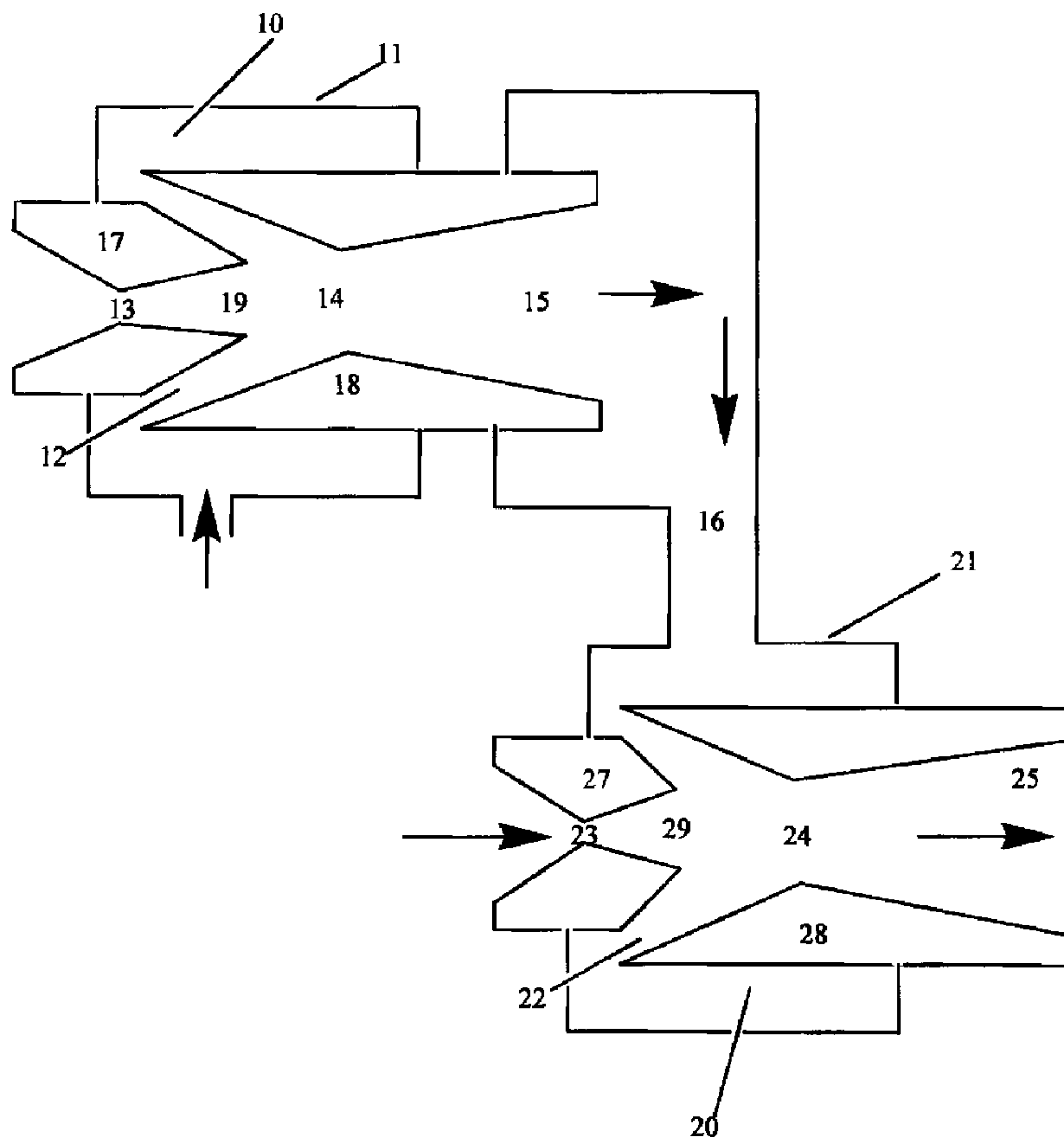




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(54) Title: TANDEM SUPERSONIC EJECTORS



(57) Abrégé/Abstract:

Low pressure off gases such as from dry gas seals may be recompressed to a higher pressure using multiple tandem supersonic ejectors in which upstream ejectors operate at a higher nozzle exit Mach number than down stream ejectors and all ejectors are operated at exit Mach numbers greater than 1.

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## **TANDEM SUPERSONIC EJECTORS**

### **ABSTRACT OF THE DISCLOSURE**

Low pressure off gases such as from dry gas seals may be recompressed to a higher pressure using multiple tandem supersonic ejectors in which upstream ejectors operate at a higher nozzle exit Mach number than down stream ejectors and all ejectors are operated at exit Mach numbers greater than 1.

## **FIELD OF THE INVENTION**

The present invention relates to the recovery and recompression of off gasses. The off gases may be from any source including a compressor station on a gas pipeline or a source in a chemical plant. The off gases  
5 are recompressed and may be fed back to the same or another chemical reaction or may be fed to a power means such as a gas fired boiler or a gas fired turbine. In accordance with the present invention the off gases may be recompressed using tandem supersonic ejectors.

## **BACKGROUND OF THE INVENTION**

10 Ejectors, sometimes called gas (e.g. steam) ejectors or venturi ejectors are well known. They may be used to maintain a vacuum or to compress a gas. The advantage of the ejector over conventional mechanical pumps such as piston pumps or compressors and diaphragm pumps is that it has no moving parts and is generally robust (subject to  
15 filtering the gas streams to reduce pitting and corrosion). Typically ejectors are subsonic, as supersonic ejectors tend to produce a low pressure exit stream. Additionally supersonic ejectors are sensitive to the design/construction parameters.

An ejector typically comprises an expansion nozzle port through  
20 which a motive gas enters the ejector through an inlet port. The gas is expanded to a lower pressure as it passes through a constricted throat section of the nozzle. Generally there is a suction port opening into an enclosed chamber about the expansion nozzle through which the gas to be captured is drawn into the ejector. Then downstream of the expander

there is a diffuser section having an inlet, a throat section and a diverging discharge section.

Conventional subsonic ejectors are commonly used to maintain a vacuum on a system such as disclosed in the following patents.

5 United States Patent 5,380,822 issued Jan. 10, 1995 discloses the use of a gas, typically steam, ejector to maintain a lower pressure in the later stages of a falling strand devolatilizer than in the down stream condenser to prevent water from freezing. This patent doesn't teach supersonic ejectors nor does it suggest ejectors could be used in  
10 sequence or in tandem.

United States patent 6,855,248 Issued Feb. 15, 2005 to Oliver et al. assigned to Sulzer Chemtech Limited teaches the use of a steam ejector to maintain a vacuum on a processing column (example 1). The patent does not teach the use of a gas ejector to recycle off gases. Further the  
15 patent does not teach tandem supersonic ejectors.

United States patent 6,330,821 issued Dec. 18, 2001 to Arnold et al. assigned to The Goodyear Tire & Rubber Company teaches the use of a gas ejector to maintain a vacuum on a part being tested. The patent does not teach the use of a gas ejector to recycle off gases. Further the  
20 patent does not teach tandem supersonic ejectors.

United States patent 4,194,924 issued March 25, 1898 assigned to The United States of America as represented by the Secretary of the Air teaches distilling a carrier solvent and JP -4 in a heated vacuum column in which the vacuum is provided by a gas (steam) ejector. The patent does

not teach the use of a gas ejector to recycle off gases. Further the patent does not teach tandem supersonic ejectors.

United States patent 4,834,343 issued May 30, 1989 to Boyles teaches a non flooded treatment column including a venturi device within  
5 the top of the column to redisperse the gas beneath the fluid level (Figure 6 and Col. 7 lines 60 to 65). However, the patent does not teach supersonic ejectors nor does it teach tandem ejectors or venturi's. Further the patent does not teach off gases which have escaped a containment zone such as a seal.

10 The present invention seeks to provide a simple and economical process to capture off gases and recompress them to be used at a higher pressure.

### **SUMMARY OF THE INVENTION**

The present invention provides a process for the repressurization of  
15 an off gas at a pressure from 68.94 kPa-g (10 psig) to 344.7 kPa-g (50 psig) and to a pressure from 2068.2 kPa-g (300 psig) to 3447 kPa-g (500 psig) comprising using two or more supersonic gas ejectors in tandem wherein an upstream supersonic ejector operates at an exit nozzle Mach number greater than the exit nozzle Mach number of the subsequent down  
20 stream supersonic ejectors and all ejectors operate at a nozzle exit Mach numbers greater than 1.

Preferably there are first and second supersonic gas ejectors in tandem wherein the first supersonic gas ejector is operated at an exit Mach number greater than 2, the second gas ejector is operated at an exit

Mach number less than 2, and both of which are operated at exit Mach numbers greater than 1.0.

### **BRIEF DESCRIPTION OF THE DRAWINGS**

Figure 1 is a schematic sectional drawing of the tandem supersonic  
5 ejectors of the present invention.

Figure 2 shows Mach number contours at the exit of a supersonic nozzle and diffuser.

Figure 3 shows Stagnation Pressure Contours at Exit of Supersonic  
Nozzle and Diffuser.

10 Figure 4 shows the overall performance of the two-Stage  
Supersonic Ejector (discharge pressure from 2<sup>nd</sup> stage vs. suction flow at  
1<sup>st</sup> stage

Figure 5 shows overall performance of the two-Stage Supersonic  
Ejector (suction pressure vs. suction flow at 1<sup>st</sup> stage).

15 Figure 6 shows overall performance of the two-Stage Supersonic  
Ejector (intermediate pressure between the two stages vs. suction flow at  
1<sup>st</sup> stage).

### **DETAILED DESCRIPTION**

In the handling of gaseous or volatile materials such as C<sub>1-8</sub>  
20 aliphatic or aromatic hydrocarbons which may be unsubstituted or further  
substituted by one or more C<sub>1-4</sub> alkyl radicals, inert gases, CO, CO<sub>2</sub>, N<sub>2</sub>,  
natural gas and mixtures thereof the gases may be used in side processes  
or be subject to leakage from joints or seals, for example drive trains or  
shafts for compressors or stirrers. One particularly useful type of seal for a  
25 shaft is a dry-gas seal.

A typical dry-gas seal is a non-contact end face seal in which the sealing mechanism is comprised of two rings: the first ring with grooves etched in the seal face, which rotates with the shaft, is known as the mating ring, while the second ring has a smooth face and is restrained  
5 from movement except along the axis of the shaft. A pair of these seals often co-exists; hence provide a two-stage sealing effect.

For successful operation, it is essential that a steady flow of clean seal gas be supplied to the gaps between the rings. The seal supply gas source must be at a higher pressure than that of the process gas that is  
10 being sealed in order for flow to occur. On overhung compressors only one dry seal is required, in which case the seal supply gas can be drawn from the compressor discharge, filtered, and supplied in a steady flow to the seal capsule. Most of the seal supply gas re-enters the process cavity, while a small volume (leakage) passes through the seal faces and is  
15 vented. The amount of gas leakage depends on the process pressure and rotating shaft diameter.

In the operation of a pipeline at a compressor station the gas used in the dry-gas seal is typically the gas being passed through the pipeline. The gas pressure is typically from 5515.2 kPa-g (800 psig) to 6894 kPa-g  
20 (1000 psig), preferably from 6204.6 kPa-g (900 psig) to 6894 kPa-g (1000 psig), and passed through the seal. The off gas (or vent gas or buffer gas) from the seal is typically at a pressure from 68.94 kPa-g (10 psig) to 344.7 kPa-g (50 psig), preferably from 207 kPa-g (30 psig) to 344.7 kPa-g (50 psig). Generally the flow rate of the off gas is low, typically from 1-3  
25 kg/hour.

For a chemical plant the off gas pressures and rates may be comparable.

The off gas or buffer gas may be captured in a housing generally air tight around the seal. However, the off gas or buffer gas is at too low a  
5 pressure to do very much with. At a chemical plant the pressure of the off gas may be too low to feed directly to a flare stack, or directed back to the chemical process or to a power generation means such as a gas fired boiler or a gas fired turbine.

Figure 1 is a sectional schematic diagram of tandem supersonic  
10 ejectors in accordance with the present invention. In the figure there are two supersonic ejectors 10, and 20 in tandem. The first supersonic ejector comprises an enclosure 11, which is airtight or substantially airtight that encloses a suction port 12. The suction port 12 of the first supersonic ejector 10 may be annular. The motive gas enters the nozzle 17 of the  
15 first supersonic ejector, is expanded through a constricted throat 13, and is further expanded through the diverging section of the nozzle to a much lower pressure and high supersonic velocity. This supersonic velocity of the motive gas as it exits nozzle 17 at exit 19 of the first supersonic ejector 10 and the resulting reduction in the pressure draws the off gas into the  
20 ejector through suction port 12. The combined motive gas and the off gas proceed to diffuser 18 of the first supersonic ejector having a larger throat 14 than that of the nozzle 17. The cross sectional area of the throat 14 of the diffuser of the first ejector is larger in size than the cross sectional area of the throat 13 of the nozzle 17. Due to the converging and then  
25 diverging sections of the cross section area of the channel through the

diffuser the speed of the motive gas and entrained off gas decreases. The mixture of the motive gas and the off gas exit the ejector at the end 15 of the diffuser 18 at higher pressure than that of the off gas. The end of the diffuser 15 exits into a conduit 16 leading to an enclosure 21, which is air  
5 tight or substantially air tight, that encloses a suction port 22 of the second supersonic ejector 20. The suction port 22 of the second supersonic ejector 20 may be annular.

The motive gas enters the nozzle 27 of the second supersonic ejector and proceeds to the constricted throat 23, is expanded through the  
10 diverging section of the nozzle 27 exits the nozzle at exit 29 and proceeds to diffuser 28 having a larger throat 24 than throat 23 of nozzle 27. The cross sectional area of the channel through the second supersonic ejector also increases in size from throat 23 of the nozzle 27 to the throat 24 of the diffuser 28. This increases the velocity of the motive gas as it passes  
15 through throat 23 and the diverging section of the nozzle 27 and reduces the pressure drawing the exit gas from the first supersonic ejector passing through port 16 into the ejector through suction port 22. Due to the converging and diverging cross section areas of the channel through the diffuser the speed of the motive gas and entrained off gas decreases in  
20 the diffuser. The mixture of the motive gas and the gas in port 16 exit the ejector at the end 25 of the diffuser 28. Exit port 25 from diffuser 28 of the second supersonic ejector feeds a pipe or line to recirculate the off gas combined with the motive gas for further processing.

In operation a motive gas at a higher pressure than the off gas, in  
25 the case of a pipeline the natural gas within the line and in the case of a

chemical plant likely process steam is injected into nozzle 17 of the first  
supersonic ejector 10. The cross section area of the ejector narrows to a  
throat section 13 of the first supersonic ejector. This increases the velocity  
of the gas as it passes through the throat and continues to expand through  
5 the diverging section of nozzle 17 to exit 19, which creates a lower  
pressure at the suction inlet 12 of the first supersonic ejector 10. This  
draws the off gas within the enclosure 11 into the first supersonic ejector.  
The off gas is drawn into and entrained with the motive gas passing  
through the first supersonic ejector. Downstream the cross section area of  
10 the throat 14 of the diffuser 18 is larger than throat 13 of the nozzle 17.  
The diffuser 18 expands to a discharge 15 into or is fed to the suction port  
22 for the second supersonic ejector 20. A second motive gas is fed to the  
nozzle 27 of the second supersonic ejector, which narrows to throat 23.  
The gas velocity increases and the pressure drops drawing the off gas into  
15 the nozzle and leaves at exit 29. The cross section area of the second  
supersonic ejector also increases to a throat 24 of the diffuser 28 and  
further then expands to a discharge port 25. The discharge port 25 then  
feeds a line (not shown) which directs the recompressed off gas to  
subsequent processing at a higher pressure.

20 In a preferred embodiment of the invention the nozzles (17 and 27)  
of the supersonic ejectors are adjustable relative to the diffusers. Typically  
this is done by having the nozzle threaded and mounted on receiving  
threads on the enclosure or on a portion of the inlet to the diffuser in a  
manner not to close the suction port.

The ejectors should be designed so that the first supersonic ejector is operated at an exit Mach number from 2.4 to 2.6 and the second supersonic ejector is operated at an exit Mach number from 1.6 to 1.8. In the first supersonic ejector the ratio of the cross section area of the nozzle exit 19 to the nozzle throat 13 may be from 2.9 to 3.2, preferably from 3.0 to 3.1. In the second supersonic ejector the ratio of the cross section area of the nozzle exit 29 to the nozzle throat 23 may be from 1.30 to 1.45, preferably from 1.35 to 1.40. The ratio of the area of the throat 14 of the diffuser 18 to the throat 13 of the nozzle 17 of the first supersonic ejector may range from about 4.60 to about 4.90, preferably from 4.70 to 4.80. The ratio of the area of the throat 24 of the diffuser 28 to the throat 23 of the nozzle 27 of the second supersonic ejector may range from about 1.70 to about 1.90, preferably from 1.80 to 1.90. Typically the ratio of the motive gas flow rate to the first supersonic gas ejector to the off gas flow rate is from 32 to 45. (e.g. either g per g or Kg per Kg as this is a unitless ratio). Typically the ratio between the motive gas flow rate to the second supersonic gas ejector and the discharge flow from the first supersonic ejector is from 20 to 25.

The design parameters for the supersonic ejectors are well known to those skilled in the art.

Without being bound by theory, the one-dimensional governing equations for the isentropic expansion of gas through a converging-diverging supersonic nozzle can be written as follows wherein in the following equations

25

A - cross sectional area

c - local speed of sound

$\dot{m}$  - mass flow rate

M - Mach number

5 P - pressure

u - local velocity

s - entropy

T - temperature

$\Psi$  - exergy

10  $\rho$  - density

Subscript:

1 - motive gas to 1<sup>st</sup> stage ejector

2 - suction gas to 1<sup>st</sup> stage ejector

3 - exit gas from 1<sup>st</sup> stage ejector

15 in - motive gas to 1<sup>st</sup> stage ejector

out- motive gas to 1<sup>st</sup> stage ejector

r reference condition for the exergy:

Energy: 
$$udu = -\frac{1}{\rho}(dP)_s \quad (1)$$

20 Continuity: 
$$\frac{dA}{A} = -\frac{d\rho}{\rho} - \frac{du}{u} \quad (2)$$

Using the isentropic relation:

$$\left(\frac{dP}{d\rho}\right)_s = c^2 \quad (3)$$

25

And substituting from (1) and (3) in (2) we can write the continuity equation in the form:

$$\begin{aligned} \frac{dA}{A} &= \frac{1}{\rho} \left( \frac{1}{u^2} - \frac{1}{c^2} \right) (dP)_s & (4) \\ &= \frac{1-M^2}{\rho u^2} (dP)_s \end{aligned}$$

5

where, subscript 's' refers to isentropic (i.e. at constant entropy).

Equations (1) and (4) are solved simultaneously along a known geometry-diverging nozzle to delineate the pressure profile along the nozzle as well as all other local flow parameters. A proper equation of state, such as American Gas Association Report Number 8 equation of state for hydrocarbon gases (AGA-8), must be utilized to account for real gas properties. For example for a supersonic nozzle with an exit to throat area ratio of 3.06, the calculated expansion pressure ratio based on the above formulation and accounting for real gas behavior is 18.7, and exit Mach number = 2.54 (for a stagnation inlet pressure of 5500 kPa-g and temperature of 283 K).

In a gas transmission line the recompressed off gas – natural gas may be fed to a power means such as a gas fired turbine. For a chemical plant the recompressed off gas may be fed to a power means such as a gas fired boiler or it could be fed to a flare gas tower or it could be recycled back to the process.

### **EXAMPLES**

The present invention is illustrated by the following non-limiting examples.

### Example 1 Computational Fluid Dynamics

In order to arrive at the optimum design of a fixed-geometry diffuser that could work with different nozzles of different throat areas, Computational Fluid Dynamics (CFD) analysis was carried out to discern the flow field. Figures 2 and 3 show an example of CFD results showing the Mach number and stagnation pressure at nozzle exit and through the diffuser. This is an example of an optimized nozzle/diffuser arrangement showing that the nozzle is clear from a standing shock wave, which is good for suction. It is also shown that the shock wave region (which is the region where Mach number drops sharply from supersonic to subsonic – Fig. 2) is located at the throat of the diffuser or slightly downstream. This is also preferable for minimum stagnation pressure losses, as is also confirmed by the stagnation pressure contours of Fig. 3.

In order to further assess the efficiency of the ejector (combined of both elements: nozzle and diffuser), the following definition of ejector efficiency was adopted:

$$\eta = \frac{(\dot{m}_1 + \dot{m}_2)\psi_3}{\dot{m}_1\psi_1} \quad (5)$$

where;  $\psi$  is the thermodynamic availability (exergy), subscript 1,2,3 refer to motive, suction and discharge conditions. Exergy is defined with reference to suction condition, hence:

$$\psi_1 = (h_1 - h_r) - T_r(s_1 - s_r) \quad (6)$$

$$\psi_3 = (h_3 - h_r) - T_r(s_3 - s_r) \quad (7)$$

The above definition was used later in assessing the various nozzle geometry and respective position with respect to the diffuser.

Example 2 – First Supersonic Ejector

Based on the above calculations, and in order to satisfy the relatively low suction pressure to match the dry-gas seal leakage pressure (~ 400 kPa-a), an ejector with a highly supersonic exit flow was employed.

- 5 A flexible prototype design was fabricated which allows for various geometry supersonic nozzles to be tested with a fixed geometry diffuser, but it also allows for fine adjustments of the position of the nozzle exit in relation to the diffuser inlet (either positively, i.e. inserted into the diffuser inlet section, or negatively, i.e. retrieved back with a gap in between nozzle  
10 exit and diffuser inlet).

The supersonic diffuser has an inlet diameter of 4 mm, a throat diameter of 3.5 mm and length of 8 mm, and an exit diameter of 18 mm. Inlet  $\frac{1}{2}$ -angle of the inlet section is  $4.7^\circ$ , while that of the exit section is  $5^\circ$ . Several supersonic converging/diverging nozzles were fabricated with  
15 different throat diameters and exit/throat area ratios as shown in Table 1.

**TABLE 1****Various Supersonic Nozzle Tested in the 1<sup>st</sup> Stage Ejector**

Throat (mm)	Exit (mm)	Area Ratio
3	4.5	2.25
	8	7.11
2	2.5	1.56
	3	2.25
	3.5	3.06
	4	4.00
	5	6.25
1.6	2.4	2.25
	2.8	3.06

20

The various nozzles were tested at the TransCanada Pipelines Limited test facility at Didsbury, Alberta. Suction gas was drawn from the

same pipeline gas, which then flows through two-step regulators that throttle the line pressure to the specified maximum suction pressure. A summary of the results of the experiment is reported in table 2.

5

TABLE 2

**Summary Results of 1<sup>st</sup> Stage Supersonic Ejector Testing**

Nozzle	Compression Ratio (P3/P2)	Expansion Ratio (P1/P2)	Efficiency	Comment
1.6x2.8	3.5	20	0.46	P1=5000 kPa
1.6x2.8	3	16	0.46	P1=4000 kPa
1.6x2.4	3.6	19	0.48	P1=5000 kPa
1.6x2.4	2.9	16	0.44	P1=4000 kPa
2x3.5	4.5	17	0.56	P1=5000 kPa
2x3.5	4.3	15	0.56	P1=4000 kPa
2x3	4.1	15	0.56	P1=4000 kPa

10 In the table the efficiency is defined by Equation (5) above.

The best performing 1<sup>st</sup> stage stand-alone configuration is given in Table 3, where the position of the nozzle exit is inserted 20.5 mm into the diffuser inlet section.

TABLE 3

15

**Characteristic Dimensions of the Best 1<sup>st</sup> Stage Ejector**

Nozzle Diameter	mm	1.6
Nozzle Exit	mm	2.8
Half Angle	deg	1
Diffuser Inlet Diameter	mm	4
Half Angle	deg	4.7
Diffuser Throat Diameter	mm	3.5
Diffuser Throat Length	mm	8
Diffuser Exit Diameter	mm	18
Diffuser Exit Half Angle	deg	5

Example 3 Second Stage Ejector

Computational Fluid Dynamics (CFD) was utilized to optimize the best supersonic diffuser dimensions (throat, inlet, and outlet diameters, as well as angles) and position of the nozzle with respect to the diffuser inlet.

5 Table 4 shows the optimum design for the second stage ejector.

**TABLE 4****Characteristic Dimensions of the 2<sup>nd</sup> Stage Ejector**

Nozzle Diameter	mm	7.4
Nozzle Exit	mm	8.7
Half Angle	deg	1
Diffuser Inlet Diameter	mm	12.4
Half Angle	deg	1
Diffuser Throat Diameter	mm	10
Diffuser Throat Length	mm	25
Diffuser Exit Diameter	mm	30
Diffuser Exit Half Angle	deg	5

Tests were conducted on the 2<sup>nd</sup> stage ejector alone in order to  
 10 optimize the position of its supersonic nozzle with respect to the diffuser inlet. The best performance was obtained with the position of the nozzle exit at 1.42 mm away from the inlet section of the supersonic diffuser in this 2<sup>nd</sup> stage ejector. At this nozzle position, the discharge pressure ( $P_{out}$ ) reaches 3400 kPa-g.

15 Example 4 Combined First and Second Stage Ejectors

Tests were then conducted at the Didsbury facility on the two-stage ejector configuration combined, by varying  $P_1$  to the 1<sup>st</sup> stage ejector (4600 kPa-g, 5000 kPa-g and 5500 kPa-g), while maintaining the motive gas pressure ( $P_{in}$ ) to the 2<sup>nd</sup> stage ejector at maximum line pressure of  
 20 approximately 6000 kPa-g. Figures 4 through 6 show the results of the integrated two-stage ejector system in terms of the discharge pressure

from the 2<sup>nd</sup> stage ejector (Fig. 4), suction flow at the 1<sup>st</sup> stage (Fig. 5), and the intermediate pressure ( $P_3$ ) for different  $P_1$  (Fig. 6). The effects of varying ( $P_1$ ) are manifested in Fig. 4, which indicates that the lower  $P_1$  the higher the suction flow, but at the expense of the overall discharge  
5 pressure ( $P_{out}$ ).

The data shows that the present optimized configuration is capable of delivering the required discharge pressure ( $P_{out}$ ) of 3300 kPa-g with a suction flow of 2-2.5 kg/hr and suction pressure ( $P_2$ ) of 340 kPa-g. These values match the requirements for this ejector to work with a dry-gas  
10 leakage and a typical fuel gas line on a typical compressor station.

The embodiments of the invention in which an exclusive property or privilege is claimed are defined as follows:

1. A process for the repressurization of an off gas at a pressure from  
5 68.94 kPa-g (10 psig) to 344.7 kPa-g (50 psig) and to a pressure from  
2068.2 kPa-g (300 psig) to 3447 kPa-g (500 psig) comprising using two or  
more supersonic gas ejectors in tandem wherein an upstream supersonic  
ejector operates at an exit nozzle Mach number greater than the exit  
nozzle Mach number of the subsequent down stream supersonic ejectors  
10 and all ejectors operate at a nozzle exit Mach numbers greater than 1.
2. A process for the repressurization of an off gas at a pressure range  
of 68.94 kPa-g (10 psig) to 344.7 kPa-g (50 psig) and to a pressure range  
2068.2 kPa-g (300 psig) to 3447 kPa (500 psig) comprising using first and  
15 second supersonic gas ejectors in tandem wherein the first supersonic gas  
ejector is operated at an exit supersonic Mach number greater than 2 and  
the second gas ejector is operated at an exit supersonic Mach number  
less than 2.
- 20 3. The process according to claim 2, wherein the pressure of the off  
gas is from 172.3 kPa-g (25 psig) to 310.1 kPa-g (45 psig).
4. The process according to claim 3, wherein the pressure of the  
repressurized gas is from 2412.9 kPa-g (350 psig) to 3102.3 kPa-g (450  
25 psig).

5. The process according to claim 4, wherein pressure of the gas exiting the first supersonic ejector is from 1034.1 kPa-g (150 psig) to 1240.9 kPa-g (180 psig).
- 5 6. The process according to claim 5, wherein the off gas is selected from the group consisting of C<sub>1-8</sub> aliphatic or aromatic hydrocarbons which may be unsubstituted or further substituted by one or more C<sub>1-4</sub> alkyl radicals, inert gases, CO, CO<sub>2</sub>, N<sub>2</sub>, natural gas and mixtures thereof.
- 10 7. The process according to claim 6, wherein the ratio of the motive gas flow rate to the first supersonic gas ejector to the off gas flow rate is from 32 to 45.
8. The process according to claim 7, wherein the ratio between the  
15 motive gas flow rate to the second supersonic gas ejector and the discharge flow from the first supersonic ejector is from 20 to 25.
9. The process according to claim 8, wherein the first supersonic  
ejector is operated at an exit Mach number from 2.4 to 2.6 and the second  
20 supersonic ejector is operated at an exit Mach number from 1.6 to 1.8.
10. The process according to claim 9, wherein the off gas is a buffer or seal gas for a dry seal.

11. The process according to claim 10, wherein the off gas is natural gas.
12. The process according to claim 11, wherein the buffer or seal gas is  
5 fed to the dry seal at a pressure from 5515.2 kPa-g (800 psig) to 6894 kPa-g (1000 psig).
13. The process according to claim 12, wherein the flow rate of off gas from the dry seal is from 1 to 3 kg/hour.  
10
14. The process according to claim 13, wherein the buffer or seal gas is fed to the first supersonic ejector at a pressure from 68.94 kPa-g (10 psig) to 344.7 kPa-g (50 psig).
- 15 15. The process according to claim 14, wherein the recompressed gas is fed to a power means.
16. The process according to claim 15, wherein said power means is selected from the group consisting of a natural gas fired turbine and a  
20 natural gas fired boiler.
17. The process according to claim 9 wherein the off gas is from a chemical reaction, or a purge bin.

18. The process according to claim 17, wherein the recompressed gas is fed to a flare stack.
19. The process according to claim 17, wherein the recompressed gas  
5 is fed back to the chemical reactor.
20. The process according to claim 17, wherein the recompressed gas is fed to power means.
- 10 21. The process according to claim 20, wherein the power means is selected from the group consisting of a natural gas fired turbine and a natural gas fired boiler.

Application number / numéro de demande: 2560814

Figures 2, 3, 4, 5, 6

Pages: 2, 3 & 4

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Figure 1

