



(12) **United States Patent**  
**Kreisel**

(10) **Patent No.:** **US 11,008,121 B2**  
(45) **Date of Patent:** **May 18, 2021**

(54) **SPACE BODY**

(71) Applicant: **Jörg Kreisel**, Remscheid (DE)

(72) Inventor: **Jörg Kreisel**, Remscheid (DE)

(73) Assignee: **iBOSS GmbH**, Aachen (DE)

(\* ) Notice: Subject to any disclaimer, the term of this patent is extended or adjusted under 35 U.S.C. 154(b) by 368 days.

(21) Appl. No.: **16/300,982**

(22) PCT Filed: **May 12, 2017**

(86) PCT No.: **PCT/DE2017/100401**

§ 371 (c)(1),

(2) Date: **Nov. 13, 2018**

(87) PCT Pub. No.: **WO2017/194058**

PCT Pub. Date: **Nov. 16, 2017**

(65) **Prior Publication Data**

US 2019/0161213 A1 May 30, 2019

(30) **Foreign Application Priority Data**

May 13, 2016 (DE) ..... 10 2016 108 951.6

(51) **Int. Cl.**

**B64G 1/12** (2006.01)

**B64G 1/64** (2006.01)

**B64G 1/10** (2006.01)

(52) **U.S. Cl.**

CPC ..... **B64G 1/12** (2013.01); **B64G 1/10** (2013.01); **B64G 1/646** (2013.01); **B64G 2001/1092** (2013.01)

(58) **Field of Classification Search**

CPC . B64G 1/10; B64G 1/12; B64G 1/646; B64G 2001/1092; B64G 1/64; B64G 1/402; B64G 1/40; B64G 1/66; B64G 1/00; E04H 1/005; E04H 1/04; E04H 1/12; E04H 2001/1283

See application file for complete search history.

(56) **References Cited**

U.S. PATENT DOCUMENTS

3,295,789 A \* 1/1967 Hill ..... B64G 1/12  
244/159.3  
3,300,162 A \* 1/1967 Maynard ..... B64G 1/12  
244/159.4  
3,332,640 A \* 7/1967 Nesheim ..... B64G 1/12  
244/159.5  
3,420,470 A \* 1/1969 Meyer ..... B64G 1/641  
244/173.3

(Continued)

FOREIGN PATENT DOCUMENTS

DE 102015002367 A1 9/2015  
DE 102014104695 A1 10/2015

(Continued)

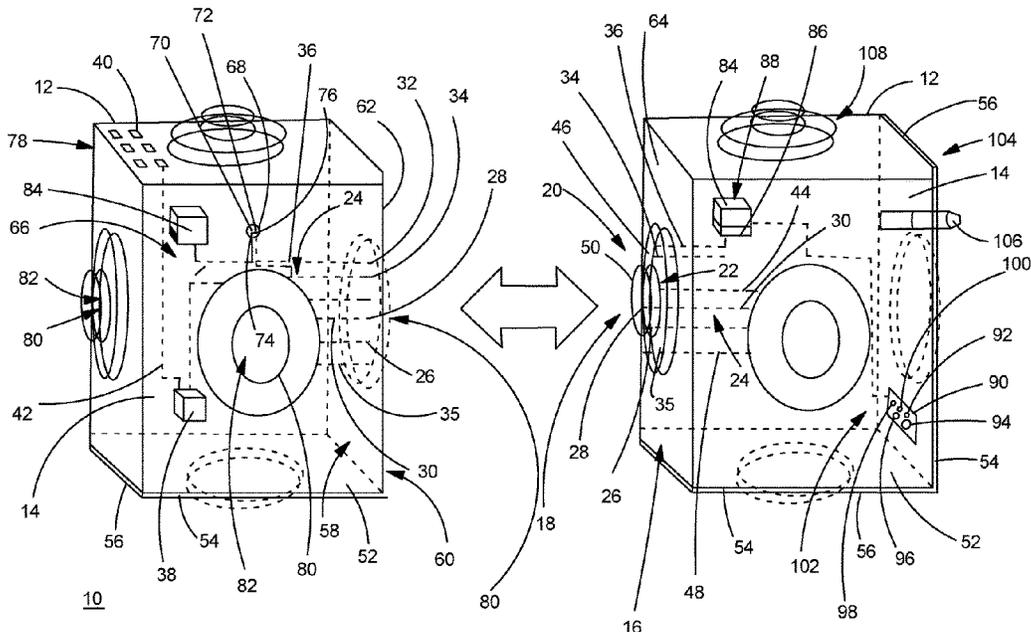
Primary Examiner — Medhat Badawi

(74) Attorney, Agent, or Firm — Smartpat PLC

(57) **ABSTRACT**

A space body has at least two space modules which can be interchangeably connected to one another. At least one closable connection opening is in each case provided on the space modules, via which a passage is produced when the space modules are connected. Furthermore, at least one coupling device connects in each case the space modules to one another. A supply interface is provided for coupling a supply line of the space modules.

**23 Claims, 2 Drawing Sheets**



(56)

References Cited

U.S. PATENT DOCUMENTS

3,478,986 A \* 11/1969 Fogarty ..... B64G 1/646  
244/159.4  
3,566,554 A \* 3/1971 Schaffer et al. .... E04B 1/34815  
52/64  
3,707,813 A \* 1/1973 Cymbrowitz ..... E04B 1/34807  
52/79.11  
3,709,447 A \* 1/1973 Devlin ..... B64G 1/12  
244/159.4  
RE27,903 E \* 1/1974 Fogarty ..... B64G 1/64  
244/172.5  
3,791,080 A \* 2/1974 Sjoberg ..... E04B 1/3412  
52/79.4  
3,792,558 A \* 2/1974 Berce ..... E04B 1/348  
52/79.7  
3,822,569 A \* 7/1974 Lautrup-Larsen .... A63H 33/04  
446/85  
3,955,328 A \* 5/1976 Lindsay ..... E04B 1/34846  
52/73  
4,057,207 A \* 11/1977 Hogan ..... B64G 1/60  
244/159.4  
4,079,904 A \* 3/1978 Groskopf ..... B64G 1/14  
244/172.5  
4,273,305 A \* 6/1981 Hinds ..... B64G 1/14  
244/172.5  
4,384,692 A \* 5/1983 Preukschat ..... B64G 1/1007  
136/292  
4,508,404 A \* 4/1985 Frawley ..... H01R 13/629  
244/135 A  
4,546,583 A \* 10/1985 Hussar ..... E04B 1/04  
52/236.1  
4,715,566 A \* 12/1987 Nobles ..... B64G 1/12  
244/159.4  
4,728,060 A \* 3/1988 Cohen ..... B64G 9/00  
244/159.4  
4,744,533 A \* 5/1988 Mullen ..... B64G 1/12  
244/159.4  
4,771,971 A \* 9/1988 Ludwig ..... B64G 1/641  
165/104.33  
4,834,325 A \* 5/1989 Faget ..... B64G 1/641  
244/159.4  
4,872,625 A \* 10/1989 Filley ..... B64G 1/12  
244/159.4  
4,878,637 A \* 11/1989 Mullen ..... B64G 1/12  
244/159.4  
4,880,187 A \* 11/1989 Rourke ..... B64G 1/1078  
244/159.4  
4,903,919 A \* 2/1990 Johnson ..... B64G 1/646  
244/172.4  
5,052,640 A \* 10/1991 Chang ..... B64G 1/222  
244/172.7  
5,094,170 A \* 3/1992 Raynaud ..... F42B 12/58  
102/489  
5,145,130 A \* 9/1992 Purves ..... B25J 5/00  
244/159.4  
5,152,482 A \* 10/1992 Perkins ..... B64G 1/22  
244/159.4  
5,199,672 A \* 4/1993 King ..... B64G 1/007  
244/164  
5,271,582 A \* 12/1993 Perkins ..... B64G 1/22  
244/159.4

5,566,909 A \* 10/1996 Lapins ..... B64G 1/105  
244/173.3  
5,791,600 A \* 8/1998 Thompson ..... B64G 1/12  
244/120  
5,806,799 A \* 9/1998 Lounge ..... B64G 1/12  
244/159.4  
6,138,951 A \* 10/2000 Budris ..... B64G 1/002  
102/393  
6,299,107 B1 \* 10/2001 Kong ..... B64G 1/646  
244/172.4  
6,354,457 B1 \* 3/2002 Aaron ..... F17C 1/00  
220/581  
6,536,712 B1 \* 3/2003 Barenett ..... B64G 1/10  
244/158.3  
6,669,148 B2 \* 12/2003 Anderman ..... B64G 1/007  
244/172.4  
6,789,767 B2 \* 9/2004 Mueller ..... B64G 1/007  
244/173.3  
7,988,096 B2 \* 8/2011 Humphries ..... B64G 1/1078  
244/158.1  
8,006,937 B1 \* 8/2011 Romano ..... B64G 1/646  
244/172.4  
8,047,473 B2 \* 11/2011 Johnson ..... B64G 9/00  
244/159.4  
8,763,326 B2 \* 7/2014 Takeshima ..... E04H 9/028  
52/236.1  
8,915,472 B2 \* 12/2014 Aston ..... B64G 1/405  
244/171.1  
D788,016 S \* 5/2017 Blincow ..... D12/320  
10,407,190 B2 \* 9/2019 Fernandez ..... B64G 1/1085  
2002/0035419 A1 \* 3/2002 Lin ..... B64G 1/244  
701/27  
2011/0210750 A1 \* 9/2011 Medelius ..... H01B 1/24  
324/543  
2011/0320429 A1 \* 12/2011 Doig ..... G06F 16/972  
707/711  
2011/0321003 A1 \* 12/2011 Doig ..... G06F 16/904  
717/107  
2012/0041634 A1 \* 2/2012 Madhavanpillai ..... G01P 5/16  
701/30.1  
2012/0054143 A1 \* 3/2012 Doig ..... G06Q 30/0269  
706/47  
2013/0334824 A1 \* 12/2013 Freda ..... F03D 1/025  
290/55  
2015/0083865 A1 \* 3/2015 Nakasone ..... B64G 1/428  
244/158.6  
2015/0375875 A1 \* 12/2015 Dula ..... B64G 1/403  
244/171.3  
2016/0053941 A1 \* 2/2016 Rebernik ..... F17C 3/02  
206/583  
2016/0130019 A1 \* 5/2016 Jaeger ..... B64G 1/402  
62/7  
2017/0015443 A1 \* 1/2017 Lakshmanan ..... B64G 1/64  
2017/0210494 A1 \* 7/2017 Blackwell ..... B64G 1/58  
2017/0228616 A1 \* 8/2017 Tasdizen ..... G06K 9/00664  
2018/0186476 A1 \* 7/2018 Poncet ..... B64G 1/646

FOREIGN PATENT DOCUMENTS

EP 0196793 A1 10/1986  
WO 9200223 A1 1/1992

\* cited by examiner

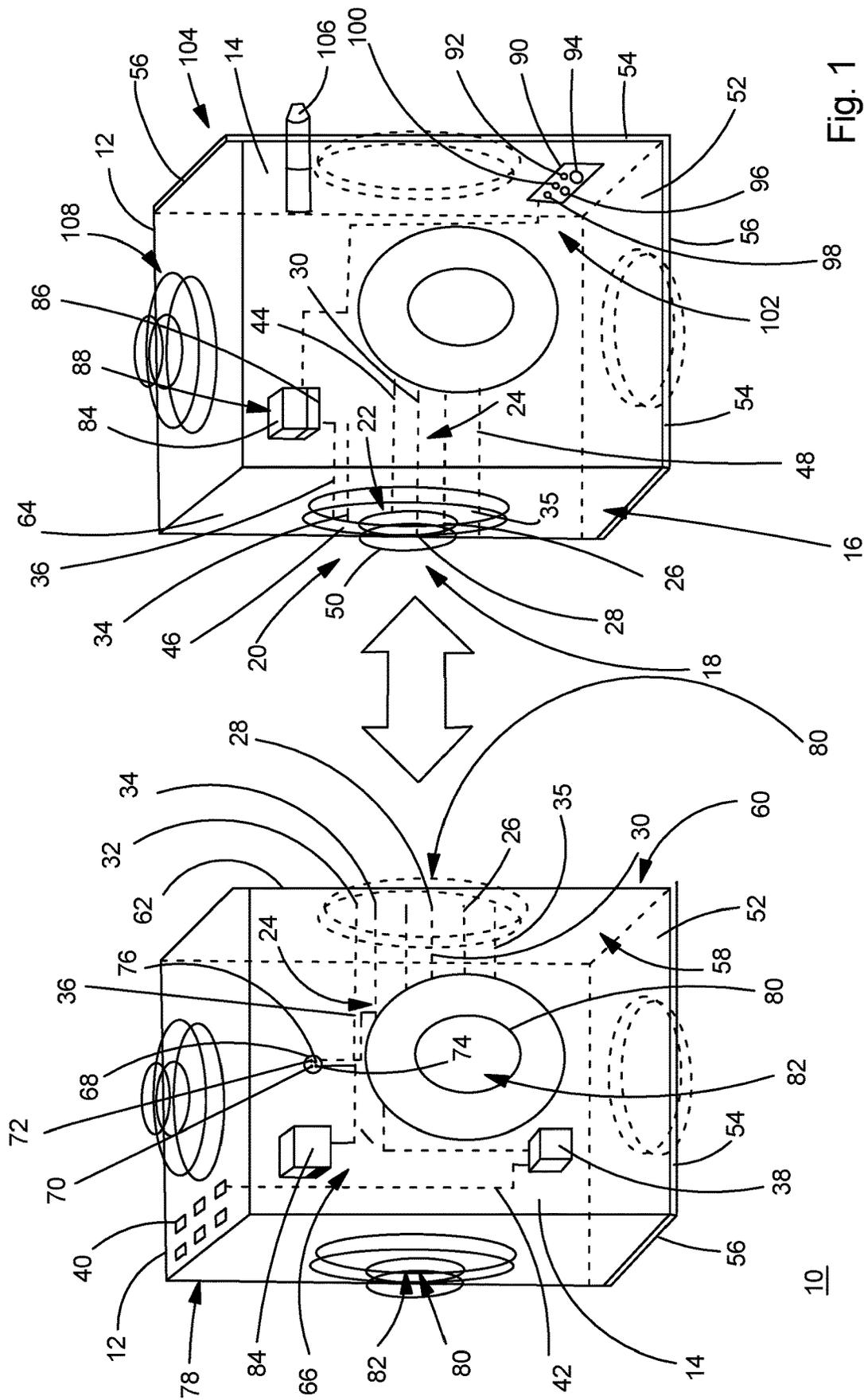
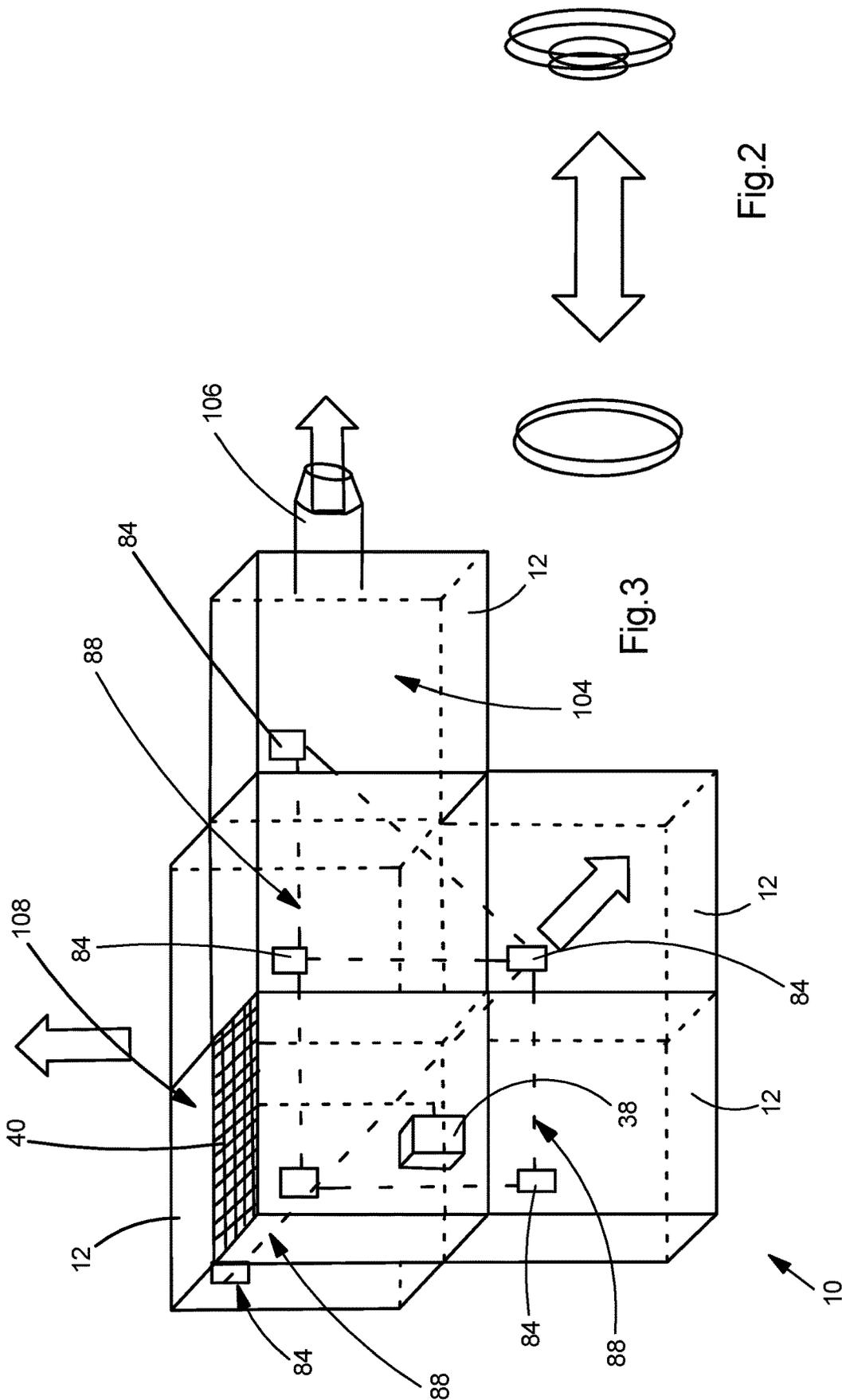


Fig. 1

10



## 1

## SPACE BODY

## TECHNICAL FIELD

The invention relates to a space body with at least two space modules, which are interchangeable connectable.

## BACKGROUND

From DE 10 2014 104 695 A1 a module with at least one coupling device is known for setting up a modular spacecraft. Furthermore, an androgynous coupling device for connecting modules is described. The coupling device is used to build up a modular constructed spacecraft, wherein the coupling device can be coupled to an identically constructed further androgynous coupling device for connecting the modules and comprises a plurality of coupling elements for this purpose. The interior of a sleeve-like linear guide element forms an expansion opening for interfaces providing network connection, data connection, hydraulic or pneumatic connection to the involved modules with each other. The sleeve-like linear guide element is surrounded by likewise sleeve-shaped nested coupling elements.

EP 0 196 793 A1 shows a modular satellite which modules comprise coupling devices for connecting the individual modules to one another. Each of these androgynous coupling modules is configured for coupling and connecting to an identically constructed coupling device and comprises a plurality of coupling elements for this purpose. With such an androgynous coupling device (androgynous interface) for orbitally maintainable satellite systems the reconfiguration of a modular satellite system composed of modules by robotic manipulation is enabled.

From WO 92/00223 an unmanned spacecraft is known. The spacecraft is configured modularly. Various subsidiary payload bodies are docked to a central mother body. Standardized interfaces are provided for this purpose, which enable the mother body to exchange for example control, data and electric energy with the subsidiary payload bodies. The subsidiary payload bodies are not self-sufficient systems and thus mostly dependent on the mother body. The subsidiary payload bodies serve only to provide further individual functions to the mother body.

The known space modules have the disadvantage that they may couple with each other, but not in any order. For example, the space modules can be coupled to a central satellite, but they cannot be assembled arbitrary in the known systems. This requires a precise planning especially concerning the control of the space modules. For this purpose, a separate central control module is required regularly for the entire space body, which takes over and coordinates the required control tasks of the individual space modules. The computer capacity of the controller is associated with the number of coupled space modules. The larger the space body is the more extensive is the control. If necessary, redundant systems are used in order to compensate controller failures. This is a complex system, which considerably increases the cost, for example for space flight.

## SUMMARY

Thus, an object of the invention is to provide a space body with different space modules, which is expandable as desired and has a high failure safety. The object is achieved in a space body with at least two space modules, which are interchangeable connectable, comprising

## 2

- a) at least one closable connection opening on each of the space modules, via which a passage is provided when connecting the space modules,
- b) at least one coupling device respectively, which connects the space modules,
- c) at least one supply interface respectively, for coupling a supply line of the space modules, and
- d) resources for supplying the space body which are distributed decentralized in each of the space modules, wherein a resource exchange between the space modules and/or a resource consumption is provided via supply lines.

Such space bodies are mainly used in aerospace, for example with satellites. In doing so, a space module can be coupled to another space module. When coupling the space modules, connection interfaces, which each space module must have, are connected to each other. Via the connection interfaces the supply lines of the space modules are connected to each other, monitored and controlled.

The invention is based on the principle of creating an intelligent space body, especially for extreme conditions as they prevail in aerospace or underwater, which is arbitrarily expandable. The more space modules are coupled, the more complex is their control. In the present invention each space module brings along its own resources, so that the supply of the space body can be configured decentralized. In doing so, a space module can be decoupled without interrupting the supply of a necessary resource. Furthermore, own redundant systems are no longer required, because in the event of a resource failure in the system the networking allows to replace these resources by one or more other space modules. Central resources are more of an obstacle as they always have to be dimensioned accordingly when expanding the space body by further space modules. In the decentralized arrangement the controller is interconnected during coupling via an appropriate network structure of suitable supply lines for the resources. The network structure requires standardized connection interfaces.

As an advantageous embodiment of the space body according to the invention with at least two space modules has been found when a processor-controlled controller for controlling the space modules is provided, wherein the controller as a resource for supplying the space body is provided decentralized in each of the space modules and is networked together via the supply interface for coordinated control of the space body. A processor-controlled controller is provided in each space module, which is for example formed by a suitable computer unit. The processor-controlled controller of a module is always able to work independently in a space module. But they are interconnected in such a manner in the compound of the space body that they provide a combined computing or controlling capacity.

Preferably the controller is then configured as a neural network structure. A neural network structure is able to learn. Thus, learned tasks can be carried out independently by the space body. Such a network structure for a space body makes it flexible and highly adaptable to extreme conditions.

An advantageous embodiment of the space body according to the invention is that gas resources are provided decentralized in the space modules respectively for supplying the space body. The gas resources are present as resource in each space module, i.e. decentralized in the space body. The gas resources can be moved via suitable supply lines between the space modules and be led to consumers if necessary.

Analogous to the gas, in a preferred embodiment of the space body according to the invention with at least two space modules liquid resources are provided decentralized in the space modules respectively for supplying the space body. The liquid resources are also present as a resource in each space module, i.e. decentralized in the space body. The liquid resources can be moved between the space modules and be passed to consumers if necessary via suitable supply lines.

A further preferred variant of the inventive space body is that thermal resources are provided decentralized in the space modules respectively for supplying the space body either for cooling or for heating. Often a thermal exchange must be possible between the space modules. For example, heat and/or cold storages are provided in the space modules for this purpose. The thermal energy is shifted between the space modules by means of a suitable energy carrier via supply lines as needed.

A further advantageous embodiment of the inventive space body consists in that resources for electric energy are provided decentralized in the space modules respectively, for supplying the space body. Both electric storages, such as batteries and accumulators or electric generator, such as photovoltaic modules or fuel cells are arranged as a resource decentralized in the space body. The electric resources are present in each space module and therefore decentralized in the space body. The electric resources can be moved via suitable electric lines between the space modules and be passed to consumers if necessary.

In a further preferred embodiment of the inventive space body with at least two space modules the supply interface comprises a standardized dockable multiple connection which is provided simultaneously for different supply lines. Often there is the problem that the space modules cannot be coupled compatible with each other. Thus, it is particularly advantageous in the invention if a standardized multiple connection is provided, which implements as far as possible all connections. There is no need to create separate connections for the individual space modules. This is all combined in the multiple connection for the different supply lines. Preferably, the multiple connection also comprises a gas connection and/or a connection for liquids, a connection for communication and/or a connection for electric energy. Especially with the networked control structure it is desirable to standardize the connection. This is achieved much easier with standardized multiple connections. An optional connection for thermal energy provides an optimal distribution of heat between the space modules on demand.

A further advantageous embodiment of the inventive space body with at least two space modules consists in that the multiple connection comprises a connection device for pressure lines. Often also pressure lines must be coupled between the space modules. For example, the pressure lines are used for pneumatic or hydraulic pressure. The multiple connection therefore preferably has a connection device which withstands the required pressure of liquids or gas. For example, hydraulic or pneumatic machines can be driven via the pressure lines. This makes it easy to drive robots or their moving parts.

A particular variant of the space body according to the invention with at least two space modules is that the supply interface is configured for mechanically and/or magnetically and/or electromagnetically coupling. This measure serves for establishing a coupling of the interfaces between two space modules as quickly as possible. This is easily solved

by a mechanical, magnetic or electromagnetic coupling mechanism. Combinations of these mechanisms are conceivable.

Furthermore, an advantageous embodiment of the inventive space body with at least two space modules consists in that that the supply interface is configured centrally lockable. This prevents the supply interface from releasing again by itself after a coupling process. Only when a locking mechanism is released again, the connection of the supply interface can be detached. The locking is preferably carried out centrally, so that all desired interfaces of a module are detached simultaneously.

A preferred variant of the inventive space body results from the fact that an energy storage and/or a solar energy supply is provided for energy supply. The energy supply of the space body is preferably ensured thereby, that an energy storage is provided, such as an accumulator. The energy storage can be charged via solar cells. Combustion processes are not given in solar cells. Solar energy is practically unlimited. Thus, the energy for a space body according to the invention is provided advantageously.

A particular embodiment of the inventive space body is achieved in that the space body is configured as a pressure body. This measure serves to ensure that the space body can withstand greater pressure differences. Greater pressure differences occur both in space and underwater. In space, the internal pressure is relatively high, while underwater, the external pressure is quite large.

In an advantageous embodiment of the space body according to the invention with at least two space modules the space modules have a stackable geometry. The advantage of the stackable geometry is that any number of space modules can be connected and stacked like containers. Thus, the space body can be extended to any size, as far as the static stability is given. In this case the space modules can comprise various components that are required for each desired requirement. Preferably, the space modules of the space body have a cubic geometry. Cubic space modules can be stacked particularly well in all possible directions.

For the stackability it is necessary, that not only one coupling device or one supply interface is provided. With one coupling devices per space module only a single space module could be docked to another, which then forms the space body. Further docking of space modules is not possible. Therefore, in another special embodiment of the inventive space body the supply interface is provided on at least two sides of a space module respectively. As a result always more than two space modules can be combined to form a space body.

Furthermore, a particular variant of the space body according to the invention can be achieved in that an inner and/or outer insulation is provided on the space modules. This advantageous measure serves to ensure that the space body is insulated against both cold and heat. The space body should be configured for extreme conditions. This includes thermal requirements. A corresponding insulation helps against this. A special insulation against radiation can also be provided as protection. The insulation can be arranged not only on the outer surfaces, but also on the inner surfaces of the space body or the space modules.

The functions and the status of the space modules must be constantly monitored in case of extreme use of the space body. A special embodiment of the inventive space body with at least two space modules results in that a processor-controlled monitoring device is provided for the space modules. The monitoring device monitors the state of the space body and signals malfunction or the like if necessary.

A further useful embodiment of the inventive space body results from the fact that the monitoring device comprises alternatively or in combination an optical sensor, an acceleration sensor, a motion sensor, a temperature sensor or a position sensor. With these sensors the monitoring device is able to detect and process a part of the environment. The evaluation of the measured signals of the sensors helps to determine the state of the space body.

Furthermore, an advantageous embodiment of the space body according to the invention can be achieved if at least one space module comprises a drive. The drive enables the space body to move. The type of drive and movement is of course dependent on the environment. The underwater drive will usually be a different drive than that used in the air or on land. An appropriate embodiment of the space body according to the invention the space module with a drive comprises a drive control. The drive control controls the drive, for example in acceleration and direction.

In a particularly appropriate variant of the inventive space body the space body is provided for aerospace and/or for use underwater or in live unfriendly environment.

Preferably, in a particular embodiment of the space body according to the invention, an internal supply access is provided within a space module of the space body. Frequently requirements to the space body are that various resources can be accessed inside, such as compressed air or water. A corresponding supply access is therefore provided in at least one space module of the space body. An advantageous aspect of the invention results from the fact that the internal supply access of the space body comprises a connection for gas, a connection for liquid, a connection for communication and/or a connection for thermal energy and/or a connection for electric energy. These supply accesses are particularly important when experiments are performed in such a space body. The experimenters must be able to access the various resources that are available.

Further embodiments and advantages will become apparent from the subject of the dependent claims, as well as the drawings with the accompanying descriptions. An exemplary embodiment is explained below in more detail and with reference to the accompanying drawings. The invention should not be limited to the exemplary embodiment. Rather, embodiments are also considered, which now and in the future will appear to the skilled person in an equivalent manner with other technical aids.

#### BRIEF DESCRIPTION OF THE DRAWINGS

FIG. 1 shows schematically a space body according to the invention with two initially separated space modules.

FIG. 2 shows a supply interface for a space module according to the invention with at least two space modules.

FIG. 3 shows a space body with several stacked space modules.

#### DETAILED DESCRIPTION

In FIG. 1 an inventive space body is designated with 10. The space body consists of space modules 12, which can be assembled detachable and interchangeable. In the present embodiment the space modules 12 are configured as two cuboids 14. The cuboids 14 are particular easy to stack and assemble. When stacking always two end faces 16 can be assembled on each other. Other shapes, such as octahedron or other stackable polyhedron may also be used as space module 12. Each space module 12 comprises a coupling

device 18 on at least one end face 16. A supply interface 20 is provided at the coupling device 18.

In the present embodiment the supply interface 20 consists of a multiple connection 22, which connects different supply lines 24. With the multiple connection 22, the supply lines 24 of the space modules 12 can be connected to each other in a standardized way. In particular, the multiple connection 22 comprises a connection 26 for gas, through which, for example, oxygen, breathing air or exhaust air can be conducted. In addition, the multiple connection 22 comprises a connection 28 for liquids. For example, water is pumped through liquid lines 30 via such a liquid connection 28. A heated liquid can be conducted via the liquid lines 30 for heating one or all space modules 12. Furthermore, the multiple connection 22 comprises a communication connection 32, an electrical connection 34 for electric energy and a connection 35 for thermal energy. Via the connection 35 for thermal energy the heat balance of the entire system is regulated and heat energy is distributed and compensated.

Data lines 36 can be coupled together by the communication connection 32. The data lines 36 are used to exchange digital data between the space modules 12. In the present embodiment the data lines 36 are configured as cable. In principle, the data exchange between the space modules 12 can also be realized via electromagnetic waves, such as radio waves or light waves. For this purpose, suitable transmitting and receiving units have to be provided.

The electric connection 34 is provided for the energy exchange of electrical energy between the space modules 12. In this case, an energy generator 38 for electric energy can be provided on or in a space module 12. The energy generator 38 is, for example, an accumulator, which is fed by solar cells 40. The energy of the energy generator 38 is distributed in the space modules 12 via electric lines 42, usually cable.

Such space modules 12 are basically also suitable for robotics. Therefore pressure lines 44 are provided for liquids and gases with which, for example, robot parts can be driven pneumatically or hydraulically. But working gases with increased pressure can also be led through the pressure lines 44.

The coupling device 18 comprises magnetic coupling mechanism 46, whereby the supply interfaces 20 can be coupled in pairs. The magnetic coupling mechanism 46 has at least one electromagnet 48 respectively, for increasing the attraction force. The space modules 12 and the coupling device 18 attract each other via the magnetic coupling mechanism 46. After the coupling devices 18 and the supply interfaces 20 of the space modules 12 are connected to each other, a mechanical locking mechanism 50 engages for security. The electro magnet 48 of the magnetic coupling mechanism 46 can be turned off now if necessary, i.e. to save energy, for example.

The space body 10 or its space modules 12, of which the space body 10 is made, are configured as pressure bodies 52. The pressure bodies 52 are particularly required for use, for example, under water. The space modules must withstand extremely high pressures under water, depending on the depth. On the other hand, the pressure difference of the space modules 12 to the vacuum, which prevails in space, has to be endured also. For extreme temperatures, both in the heat range, as well as in the cold range a suitable thermal insulation 54 is provided. The insulation 54 comprises a protection layer 56, which protects the space body 10 against radiation. The insulation 54 may be provided on the inside 58 of the space modules 12 or on the outside 60 of the space modules 12.

The space modules **12** of the space body **10** have a cubic geometry, because such geometric bodies can be stacked very well. Thereby, a different space module **12** can be coupled to outer surface. Basically, other geometries of space modules are also conceivable, but they must be well stackable. Stacking in this context means that at least one plan outside surface **62** of a space module **12** is parallel to the plan outside surface **64** of another space module **12** and can be joined together. The stackability allows expanding the space body **10** in size as desired. Each space module **12** can be assigned its own function. In doing so, any constellation of space bodies **10** can be created, as can be seen well in FIG. 3.

In the interior **66** of the space module **12** internal supply accesses **68** are provided. In this case, an internal connection **70** for gas, a connection for liquid **72**, a data connection **74** and a connection **76** for electric energy are provided. The supply accesses **68** also make it possible to be supplied with appropriate resources in the interior **66** of the space modules **12**. If a space module **78** is used, for example, for experiments, an experimenter can use the supply accesses **68**.

The space modules **12** have closable openings **80**, which serve as passages **82**. Due to the closable openings **80** there is a continuous spatial connection between the space modules **12**. On the one hand one can move through the openings **80** or on the other hand material or lines can be passed through. Of course the type of use depends on the size of the openings **80** and the passages **82**.

Each of the space modules **12** has its own resources, which it provides to the entire system of the space body **10** in the coupled state. The space modules **12** comprise as a resource in particular a processor-controlled controller **84** with memory **86** and program or function structures. The controller **84** controls the processes to be controlled in each space modules **12** if they are self-sufficient. Once the space modules **12** are coupled to each other to form a space body **10** the controller **84** of the individual space modules **12** are networked. The data lines **36** connect the controller **84** via the communication connections in a suitable manner. As a result, the computing capacity of the space body **10** is increased according to the number of space modules **12**. Instead of a central computer unit the space body **10** is now controlled by many decentralized controller **84** as a control network **88**. In case of a failure of a controller **84** of a space module **12**, the tasks and functions can be taken over by other controller **84**. The control network **88** is preferably configured as a neural network. Thus, the control network **88** is able to learn and independently adapt itself to different circumstances. This adaptability is particularly required when the space body **10** is unmanned on its way.

The control network **88** controls and monitors all operations in the space body **10**. Suitable sensors **90** are provided in order to detect the outside world by the control network **88**. Thus, the control network **88** is connected to at least one optical sensor **92**, such as a digital camera. In addition, at least one acceleration sensor **94** provides motion data to the control network **88**. A position sensor **96** continuously detects the position of the body **10** and provides appropriate data to the control network **88**. Motion sensors **98** monitor the motion within the space body **10**. The evaluation of the motion sequences is carried out via the control network **88**. Pressure sensors **100** detect the pressure within the space body **10**. The control network **88** can regulate the pressure in the space body **10** to a fixed setpoint. Together with the sensors **90** the control network **88** forms a monitoring device **102**, with which the space body **10** can be monitored in its functions.

In particularly, the coupling device **18**, the closeable openings **80** and the supply interfaces **20** are controlled via the control network **88**. The control network **88** coordinates all the work and the total functions of the space modules **12** with each other.

One space module **104** comprises a drive **106**. The drive control is taken over by the control network **88** in the present embodiment. In principle, a separate drive control for the drive **106** in the space module **104** is also possible. The drive **106** serves for moving the space body **10**. The drive can be configured in various ways, such as a jet drive or a rocket drive.

Other resources like gas resources, liquid resources, thermal and electric resources are included in each space module **12**. In the coupled state, the resources are available decentralized in the entire space body **10**. The resources are always available to the space body **10** as a whole. Through the supply lines **24** the resources can finally be moved through the space body, as it is required for the function and work respectively. The failure of a resource of a space module **12** is taken over by the other space modules **12** which have no malfunction. Thus, the space body **10** as a whole remains fully functional.

In FIG. 2 the supply interface **20** for the space module **10** according to the invention with at least two space modules is shown schematically. The supply interface **20** comprises the multiple connection **22**, which has already been described with reference to FIG. 1. The multiple connection **22** connects the different supply lines **24** in a standardized manner. The multiple connection **22** comprises in particular the gas connection **26**, the connection **28** for liquids, the communication connection **32** and the electric connection **34** for electric energy.

The communication connection **32** is particularly important for networking the controller **84** to a control network **88**. The space modules **12** can exchange data or control commands via the control network **88**. Liquids and gases are led through the pressure lines **44**.

FIG. 3 shows the space body **10** in a schematic and perspective view, in which a plurality of stacked space modules **12** are composed container-like. From this it becomes clear how important the geometry of the space modules **12** is for the stackability. In this space body **10** the solar cells **40** are arranged on an outer surface **108** for the power supply, in particular for charging an accumulator.

#### LIST OF REFERENCE NUMERALS

- 10** space body
- 12** space modules
- 14** cuboid
- 16** end faces
- 18** coupling device
- 20** supply interface
- 22** multiple connection
- 24** supply lines
- 26** gas connection
- 28** liquid connection
- 30** liquid lines
- 32** communication connection
- 34** electrical connection
- 35** connection for thermal energy
- 36** data lines
- 38** energy generator
- 40** solar cells
- 42** electric lines
- 44** pressure lines

46 magnetic coupling mechanism  
 48 electromagnet  
 50 locking mechanism  
 52 pressure body  
 54 insulation  
 56 protection layer  
 58 inside of the space module  
 60 outside of the space module  
 62, 64 outside surface of a space module  
 66 interior of a space module  
 68 internal supply accesses  
 70 gas connection  
 72 liquid connection  
 74 data connection  
 76 connection for electric energy  
 78 space module for experiments  
 80 closeable opening  
 82 passages  
 84 processor-controlled controller  
 86 memory  
 88 control network  
 90 sensors  
 92 optical sensor  
 94 acceleration sensor  
 96 position sensor  
 98 motion sensor  
 100 pressure sensor  
 102 monitoring device  
 104 space module with drive  
 106 drive  
 108 outer surface

The invention claimed is:

1. A space body with at least two interchangeably connectable space modules, comprising:
  - at least one closable connection opening on each of the space modules, via which a passage is provided when connecting the space modules;
  - at least one coupling device on each of the space modules for connecting the space modules;
  - at least one supply interface with a magnetic coupling mechanism on each of the space modules, for magnetically coupling a supply line which extends through the connection openings of two connected ones of the space modules;
  - a mechanical locking mechanism for mechanically coupling the supply line; and
  - decentralized resources for supplying the space body distributed in each of the space modules, wherein a resource exchange between the space modules is provided via the supply line,
  - wherein the magnetic coupling mechanism comprises an electromagnet for magnetically coupling the supply line,
  - wherein the supply interface is configured to engage the mechanical locking mechanism and thereby mechanically couple the supply line after the supply line has been magnetically coupled with support of the electromagnet, and
  - wherein the supply interface is configured to turn off the electromagnet after the mechanical locking mechanism is engaged.
2. The space body according to claim 1, wherein a processor-controlled controller for controlling the space modules is provided, wherein the controller as a resource for supplying the space body is provided decentralized in each of the

- space modules and is networked together via the supply interface for coordinated control of the space body.
- 3. The space body according to claim 2, wherein the controller is configured as a neural network structure.
- 5 4. The space body according to claim 1, wherein gas resources are provided decentralized in the space modules respectively for supplying the space body.
- 5. The space body according to claim 1, wherein liquid resources are provided decentralized in the space modules respectively for supplying the space body.
- 10 6. The space body according to claim 1, wherein thermal resources are provided decentralized in the space modules respectively for supplying the space body either for cooling or for heating.
- 15 7. The space body according to claim 1, wherein resources for electric energy are provided decentralized in the space modules respectively for supplying the space body.
- 8. The space body according to claim 1, wherein the at least one supply interface comprises a standardized dockable multiple connection, for magnetically coupling the supply line and at least one further supply line.
- 20 9. The space body according to claim 8, wherein the multiple connection comprises
  - a) a connection for gas and/or
  - 25 b) a connection for liquid and/or
  - c) a connection for communication and/or
  - d) a connection for thermal energy and/or
  - e) a connection for electric energy.
- 10. The space body according to claim 8, wherein the multiple connection comprises a connection device for pressure lines.
- 30 11. The space body according to claim 1, wherein an energy storage and/or a solar energy supply is provided for energy supply.
- 35 12. The space body according to claim 1, wherein the space body is configured as a pressure body.
- 13. The space body according to claim 1, wherein the space modules have a stackable geometry.
- 40 14. The space body according to claim 1, wherein the space modules have a cubic geometry.
- 15. The space body according to claim 1, wherein the supply interface is provided on at least two sides of a space module respectively.
- 45 16. The space body according to claim 1, wherein an inner and/or outer insulation is provided on the space modules.
- 17. The space body according to claim 1, wherein a processor-controlled monitoring device is provided for the space modules.
- 50 18. The space body according to claim 17, wherein the monitoring device comprises
  - a) an optical sensor and/or
  - b) an acceleration sensor and/or
  - c) a motion sensor and/or
  - d) a temperature sensor and/or
  - 55 e) a position sensor.
- 19. The space body according to claim 1, wherein at least one space module comprises a drive.
- 20. The space body according to claim 19, wherein the space module with the drive comprises a drive control.
- 60 21. The space body according to claim 1, wherein the space body is provided for aerospace and/or for use underwater or in a life-unfriendly environment.
- 22. The space body according to claim 1, wherein an internal supply access is provided within a space module of the space body.
- 65 23. The space body according to claim 22, wherein the internal supply access comprises

- a) a connection for gas and/or
- b) a connection for liquid and/or
- c) a connection for communication and/or
- d) a connection for thermal energy and/or
- e) a connection for electric energy.

5

\* \* \* \* \*