

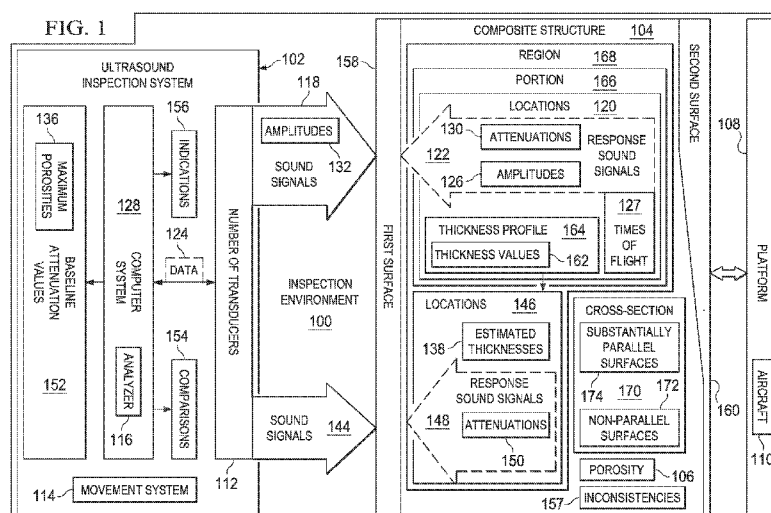


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|---|--|
| <p>(51) International Patent Classification:
 G01N 29/04 (2006.01) G01N 15/08 (2006.01)
 G01N 29/11 (2006.01)</p> | <p>(81) Designated States (<i>unless otherwise indicated, for every kind of national protection available</i>): AE, AG, AL, AM, AO, AT, AU, AZ, BA, BB, BG, BH, BN, BR, BW, BY, BZ, CA, CH, CL, CN, CO, CR, CU, CZ, DE, DK, DM, DO, DZ, EC, EE, EG, ES, FI, GB, GD, GE, GH, GM, GT, HN, HR, HU, ID, IL, IN, IR, IS, JP, KE, KG, KN, KP, KR, KZ, LA, LC, LK, LR, LS, LT, LU, LY, MA, MD, ME, MG, MK, MN, MW, MX, MY, MZ, NA, NG, NI, NO, NZ, OM, PA, PE, PG, PH, PL, PT, QA, RO, RS, RU, RW, SA, SC, SD, SE, SG, SK, SL, SM, ST, SV, SY, TH, TJ, TM, TN, TR, TT, TZ, UA, UG, US, UZ, VC, VN, ZA, ZM, ZW.</p> |
| <p>(21) International Application Number:
 PCT/US2014/034637</p> | |
| <p>(22) International Filing Date:
 18 April 2014 (18.04.2014)</p> | |
| <p>(25) Filing Language:
 English</p> | |
| <p>(26) Publication Language:
 English</p> | |
| <p>(30) Priority Data:
 13/897,007 17 May 2013 (17.05.2013) US</p> | |
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 — with international search report (<i>Art. 21(3)</i>)</p> |

Published:

— with international search report (Art. 21(3))

(54) Title: POROSITY INSPECTION SYSTEM FOR COMPOSITE STRUCTURES WITH NON-PARALLEL SURFACES



(57) Abstract: A method and apparatus for inspecting a composite structure. A thickness profile of a portion of a region of a composite structure is identified. The region has a cross-section with non-parallel surfaces. An estimated thickness for a location within the region and outside of the portion is identified using the thickness profile. An indication of whether the location has undesirable porosity is generated based on information about attenuation of response sound signals and the estimated thickness for the location.

**POROSITY INSPECTION SYSTEM FOR COMPOSITE STRUCTURES
WITH NON-PARALLEL SURFACES**

BACKGROUND INFORMATION

1. Field:

The present disclosure relates generally to composite structures and, in particular, to inspecting composite structures. Still more particularly, the present disclosure relates to a method and apparatus for inspecting composite structures to identify porosity in the composite structures.

2. Background:

Aircraft are being designed and manufactured with greater and greater percentages of composite materials. Composite materials are used in aircraft to decrease the weight of the aircraft. This decreased weight improves performance features such as payload capacity and fuel efficiency. Further, composite materials provide longer service life for various components in an aircraft.

Composite materials may be tough, light-weight materials created by combining two or more functional components. For example, a composite material may include reinforcing fibers bound in a polymer resin matrix. The fibers may be unidirectional or may take the form of a woven cloth or fabric. The fibers and resins may be arranged and cured to form a composite structure.

Using composite materials to create aerospace composite structures may allow for portions of an aircraft to be manufactured in larger pieces or sections. For example, a fuselage in an aircraft may be created in cylindrical sections to form the fuselage of the aircraft. Other examples include, without limitation, wing sections joined to form a wing or stabilizer sections joined to form a stabilizer.

In manufacturing composite structures, layers of composite material may be laid up on a tool. The layers of composite material may be comprised of fibers in sheets. These sheets may take the form of, for example, without limitation, fabrics, tape, tows, or other suitable configurations for the sheets. In some cases, resin may be infused or pre-impregnated into the sheets. These types of sheets are commonly referred to as prepreg.

The different layers of prepreg may be laid up in different orientations and different numbers of layers may be used depending on the desired thickness of the composite structure being manufactured. These layers may be laid up by hand or by using automated lamination equipment such as a tape laminating machine or a fiber placement system.

After the different layers have been laid up on the tool, the layers may be consolidated and cured upon exposure to temperature and pressure, thus forming the final composite structure. Thereafter, the composite structure may be inspected to ensure that inconsistencies are not present. The inspection may be performed using ultrasound testing, infrared testing, visual inspections, and other suitable types of testing.

This testing may be performed to identify various potential inconsistencies in the composite structure. For example, inconsistencies may include delamination, voids, undesired levels of porosity, and other types of inconsistencies.

With respect to porosity levels, increasing porosity in a composite structure may result in a less than desired load that the composite structure is able to withstand and perform as desired. Thus, a composite structure may be inspected using ultrasound testing to ensure that the porosity level is within a desired level for the composite structure.

Currently, when ultrasound testing is used to determine the porosity of the composite structure, the entire surface of the composite structure may be inspected. For example, an ultrasound system may perform ultrasound testing to gather data from all of the surfaces of each composite structure. This data may then be analyzed by an operator to identify the porosity of the composite structure. This inspection of the entire surface of the composite structure may take more time and manpower than desired.

Further, the analysis of the data of cross-sections with non-parallel surfaces may add to the time and manpower needed to inspect these composite structures. As a result, ultrasound testing of composite structures may take more time and may cost more than desired. Therefore, it would be desirable to have a method and apparatus that take into account at least some of the issues discussed above, as well as possibly other issues.

SUMMARY

In an illustrative embodiment, a method is provided. A thickness profile of a portion of a region of a composite structure is identified. The region has a cross-section with non-parallel surfaces. An estimated thickness for a location within the region and outside of the portion is identified using the thickness profile. An indication of whether the location has undesirable porosity is generated based on information about attenuation of response sound signals and the estimated thickness for the location.

In another illustrative embodiment, a method for inspecting a composite structure is provided. Information about attenuation of response sound signals in the composite structure is identified. An estimated thickness for a location in a region in the composite structure is identified from a thickness profile of the region. The region has a cross-section with non-parallel surfaces. A porosity of the composite structure at the location is identified based on the estimated thickness for the location and the information about the attenuation of the response sound signals at the location in the composite structure.

In a further illustrative embodiment, an ultrasound inspection system for a composite structure is provided. A number of transducers are configured to send sound signals into the composite structure and detect response sound signals to the sound signals sent into the composite structure. An analyzer is in communication with the number of transducers. The analyzer is configured to identify a thickness profile of a portion of a region of the composite structure, the region having a cross-section with non-parallel surfaces. The analyzer is further configured to identify an estimated thickness for a location within the region and outside of the portion using the thickness profile. The analyzer is still further configured to

generate an indication of whether the location has undesirable porosity using the estimated thickness for the location and information about attenuation of the response sound signals detected by the number of transducers.

According to an aspect of the present disclosure there is provided a method comprising: identifying (600) a thickness profile (164) of a portion (166) of a region (168) of a composite structure (104), the region (168) having a cross-section with non-parallel surfaces (172), identifying (602) an estimated thickness (138) for a location within the region (168) and outside of the portion (166) using the thickness profile (164), and generating (604) an indication of whether the location has undesirable porosity (106) based on information about attenuation of response sound signals (122) and the estimated thickness (138) for the location. Advantageously, the method wherein identifying the estimated thickness (138) for the location within the region (168) and outside of the portion (166) using the thickness profile (164) comprises applying the thickness profile (164) to the region (168) in a direction perpendicular to a direction of thickness variation in the region (168). Advantageously, the method wherein the thickness profile (164) comprises thickness values (162) for each location along a line, the line extending parallel to a direction of thickness variation in the region (168). Advantageously, the method further comprising: identifying the thickness values (162) using ultrasound inspection of the portion (166). Advantageously, the method further comprising: identifying the thickness values (162) from a model (210). Advantageously, the method further comprising: identifying an area having inconsistencies (157) within the region (168) of the composite structure (104) based on the information about the attenuation of the response sound signals (122) and the thickness profile (164). Advantageously, the method wherein the portion (166) of the region (168) has desirable reflection from a second surface (160) of the composite structure (104).

According to a further aspect of the present disclosure there is provided a method for inspecting a composite structure (104), the method comprising: identifying information about attenuation (150) of response sound signals (148) in the composite structure (104), identifying an estimated thickness (138) for a location in a region (168) in the composite structure (104) from a thickness profile (164) of the region (168), the region (168) having a cross-section with non-parallel surfaces (172), and identifying a porosity (106) of the composite structure (104) at the location based on the estimated thickness (138) for the location and the information about the attenuation (150) of the response sound signals (148) at the location in the composite structure (104). Advantageously, the method wherein the thickness profile (164) of the region (168) is for a portion (166) of the region (168), and wherein the location is outside of the portion (166). Advantageously, the method wherein identifying the estimated thickness (138) for the location within the region (168) from the thickness profile (164) comprises applying the thickness profile (164) to the region (168) in a direction perpendicular to a direction of thickness variation in the region (168). Advantageously, the method wherein the thickness profile (164) comprises thickness values (162) for each location along a line, the line extending parallel to a direction of thickness variation in the region (168). Advantageously, the method further comprising identifying the thickness values (162) using

ultrasound inspection. Advantageously, the method further comprising identifying the thickness values (162) from a model (210). Advantageously, the method further comprising identifying an area having inconsistencies (157) within the region (168) of the composite structure (104) based on the information about the attenuation of the response sound signals (122) and the thickness profile (164).

- 5 Advantageously, the method wherein the portion (166) of the region (168) has desirable reflection from a second surface (160) of the composite structure (104).

According to a yet further aspect of the present disclosure there is provided an ultrasound inspection system (102) for a composite structure (104), the ultrasound inspection system (102) comprising: a number of transducers (112) configured to send sound signals (144) into the composite structure (104) and detect response sound signals (148) to the sound signals (144) sent into the composite structure (104), and an analyzer (116) in communication with the number of transducers (112), wherein the analyzer (116) is configured to identify a thickness profile (164) of a portion (166) of a region (168) of the composite structure (104), the region (168) having a cross-section with non-parallel surfaces (172); identify an estimated thickness (138) for a location within the region (168) and outside of the portion (166) using the thickness profile (164); and generate an indication of whether the location has undesirable porosity (106) using the estimated thickness (138) for the location and information about attenuation (150) of the response sound signals (148) detected by the number of transducers (112). Advantageously the ultrasound inspection system (102) wherein identifying the estimated thickness (138) for the location within the region (168) and outside of the portion (166) using the thickness profile (164) comprises applying the thickness profile (164) to the region (168) in a direction perpendicular to a direction of thickness variation in the region (168). Advantageously the ultrasound inspection system (102) wherein identifying the thickness profile (164) comprises identifying thickness values (162) for each location along a line, the line extending parallel to a direction of thickness variation in the region (168). Advantageously the ultrasound inspection system (102) wherein identifying the thickness values (162) for the each location along the line comprises identifying the thickness values (162) using ultrasound inspection of the portion (166). Advantageously the ultrasound inspection system (102) wherein identifying the thickness values (162) for the each location along the line comprises identifying the thickness values (162) from a model (210).

- 30 The features and functions can be achieved independently in various embodiments of the present disclosure or may be combined in yet other embodiments in which further details can be seen with reference to the following description and drawings.

BRIEF DESCRIPTION OF THE DRAWINGS

- 35 The novel features believed characteristic of the illustrative embodiments are set forth in the appended claims. The illustrative embodiments, however, as well as a preferred mode of use, further objectives and features thereof, will best be understood by reference to the following detailed description

of an illustrative embodiment of the present disclosure when read in conjunction with the accompanying drawings, wherein:

Figure 1 is an illustration of a block diagram of an inspection environment in accordance with an illustrative embodiment;

Figure 2 is an illustration of data flow in an analysis of data for response sound signals in accordance with an illustrative embodiment;

Figure 3 is an illustration of data flow in creating baseline attenuation values in accordance with an illustrative embodiment;

Figure 4 is an illustration of an inspection environment in accordance with an illustrative embodiment;

Figure 5 is another illustration of an inspection environment in accordance with an illustrative embodiment;

Figure 6 is an illustration of a flowchart of a process for detecting porosity in composite structures in accordance with an illustrative embodiment;

Figure 7 is another illustration of a flowchart of a process for detecting porosity in composite structures in accordance with an illustrative embodiment;

Figure 8 is an illustration of a data processing system in the form of a block diagram in accordance with an illustrative embodiment;

Figure 9 is an illustration of an aircraft manufacturing and service method in the form of a block diagram in accordance with an illustrative embodiment; and

Figure 10 is an illustration of an aircraft in the form of a block diagram in which an illustrative embodiment may be implemented.

DETAILED DESCRIPTION

The different illustrative embodiments recognize and take into account different considerations. For example, the different illustrative embodiments recognize and take into account that evaluating the level of porosity in a composite structure may be the most time-consuming aspect of an inspection. The illustrative embodiments recognize and take into account that a manufacturing rate may be improved by reducing the inspection time.

Further, the different illustrative embodiments recognize and take into account that composite structures having a cross-section with non-parallel surfaces may have increased inspection times. The different illustrative embodiments recognize and take into account that locations within composite structures may be returned as a “no-test.” To evaluate porosity of a location of a composite structure, a thickness at the location is used. An area may be returned as “no-test” due to a thickness which cannot be determined from a response signal to sound signals sent into the composite structure. “No-test” locations may be automatically rejected.

Further, an operator may spend additional time during an analysis compensating for “no-test” locations. Yet further, “no-test” locations may require further analysis. This additional analysis may be more time-consuming and tedious than desired. Consequently, the manufacturing and testing of composite structures may be more expensive and less efficient than desired.

5 The different illustrative embodiments recognize and take into account that an area of composite structures having a cross-section with non-parallel surfaces may be inspected using a designated thickness. Each location in the area having the cross-section with non-parallel surfaces may be inspected using this same designated thickness.

10 The designated thickness may be a smallest thickness prior to the cross-section with non-parallel surfaces. The designated thickness may instead be a smallest estimated thickness of the cross-section with non-parallel surfaces. The designated thickness may be smaller than any thickness within the area having the cross-section with non-parallel surfaces.

15 The different illustrative embodiments recognize and take into account that by using a smaller thickness than an actual thickness at a location, the allowed porosity evaluation at the location may be more stringent than necessary. By using a designated thickness, a more stringent porosity evaluation than necessary may be applied to locations within the area having the non-parallel surfaces. As a result, composite structures having a cross-section with non-parallel surfaces may have a greater percentage of rejected material. Further, using a more stringent porosity evaluation may result in rejecting composite structure which is of acceptable porosity.

20 Thus, in one illustrative embodiment, a thickness profile of a portion of a region of a composite structure is identified. The region has a cross-section with non-parallel surfaces. An estimated thickness for a location within the region and outside of the portion is identified using the thickness profile. An indication of whether the location has undesirable porosity is generated based on information about attenuation of response sound signals and the estimated thickness for the location.

25 With reference now to the figures and, in particular, with reference to **Figure 1**, an illustration of a block diagram of an inspection environment is depicted in accordance with an illustrative embodiment. In this depicted example, inspection environment **100** includes ultrasound inspection system **102**. Ultrasound inspection system **102** may be used to perform an inspection of composite structure **104**. In particular, an inspection may be conducted to determine porosity **106** in composite structure **104**.

30 In this illustrative example, composite structure **104** may be part of platform **108**. As depicted, platform **108** may take a number of different forms. In particular, platform **108** may take the form of aircraft **110**. Composite structure **104** in aircraft **110** may be, for example, without limitation, one of a skin panel, a fuselage barrel, a stringer, a panel, a flap, a door, a wing box, or some other suitable type of composite structure.

35 Composite structure **104** may have first surface **158** and second surface **160**. As depicted, first surface **158** and second surface **160** are substantially parallel to each other in some of composite structure

104 and first surface **158** and second surface **160** are non-parallel in some of composite structure **104**. As a result, cross section **170** of composite structure **104** has non-parallel surfaces **172** and substantially parallel surfaces **174**.

As depicted, ultrasound inspection system **102** may be comprised of number of transducers **112**, movement system **114**, analyzer **116**, and other suitable components. Of course, ultrasound inspection system **102** may include other components not shown in this depicted example.

In this illustrative example, number of transducers **112** may be configured to send sound signals **118** into composite structure **104** at locations **120**. A "number of," as used herein with reference to items, means one or more items. For example, number of transducers **112** may be one or more transducers.

Locations **120** may form portion **166** of region **168** of composite structure **104** selected for inspection. In other words, region **168** of composite structure **104** may be selected for inspection. To begin inspection of region **168**, number of transducers **112** may send sound signals **118** into locations **120** which form portion **166**. Portion **166** may be a fraction of region **168** or may be all of region **168**. In one illustrative example, portion **166** may have non-parallel surfaces **172**, substantially parallel surfaces **174**, or a combination thereof. In some illustrative examples, portion **166** of region **168** may have a desirable reflection of sound signals **118** from second surface **160** of composite structure **104**.

Sound signals **118** may be generated by number of transducers **112** under the control of analyzer **116** in this illustrative example. Sound signals **118** may have different frequencies. As depicted, sound signals **118** may be, for example, ultrasound signals. Without limitation, sound signals **118** may have a frequency from about 20 kilohertz to about 200 megahertz. Of course, other frequencies may be used for sound signals **118** depending on the particular implementation.

In response to sound signals **118** traveling into composite structure **104** at locations **120**, response sound signals **122** may be generated. In this illustrative example, response sound signals **122** may be detected by number of transducers **112**. In response to detecting response sound signals **122**, number of transducers **112** may generate and send data **124** to analyzer **116** for processing. In some illustrative examples, data **124** may be, for example, electrical signals indicating amplitudes **126** of response sound signals **122**. The electrical signals may be analog, digital, or some combination thereof. In some illustrative examples, data **124** may be, for example, electrical signals indicating times of flight **127** of response sound signals **122**. Times of flight **127** may be periods of time from when sound signals **118** are sent into locations **120** and response sound signals **122** are detected by number of transducers **112**.

In this illustrative example, analyzer **116** may be implemented using hardware, software, or a combination of the two. When software is used, the operations performed by analyzer **116** may be implemented in program code configured to be run on a processor unit. When hardware is employed, the hardware may include circuits that operate to perform operations in analyzer **116**.

The hardware may take the form of a circuit system, an integrated circuit, an application specific integrated circuit (ASIC), a programmable logic device, or some other suitable type of hardware

configured to perform a number of operations. With a programmable logic device, the device may be configured to perform the number of operations. The device may be reconfigured at a later time or may be permanently configured to perform the number of operations. Examples of programmable logic devices include, for example, a programmable logic array, a programmable array logic, a field programmable logic array, a field programmable gate array, and other suitable hardware devices. Additionally, the processes may be implemented in organic components integrated with inorganic components and/or may be comprised entirely of organic components excluding a human being. For example, the processes may be implemented as circuits in organic semiconductors.

In this illustrative example, analyzer **116** may be implemented in computer system **128**.

Computer system **128** may be one or more computers. When more than computer is present in computer system **128**, those computers may be in communication with each other via a communication medium such as a network.

In this illustrative example, analyzer **116** may identify thickness values using ultrasound inspection of portion **166**. Specifically, in this illustrative example, analyzer **116** may identify thickness values **162** for locations **120** from times of flight **127**. Thickness values **162** may be thickness measurements at each location in locations **120**. Thickness values **162** may form thickness profile **164** for locations **120** within portion **166**. In some illustrative examples, portion **166** may comprise a line extending parallel to a direction of thickness variation in region **168**. In these illustrative examples, thickness profile **164** may comprise thickness values for each location in locations **120** along the line comprising portion **166**.

Thickness profile **164** may be applied to region **168**. Thickness profile **164** may be applied to region **168** in a direction perpendicular to a direction of thickness variation in region **168**. In other words, thickness profile **164** may be applied to region **168** in a direction perpendicular to a direction of variation of second surface **160**.

Estimated thicknesses **138** of locations **146** within region **168** but outside of portion **166** may be identified by applying thickness profile **164** to region **168**. Estimated thicknesses **138** may correspond to values in thickness values **162**.

In this illustrative example, analyzer **116** may identify attenuations **130** of response sound signals **122**. Attenuations **130** may be identified by comparing amplitudes **126** of response sound signals **122** with amplitudes **132** of sound signals **118**.

Analyzer **116** may compare attenuations **130** with baseline attenuation values **152** to identify if porosity **106** in composite structure **104** at locations **120** is undesirable. Baseline attenuation values **152** represent maximum porosities **136** that are desired for thickness values **162** of composite structure **104** at locations **120**.

In this illustrative example, thickness values **162** may be representative of distances from first surface **158** to second surface **160** at locations **120**. Locations **120** where sound signals **118** enter

composite structure **104** may be located on first surface **158**. Second surface **160** may be substantially opposite to first surface **158**. In these illustrative examples, baseline attenuation values **152** may be identified in a number of different ways.

For example, baseline attenuation values **152** may be baseline values taken for thicknesses of a test object that may be a reference standard, a requirements value, a value from a database, or some other suitable type of value for baseline attenuation values **152**. As depicted, baseline attenuation values **152** may be threshold values.

Comparing attenuations **130** of response sound signals **122** to baseline attenuation values **152** may result in comparisons **154**. If an attenuation in attenuations **130** is greater than the baseline attenuation value in baseline attenuation values **152** for a location in locations **120**, then porosity **106** at the location in locations **120** may be greater than desired. As a result, additional evaluation of the location in locations **120** may be needed.

In this illustrative example, indications **156** may be generated by analyzer **116**. Indications **156** may be generated based on comparisons **154** of attenuations **130** in response sound signals **122** and baseline attenuation values **152** for porosity **106** for thickness values **162** of locations **120** of composite structure **104**.

Indications **156** may indicate whether additional evaluation of composite structure **104** may be needed at locations **120**. Indications **156** may indicate that additional evaluation of composite structure **104** may be needed at locations **120** or that additional evaluation of composite structure **104** may not be needed at locations **120**.

Further, analyzer **116** may control number of transducers **112** to send sound signals **144** into locations **146** in composite structure **104**. As depicted, analyzer **116** may control movement system **114** to move number of transducers **112** with respect to composite structure **104** to locations **146** on at least one of first surface **158** and second surface **160**. Number of transducers **112** may then send sound signals **144** into locations **146** of region **168** on composite structure **104**. In this manner, ultrasound inspection system **102** may test all of region **168** of composite structure **104**.

In response to sound signals **144** being sent into locations **146**, response sound signals **148** may be generated. Number of transducers **112** may detect response sound signals **148** and generate data **124** about response sound signals **148**.

Analyzer **116** may receive data **124** about response sound signals **148** from number of transducers **112** and identify attenuations **150** for response sound signals **148**. Analyzer **116** may compare attenuations **150** with baseline attenuation values **152** to identify if porosity **106** in composite structure **104** at locations **146** is undesirable. As each estimated thickness in estimated thicknesses **138** corresponds to a thickness value in thickness values **162**, baseline attenuation values **152** represent maximum porosities **136** that are desired for estimated thicknesses **138** of composite structure **104** at locations **146**. In this illustrative example, a baseline attenuation value in baseline attenuation values **152**

is selected for each location in locations **146** based on a corresponding estimated thickness in estimated thicknesses **138**. In comparing attenuations **150** with baseline attenuation values **152**, analyzer **116** forms comparisons **154**. Based on comparisons **154** of attenuations **150** with baseline attenuation values **152**, analyzer **116** may generate indications **156**. Indications **156** may indicate that porosity **106** at locations **146** is undesirable. If porosity **106** at a location in locations **146** is undesirable, this may be an inconsistency in inconsistencies **157**. Accordingly, an area having inconsistencies in inconsistencies **157** within region **168** of composite structure **104** may be identified based on attenuations **150** of response sound signals **148** and thickness profile **164**.

Indications **156** may indicate whether additional evaluations of locations **146** are needed at locations **146**. These additional evaluations may be additional evaluations for areas of composite structure **104** that may show disbonding or delamination. When disbonding and/or delamination may be present, indications **156** may show heavier levels of porosity than desired. In this illustrative example, these areas of porosity, indicated by indications **156**, may be retested.

In other illustrative examples, additional ultrasound testing may be performed on composite structure **104** using a different transducer in number of transducers **112** than was used to send sound signals **118** into composite structure **104** during the first test. Further, areas of composite structure **104** may be retested using a different frequency from number of transducers **112**. In this manner, ultrasound inspection system **102** and/or other types of inspection systems may be used to further inspect areas of composite structure **104**.

Thus, the illustrative embodiments provide a method to identify the porosity of composite structure **104** using ultrasound inspection system **102**. With the comparison of attenuations **130** to response sound signals **122** and baseline attenuation values **152**, an operator may use the illustrative embodiments to quickly and efficiently determine porosity **106** within region **168** of composite structure **104**.

The illustration of inspection environment **100** in **Figure 1** is not meant to imply physical or architectural limitations to the manner in which an illustrative embodiment may be implemented. Other components in addition to or in place of the ones illustrated may be used. Some components may be unnecessary. Also, the blocks are presented to illustrate some functional components. One or more of these blocks may be combined, divided, or combined and divided into different blocks when implemented in an illustrative embodiment.

For example, in some illustrative examples, comparisons **154** may not be created based on response sound signals **122**. In these illustrative examples, thickness values **162** may be identified based on response sound signals **122**. However, comparisons **154** may be created based on a second set of response sound signals in response to a second set of sound signals sent into locations **120**.

As another illustrative example, thickness values **162** may not be identified based on times of flight **127** of response sound signals **122**. Thickness values **162** may instead be identified based on thicknesses of corresponding locations in a model of composite structure **104**.

Turning now to **Figure 2**, an illustration of data flow in an analysis of data for response sound signals is depicted in accordance with an illustrative embodiment. As depicted, analyzer **116** receives data **124** about response sound signals **122** and response sound signals **148** generated in response to sound signals **118** and sound signals **144** being sent into composite structure **104** at locations **120** and locations **146** as shown in **Figure 1**. Data **124** may be used to identify attenuations **130** and attenuations **150** in response sound signals **122** and response sound signals **148** as shown in **Figure 1**.

With attenuations **130**, database **200** may be used to identify baseline attenuation values **152** from baseline attenuation values **202**. In these illustrative examples, baseline attenuation values **202** may be indexed in a number of different ways. For example, thicknesses **204** may be used as an index in baseline attenuation values **202**. Thicknesses **204** may correspond to thicknesses in composite structure **104** in **Figure 1**.

In one illustrative example, analyzer **116** may use thickness values **162** of composite structure **104** at locations **120** as shown in **Figure 1** as an index to identify baseline attenuation values **152** that correspond to thickness values **162** in thicknesses **204** in database **200**. In other words, thicknesses **204** serves as an index to identify baseline attenuation values **152** for thickness values **162** at locations **120**.

Thickness values **162** may be identified in a number of different ways. For example, locations **120** may be used to identify corresponding locations **208** in model **210** from which an estimate of thickness values **162** may be identified using geometric information **212** in model **210**.

In this illustrative example, model **210** may be a model of composite structure **104**. As another example, model **210** may be a model of platform **108** including composite structure **104** as shown in **Figure 1**. In this illustrative example, model **210** takes the form of computer-aided design (CAD) model **214**.

As depicted, geometric information **212** may be, for example, without limitation, thicknesses, lengths, orientations, or other information regarding features of composite structure **104**. Corresponding locations **208** in model **210** may correspond to locations **120** on composite structure **104**. In other words, corresponding locations **208** may be representations of locations **120** on composite structure **104** in model **210**.

With corresponding locations **208** in model **210**, analyzer **116** may identify thicknesses **216** for corresponding locations **208** from geometric information **212** for composite structure **104** in model **210**. However, thicknesses **216** may only be estimates of thickness values **162** for composite structure **104** at locations **120**. Variances may occur in thickness values **162** relative to thicknesses **216** during manufacturing of composite structure **104**. Depending on the implementation, thicknesses **216** may be sufficiently accurate for use by analyzer **116**.

In another illustrative example, analyzer **116** may identify thickness values **162** without using model **210**. For example, without limitation, thicknesses **204** may be identified based on times of flight **127**. Times of flight **127** may be periods of time from when sound signals **118** are sent into locations **120** and response sound signals **122** are detected in **Figure 1**.

5 With times of flight **127**, analyzer **116** may identify thickness values **162** using other parameters of sound signals **118**. For example, the speed of sound signals **118** may be used in determining thicknesses **204** when times of flight **127** are used.

After baseline attenuation values **152** are identified, analyzer **116** may then compare baseline attenuation values **152** with attenuations **130** and attenuations **150** to generate comparisons **154** in **Figure**
10 **1**. In comparing baseline attenuation values **152** with attenuations **130**, thickness values **162** are used. In comparing baseline attenuation values **152** with attenuations **150**, estimated thicknesses **138** are used. With comparisons **154**, indications **156** may be generated indicating whether additional evaluation of locations **120** and locations **146** is needed.

By applying thickness profile **164** to region **168**, thickness evaluations need not be done for
15 locations **146** in region **168**. As a result, ultrasound testing on composite structure **104** may be conducted efficiently and with less time-consuming data analysis. Further, by using estimated thicknesses **138**, locations **146** may not be labeled as “no-test” areas. Thus an operator may focus on areas of composite structure **104** that need additional evaluation and analysis instead of analyzing all data over the entire surface of composite structure **104** in **Figure 1**.

20 Turning now to **Figure 3**, an illustration of data flow in creating baseline attenuation values is depicted in accordance with an illustrative embodiment. This figure illustrates the generation of baseline attenuation values **202** for database **200**.

In this illustrative example, analyzer **116** may receive data **300**. Data **300** may be generated using reference standard **302**. In particular, data **300** may include amplitudes **304** of response sound signals **306**
25 generated in response to sending sound signals **308** into reference standard **302** using number of transducers **112**.

Reference standard **302** may be composite structure **310**. As depicted, reference standard **302** may have the same or different dimensions as composite structure **104** in **Figure 1**. Reference standard **302** has porosity **312** at a desired level throughout composite structure **310** for thicknesses **314**.

30 With amplitudes **304** in data **300**, attenuations **316** in response sound signals **306** may be identified for reference standard **302** for locations **318** in reference standard **302**. For example, attenuations **316** may be identified by comparing amplitudes **304** in response sound signals **306** with amplitudes **320** for sound signals **308**.

In particular, attenuations **316** may be values for thicknesses **314** at locations **318** in reference
35 standard **302**. Attenuations **316** may represent values for porosity **312** in reference standard **302**.

Additionally, analyzer **116** may identify tolerances **322** for porosity **312**. Tolerances **322** may be based on a number of different factors. For example, tolerances **322** may be based on at least one of a desired load for reference standard **302**, safety margins, and other suitable factors. With tolerances **322**, attenuations **316** may be adjusted to obtain baseline attenuation values **202**.

5 In these illustrative examples, tolerances **322** may be an allowable adjustment in porosity **312** for reference standard **302**. Tolerances **322** may be the same for locations **318** or may be different for different locations in locations **318**. This adjustment may be an increase in porosity **312**. This adjustment may be made by increasing attenuations **316** that reflect the increase in porosity **312** that may be allowable.

10 As depicted, baseline attenuation values **202** in database **200** may take a number of different forms. For example, without limitation, baseline attenuation values **202** may take the form of graphs of curves, tables, length lists, flat files, and other suitable forms.

In this manner, ultrasound inspection of reference standard **302** may provide baseline attenuation values **202** for use in ultrasound inspection of composite structure **104** in **Figure 1**. Further, acceptable
15 tolerances for porosity **106** in composite structure **104** may be identified. With the identification of tolerances **322**, database **200** of baseline attenuation values **202** may be generated for different thicknesses in thicknesses **204** of composite structure **104** and used for inspecting composite structure **104** in **Figure 1**.

The illustrations of data flows in **Figures 2** and **3** are not meant to imply physical or architectural
20 limitations to the manner in which an illustrative embodiment may be implemented. Other components in addition to or in place of the ones illustrated may be used. Some components may be unnecessary. Also, the blocks are presented to illustrate some functional components. One or more of these blocks may be combined, divided, or combined and divided into different blocks when implemented in an illustrative embodiment.

25 For example, baseline attenuation values **202** may be indexed to locations in database **200**. The locations in database **200** may correspond to locations in composite structure **104** in **Figure 1**.

Turning now to **Figure 4**, an illustration of an inspection environment is depicted in accordance with an illustrative embodiment. In this depicted example, inspection environment **400** is an example of a physical implementation for inspection environment **100** and components in inspection environment **100**
30 in **Figure 1** and **Figure 2**.

In this illustrative example, inspection environment **400** includes ultrasound inspection system **402**, which may be configured to inspect composite structure **404**. Composite structure **404** has first surface **403** and second surface **405**. As can be seen in this illustrative example, composite structure **404** may have thickness **406** at location **408**, thickness **410** at location **412**, thickness **414** at location **416**, and
35 thickness **418** at location **420**. As can be seen, thickness **406**, thickness **410**, thickness **414**, and **418** are each different in composite structure **404**.

Thickness **406** is located in parallel surface area **422**. Thicknesses within parallel surface area **422** are substantially the same as thickness **406**. Thickness **410** and thickness **414** are located in non-parallel surface area **424**. Thicknesses within non-parallel surface area **424** vary. As depicted, second surface **405** angles upward at a substantially constant angle. In some other illustrative examples, second surface **405** may instead angle upward at a varying angle. In some other illustrative examples, second surface **405** may angle downwards in a constant or varying angle. In yet other illustrative examples, second surface **405** may be a combination of upward and downward angles. Thickness **418** is located in parallel surface area **426**. Thicknesses within parallel surface area **426** are substantially the same as thickness **418**.

In this depicted example, ultrasound inspection system **402** may include number of transducers **434**, movement system **430**, and analyzer **432**. As depicted, number of transducers **434** may include transducer **428**. Transducer **428** may be configured to be moved by movement system **430**.

In this example, movement system **430** may include robotic arm **436**. Robotic arm **436** may be configured to be controlled by analyzer **432**. In this illustrative example, robotic arm **436** may be configured to move transducer **428** to different locations on first surface **403** of composite structure **404**.

In this illustrative example, transducer **428** may be configured to send sound signal **440** into composite structure **404** at location **416**. Additionally, transducer **428** may also be configured to detect response sound signal **442**. Transducer **428** generates data and sends the data to analyzer **432** in response to detecting response sound signal **442**.

In this illustrative example, analyzer **432** may be implemented in computer **438**. Computer **438** may be one implementation for a computer in computer system **128** shown in block form in **Figure 1**. Analyzer **432** may identify the attenuation for response sound signal **442** and determine whether an indication should be generated for location **416**. This determination may be based on a desired level of porosity for thickness **414** at location **416**. The attenuation may be used to identify the amount of porosity based on thickness **414** at location **416**.

In these illustrative examples, transducer **428** may be moved by robotic arm **436** to other locations on first surface **403** of composite structure **404** to send additional sound signals and detect additional response sound signals. This information from other locations may be used to generate indications of whether additional evaluation is needed for those locations.

Turning now to **Figure 5**, another illustration of an inspection environment is depicted in accordance with an illustrative embodiment. In this depicted example, inspection environment **400** is an example of a physical implementation for inspection environment **100** and components in inspection environment **100** in **Figure 1** and **Figure 2**. **Figure 5** is a depiction of inspection environment **400**. In this illustrative example, **Figure 4** is a view of **Figure 5** along cross-section line **4**.

In this illustrative example, inspection environment **400** includes ultrasound inspection system **402** which may be configured to inspect composite structure **404**. Composite structure **404** has parallel

surface area **422**, non-parallel surface area **424**, and parallel surface area **426**. Parallel surface area **422** and non-parallel surface area **424** are separated by line **502**. Line **502** may represent the beginning of an angle in second surface **405** of composite structure **404**. Non-parallel surface area **424** and parallel surface area **426** are separated by line **504**. Line **504** may represent the end of an angle in second surface **405** of composite structure **404**. Composite structure **404** may have area of undesirable porosity **506**. Area of undesirable porosity **506** may be identified by inspecting composite structure **404** using ultrasound inspection system **402**.

In this illustrative example, ultrasound inspection system **402** may be configured to inspect region **508** of composite structure **404**. Region **508** may be identified for inspection by a human operator or by computer **438**. To inspect region **508** of composite structure **404**, ultrasound inspection system **402** may send signals into locations within region **508**. Attenuations of response sound signals received from locations within region **508** may be used to identify porosity of region **508**.

To determine porosity of a location in region **508**, attenuation for the location may be compared to a baseline attenuation for a thickness of the location. Region **508** has portion **510**. As depicted, portion **510** is a line extending across region **508**. In some illustrative examples, portion **510** may instead be an area, a point, or other suitable section of region **508**.

In this illustrative example, thicknesses for locations in region **508** may be identified by applying thickness profile **511** of portion **510** to region **508**. Thickness values in thickness profile **511** may be identified in a variety of ways.

In one illustrative example, thickness values in thickness profile **511** may be identified using ultrasound inspection system **402**. To identify thicknesses of locations in portion **510**, ultrasound inspection system **402** may send sound signals such as sound signals **118** of **Figure 1** into locations **509** in portion **510**. The thickness values for each location in locations **509** may be identified from times of flight such as times of flight **127** of **Figure 1** of response sound signals such as response sound signals **122** of **Figure 1** received from locations **509**. The thickness values for portion **510** form thickness profile **511** of region **508**.

In other illustrative examples, thickness values in thickness profile **511** may be identified from a model, such as model **210** of **Figure 2**. In these illustrative examples, thicknesses for corresponding locations in the model may be applied to locations within portion **510**. The thickness values for portion **510** form thickness profile **511** of region **508**.

Thickness profile **511** may next be applied to region **508**. In this illustrative example, thickness profile **511** is applied to region **508** in a direction perpendicular to the direction of thickness variation in region **508**. In this illustrative example, thickness profile **511** is applied to region **508** in direction **512** and direction **514**.

By applying thickness profile **511** to region **508**, estimated thicknesses are identified for locations within region **508** that are outside of portion **510**. For example, location **516** is within portion **510**.

Location **516** will have a thickness value within thickness profile **511**. By applying thickness profile **511** along direction **514**, the thickness value of location **516** will be applied to location **518**. As a result, the estimated thickness of location **518** will be the same value or substantially the same value as the thickness value of location **516**.

5 Likewise, location **520** of portion **510** will have a thickness value within thickness profile **511**. By applying thickness profile **511** along direction **514**, the thickness value of location **520** will be applied to location **522**.

10 In one illustrative example, analyzer **432** may identify the attenuation for a response sound signal received from location **518**. Analyzer **432** may determine whether an indication should be generated for location **518**. This determination may be based on a desired level of porosity for the estimated thickness of location **518**, which is equivalent to the thickness value of location **516**.

15 In one illustrative example, analyzer **432** may identify the attenuation for a response sound signal received from location **522**. Analyzer **432** may determine whether an indication should be generated for location **522**. This determination may be based on a desired level of porosity for the estimated thickness of location **522**, which is equivalent to the thickness value of location **520**.

20 The different components illustrated for inspection environment **400** are not meant to provide architectural limitations to the manner in which different embodiments may be implemented. The different illustrative embodiments may be implemented in an inspection environment including components in addition to and/or in place of those illustrated for inspection environment **400**. Other components shown in **Figures 4** and **5** can be varied from the illustrative examples shown. For example, computer **438** may communicate with number of transducers **434** over a wireless connection.

25 The different components shown in **Figures 4** and **5** may be combined with components in **Figures 1-3**, used with components in **Figures 1-3**, or a combination of the two. Additionally, some of the components in **Figures 4** and **5** may be illustrative examples of how components shown in block form in **Figures 1-3** can be implemented as physical structures.

Turning now to **Figure 6**, an illustration of a flowchart of a process for detecting porosity in composite structures is depicted in accordance with an illustrative embodiment. The process illustrated in **Figure 6** may be implemented using ultrasound inspection system **102** in **Figure 1**. Further, this process may be implemented to inspect composite structure **104** for porosity **106** in **Figure 1**.

30 The process may begin by identifying a thickness profile of a portion of a region of a composite structure, the region having a cross-section with non-parallel surfaces (operation **600**). The process may then identify an estimated thickness for a location within the region and outside of the portion using the thickness profile (operation **602**). The process may then generate an indication of whether the location has undesirable porosity based on information about attenuation of response sound signals and the estimated thickness for the location (operation **604**). Afterwards, the process terminates.

35

Turning now to **Figure 7**, another illustration of a flowchart of a process for detecting porosity in composite structures is depicted in accordance with an illustrative embodiment. The process illustrated in **Figure 7** may be implemented using ultrasound inspection system **102** in **Figure 1**. Further, this process may be implemented to inspect composite structure **104** for porosity **106** in **Figure 1**.

5 The process may begin by identifying information about attenuation of response sound signals in the composite structure (operation **700**). The process may then identify an estimated thickness for a location in a region in the composite structure from a thickness profile of the region, the region having a cross-section with non-parallel surfaces (operation **702**). The process may then identify a porosity of the composite structure at the location based on the estimated thickness for the location and the information
10 about the attenuation of the response sound signals at the location in the composite structure (operation **704**). Afterwards, the process terminates.

The flowcharts and block diagrams in the different depicted embodiments illustrate the architecture, functionality, and operation of some possible implementations of apparatuses and methods in an illustrative embodiment. In this regard, each block in the flowcharts or block diagrams may represent
15 a module, segment, function, and/or a portion of an operation or step. For example, one or more of the blocks may be implemented as program code, in hardware, or a combination of the program code and hardware. When implemented in hardware, the hardware may, for example, take the form of integrated circuits that are manufactured or configured to perform one or more operations in the flowcharts or block diagrams.

20 For example, in **Figure 7**, information about attenuation of response sound signals is identified. All of the response sound signals may be received in operation **700** before identifying attenuations for the response sound signals. This type of processing may be referred to as a post-processing of data. In some illustrative examples, the operations may be performed each time a response sound signal is received rather than after all of the response sound signals have been received. This type of processing may be
25 referred to as dynamic processing of data.

Turning now to **Figure 8**, an illustration of a data processing system is depicted in the form of a block diagram in accordance with an illustrative embodiment. Data processing system **800** may be used to implement one or more computers in computer system **128** in **Figure 1**. In this illustrative example, data processing system **800** includes communications framework **802**, which may provide
30 communications between processor unit **804**, memory **806**, persistent storage **808**, communications unit **810**, input/output (I/O) unit **812**, and display **814**. In this example, communications framework **802** may take the form of a bus system.

Processor unit **804** may execute instructions for software that may be loaded into memory **806**. Processor unit **804** may be a number of processors, a multi-processor core, or some other type of
35 processor, depending on the particular implementation.

Memory **806** and persistent storage **808** may be examples of storage devices **816**. A storage device may be any piece of hardware that is capable of storing information, such as, for example, without limitation, data, program code in functional form, and/or other suitable information either on a temporary basis and/or a permanent basis. Storage devices **816** may also be referred to as computer readable storage
5 devices in these illustrative examples. Memory **806**, in these examples, may be, for example, a random access memory or any other suitable volatile or non-volatile storage device. Persistent storage **808** may take various forms, depending on the particular implementation.

For example, persistent storage **808** may contain one or more components or devices. For example, persistent storage **808** may be a hard drive, a flash memory, a rewritable optical disk, a
10 rewritable magnetic tape, or some combination of the above. The media used by persistent storage **808** also may be removable. For example, a removable hard drive may be used for persistent storage **808**.

Communications unit **810**, in these illustrative examples, may provide for communications with other data processing systems or devices. In these illustrative examples, communications unit **810** may be a network interface card.

Input/output (I/O) unit **812** may allow for input and output of data with other devices that may be
15 connected to data processing system **800**. For example, input/output (I/O) **812** may provide a connection for user input through a keyboard, a mouse, and/or some other suitable input device. Further, input/output (I/O) unit **812** may send output to a printer. Display **814** provides a mechanism to display information to a user.

Instructions for the operating system, applications, and/or programs may be located in storage
20 devices **816**, which are in communication with processor unit **804** through communications framework **802**. The processes of the different embodiments may be performed by processor unit **804** using computer-implemented instructions, which may be located in a memory, such as memory **806**.

These instructions may be referred to as program code, computer usable program code, or
25 computer readable program code that may be read and executed by a processor in processor unit **804**. The program code in the different embodiments may be embodied on different physical or computer readable storage media, such as memory **806** or persistent storage **808**.

Program code **818** may be located in a functional form on computer readable media **820** that is selectively removable and may be loaded onto or transferred to data processing system **800** for execution
30 by processor unit **804**. Program code **818** and computer readable media **820** may form computer program product **822** in these illustrative examples. In one example, computer readable media **820** may be computer readable storage media **824** or computer readable signal media **826**.

In these illustrative examples, computer readable storage media **824** is a physical or tangible storage device used to store program code **818** rather than a medium that propagates or transmits program
35 code **818**.

Alternatively, program code **818** may be transferred to data processing system **800** using computer readable signal media **826**. Computer readable signal media **826** may be, for example, a propagated data signal containing program code **818**. For example, computer readable signal media **826** may be an electromagnetic signal, an optical signal, and/or any other suitable type of signal. These signals may be transmitted over communications links, such as wireless communications links, optical fiber cable, coaxial cable, a wire, and/or any other suitable type of communications link.

The different components illustrated for data processing system **800** are not meant to provide architectural limitations to the manner in which different embodiments may be implemented. The different illustrative embodiments may be implemented in a data processing system including components in addition to and/or in place of those illustrated for data processing system **800**. Other components shown in **Figure 8** can be varied from the illustrative examples shown. The different embodiments may be implemented using any hardware device or system capable of running program code **818**.

Illustrative embodiments of the disclosure may be described in the context of aircraft manufacturing and service method **900** as shown in **Figure 9** and aircraft **1000** as shown in **Figure 10**. For example, ultrasound inspection system **102** in **Figure 1** may be used during at least one of the steps shown in this figure.

Turning first to **Figure 9**, an illustration of an aircraft manufacturing and service method is depicted in the form of a block diagram in accordance with an illustrative embodiment. During pre-production, aircraft manufacturing and service method **900** may include specification and design **902** of aircraft **1000** in **Figure 10** and material procurement **904**.

During production, component and subassembly manufacturing **906** and system integration **908** of aircraft **1000** in **Figure 10** takes place. Thereafter, aircraft **1000** in **Figure 10** may go through certification and delivery **910** in order to be placed in service **912**. While in service **912** by a customer, aircraft **1000** in **Figure 10** is scheduled for routine maintenance and service **914**, which may include modification, reconfiguration, refurbishment, and other maintenance or service.

Each of the processes of aircraft manufacturing and service method **900** may be performed or carried out by a system integrator, a third party, and/or an operator. In these examples, the operator may be a customer. For the purposes of this description, a system integrator may include, without limitation, any number of aircraft manufacturers and major-system subcontractors; a third party may include, without limitation, any number of vendors, subcontractors, and suppliers; and an operator may be an airline, a leasing company, a military entity, a service organization, and so on.

With reference now to **Figure 10**, an illustration of an aircraft is depicted in the form of a block diagram in which an illustrative embodiment may be implemented. Aircraft **1000** may be one implementation for platform **108** in **Figure 1**. Specifically, aircraft **1000** may be one implementation for aircraft **110** in **Figure 1**.

In this example, aircraft **1000** is produced by aircraft manufacturing and service method **900** in **Figure 9** and may include airframe **1002** with plurality of systems **1004** and interior **1006**. Examples of systems **1004** include one or more of propulsion system **1008**, electrical system **1010**, hydraulic system **1012**, and environmental system **1014**. Any number of other systems may be included. Although an aerospace example is shown, different illustrative embodiments may be applied to other industries, such as the automotive industry.

Apparatuses and methods embodied herein may be employed during at least one of the stages of aircraft manufacturing and service method **900** in **Figure 9**. In one illustrative example, composite structures manufactured during component and subassembly manufacturing **906** may be inspected using ultrasound inspection system **102** in **Figure 1** as well as composite structures that may be manufactured while aircraft **1000** is in service **912** or during maintenance and service **914**. One or more apparatus embodiments, method embodiments, or a combination thereof may be utilized while aircraft **1000** is in service **912** and/or during maintenance and service **914** in **Figure 9**. The use of a number of the different illustrative embodiments may substantially expedite the assembly of and/or reduce the cost of aircraft **1000**.

Thus, the illustrative embodiments provide a method and apparatus for inspecting composite structures. By using ultrasound inspection system **102**, inspections of composite structures may be made more quickly and with less effort from operators. Ultrasound inspection system **102** is configured to provide indications of whether additional evaluations are needed by an operator. These indications may be based on a comparison of the attenuation in response sound signals to baseline attenuation values. These baseline attenuation values may be selected to provide an indication of when the porosity is greater than a desired level. Thus, when the attenuation is greater than a baseline attenuation value, additional evaluation of the location may be needed.

In inspecting the composite structure, thickness values may be identified for locations within a portion of a region being inspected. A thickness profile may comprise these thickness values. The thickness profile may be applied to the region. Accordingly, estimated thicknesses may be applied to locations within the region but outside of the portion. The estimated thicknesses may have the same or substantially the same values as the thickness values in the thickness profile. Baseline attenuation values for locations in the region but outside of the portion may be selected based on estimated thicknesses. Thus thicknesses for each location need not be individually identified.

By using estimated thicknesses for non-parallel areas, less stringent porosity requirements may be applied than using a single designated thickness. Accordingly, using estimated thicknesses for non-parallel areas may result in less rejected material. Further, by using estimated thicknesses for non-parallel areas, these areas may not be labelled as “non-test” areas. Thus, by using estimated thicknesses for non-parallel areas, operators may spend less time compensating for “no-test” regions.

The description of the different illustrative embodiments has been presented for purposes of illustration and description, and is not intended to be exhaustive or limited to the embodiments in the form disclosed. Many modifications and variations will be apparent to those of ordinary skill in the art. Further, different illustrative embodiments may provide different features as compared to other

5 illustrative embodiments. The embodiment or embodiments selected are chosen and described in order to best explain the principles of the embodiments, the practical application, and to enable others of ordinary skill in the art to understand the disclosure for various embodiments with various modifications as are suited to the particular use contemplated.

CLAIMS:

What is claimed is:

- 5
1. A method for inspecting a composite structure (104), the method comprising:
identifying information about attenuation (150) of response sound signals (148) in the composite structure (104);
identifying an estimated thickness (138) for a location in a region (168) in the composite structure
10 (104) from a thickness profile (164) of the region (168), the region (168) having a cross-section with non-parallel surfaces (172); and
identifying a porosity (106) of the composite structure (104) at the location based on the estimated thickness (138) for the location and the information about the attenuation (150) of the response sound signals (148) at the location in the composite structure (104).
15
 2. The method of claim 1, wherein the thickness profile (164) of the region (168) is for a portion (166) of the region (168), and wherein the location is outside of the portion (166).
 3. The method of claim 2, wherein identifying the estimated thickness (138) for the location within
20 the region (168) from the thickness profile (164) comprises applying the thickness profile (164) to the region (168) in a direction perpendicular to a direction of thickness variation in the region (168).
 4. The method of claim 1, 2, or 3, wherein the thickness profile (164) comprises thickness values (162) for each location along a line, the line extending parallel to a direction of thickness variation in the
25 region (168).
 5. The method of claim 4 further comprising:
identifying the thickness values (162) using ultrasound inspection.
 - 30 6. The method of claim 4 further comprising:
identifying the thickness values (162) from a model (210).
 7. The method of any of the preceding claims further comprising:
identifying an area having inconsistencies (157) within the region (168) of the composite
35 structure (104) based on the information about the attenuation of the response sound signals and the thickness profile (164).

8. The method of claim 2, wherein the portion (166) of the region (168) has desirable reflection from a second surface (160) of the composite structure (104).

5 9. An ultrasound inspection system (102) for a composite structure (104), the ultrasound inspection system (102) comprising:

a number of transducers (112) configured to send sound signals (144) into the composite structure (104) and detect response sound signals (148) to the sound signals (144) sent into the composite structure (104); and

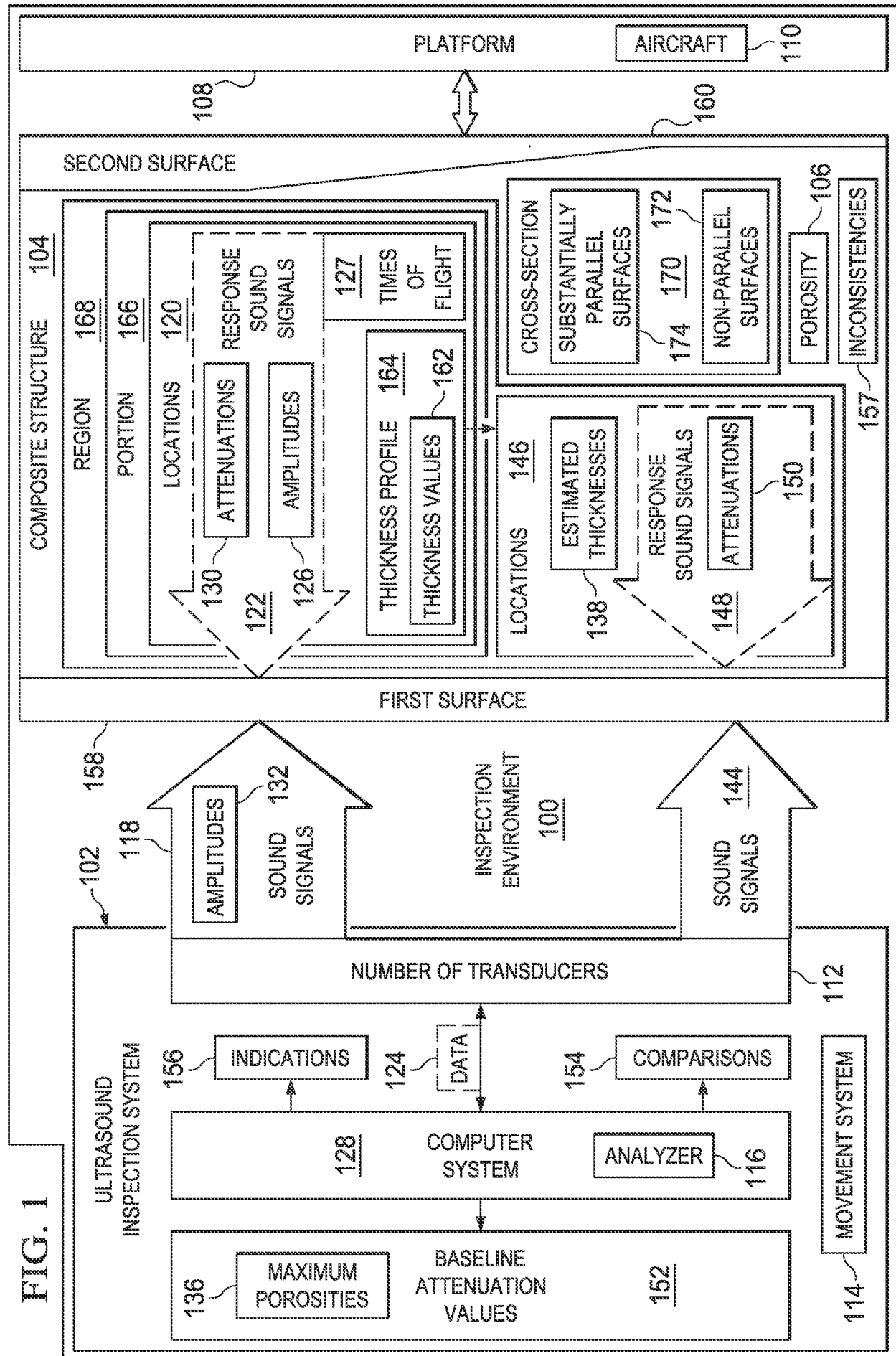
10 an analyzer (116) in communication with the number of transducers (112), wherein the analyzer (116) is configured to identify a thickness profile (164) of a portion (166) of a region (168) of the composite structure (104), the region (168) having a cross-section with non-parallel surfaces (172); identify an estimated thickness (138) for a location within the region (168) and outside of the portion (166) using the thickness profile (164); and generate an indication of whether the location has undesirable
15 porosity (106) using the estimated thickness (138) for the location and information about attenuation (150) of the response sound signals (148) detected by the number of transducers (112).

10. The ultrasound inspection system (102) of claim 9, wherein identifying the estimated thickness (138) for the location within the region (168) and outside of the portion (166) using the thickness profile (164) comprises applying the thickness profile (164) to the region (168) in a direction perpendicular to a
20 direction of thickness variation in the region (168).

11. The ultrasound inspection system (102) of claim 9 or 10, wherein identifying the thickness profile (164) comprises identifying thickness values (162) for each location along a line, the line extending
25 parallel to a direction of thickness variation in the region (168).

12. The ultrasound inspection system (102) of claim 11, wherein identifying the thickness values (162) for the each location along the line comprises identifying the thickness values (162) using
ultrasound inspection of the portion (166).

30 13. The ultrasound inspection system (102) of claim 11, wherein identifying the thickness values (162) for the each location along the line comprises identifying the thickness values (162) from a model (210).



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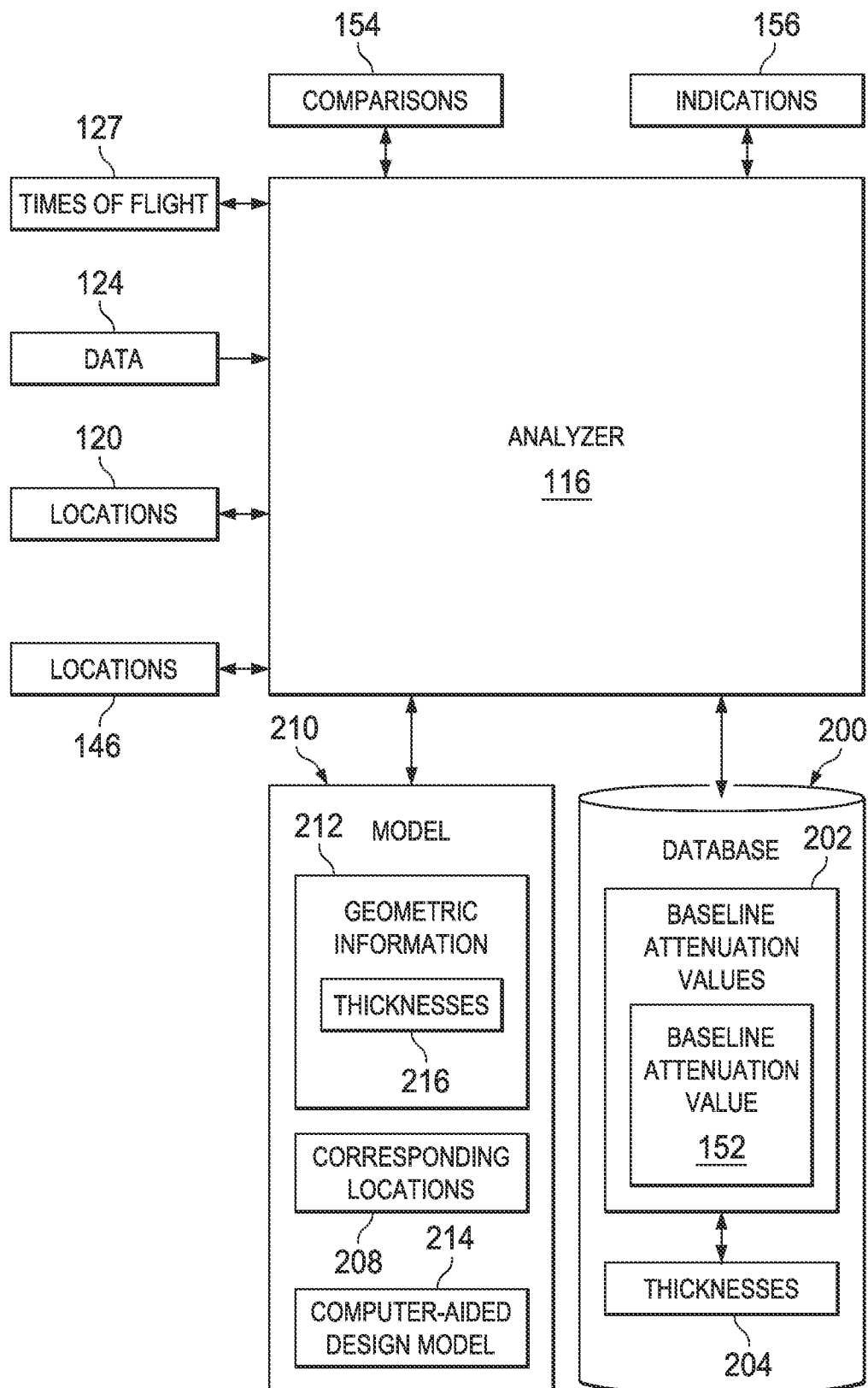


FIG. 2

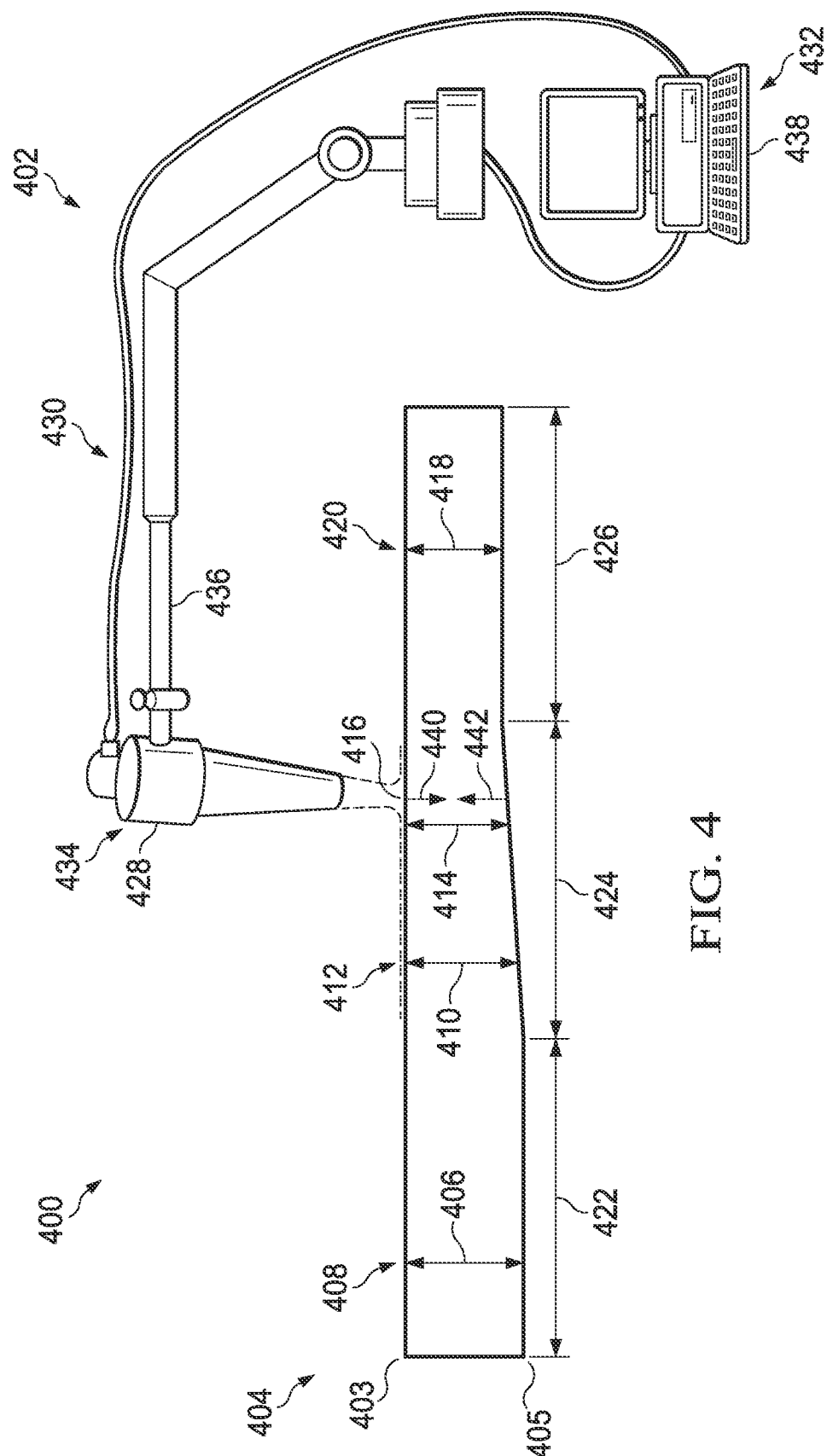
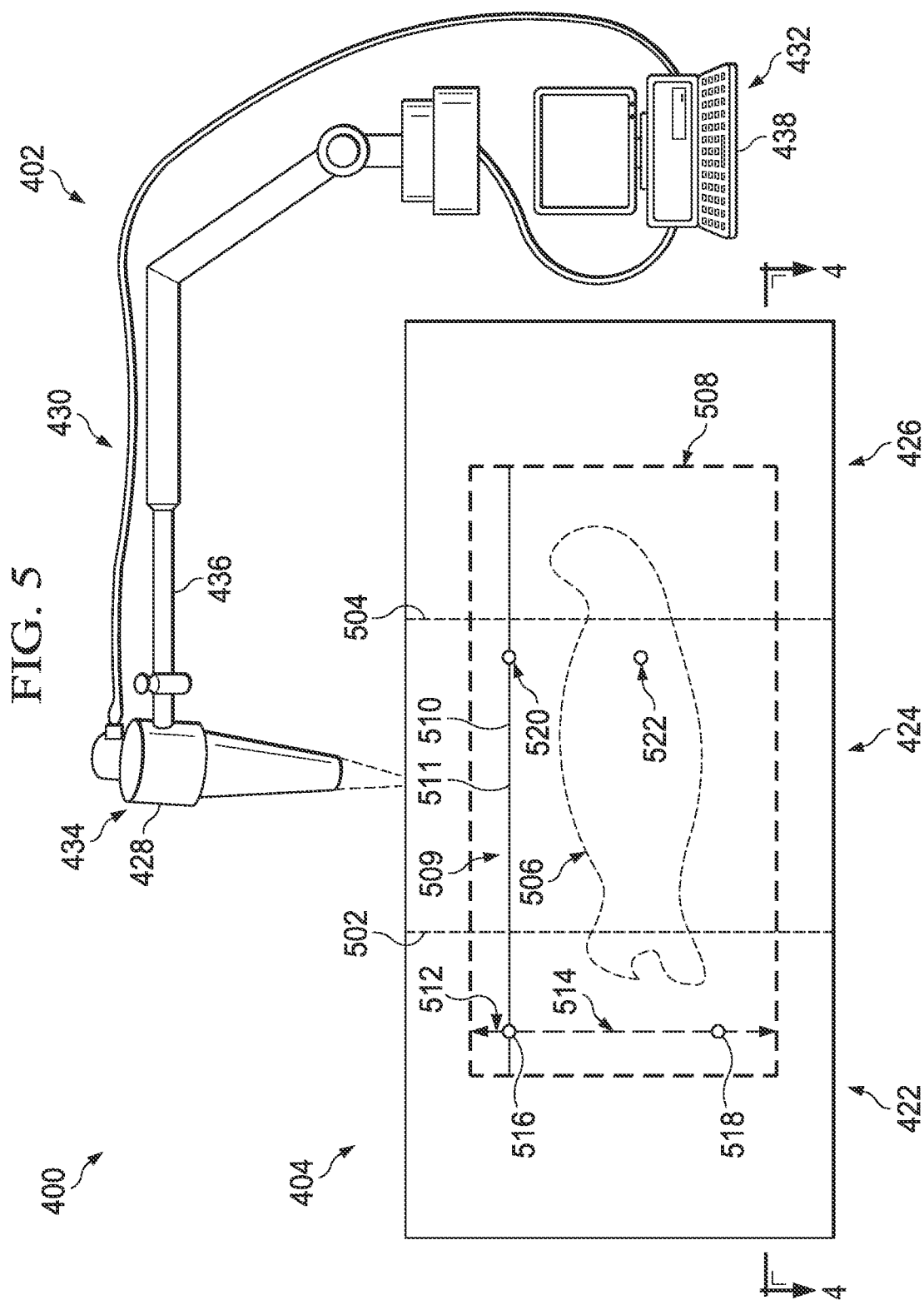


FIG. 4



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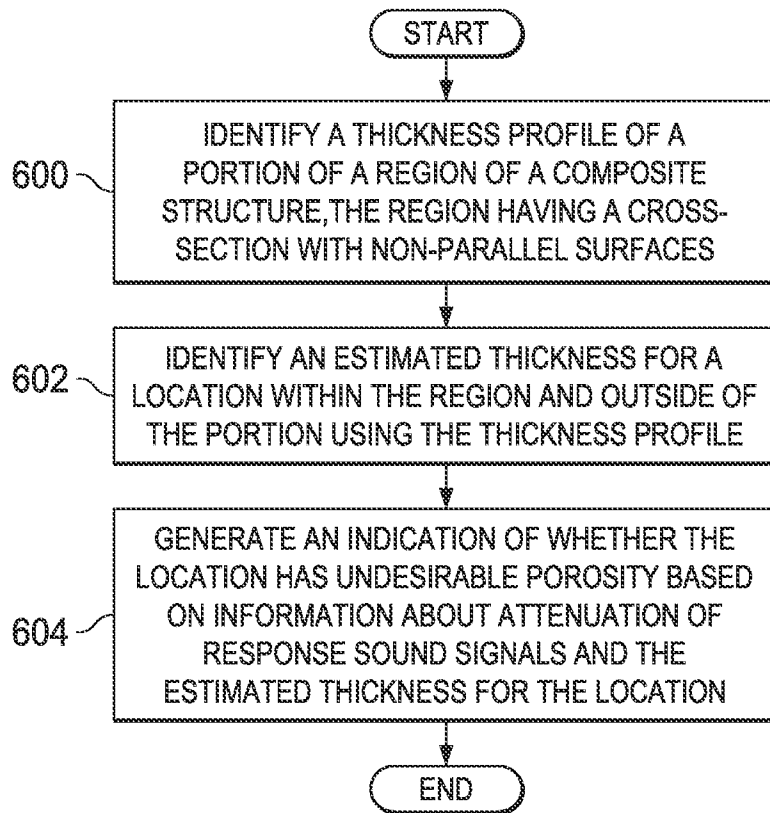


FIG. 6

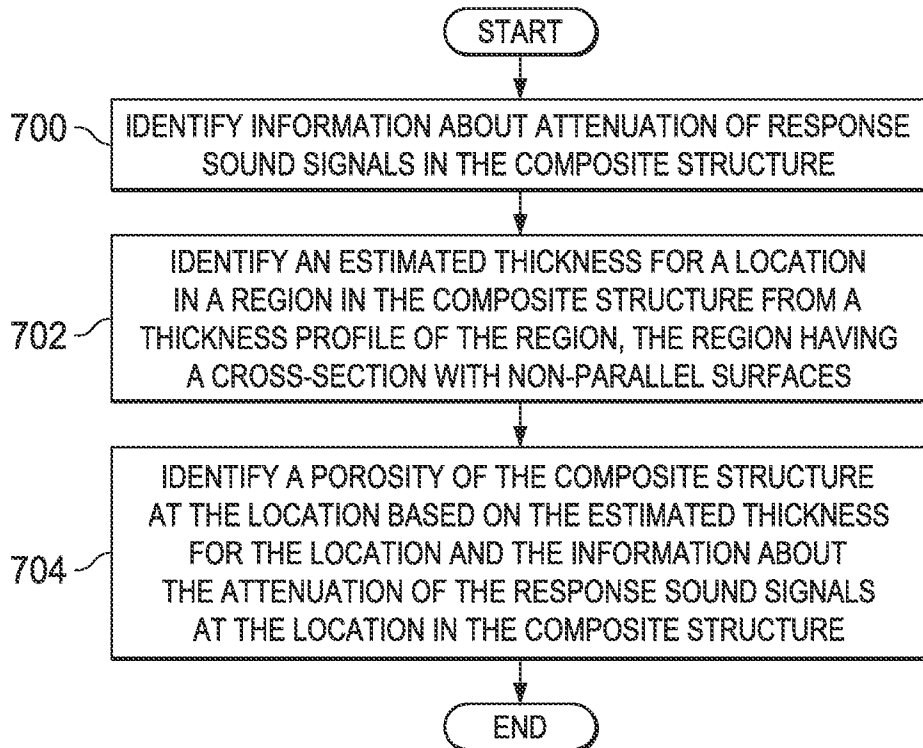
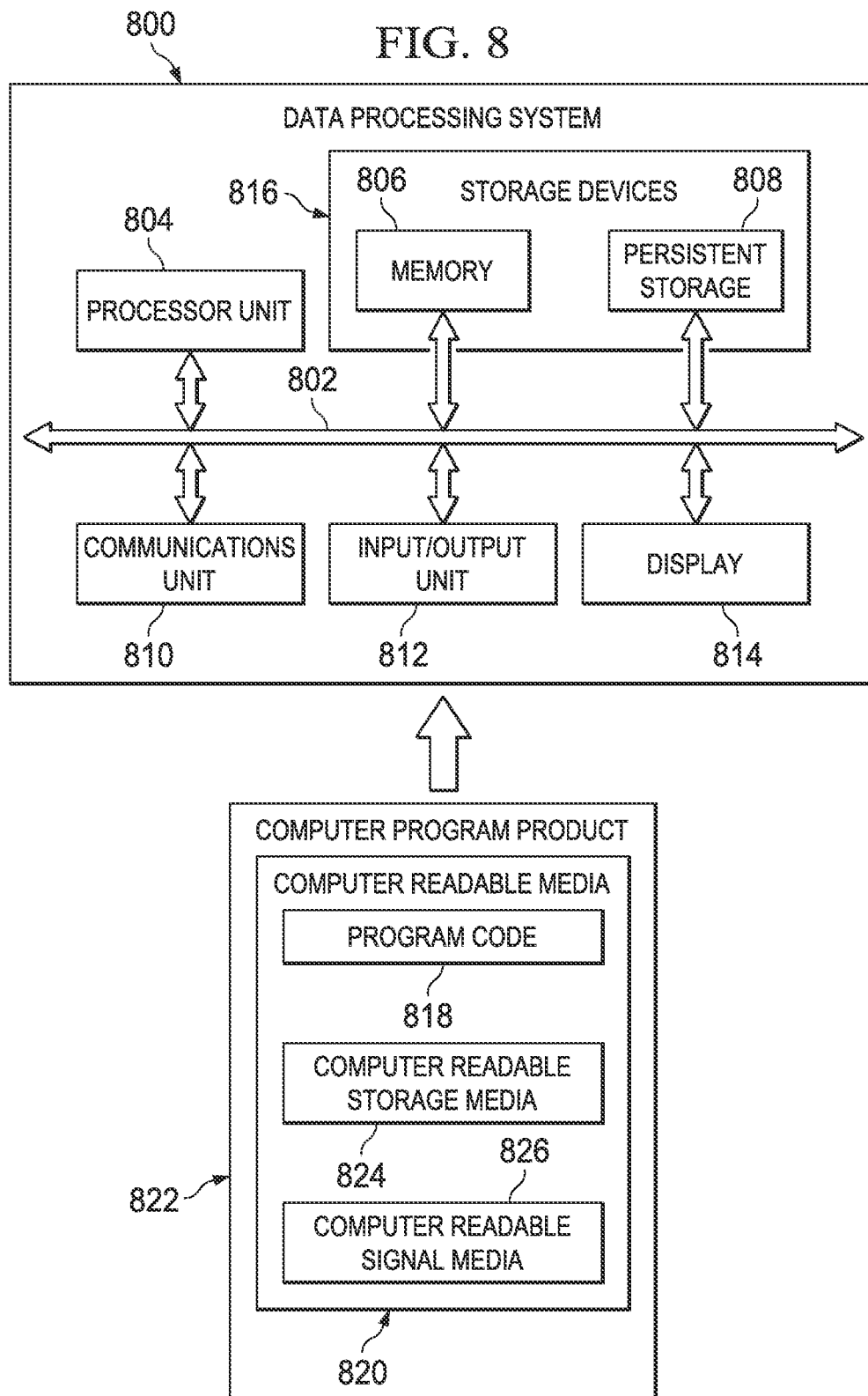
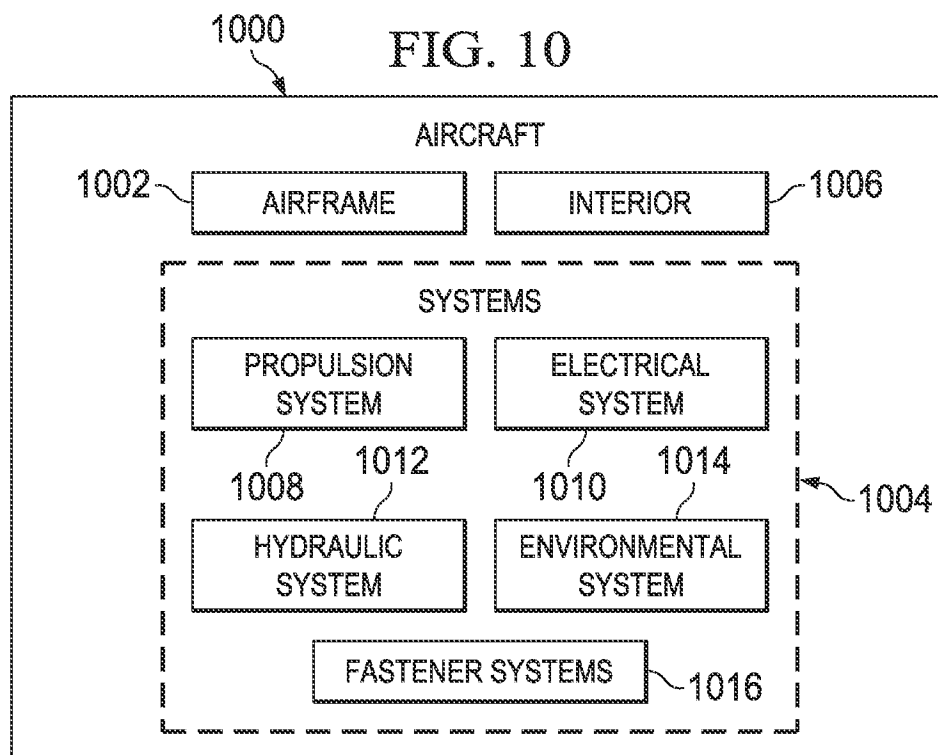
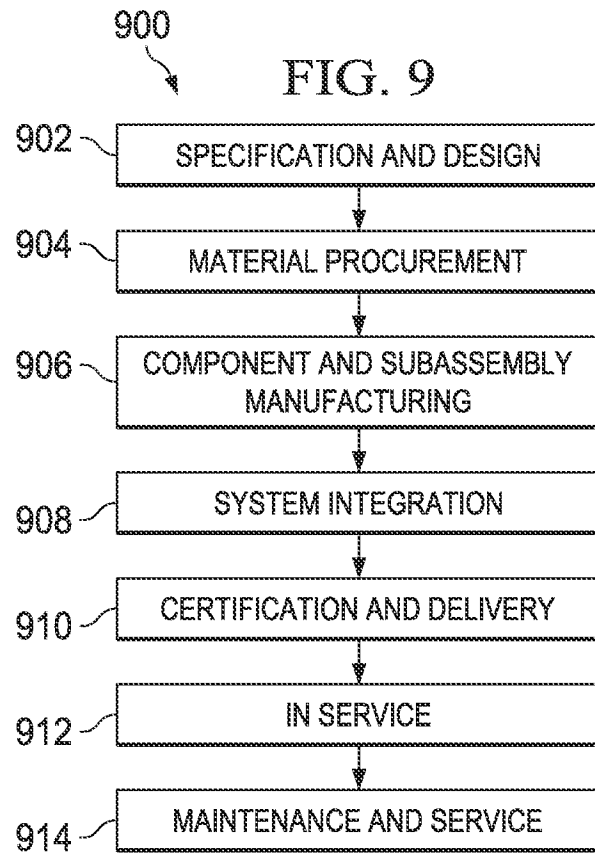


FIG. 7

FIG. 8



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INTERNATIONAL SEARCH REPORT

International application No
PCT/US2014/034637

A. CLASSIFICATION OF SUBJECT MATTER
INV. G01N29/04 G01N29/11 G01N15/08
ADD.

According to International Patent Classification (IPC) or to both national classification and IPC

B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)
G01N

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

Electronic data base consulted during the international search (name of data base and, where practicable, search terms used)

EP0-Internal

C. DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
X	EP 2 487 487 A1 (KAWASAKI HEAVY IND LTD [JP]) 15 August 2012 (2012-08-15) abstract paragraph [0012] paragraph [0057] - paragraph [0063] figures 1, 2, 12 ----- -/--	1-13

☒ Further documents are listed in the continuation of Box C.

☒ See patent family annex.

* Special categories of cited documents :

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Date of the actual completion of the international search

24 July 2014

Date of mailing of the international search report

04/08/2014

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Authorized officer

Rouault, Patrick

INTERNATIONAL SEARCH REPORT

International application No
PCT/US2014/034637

C(Continuation). DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
A	HSU D K ET AL: "SIMULTANEOUS ULTRASONIC VELOCITY AND SAMPLE THICKNESS MEASUREMENT AND APPLICATION IN COMPOSITES", THE JOURNAL OF THE ACOUSTICAL SOCIETY OF AMERICA, AMERICAN INSTITUTE OF PHYSICS FOR THE ACOUSTICAL SOCIETY OF AMERICA, NEW YORK, NY, US, vol. 92, no. 2 PT 01, 1 August 1992 (1992-08-01), pages 669-675, XP000297950, ISSN: 0001-4966, DOI: 10.1121/1.405279 abstract Part "B. Application to composites containing porosity using immersion mode measurement"	1,5,9, 11,12
A	----- US 2007/006651 A1 (KRUGER SILVIO E [CA] ET AL) 11 January 2007 (2007-01-11) abstract paragraph [0019] - paragraph [0020] paragraph [0024] paragraph [0060] -----	1,9

INTERNATIONAL SEARCH REPORT

Information on patent family members

International application No

PCT/US2014/034637

Patent document cited in search report	Publication date	Patent family member(s)	Publication date
EP 2487487	A1	15-08-2012	EP 2487487 A1 15-08-2012
			JP 5392731 B2 22-01-2014
			US 2012250970 A1 04-10-2012
			WO 2011043050 A1 14-04-2011

US 2007006651	A1	11-01-2007	NONE
