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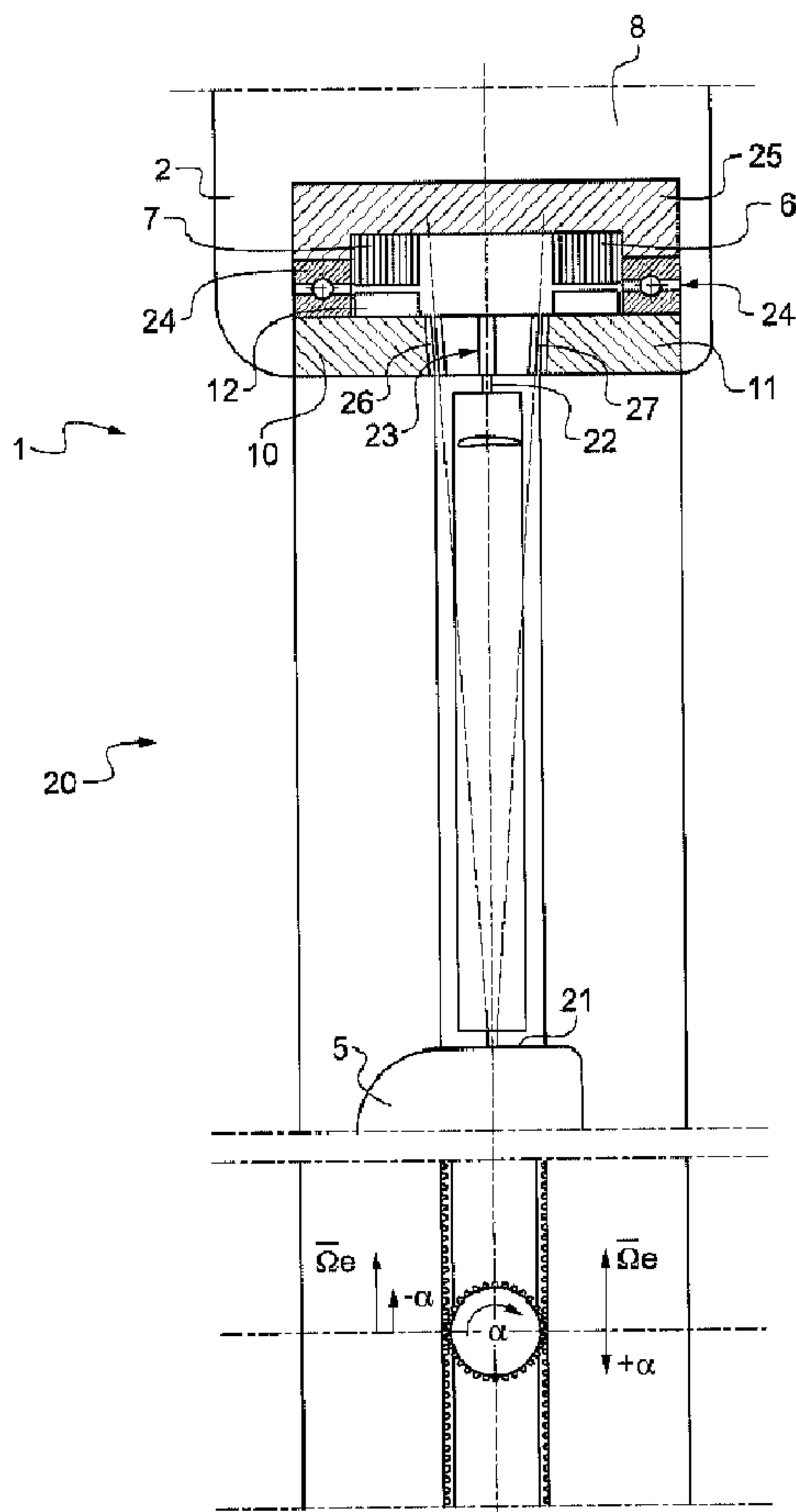
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(54) Title: ELECTRICAL POWERED TAIL ROTOR OF A HELICOPTER



(57) Abrégé/Abstract:

The invention is related to an electrical powered tail rotor (1) of a helicopter comprising a housing (2) around the tail rotor (1), at least one permanent magnet energized synchronous motor with a stator (6, 7) with a plurality of poles (9), the at least one

(57) Abrégé(suite)/Abstract(continued):

synchronous motor being integrated as a torus (8) around an opening of the housing (2) encompassing the tail rotor (1), blades (4) of the tail rotor (1) being fixed to at least one rotating component (10, 11) of the at least one synchronous motor and supply means for electric energy to the at least one synchronous motor characterized in that blade pitch control means are provided at the torus (8), characterized in that two coaxial synchronous motors with a stator (6, 7) each and a coaxial rotating component (10, 11) each are provided and in that the blades (4) of the tail rotor (1) are linked to each of the coaxial rotating components (10, 11) of said synchronous motors.

“Electrical powered tail rotor of a helicopter”

ABSTRACT

The invention is related to an electrical powered tail rotor (1) of a helicopter comprising a housing (2) around the tail rotor (1), at 5 least one permanent magnet energized synchronous motor with a stator (6, 7) with a plurality of poles (9), the at least one synchronous motor being integrated as a torus (8) around an opening of the housing (2) encompassing the tail rotor (1), blades (4) of the tail rotor (1) being fixed to at least one rotating 10 component (10, 11) of the at least one synchronous motor and supply means for electric energy to the at least one synchronous motor characterized in that blade pitch control means are provided at the torus (8), characterized in that two coaxial synchronous motors with a stator (6, 7) each and a coaxial rotating component 15 (10, 11) each are provided and in that the blades (4) of the tail rotor (1) are linked to each of the coaxial rotating components (10, 11) of said synchronous motors.

Electrical powered tail rotor of a helicopter

Technical Field

The invention is related to an electrical powered tail rotor of a helicopter.

5 Background of the Invention

The power consumed by a tail rotor of state of the art helicopters is supplied from a central energy generator via a main gear box, a plurality of intermediate gears and a tail rotor shaft. By removing the main gear box and the rigid mechanical coupling 10 between energy generator and tail rotor more design flexibility for the helicopter may be attained. One of the keys to realise an electrical powered tail rotor of a helicopter is a suitable electrical motor.

The document US 2004051401 A1 discloses an electric motor 15 for rotating an object around a central axis. The electric motor includes a motor casing. A circular segmented rail element is disposed within the motor casing about the central axis. The circular segmented rail element includes metallic non-ferrous segments interleaved with non-metallic segments. Each of the 20 metallic non-ferrous segments has a predetermined segment length. At least one coil element is connected to the motor casing. The circular segmented rail element is disposed adjacent the at least one coil element. The at least one coil element has a predetermined coil length that is less than or equal to the 25 predetermined segment length. The at least one-coil element is configured to apply electromagnetic energy to the circular segmented rail element, such that the circular segmented rail element rotates around the central axis.

The document WO 0184063 A2 discloses a stator assembly for a brushless DC ring motor for a cooling fan piloted on the stator assembly. A ring supports a plurality of fan blades for sweeping an area inside the shroud. A rotor assembly for the brushless DC ring 5 motor is attached to the ring of the cooling fan. The rotor assembly confronts the stator assembly around an outer diameter of the stator assembly. The cooling system is controlled by an electronic controller to rotate the cooling fan to provide appropriate cooling for the vehicle.

10 A hybrid helicopter drive has been proposed in the document "The Hybrid Helicopter Drive, ... by Peter Jänker et al. at Europ. Rotorcraft Forum, Sept. 2010" with an integration of an electrical motor for a Fenestron tail rotor. The electrical motor is realised by so called disc shaped electrical "Trans-Flux-Motors" with increased 15 pole numbers. The electrical "Trans-Flux-Motor" for the Fenestron tail rotor is conceived as a torus around the Fenestron opening, the blade tips of the tail rotor being fixed to its rotating component. An electrical "Trans-Flux-Motor" is presented in document DE 10 2007 013 732 A1.

20 The document DE 102007013732 A1 discloses a direct drive with a stator and one or multiple support rings that are made of plastic. The support rings supports the permanent magnets that are arranged in two or more concentric rings. The annular or sector shaped stator logs, made of plastic, are arranged in axial direction 25 adjacent to the concentric rings of the support rings in such a way that a magnetic flux is allowed in radial direction between adjacent concentric rings.

30 The document "The Hybrid Helicopter Drive,... by Peter Jänker et al. at Europ. Rotorcraft Forum, Sept. 2010" further discloses such electrical "Trans-Flux-Motors" with two disks for the main rotor.

The document WO 2005/100154 A1 discloses a rotor for mounting on a helicopter drive shaft, comprising a hub for location on the shaft and a plurality of blades mounted to and extending out from the hub, wherein a pitch angle ($[\phi]$) of at least one of the 5 blades is controllable with respect to each other blade by an electrical stepper motor mechanism arranged at the hub. Also disclosed is a method for determining a pitch angle ($[\phi]$) of the blades of the rotor, a computer program arranged to, when loaded onto a computing system, utilise an algorithm for determining blade 10 pitch angle ($[\phi]$) values for the blades, an alternator for providing power to motors that control the pitch of the blades and a control method for implementation by a computer in controlling the pitch of the blades in real time.

The document US 2009/140095 A1 discloses a rotary-wing 15 aircraft with an electric motor mounted along an axis of rotation to drive a rotor system about the axis of rotation

The document US 4953811 A discloses a helicopter engine turning a tail rotor while it is turning the main rotor. Tail rotors, while essential components, take power from the engine, introduce 20 a drag force, add weight, and increase rotor noise. Since the engine is as close as possible to the main rotor, the complexity, number of parts, weight and efficiency of the remote tail rotor have gone unchanged. Those parts and hence their added weights have been eliminated. A self-driving tail rotor for a helicopter is 25 provided.

The document WO 2009/129309 A2 discloses a wind 30 generator in which superconducting ring generators are utilized without the need for a load bearing drive shaft and other mechanical components allowing for the use of variable geometry blades, a decrease in the overall weight, and an increase in the overall efficiency of the wind generator system.

The document US 2006/049304 A1 discloses a hover aircraft with an air impeller engine having an air channel duct and a rotor with outer ends of its blades fixed to an annular impeller disk that is driven by magnetic induction elements arrayed in the air channel duct. The air-impeller engine is arranged vertically in the aircraft frame to provide vertical thrust for vertical takeoff and landing. Preferably, the air-impeller engine employs dual, coaxial, counter-rotating rotors for increased thrust and gyroscopic stability. An air vane assembly directs a portion of the air thrust output at a desired angle to provide a horizontal thrust component for flight maneuvering or translation movement. The aircraft can employ a single engine in an annular fuselage, two engines on a longitudinal fuselage chassis, three engines in a triangular arrangement for forward flight stability, or other multiple engine arrangements in a symmetric, balanced configuration. Other flight control mechanisms may be employed, including side winglets, an overhead wing, and/or air rudders or flaps. An integrated flight control system can be used to operate the various flight control mechanisms. Electric power is supplied to the magnetic induction drives by high-capacity lightweight batteries or fuel cells. The hover aircraft is especially well suited for applications requiring VTOL deployment, hover operation for quiet surveillance, maneuvering in close air spaces, and long duration flights for continuous surveillance of ground targets and important facilities requiring constant monitoring.

25 Summary of the Invention

It is an object of the invention to provide for an electrical powered tail rotor of a helicopter with improved efficiency.

A solution is provided with an electrical powered tail rotor of a helicopter with the features of claim 1.

According to the invention an electrical powered tail rotor of a helicopter (H) comprises a housing around the tail rotor and at least one permanent magnet energized synchronous motor with an increased pole number. Said at least one synchronous motor is 5 integrated as a torus in the housing around an opening of the housing encompassing the tail rotor. Blades of the tail rotor are fixed to at least one rotating component of said at least one synchronous motor. Supply means provide electric energy to said at least one synchronous motor. The invention is characterized in 10 that blade pitch control means are provided at the torus. As a major advantage the inventive electrical powered tail rotor of the helicopter allows replacement of the tail drive shaft by essentially less weighing cable to the tail rotor. Drive and control of the inventive electrical powered tail rotor can be integrated and 15 deletion of a stator at the tail rotor results in considerable less noise. A further advantage the inventive electrical powered tail rotor of the helicopter is separation of the propulsion of main rotor and tail rotor and thus independence between main rotor and tail rotor in terms of rotational speed allowing higher forward speeds 20 for the helicopter as the main rotor speed can be adapted over a wide range to the optimum required. According to a still further advantage of the inventive electrical powered tail rotor of the helicopter noise emitted by the rotors can be reduced through setting of the respective rotor speeds such, that the noise 25 interaction between main and tail rotor is minimizing the total emission. A gearbox for the tail rotor is not needed any longer with the subsequent advantage of a high potential for less weight. There is as well high potential to ease the process of adjustment for the tail rotor unit and drastically reduce production cost as the 30 installation of the drive-shaft, as well as its production, requires considerable efforts. The inventive electrical powered tail rotor of the helicopter allows independent control of the tail rotor thrust by

means of rotational speed, blade pitch and tail rotor torque, thus making power available independent from the main rotor system for a wide range of power settings, airspeeds, altitudes and temperature. As an example the rotor speed can be used to 5 compensate for altitude effects more efficiently than can be done by tail rotor blade pitch. According to the invention it would even be possible to completely stop the tail-rotor during forward flight and thus reduce the drag and the power demand of the helicopter as the tail-rotor is not requiring power. The use of an electrical 10 motor for the drive of the tail rotor of the helicopter allows more possibilities to shape the core of the ducted tail rotor, and to have more possibilities to optimize the aerodynamic shape of the complete tail unit, especially the tail boom. The permanent magnet energized synchronous motor has excellent control characteristics 15 and excellent efficiency for the transformation of electrical power into mechanical power. The electrical motors of the invention have low weight at high power output with balanced efficiency over a wide range of speed and power settings and are not less reliable than traditional mechanical drive trains and engines.

20 According to a preferred embodiment of the invention an electrical powered tail rotor of a helicopter comprises two coaxial synchronous motors with two coaxial rotating components and the blades of the tail rotor are linked to each of said respective coaxial rotating components of the two synchronous motors. The two 25 synchronous motors operate at essentially the same rotational speed allowing a relative twist between the two coaxial rotating components for control of the blade pitch of the blades of the tail rotor by means of a suitable mechanism. The phase shift between the two coaxial rotating components results in a collective change 30 of the blade pitch for all blades and thus allows control of the thrust of the tail rotor. The two synchronous motors of the invention each have a big diameter and each have little axial length. The two

5 synchronous motors for the tail rotor are arranged in a fail safe concept, i. e. the failure of one will leave at least half of the power available. The excellent control characteristics of the permanent magnet energized synchronous motor provide for the precise control of the blade pitch of the blades of the tail rotor.

10 According to a further preferred embodiment of the invention each stator is provided on an inner circumference with a plurality of poles and each of said two coaxial rotating components with a corresponding plurality of permanent magnets on an outer circumference of each of the rotating components. The poles and permanent magnets are each preferably arranged regularly in pairs, with the permanent magnets in pairs out of phase to provide for continuous interference of at least a part of the permanent magnets.

15 According to a further preferred embodiment of the invention the two coaxial rotating components are supported by a magnetic bearing integrated in the torus around the housing opening encompassing the tail rotor. The magnetic bearing allows contact free rotation of the mobile parts of the tail rotor.

20 According to a further preferred embodiment of the invention a retainer is provided at the torus around the housing opening encompassing the tail rotor to withhold said one or two coaxial rotating components at start or in case of a failure of the magnetic bearing.

25 According to a further preferred embodiment of the invention said two coaxial rotating components are supported by a ball bearing between a stator and the rotating components of each of said two synchronous motors.

According to a further preferred embodiment of the invention said two coaxial rotating components are linked by a bevel gear to the blades of the tail rotor.

According to a further preferred embodiment of the invention 5 said two synchronous motors are asymmetric. One of said two synchronous motors may be conceived to provide all the driving power for the blades of the tail rotor whereas the other of said two synchronous motors solely controls the blade pitch. Blades of this inventive embodiment are preferable connected without azimuthal 10 tolerance to the synchronous motors conceived to provide the driving power. The phase shift of said two synchronous motors relative to each other solely controls the blade pitch of this further preferred embodiment of the invention.

According to a further preferred embodiment of the invention 15 a sliding sleeve is provided at the torus for control of the pitch of the blades of said one coaxial rotating component of said one synchronous motor.

Brief Description of the Drawings

Preferred embodiments of the invention are presented by 20 means of the following description with reference to the attached drawings, from which

Fig. 1 shows a spatial view of a tail rotor of a helicopter according to the invention,

Fig. 2 shows a spatial view of stators of electrical motors 25 around a tail rotor according to the invention,

Fig. 3 shows a spatial view of two coaxial rotating components of electrical motors of a tail rotor according to the invention, and

Fig. 4 shows a cross sectional view of the electrical motor according to the invention.

Description of Preferred Embodiments

According to Fig. 1 a tail rotor 1 is arranged within a housing 5 2 of a helicopter's tail boom 3. Blades 4 of the tail rotor 1 are centrally supported by a hub 5. Hub 5 is essentially ball shaped towards an inlet side and essentially flat towards an outlet side of the tail rotor 1.

According to Fig. 2 two coaxial stators 6, 7 are provided on 10 the inner circumference of a torus 8 of a brushless electrical motor assembly composed of two permanent magnet energized synchronous motors with a plurality of poles 9 on each of the two coaxial stators 6, 7.

The poles 9 on each of the two coaxial stators 6, 7 are 15 connected to supply means (not shown) for electrical power. Power semiconductors (not shown) and microcontrollers (not shown) provide for two multiphase inverters (not shown) for precise control of the two brushless synchronous motors.

According to Fig. 3 the electrical motor assembly is 20 composed of one rotating component 10, 11 for each of said two synchronous motors. The two rotating components 10, 11 are coaxial. A plurality of permanent magnets 12 arranged regularly on an outer circumference of each of the rotating components 10, 11 correspond to the plurality of poles 9 on each of the two coaxial 25 stators 6, 7.

Blade tips of the blades 4 of the tail rotor 1 are held in between the two rotating components 10, 11.

The blade pitch control means includes the provision of

phase shift between the two rotating components 10, 11 of the synchronous motors by rotating the two rotating components 10, 11 coaxially relative to each other. The control of the respective angular positions of the two rotating components 10, 11 relative to 5 each other may be effected with special sensors, e. g. on the basis of Hall effect due to the passing permanent magnets 12 or without sensors by detecting any voltages induced in momentarily current free coils.

According to Fig. 4 corresponding features are referred to 10 with the references of Fig. 1 – 3. The anti-torque system, i. e. the profiled blades 4 of the tail rotor 1 of a housed concept like the “Fenestron” is powered electrically by the two coaxial synchronous motors fitted into the housing 2 of the tail rotor 1.

The tail rotor 1 comprises an inlet fairing 20 for the blades 4 15 supported in a pivoting mechanism 21 of the hub 5 of said ducted fan. The blades 4 of the tail rotor 1 are held at their respective blade tips 22 in a bevel gear 23. The bevel gear 23 is mounted between the two rotating components 10, 11 of the two coaxial synchronous motors as a link for the blades 4 to each of the two 20 rotating components 10, 11. The blades 4 are rotated via the blade tips 22 with the bevel gear 23. The bevel gear 23 engages with gears 26, 27 preferably all along inner lateral faces of the rotating components 10, 11. The gears are provided with small moduli to allow tolerances for the blade pitch.

25 The two coaxial synchronous motors comprise the permanent magnets 12 arranged in pairs regularly on the outer circumference of each of the rotating components 10, 11 as passive flux rings. The rotating components 10, 11 are respectively supported by outer thin annular roller bearings 24 or by magnetic bearings (not 30 shown) as retainer. The roller bearings 24 support the two rotating components 10, 11 against a frame type casing 25 of the torus 8.

Layered sheet metal packages for the respective poles 12 of the coaxial stators 6, 7 are mounted against the frame type casing 25 on the inner circumference of the torus 8. The layered sheet metal packages for the respective poles 12 abut laterally against the 5 inside of the thin annular roller bearings 24 and the inside of flanks of the frame type casing 25.

A relative rotation of the two rotating components 10, 11 with regard to each other rotates the bevel gear 23 of the blade pitch control means. The rotation of the bevel gear 23 is transmitted to 10 the blade tip 22 of the profiled blade 4 for control of the blade pitch and thus the thrust of the operating tail rotor 1. The blades 4 of the tail rotor 1 may be irregularly distributed along the circumference of the two rotating components 10, 11 for less sound emission of the tail rotor 1.

15 The two coaxial synchronous motors may be asymmetric with regard to drive power, i. e. one of the two coaxial synchronous motors may take over all of the drive power while the other is the blade pitch control means exclusively taking care of the control of the blade pitch of the blades 4. The blades 4 would be linked to the 20 coaxial synchronous motor taking over all of the drive power in such a way that there would be no azimuthal move any more.

The provision of phase shift between the two rotating components 10, 11 of the two coaxial synchronous motors allows a further control of the thrust provided by the tail rotor 1, namely 25 supplemental to the control of the thrust by varying solely the rotational speed of the tail rotor 1.

In case of stationary flight the two coaxial synchronous motors have exactly the same rotational speed and essentially the same power rate. The profiles of the blades 4 may be selected with 30 a so called S-lay-out for a positive zero moment coefficient or this

moment may be used as retroactive moment by a selection of the pivot axis of the blades 4 with a few percents before a quarter of the blade chord. Any of said selections would allow a safe landing in case of a failure of the control due to a neutral positioning of the
5 blades 4.

For a tail rotor 1 with only one synchronous motor a sliding sleeve as the blade pitch control means may be provided at the torus 8 for control of the pitch from the outer radius of the blades 4 and thus for supplemental control of the thrust of said tail rotor 1
10 with one coaxial rotating component 10.

WHAT IS CLAIMED IS:

1. An electrical powered tail rotor (1) of a helicopter comprising
a housing (2) around the tail rotor (1),
5 at least one permanent magnet energized synchronous motor with a stator (6, 7) with a plurality of poles (9), said at least one synchronous motor being integrated as a torus (8) around an opening of the housing (2) encompassing the tail rotor (1), blades (4) of the tail rotor (1) being fixed to at least one rotating component (10, 11) of said at least one synchronous motor and supply means for electric energy to said at least one synchronous motor characterized in that blade pitch control means are provided at the torus (8), characterized in that two coaxial synchronous motors with a stator (6, 7) each and a coaxial rotating component
10 (10, 11) each are provided and in that the blades (4) of the tail rotor (1) are linked to each of said coaxial rotating components (10, 11) of said synchronous motors.
15
2. The tail rotor (1) according to claim 1,
characterized in that each stator (6, 7) is provided on an inner circumference with a plurality of poles (9) and each of said two coaxial rotating components (10, 11) with a corresponding plurality of permanent magnets (12) on an outer circumference of each of the rotating components (10, 11), the poles (9) being preferably
20 arranged regularly in pairs on the inner circumference of each stator (6, 7) and the permanent magnets (12) being preferably arranged regularly in pairs on the outer circumference of the rotating components (10, 11).
25
3. The tail rotor (1) according to claim 2,

characterized in that a retainer is provided at the torus (8) around the housing opening encompassing the tail rotor (1).

4. The tail rotor (1) according to claim 3,

characterized in that said two coaxial rotating components (10, 11) 5 are supported by a magnetic bearing integrated in the torus (8) around the housing opening encompassing the tail rotor (1).

5. The tail rotor according to claim 3,

characterized in that said two coaxial rotating components are supported by a ball bearing between a stator and the rotating 10 component of each of said two synchronous motors.

6. The tail rotor (1) according to claim 1,

characterized in that said two coaxial rotating components (10, 11) are linked by a bevel gear (23) to the blades (4) of the tail rotor (1).

15 7. The tail rotor (1) according to claim 1,

characterized in that said two synchronous motors are asymmetric to each other.

8. The tail rotor (1) according to claim 1,

characterized in that a sliding sleeve is provided at the torus (8) for 20 control of the pitch of the blades (4) of one coaxial rotating component (10, 11) of one synchronous motor.

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Fig.1

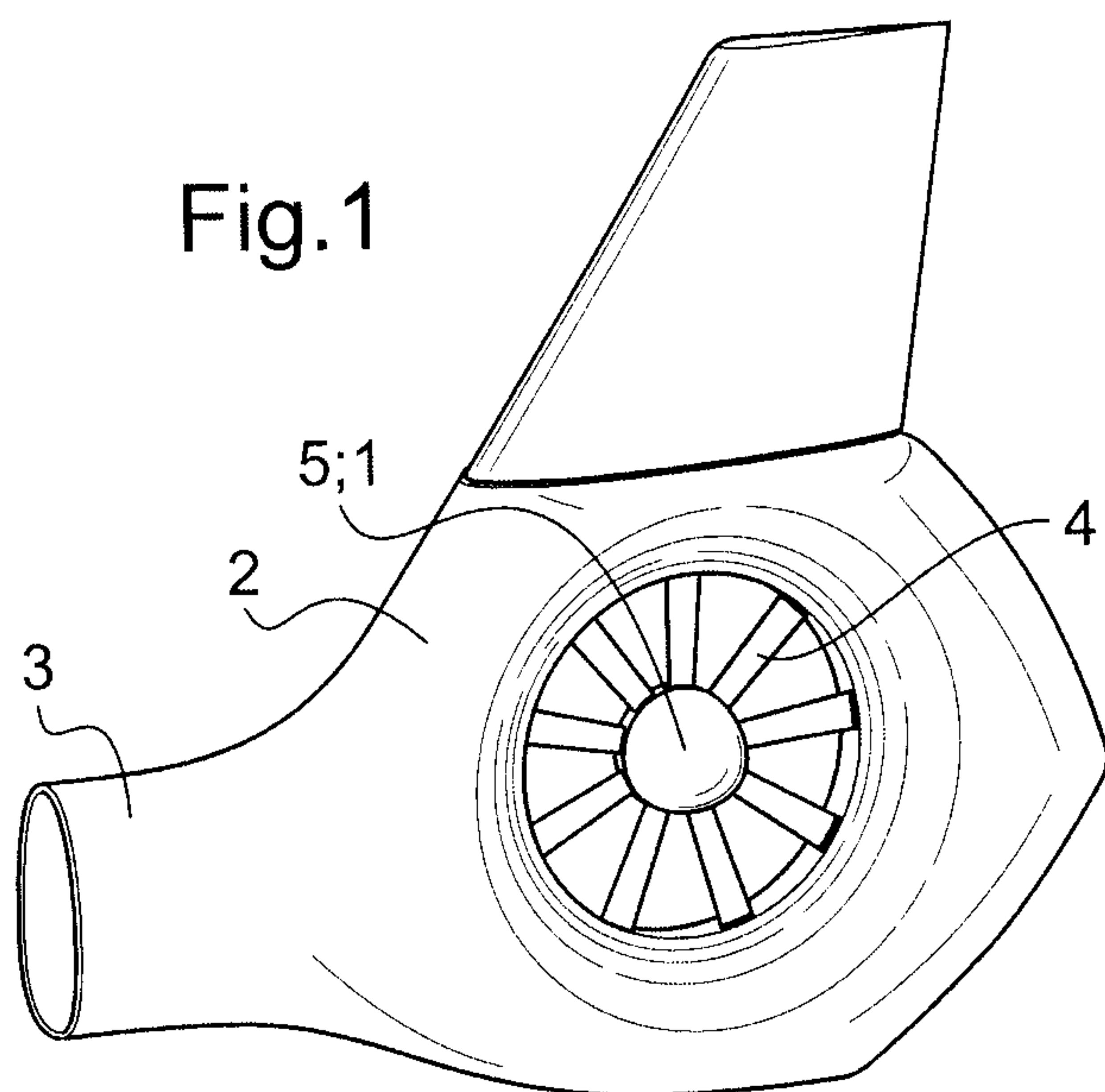
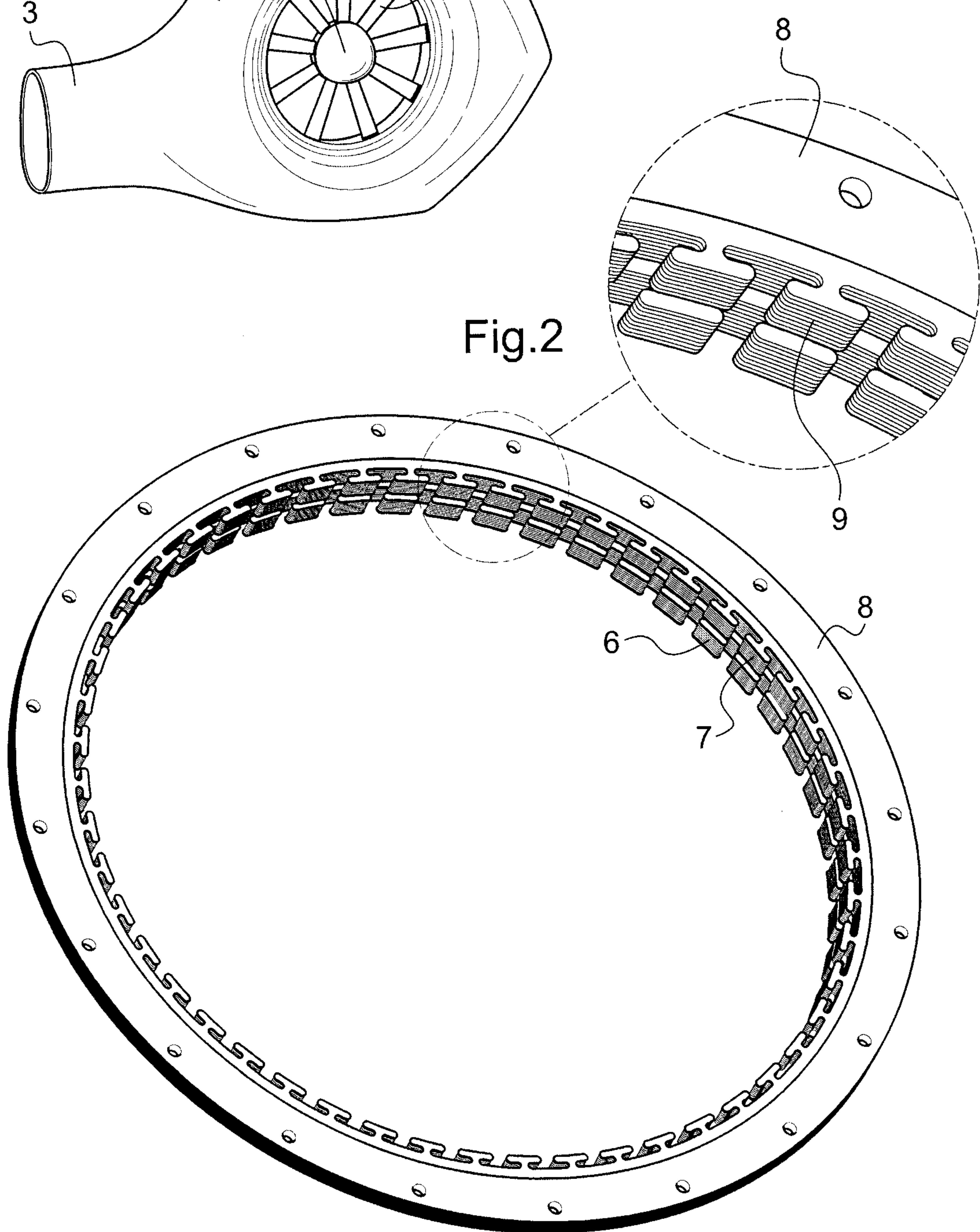
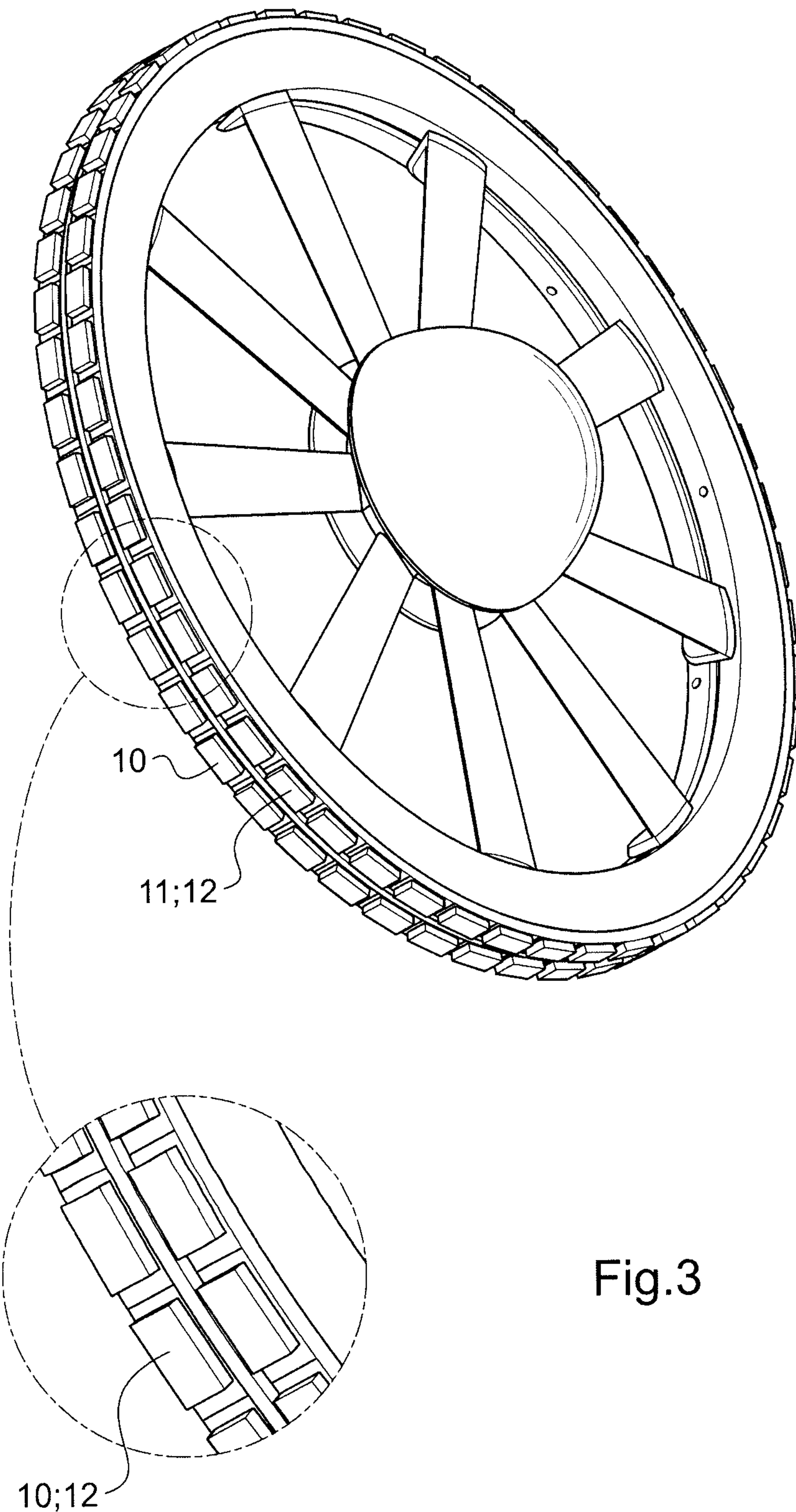


Fig.2



2/3



3/3

Fig.4

