30**
Aluminum	0.75-2.75	0.75-2.75	2.3-4.4	1-3	0-12**
Titanium	-	-	-	-	0-6**
Tungsten	3.0-5.0	3.0-5.0	0.3-2.4	4.0-6.0	0-15**
Molybdenum	-	-	-	-	0-15**
Niobium	-	-	-	-	0-12**
Ruthenium	0.001-12*	0.001-12*	-	0.001-12*	0-1**
Iridium	0.001-12*	0.001-12*	-	0.001-12*	0-12*
Rhenum	0.001-12*	0.001-15*	0.001-1.5	0.001-15*	0-15**‡
Hafnium	-	-	0.001-1.5	-	0-6**
Tantalum	-	5.5-7.5	7.8-9.8	5-7	0-15**
Platinum	16-40†	16-40†	0-40	16-40†	0-40**
Palladium	16-40†	16-40†	2-4	16-40†	0-40**
Iron	-	-	-	-	0-3**
Manganese	-	-	-	-	0-1**
Rare Earth/ Lanthanide	0.001-5	0.001-5	0.001-5	0.001-5	0-5**
Carbon	0.05- 1.05‡	0.05- 1.05‡	-	0.05-1.05	0-2**
Boron	-	-	1.3-3.4	-	0.001-6†
Silicon	-	-	2.3-4.4	-	0.001-10†
Zirconium	0-1.5	-	-	0-1.5	0-2**
Cobalt	Balance	Balance	Balance	Balance	Balance

\*At least one of the selected metals

\*\*At least one of the selected metals

†At least one of the selected metals

‡These elements are optional in selected alloys.

While the invention has been described in detail in the foregoing description, the same is to be considered as illustrative and not restrictive in character, it being understood that only the preferred embodiments have been shown and described, and that all changes and modifications that come within the spirit of the invention are desired to be protected.

## WHAT IS CLAIMED IS:

1. A cobalt-base alloy composition, consisting essentially of, in percent by weight:

Nickel from about 0.001% to less than the weight percent of cobalt;

5 at least one element selected from the group consisting of:

Iridium up to about 12%,

Ruthenium up to about 12%;

at least one element selected from the group consisting of:

Silicon from about 0.001% to about 10%,

10 Boron from about 0.001% to about 6%;

at least one element selected from the group consisting of:

Chromium up to about 40%,

Aluminum up to about 12%,

Titanium up to about 6%,

15 Tungsten up to about 15%,

Molybdenum up to about 15%,

Niobium up to about 12%,

Rhenium up to about 15%,

Hafnium up to about 6%,

20 Tantalum up to about 15%,

Palladium up to about 40%,

Platinum up to about 40%,

Iron up to about 3%,

Manganese up to about 1%,

25 Rare earth/Lanthanide elements up to about 5%,

Carbon up to about 2%,

Zirconium up to about 2%; and

the balance cobalt.

2. The cobalt-base alloy composition of claim 1, wherein said composition  
30 consists essentially of, in percent by weight:

Nickel from about 9.5% to about 11.5%,

Chromium from about 22% to about 24%,

Aluminum from about 0.5% to about 2.5%,

Titanium from about 0.75% to about 2.75%,

35 Tungsten from about 2% to about 4%,

Platinum up to about 40%,

Palladium up to about 40%,  
Rhenium from about 0.001% to about 2%,  
Rare earth/Lanthanide series addition up to about 5%,  
Tantalum from about 5% to about 7%,  
Carbon up to about 1.05%,  
Boron from about 0.5% to about 2.5%, and  
Silicon from about 4% to about 6%; and

the balance cobalt.

3. The cobalt-base alloy composition of claim 1, wherein said composition consists essentially of, in percent by weight:

Nickel from about 9% to about 11%,  
Chromium from about 21.5% to about 23.5%,  
Titanium from about 0.001% to about 0.25%,  
Tungsten from about 6% to about 8%,  
Rhenium from about 0.001% to about 15%,  
Tantalum from about 2.5% to about 4.5%,  
Platinum up to about 40%,  
Palladium up to about 40%,  
Rare earth/Lanthanide series addition up to about 5%,  
Carbon up to about 1.1%,  
Boron from about 0.5% to about 2.5%,  
Silicon from about 4% to about 6%,  
Zirconium from about 0.001% to about 1.5%; and

the balance cobalt.

4. The cobalt-base alloy composition of claim 1, wherein said composition consists essentially of, in percent by weight:

Nickel from about 29% to about 32%,  
Chromium from about 13.75% to about 15.75%,  
Aluminum from about 2.3% to about 4.4%,  
Tungsten from about 0.3% to about 2.4%,  
Rhenium from about 0.001% to about 1.5%,  
Tantalum from about 7.8% to about 9.8%,  
Hafnium from about 0.001% to about 1.5%,  
Rare earth/Lanthanide series addition up to about 5%,  
Platinum up to about 40%,  
Palladium from about 2% to about 4%,

Carbon up to about 0.8%,  
Boron from about 1.3% to about 3.4%, and  
Silicon from about 2.3% to about 4.4%; and

the balance cobalt.

5           5.       The cobalt-base alloy composition of claim 1 overlaid with an  
environmental coating.

6.       The cobalt-base alloy composition of claim 1, wherein said environmental  
coating is selected from the group consisting of simple aluminides, platinum aluminides,  
and diffusion braze alloys.

10          7.       The cobalt-base alloy composition of claim 1, wherein the cobalt-base  
alloy composition overlays a superalloy substrate.

8.       A cobalt-base alloy composition, consisting essentially of, in percent by  
weight:

Nickel from about 0.001% to less than the weight percent of cobalt;  
15       at least one element selected from the group consisting of:

Iridium from about 0.001% to about 12%;

Ruthenium from about 0.001% to about 12%; and

the balance cobalt; wherein said braze alloy composition is a eutectic alloy.

20          9.       The cobalt-base alloy composition of claim 8 overlaid with an  
environmental coating.

10.       The environmental coating of claim 9, wherein said environmental coating  
is selected from the group consisting of simple aluminides, platinum aluminides, and  
diffusion braze alloys.

25          11.       The cobalt-base alloy composition of claim 8, wherein the cobalt-base  
alloy composition overlays a superalloy substrate.

12.       A cobalt-base metal alloy composition, consisting essentially of, in  
percent by weight:

Nickel from about 0.001% to less than the weight percent of cobalt;

at least one element selected from the group consisting of:

30           Iridium from about 0.001% to about 12%,

Ruthenium from about 0.001% to about 12%, and

Rhenium from about 0.001% to about 15%;

at least one element selected from the group consisting of:

Palladium from about 16% to about 40%, and

35           Platinum from about 16% to about 40%;

at least one element selected from the group consisting of:

Chromium up to about 30%,  
Aluminum up to about 12%,  
Titanium up to about 6%,  
5 Tungsten up to about 15%  
Molybdenum up to about 15%,  
Niobium up to about 12%,  
Hafnium up to about 6%,  
Tantalum up to about 15%,  
10 Iron up to about 3%,  
Manganese up to about 1%,  
Rare earth/Lanthanide series element(s) up to about 5%,  
Carbon up to about 2%, and  
Zirconium up to about 2%; and

15 the balance Cobalt.

13. The cobalt-base, base metal alloy composition according to claim 12,  
wherein said composition consists essentially of, in percent by weight:

Nickel from about 9.0% to about 11.0%;  
Chromium from about 21.5% to about 23.5%;  
20 Aluminum from about 1.0% to about 3.0%;  
Tungsten from about 4.0% to about 6.0%;  
Tantalum from about 5.0% to about 7.0%;  
Rare earth/Lanthanide series addition up to about 5%;  
Carbon from about 0.05% to about 1.05%;  
25 Zirconium from about 0.001% to about 1.5%;

at least one element selected from the group consisting of:

Iridium from about 0.001% to about 12%,  
Ruthenium from about 0.001% to about 12%, and  
Rhenium from about 0.001% to about 15%;

30 at least one element selected from the group consisting of:

Platinum from 16% to about 40% and  
Palladium from 16% to about 40%; and

the balance Cobalt.

14. The cobalt-base, base metal alloy composition according to claim 12,  
35 wherein said composition consists essentially of, in percent by weight:

Nickel from about 9.5% to about 11.5%,

Chromium from about 21% to about 23%,

Aluminum from about 0.75% to about 2.75%,

Tungsten from about 3.0% to about 5.0%,

Tantalum from about 5.5% to about 7.5%,

5 Rare earth/Lanthanide series element(s) up to about 5%,

Carbon from about 0.05% to about 1.05%; and

at least one element selected from the group consisting of:

Iridium from about 0.001% to about 12%,

Ruthenium from about 0.001% to about 12%, and

10 Rhenium from about 0.001% to about 15%;

At least one element selected from the group consisting of:

Platinum from 16% to about 40%;

Palladium from 16% to about 40%; and

the balance Cobalt.

15 15. A method of repairing a damaged region of a corrosion resistant superalloy substrate comprising:

preparing a repair mixture comprising a cobalt-base braze alloy, a base metal alloy composition and an organic binder, said cobalt-base braze alloy consisting essentially of, in percent by weight:

20 Nickel from about 0.001% to less than the weight percent of cobalt;

at least one element selected from the group consisting of:

Rhenium from about 0.001% to about 15%,

Palladium from about 0.001% to about 40%,

Platinum from about 0.001% to about 40%,

25 Iridium from about 0.001% to about 12%, and

Ruthenium from about 0.001% to about 12%;

at least one element selected from the group consisting of:

Boron from about 0.001% to about 6% and

Silicon from about 0.001% to about 10%; and

30 at least one element selected from the group consisting of:

Chromium, Aluminum, Titanium, Tungsten, Molybdenum, Niobium,

Rhenium, Hafnium, Tantalum, Iron, Manganese, Rare earth/Lanthanide elements,

Carbon, and Zirconium; and

the balance Cobalt;

depositing said repair mixture on at least a portion of the damaged region; and

joining the deposited repair mixture to the superalloy substrate.

16. The method of claim 15, and further including:  
5 applying an environmental coating to said superalloy substrate.

17. The method of claim 15 wherein said joining includes heating the deposited repair mixture to a temperature of between about 800 °F (and less than the incipient melting temperature of the superalloy).

18. The method of claim 15, wherein said joining includes heating the  
10 deposited repair mixture under an inert atmosphere or under subatmospheric conditions.

19. The method of claim 15 wherein said joining includes heating the deposited repair mixture to a temperature sufficient to form a solid solution matrix.

20. The method of claim 15, wherein said joining includes a long term  
15 diffusion heat treatment.

21. The method of claim 20, wherein the long term diffusion heat treatment includes:

heating said deposited repair mixture to a first temperature between about 2000°F (1093°C) and about 2100°F (1149°C);

20 heating the deposited repair mixture to a second temperature greater than or equal to about 2100°F (1149°C);

maintaining the deposited repair mixture at a temperature greater than or equal to about 2100°F (1149°C) for between about 20 hours and about 32 hours; and

lowering the temperature of the repair mixture to about 250°F (121°C).

22. The method of claim 21 wherein the deposited repair mixture is  
25 maintained at a temperature between about 2000°F (1093°C) and about 2100°F (1149°C) for about 2 hours.

23. The method of claim 20, wherein the long term diffusion heat treatment includes:

30 heating said mixture deposited on said damaged region to a temperature of greater than or equal to about 2000°F (1093°C);

decreasing the temperature of the deposited repair mixture from about 2000°F (1093°C) to a temperature between about 1999°F (1093°C) and about 1600°F (871°C);

maintaining the deposited repair mixture at a temperature between about 1999°F (1093°C) and about 1600°F (871°C) for less than about 20 hours; and

lowering the temperature of the repair mixture to about 250°F (121°C).

5           24.     The method of claim 23 wherein the deposited repair mixture is maintain at a temperature greater than or equal to about 2000°F (1093°C) for about 2 hours.

25.     The method of claim 15, wherein said joining includes:

10           heating the deposited repair mixture to a first temperature of between about 800°F (427°C) and about 1800°F (982°C);

heating the deposited repair mixture to a second temperature greater than or equal to about 1800°F (982°C);

15           heating the deposited repair mixture to a third temperature between about 1800°F (982°C) and less than about the incipient melting temperature of the superalloy substrate for between about 15 and about 45 minutes; and

cooling the deposited repair mixture to a temperature of less than or equal to about 1800° F (982°C).

26.     The method of claim 25 wherein the deposited repair mixture is maintained at a temperature between about 800°F (427°C) and about 1800°F (982°C) for about 15 minutes.

27.     The method according to claim 15, wherein the cobalt-base braze alloy composition is prealloyed, and wherein the base metal alloy composition is prealloyed.

28.     The method of claim 15, wherein the cobalt-base braze alloy composition of consists essentially of, in percent by weight:

25           Nickel from about 9.5% to about 11.5%,  
Chromium from about 22% to about 24%,  
Aluminum from about 0.5% to about 2.5%,  
Titanium from about 0.75% to about 2.75%,  
Tungsten from about 2% to about 4%,  
30           Platinum up to about 40%,  
Palladium up to about 40%,  
Rhenium from about 0.001% to about 2%,  
Rare earth/Lanthanide series addition up to about 5%,  
Tantalum from about 5% to about 7%,



Carbon up to about 1.05%,  
Boron from about 0.5% to about 2.5%, and  
Silicon from about 4% to about 6%; and  
the balance cobalt.

- 5        29.     The method of claim 15, wherein the cobalt base braze composition consists essentially of, in percent by weight:

          Nickel from about 9% to about 11%,  
          Chromium from about 21.5% to about 23.5%,  
          Titanium from about 0.001% to about 0.25%,  
10        Tungsten from about 6% to about 8%,  
          Rhenium from about 0.001% to about 15%,  
          Tantalum from about 2.5% to about 4.5%,  
          Platinum up to about 40%,  
          Palladium up to about 40%,  
15        Rare earth/Lanthanide series addition up to about 5%,  
          Carbon up to about 1.1%,  
          Boron from about 0.5% to about 2.5%,  
          Silicon from about 4% to about 6%,  
          Zirconium from about 0.001% to about 1.5%; and  
20        the balance cobalt.

30.     The method of claim 15, wherein the cobalt base braze composition consists essentially of, in percent by weight:

          Nickel from about 29% to about 32%,  
          Chromium from about 13.75% to about 15.75%,  
25        Aluminum from about 2.3% to about 4.4%,  
          Tungsten from about 0.3% to about 2.4%,  
          Rhenium from about 0.001% to about 1.5%,  
          Tantalum from about 7.8% to about 9.8%,  
          Hafnium from about 0.001% to about 1.5%,  
30        Rare earth/Lanthanide series addition up to about 5%,  
          Platinum up to about 40%,  
          Palladium from about 2% to about 4%,  
          Carbon up to about 0.8%,  
          Boron from about 1.3% to about 3.4%, and

Silicon from about 2.3% to about 4.4%; and  
the balance cobalt.

31. The method of claim 15, wherein the repair mixture comprises at least one additional base metal alloy composition.

5 32. The method of claim 15, wherein said at least one additional base alloy composition is not a eutectic alloy.

33. The method of claim 15, wherein the repair mixture is provided as a powder metal slurry.

34. The method of claim 15, wherein the repair mixture is provided as a  
10 pre-sintered powdered metal alloy tape.

35. The method of claim 15, wherein the repair mixture is provided as a plasticized powdered metal alloy tape.

36. The method of claim 15, wherein the repair mixture is provided as a pre-sintered alloy preform.

15 37. The method of claim 15, wherein the repair mixture comprises less than or equal to about 50%, by weight based on the total weight of the repair mixture, of the cobalt-base braze alloy composition.

38. The method of claim 15, wherein the repair mixture comprises less than or equal to about 30%, by weight based on the total weight of the repair mixture,  
20 of the cobalt base braze alloy composition.

39. A method of repairing a damaged region of a corrosion resistant superalloy substrate comprising:

preparing a repair mixture comprising a cobalt-base braze alloy composition, a base metal alloy composition; and an organic binder; wherein the  
25 cobalt-base braze alloy composition is a eutectic alloy comprising, in percent by weight:

Nickel from about 0.001% to less than the weight percent of cobalt;

at least one element selected from the group consisting of:

30 Rhenium from about 0.001% to about 15%,  
Palladium from about 0.001% to about 40%,  
Platinum from about 0.001% to about 40%,  
Iridium from about 0.001% to about 12%, and  
Ruthenium from about 0.001% to about 12%; and

the balance Cobalt;  
depositing said mixture on said damaged region of said corrosion  
resistant superalloy substrate; and  
joining the repair mixture to the superalloy substrate.

- 5           40.     A method of long term diffusion heat treating a superalloy mixture of  
at least one braze alloy and at least one base metal alloy comprising:  
                providing a superalloy mixture of at least one braze alloy and at least  
one base metal alloy, said superalloy mixture including an amount of brittle phases;  
                heating said superalloy mixture to a temperature of at least about  
10   2000°F (1093°C);  
                increasing the temperature of the superalloy mixture to a temperature  
of at least about 2100°F (1149°C);  
                maintaining the temperature of the superalloy mixture at a temperature  
of at least about 2100°F (1149°C) for between about 20 and about 32 hours; and  
15           decreasing the temperature of the superalloy mixture a temperature less  
than or equal to about 250°F (121°F);  
                wherein said method reduces the amount of brittle phases.
41.     The method of claim 40 wherein the deposited repair mixture is  
maintained at a temperature of at least about 2000°F (1093°C) for about 2 hours.
- 20           42.     A method of long term diffusion heat treating a superalloy mixture of  
at least one braze alloy and at least one base metal alloy comprising:  
                providing a superalloy mixture of at least one braze alloy and at least  
one base metal alloy, wherein the superalloy mixture includes an amount of brittle  
phases;  
25           heating the superalloy mixture to a temperature of at least about  
2000°F (1093°C);  
                maintaining the temperature of the superalloy mixture at a temperature  
between about 1999°F (1093°C) and about 1600°F (871°C) for less than about 20  
hours; and  
30           decreasing the temperature of the superalloy mixture to a temperature  
of less than or equal to about 250°F (121°C);  
                wherein said method reduces the amount of brittle phases.
43.     The method of claim 42 wherein the deposited repair mixture is  
maintained at a temperature of at least about 2000°F (1093°C) for about 2 hours.