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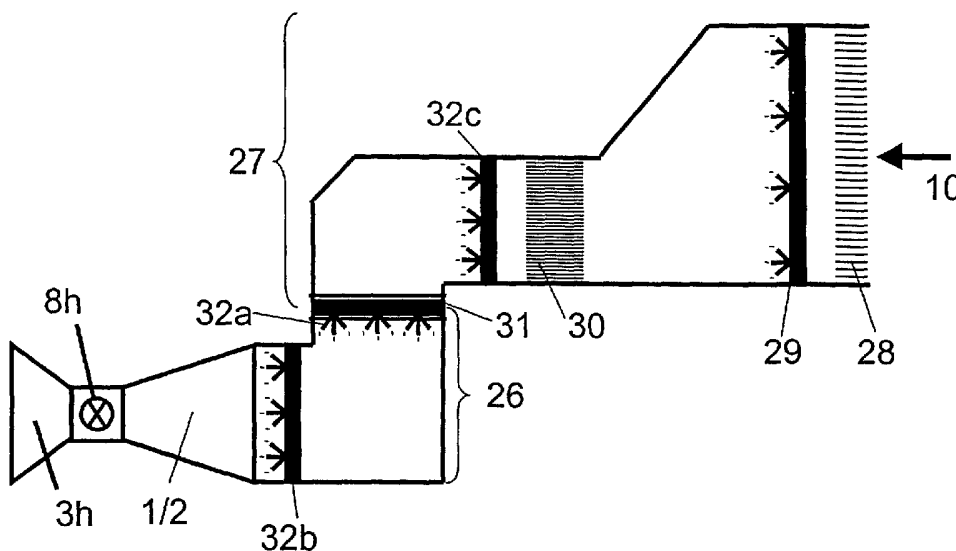
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(54) Title: METHOD AND APPARATUS FOR ACHIEVING POWER AUGMENTATION IN GAS TURBINES USING WET COMPRESSION



(57) Abstract: A gas turbine unit as well as a method for operating a gas turbine with high-pressure turbine (3h) and a low-pressure turbine (31) unit is proposed. A very quick and at the same time easily controllable augmentation or reduction of shaft power being produced by the gas turbine can be achieved by providing at least one liquid droplet injection device (32) on the upstream side of compressor (1, 2) for injecting liquid into the stream of intake air (10) in order to increase the shaft power generated by the gas turbine unit. The amount of water mass flow corresponding to the desired increase or decrease of shaft power output of the gas turbine unit is added or reduced in the form of liquid droplets in a substantially stepless manner and immediately within a time interval that is determined by the design characteristics of the liquid droplet injection device (32).

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## DESCRIPTION

## TITLE

Method and apparatus for achieving power augmentation in gas turbines using wet compression

## FIELD OF THE INVENTION

The present invention relates to a gas turbine unit as well as to a method of operating a gas turbine unit both using over-fogging. More particularly it concerns a gas turbine unit with sequential combustion chambers essentially comprising: a compressor for compressing intake air supplied thereto and discharging the compressed air; a high-pressure combustion chamber in which fuel is combusted with the compressed air discharged from said compressor; a high-pressure turbine driven by the hot combustion air discharged from said high-pressure combustion chamber; a low-pressure combustion chamber in which fuel is combusted with the expanded combustion air discharged from said high-pressure turbine; and a low-pressure turbine driven by the hot combustion air discharged from said low-pressure combustion chamber.

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## BACKGROUND OF THE INVENTION

It is well known that the addition of water, or other suitable liquids or mixtures of liquids, into the working medium of gas turbines can be used for increasing the power that can be generated by gas turbine units. The increase of available power is due to, among other reasons, the cooling effect of the added water reducing the compressor power consumption and allowing an increased firing of the gas turbine unit, as well as due to the increased mass flow passing the turbine blades.

Water can be added either in the form of vapour, that is in the form of humidified air or, and this is the object of the present invention, in the form of small liquid droplets, i.e. water can be added above the saturation level of water in the air. This technique, known as over-fogging, is usually carried out by adding liquid droplets of particular size to the air stream entering the compressor of the gas turbine unit (so-called 'wet compression'). 'Wet compression' allows to increase the power available due to the reduction of work required for compression of the inlet air, as the latent heat for evaporation of this water cools the inlet air stream when it passes the compressor stages.

There are a number of documents describing particular designs of gas turbine units and methods for their operation using 'wet compression'. For example US 5,930,990 as well as its continuation-in-part US 5,867,977 describe an apparatus for 'wet compression' as well as a method for increasing the power available by using 'wet compression'. These documents concern gas turbine units with one single combustion chamber, where small droplets of water are being added to the intake air entering the compressor. The increase of added water is carried out in a stepwise, incremental manner, and due to problems of thermal non-equilibrium induced in the components of the gas turbine unit downstream of the injection (thermal shock leading to cracking in certain alloyed components and to undesired reduction of clearance between parts which are in motion relative to each other during operation), this stepwise addition of water is proposed to be carried out on a timescale of 10-20 minutes. Additionally the proposed means for adding the water are designed like a grid of nozzles lying in one plane and being arranged substantially perpendicular to the direction of flow of the intake air stream. This stepwise increase of water added to the intake air can, according to this document, be achieved by either increasingly feeding the nozzles with water or by systematically feeding more and more nozzles with water (sequential increase in time or in position).

In order to control and to optimise the addition of water when using wet compression, WO 00/50739 proposes a particular apparatus for monitoring destructive 'wet compression'

power augmentation-related casing distortions in gas turbine units. Also this document points out that in order to avoid thermal stress when activating 'wet compression', the addition of water has to be carried out smoothly under a careful monitoring of the associated distortions of the casing and other components of the gas turbine unit. Another particular device to be used for 'wet compression' is proposed in US 6,216,443. The water is added by means of a liquid droplet injection device provided on the upstream side of a compressor, and on the downstream side of a silencer. The addition of water is controlled by means of a valve, and it is also pointed out that due to thermal stress and due to the fact that concomitantly other parameters have to be adjusted when adding water to the intake air (for example the addition of fuel has to be adjusted), the addition has to be performed sufficiently slowly.

#### SUMMARY OF THE INVENTION

It is therefore the aim of the present invention, to provide a gas turbine unit as well as a method for its operation allowing a simple and constructively easy augmentation of the available shaft power of the gas turbine unit by using over-fogging, i.e. by injecting liquid droplets into the stream of intake air upstream of the compressor. More particularly, this aim shall be achieved for a gas turbine unit, essentially comprising: a compressor for compressing intake air supplied thereto and discharging the compressed air; a high-pressure combustion chamber in which fuel is combusted with the compressed air discharged from said compressor; a high-pressure turbine driven by the hot combustion air discharged from said high-pressure combustion chamber; a low-pressure combustion chamber in which fuel is combusted with the expanded combustion air discharged from said high-pressure turbine; and a low-pressure turbine driven by the hot combustion air discharged from said low-pressure combustion chamber.

This aim is achieved by the technical features as given in independent claims 1 and 2, as well as by the process steps according to claims 16 and 17. More particularly, this aim is achieved for gas turbine units, the compressor of which is working with a pressure ratio larger than 15 bar (i.e. the process is aiming mainly at, but not limited to large gas turbine units, and preferentially, the pressure ratio is even larger than 20 bar), by providing at least one liquid droplet injection device on the upstream side of said compressor for injecting liquid into the stream of intake air in order to increase the shaft power generated by the gas turbine unit, wherein the amount of liquid mass flow corresponding to the desired increase or decrease of shaft power output of the gas turbine unit is added or reduced in the form of liquid droplets immediately, i.e. within a time interval that is

determined by the design characteristics of the liquid droplet injection device to increase or decrease the liquid droplet mass flow, and in a substantially stepless manner, and wherein preferentially said liquid mass flow is injected substantially across the entire cross-section of the air intake.

The liquid injected preferentially consists substantially or completely of water. The water may be demineralized to avoid detrimental formation of deposit and scale, and/or may contain additives in particular to prevent the formation of deposit. For simplicity and clarity, in the following the term water shall be used meaning generally liquid, which leads to the desired effect when injected upstream of the compressor.

When talking about "immediately, i.e. within a time interval that is determined by the design characteristics of the liquid droplet injection device", that "immediately" means substantially as quickly as possible within the bounds predetermined by the design (tube widths, control and valve speed etc.) of the injection device. The term "immediately" or "immediate" shall in the following stand in the above sense.

Contrary to all expectations of the person skilled in the art and contrary to all statements found in the state of the art, an immediate and stepless addition or reduction of water mass flow when using over-fogging is possible without the deleterious effects mentioned in the above cited documents. Surprisingly, immediate and complete switching-on or switching-off of the liquid droplet injection device is possible without waiting for steady state conditions in the gas turbine by using slowly increasing or decreasing amounts of water. If it is additionally preferentially made sure that the droplets are added to the stream of intake air substantially across the entire cross-section of the intake air guide, the switching-on and switching-off can basically be carried out in an on/off-manner. This unexpectedly possible fast and simple switching leads to a number of possible constructional simplifications and other advantages. On the one hand no complicated control and regulation means need to be provided for the liquid droplet injection device, a simple on/off-control is sufficient thereby reducing costs as well as possible sources of failure. On the other hand the shaft power-augmentation/reduction effect of the gas turbine unit is available much more quickly than when using some particular, slow (stepwise) slope for increasing or decreasing the water addition.

According to a first preferred embodiment of the present invention, the gas turbine unit additionally comprises a cooling system which uses compressed or partially compressed air discharged from the compressor for cooling components of the gas turbine unit, wherein the cooling system comprises a cooling unit which is controlled such as to ensure substantially constant quality of the cooled cooling air. Cooling systems for gas turbine

units are highly sophisticated systems that have to be carefully controlled. Any change in the mode of operation of the gas turbine leads to a consequential change of the conditions within the compressor and to a corresponding change of the air entering the cooling system. To avoid flow-back of the cooling medium and to assure effective and sufficient cooling of the components that are being cooled by the cooling system, the cooling system is usually controlled with respect to temperature as well as pressure. Using conventional 'wet compression' start-up schemes, the cooling system has to be heavily controlled and actively guided. In particular, the cooling unit has to be controlled in reaction to the slow (stepwise) increase of water addition in order to keep pressure and temperature of the cooling medium downstream of the cooling unit at the desired levels i.e. within the desired limits. Surprisingly, the simplified immediate and stepless addition or reduction of water for 'wet compression' as proposed according to the present invention also considerably simplifies the control of the cooling system. The control of the cooling system can basically be reduced to a single feed forward signal synchronised to the on/off-control of the liquid droplet injection device (if need be the signals of the two systems can be slightly displaced relative to each other in order to take account of hysteresis-effects of the gas turbine unit). In other words it is possible to design the gas turbine unit such that the at least one injection device can be controlled in an on/off-manner only, and that preferably the cooling unit can be controlled with a single signal. It is however also possible to have certain discrete levels of desired power augmentation or reduction of the gas turbine unit and to have on/off-possibilities corresponding to these levels for the two systems only.

As the single feed forward signal for the cooler is necessary anyway for the case of an emergency shut down of the injection system, the proposed gas turbine unit clearly is a simplification of the turbine control system thus increasing reliability and reducing the occurrence of possible errors.

According to another preferred embodiment of the present invention, the gas turbine unit comprises a fuel (gas or oil) control valve that adjusts the fuel mass flow in order to maintain the desired firing temperatures of the gas turbine unit. The fuel control valve is subject to a complicated control mechanism of the gas turbine unit control system, its actual position depending on numerous parameters, for example, but not limited to the compressor discharge conditions. Accordingly, to prevent over-firing of the gas turbine unit with subsequent deleterious effects on the combustion chamber and turbine blade components, the fuel control valve needs to be heavily controlled and actively guided, if water is added in a slow, stepwise manner, since in reaction to such addition of liquid

water droplets results in a significant change of the compressor discharge temperature. In this regard it shall be mentioned that over-firing of the gas turbine unit may not only occur when decreasing the liquid injection mass flow, resulting in an increase of the compressor discharge temperature, but also when increasing the liquid injection mass flow, resulting in a decrease of the compressor discharge temperature, due to overshooting effects of the fuel valve control system. The present invention allows to reduce the control interventions of the existing control of the fuel valve using one simple feed forward signal.

According to a further preferred embodiment of the present invention, the gas turbine unit additionally comprises the ability to rapidly increase the power output of the unit using the immediate and stepless addition or reduction of liquid mass flow. This is particularly applicable, but not limited to the case of the power generated by the gas turbine unit being converted into electricity with a generator and being fed into an electricity grid of electricity generators and consumers. When the electricity grid is subject to a fast increase or decrease in power demand caused by e.g. switching on or off of large consumers or a fast decrease in power generation capacity caused by e.g. the emergency shut-down of a large power generation unit, the frequency of the electricity grid drops or rises and an immediate increase or decrease in power generation capacity within seconds or a few minutes is needed in order to maintain the electricity grid frequency reasonably stable and to prevent breakdown of parts of or even of the complete electricity grid. The ability of a particular gas turbine unit to rapidly increase or decrease the power output in a situation of low or high electricity grid frequency is also known as its ability to be operated in 'frequency response' mode. The current invention provides means to rapidly, i.e. on a seconds time scale increase or decrease the power output of the gas turbine unit significantly, e.g. by 10% of its full load capacity, by immediate and stepless addition of water mass flow into the intake air stream. Even more particularly, when the electricity grid frequency drops, also the shaft speed of the gas turbine unit is reduced by the corresponding amount, assuming the gas turbine unit comprises a single shaft arrangement, reason being that the shaft of the gas turbine unit is coupled with the shaft of the electricity generator, and that the shaft speed of the generator is synchronized to the frequency of the electricity grid. In case of a reduced shaft speed, the surge margin, which defines the limit for stable operation of the compressor of gas turbine units, tends to be reduced, limiting the ability of the gas turbine unit to increase its power output or even forcing it to reduce its power output in order to prevent destructive compressor surge, thus worsening the generation capacity situation of the electricity grid. The liquid injection means according to the present invention substantially improves the ability of a gas

turbine unit to rapidly increase power output even in case of a reduced shaft speed in that way that by immediate and stepless addition of water, the surge margin of the compressor of the gas turbine unit is increased by cooling the compressor blade section.

According to another preferred embodiment of the present invention the gas turbine unit additionally comprises an intake manifold situated upstream of said compressor and an intake duct situated upstream of said intake manifold connected to said intake manifold by means of an expansion joint, and the liquid droplet injection device is essentially situated at the expansion joint between the intake duct and the intake manifold, wherein preferably the intake duct additionally comprises a silencer located upstream of said liquid droplet injection device and a filter located essentially at the intake opening of the intake duct, and wherein preferably additional cooling means for cooling the intake air are situated downstream of the filter. Locating the liquid droplet injection device close to the or at the expansion joint proves to be particularly advantageous, as the droplets can be evenly distributed across the cross-section, the droplets can be generated in a very small size and the liquid droplet injection device can be easily installed and maintenance is kept simple. According to still another preferred embodiment, the liquid droplet injection device consists of a grid of fogging water ducts, preferably arranged in an essentially parallel manner on a carrying rack, on the downstream side of which fogging water ducts fogging nozzles are mounted for injecting droplets into the stream of intake air, wherein preferably the liquid droplet size injected by the liquid droplet injection device is in the range of 2 to 40  $\mu\text{m}$ , preferably around 10  $\mu\text{m}$ . Preferably, the spacing of the fogging water ducts as well as the spacing of the fogging nozzles mounted on said fogging water ducts is adapted to the flow of intake air to achieve even droplet distribution in the stream of intake air. Preferentially the nozzles are binary nozzles fed with gas or quasi-gas and liquid. According to yet another preferred embodiment of the present invention, the liquid droplet injection device is located even closer to the compressor inlet, preferably at the compressor bellmouth. This minimization of the distance of the position of droplet injection to the compressor inlet is advantageous, since it widely prevents secondary droplet formation, growing size of the injected droplets due to conglomeration, water loss on the walls or other fixed equipment in the intake manifold as well as centrifugal effects due to deflection of the inlet air stream.

As concerns the cooling system, according to another preferred embodiment, the cooled cooling air is controlled to have a temperature around 300 and 400 degree Celsius, preferably between 330 and 380 degree Celsius allowing a tolerance of less than +/-10 degree Celsius, wherein a pressure in the range of 15 to 40 bar, preferably in the range of

20 to 30 bar is maintained. Even for cooling systems with such small tolerances, the switching on/off of the over-fogging according to the present invention is possible. Further preferred embodiments of the gas turbine unit according to the present invention are described in the dependent claims.

Additionally, the present invention concerns a process for achieving an increase or decrease in shaft power production from a gas turbine unit, which gas turbine unit essentially comprises: a compressor for compressing intake air supplied thereto and discharging the compressed air; a high-pressure combustion chamber in which fuel is combusted with the compressed air discharged from said compressor; a high-pressure turbine driven by the hot combustion air discharged from said high-pressure combustion chamber; a low-pressure combustion chamber in which fuel is combusted with the expanded combustion air discharged from said high-pressure turbine; and a low-pressure turbine driven by the hot combustion air discharged from said low-pressure combustion chamber. In accordance with the above gas turbine unit according to the invention, said process is characterised in that said compressor is working with a pressure ratio larger than 15 bar (i.e. the process is aiming mainly at, but not limited to large gas turbine units, and preferentially the pressure ratio is even 20 bar), and that by means of at least one liquid droplet injection device provided on the upstream side of said compressor liquid is injected into the stream of intake air in order to increase the shaft power generated by the gas turbine unit, wherein said injection is carried out by means of an immediate and stepless addition or reduction of liquid mass flow in the form of liquid droplets corresponding to the desired increase or decrease respectively of power of the gas turbine unit, and wherein preferentially said water mass flow is being injected substantially across the entire cross-section of the intake air guide. As pointed out above, surprisingly, immediate and complete switching-on or switching-off of the liquid droplet injection device is possible without waiting for steady state conditions in the gas turbine by using slowly increasing or decreasing amounts of liquid. Provided that the droplets are added to the stream of intake air substantially across the entire cross-section of the intake air guide, the switching-on/off can even more easily basically be carried out in an on/off-manner. According to the first preferred embodiment of the process according to the present invention, the gas turbine unit additionally comprises a cooling system which uses compressed or partially compressed air discharged from the compressor for cooling components of the gas turbine unit, wherein the cooling system comprises a cooling unit which is being controlled depending on the quality of the cooled cooling air. Cooling systems of large gas turbine units are highly sophisticated devices which necessitate very

accurate control in order to ensure that constant conditions can be maintained at the locations where the cooling medium is used. Accordingly, if the liquid droplet injection device is switched on using an incremental scheme, also the cooling device has to be controlled in accordance with the incrementally changing conditions in the compressor where the cooling air is branched off from. Preferably said process is being carried out using a gas turbine unit as it has been described above.

According to another preferred embodiment of said process, between 0.5 and 5 mass.%, preferably between 1.0 and 3.0 mass % of water are injected into the intake air to achieve an increase of power of the gas turbine unit of e.g. up to 10%. Preferably, water is injected at a pressure of up to 250 bar, preferentially between 100 to 180 bar, and even more preferably at a pressure of around 140 bar water pressure, and this pressure is built up immediately within 1 s and 60 s, preferably within in the range of 10 – 30 s.

For starting up of the gas turbine unit, it is proposed to in a first step start up the gas turbine to full load, in case of additional cooling systems to switch on these cooling systems, and to, after the expiration of a delay of in the range of 15 to 45 minutes, preferably in the range of 30 minutes, add water by means of the liquid droplet injection device.

Further preferred embodiments related to the process for achieving an increase in power production from a gas turbine unit are described in the dependent claims.

The present invention additionally comprises uses of the above-mentioned gas turbine unit in the field of electricity generation and in the field of production of mechanical power generation.

#### BRIEF DESCRIPTION OF THE DRAWINGS

The invention, its nature as well as its advantages, shall be described in more detail below with the aid of the accompanying drawings. Referring to the drawings:

Figure 1 is a schematic view of a gas turbine unit with sequential combustion chambers and a heat recovery steam generator, showing the location of the liquid droplet injection device as well as the possible cooling ducts;

Figure 2 a/b is a schematic view of the intake region of a gas turbine unit;

Figure 3 is a detailed axial cut (upper half) through the intake region of a gas turbine unit with sequential combustion showing both combustion chambers and both turbine stages;

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Figure 4 is a drawing of the intake manifold of a gas turbine unit with a liquid droplet injection device installed in the expansion joint as according to the present invention; and

Figure 5 is a view of a liquid droplet injection device as it is being used according to the present invention.

#### DETAILED DESCRIPTION OF THE DRAWINGS AND THE INVENTION

Figure 1 shows a schematic view of a gas turbine unit with sequential combustion chambers. Intake air 10 enters the compressor 1/2, which in this particular case is divided into a first stage 1 operating at low-pressure and a second stage 2 operating at high-pressure. The partially compressed intake air 11 exits the first stage 1 to be fed into the second, high-pressure stage 2. The partially compressed air 11 can be cooled prior to entering the high-pressure stage 2 by cooling means 21 in order to increase the power of the gas turbine unit (so-called inter-cooling). After full compression of the intake air, the compressed air 12h enters the high-pressure combustion chamber 8h. This high-pressure combustion chamber 8h is fuelled by a duct 9, and the hot air 13h discharged from the high-pressure combustion chamber 8h is fed into the first, high-pressure turbine stage 3h. After partial expansion of the combustion air 13h, this air 12l enters the second, low-pressure combustion chamber 8l, where it is again heated and then discharged to the second, low-pressure turbine 3l. The exhaust air 14 discharged by the low-pressure turbine 3l can then either be discharged directly via a chimney to the surrounding or it can, as displayed in figure 1, be used in a heat recovery steam generator 15 to generate steam for a steam turbine 4 and then be discharged via a chimney 16. In the latter case water 17 is fed into a duct system in the heat recovery steam generator (steam drums, evaporators, economizers etc) for the generation of steam 18 to be fed into the steam turbine 4. The steam exiting the steam turbine 4 is then either discharged to the surrounding or can be recycled by means of a condenser to re-enter the heat recovery steam generator 15.

Figure 1 displays a so-called single shaft gas turbine, i.e. a gas turbine where compressor 1/2, gas turbine(s) 3h/3l as well as a generator 5 and a steam turbine 4 are mounted on one single shaft 6. Possibly, clutches allow the de-coupling of some of these units from each other.

Also displayed in this schematic diagram are the cooling means for cooling components of the gas turbine unit. Usually, partially or fully compressed cooling air 22 is branched off from either a stage of the compressor within the compressor or at the end of the

compressor. As the compression process in the compressor results in a substantial heating of the compressed air, this air has to be, for cooling purposes, cooled down to the desired temperature levels. To this end, a cooling unit 24 is provided in the cooling air ducts upstream of the locations where the cooling air 25 is to be injected. The cooling unit 24 can have various designs, like for example conventional heat exchangers. In order to avoid the deposition and/or condensation of liquid (particularly critical if heavy over-fogging is being applied upstream of the compressor) in the cooling air ducts, a droplet separator 23 (e.g. centrifugal separator) may be provided in the ducts preferably upstream of the cooling unit 24, but also at any other location where the deposition of liquid shall be particularly prevented. The cooling air can be used at various positions as displayed in Figure 1, e.g. for cooling the combustion chambers 8h and 8l, for cooling the turbines 3h and 3l as well as 4 etc. If need be, the cooling air can be mixed with steam to enhance the cooling effect.

In order for the cooling system to work properly, the cooling system has to be balanced such as to meet certain conditions. First the pressure in the cooling ducts has to be larger than the pressure at the location where the cooling air is to be injected to avoid flow-back of the cooling air into the cooling ducts. That is the reason why usually cooling air is not just taken from the surrounding but rather branched off from the compressor where high pressure is already available (this, as mentioned above, with the inherent drawback that the air is not available at ambient temperature due to the compression process).

Secondly the temperature of the cooling air may not exceed a certain level in order to allow effective cooling of the components where the cooling air is injected. Usually, the tolerances as concerns temperature as well as pressure in these cooling systems are very low necessitating sophisticated control mechanisms to stabilize and balance temperature as well as pressure of the cooling air.

Usually the cooling air has a temperature of 300 to 600 degrees Celsius and a pressure in the range of 10 to 30 bar at the point of discharge from the compressor. Typical are pairs of 400 degrees Celsius at 20 bar or 500 degrees Celsius at 30 bar (to compare: the compressed air 12h fed into the high-pressure combustion chamber usually has a temperature of about 500 degrees Celsius). The cooling air downstream of the cooling unit 24 shall have a temperature in the range of 300 to 400 degrees Celsius, preferably temperatures in the range of 330 and 380 degrees Celsius. The tolerances related to temperature are usually required to be less than +/-10 degrees Celsius to make sure the cooling system works properly. The pressure of the cooling air stream shall be in the range of 15 to 40 bar, preferably in the range of 20 to 30 bar. Tolerances related to

pressure are usually required to be less than +/- 0.5 bar. In order to fulfil these pressure and temperature requirements, the cooling unit 24 is controlled depending on measurements of temperature and pressure at various locations (see below).

Figure 2a/b shows the intake region of a gas turbine unit. Usually intake air 10 is aspirated by a duct with large cross-section, which at the entrance is provided with a filter 28 to avoid deleterious entrance of dust and other particles. Right downstream of said filter a first evaporative air inlet cooling unit 29/33 is provided meaning any system adding liquid to the air inlet mass flow, which substantially evaporates before entering the compressor of the gas turbine unit, resulting in a cooling effect and in an increase of the air intake mass flow into the gas turbine unit. This may for example be in the form of a liquid atomisation spray system ('fogging' system) 29 in figure 2a or in the form of a porous medium (evaporative cooler) 33 in figure 2b. The respective evaporative cooling system shall increase the humidity of the intake air as close to saturation as possible.

Downstream of an evaporative cooler 33 a droplet catcher 34 preferably is provided to prevent excessively large droplets from entering the compressor of the gas turbine unit. It shall be mentioned that the injection spray device 29 can also be used to inject more liquid than needed for saturating the intake air mass flow to achieve the same effect as with using the over-fogging means 32 described below. Subsequently, the cross-section of the intake duct 27 is reduced thereby increasing the flow velocity of the intake air 10. Usually downstream of this reduction of cross-section there is a silencer installed in the intake duct. Downstream of the silencer the intake duct 27 is connected with the intake manifold 26 by means of an expansion joint 31. The intake manifold 26 is fixedly connected with the gas turbine unit and is therefore subject to vibrations of the gas turbine. By contrast, the intake duct 27 is directly connected to the ground and shall be affected by the gas turbine's vibrations as little as possible, which is achieved by providing an expansion joint 31 between the air intake manifold and the air intake duct. The location of the expansion joint 31 has been found to be quite a favourable position for placing the liquid droplet injection device 32a, but is not the exclusive location for the liquid droplet injection device (see below). Downstream of the liquid droplet injection device 32a, the intake manifold 26, which guides the intake air into the compressor as smoothly as possible, is connected to the entrance of the compressor 1/2, which in this case is a compressor without inter-cooling. Alternative locations for the liquid droplet injection device are e.g. at the compressor bellmouth as close to the compressor entrance as possible (32b) or behind the silencer (32c).

Both elements 29 and 32a/b/c preferably inject demineralized water in order to avoid depositions in the gas turbine unit. Typically 1.0 to 3.0 mass-% of water are added to the stream of intake air, and a water pressure of >100 bar is used.

To visualise in more detail the conditions in the intake region of a gas turbine unit, figure 3 shows an axial cut through a gas turbine unit and in particular through its intake manifold 26. It can be seen that the liquid droplet injection device 32a is arranged substantially perpendicular to the flow of intake air 10 at the interface between the intake duct 27 and the intake manifold 26. Preferably, the liquid droplet injection device 32a is positioned on the intake duct side of the expansion joint 31. This positioning proves to be advantageous as it shows minimum exposure to vibration, low pressure loss of the equipment as well as even droplet distribution with small droplet sizes. The intake air is over-saturated by the liquid droplet injection device 32a, is then deviated by a particular, flow adapted housing (the intake manifold, figure 4) into the compressor 1/2, where the air is compressed in several steps but without inter-cooling. The temperature profile in the casing of the gas turbine unit should be monitored to avoid damages in case of irregularities.

Figure 5 shows a nozzle rack as it can be used as the liquid droplet injection device 32a. The rack consists of a carrying frame 37 onto which liquid ducts 35 are mounted, typically with a variable spacing in the range of approximately 20-30 cm. On these water ducts 35, liquid atomisation nozzles 36 are mounted which allow a high flow capacity, good droplet cone and small droplet sizes thereby reducing blading erosion rates. Usually 5-15 nozzles are mounted on one water duct. As one can see from figure 5, the spacing of the nozzles as well as of the pipes is not necessarily regular. The positioning of the water ducts 35 as well as of the liquid atomisation nozzles 36 should be in a flow weighted manner to provide even droplet distribution. For example in the case of an intake duct 27 and intake manifold 26 as displayed in figure 2a/b, where the intake air is diverted almost in a rectangular angle from a horizontal direction to a vertical direction in order to enter the intake manifold 26, the maximum of flow velocity in the region of the expansion joint measured substantially perpendicular to the flow of intake air is located closer to the outer side of the bend, with the velocity maximum usually located within the third proximal to the outer side of the bend. Usually the nozzle density can be set substantially proportional to the velocity at a particular position in order to achieve homogeneous distribution of fine droplets across the cross-section of the intake air stream. The above applies in the case where all the nozzles are identically fed with water so that when the nozzles are distributed according to the flow of the air, more water is injected in regions where there is a high flow velocity and where there is a higher density of nozzles. It is generally easier to

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achieve homogeneously high droplet quality if all nozzles are fed identically. However, the same effect can be achieved with evenly distributed nozzles and feeding nozzles, that are located in regions of high flow velocity, with more water. Also a combination of flow-adapted distribution of the nozzles with individual supply of the nozzles is possible and can be advantageous if the flow conditions in the duct vary significantly depending on the mode of operation of the gas turbine unit. The nozzles may be designed as binary nozzles fed with gas or quasi-gas and with liquid to provide good droplet quality.

As mentioned above, the over-fogging system can be combined with conventional evaporative air inlet cooling units as for example displayed by means 29 in figure 2a or 33 in figure 2b. The conditions for start-up of the over-fogging pumps are as follows: The gas turbine unit should be at or near full load and the ambient wet bulb temperature should be above a certain value, typically above 0° Celsius. If conventional evaporative inlet air cooling equipment is installed, the inlet air cooling devices should be operating when the over-fogging is being activated if ambient conditions allow. Additionally, the full or near full load operation of the gas turbine unit should have been active for a certain warm-up period, which should be in the range of preferably about 30 minutes. The compressor inlet temperature has to be monitored. If this value drops below a certain level, usually given by about 0° Celsius, an automatic shutdown of the liquid injection system is necessary. Also the over-fogging flow capacity has to be supervised to detect e.g. nozzle clogging, change in the nozzle flow capacity, leaks, differences between measured system water flow capacity etc., which would lead to uncontrolled water flows which should be avoided. As mentioned above, also the quality of feed water should be supervised. Generally this quality is monitored by measuring the conductivity which should not exceed a certain limit value. In addition, any freezing or blocking of the pipes system also should be monitored.

For plants with inlet cooling systems the following start-up scheme proves to be suitable:

1. Gas turbine unit is loaded up to or near to full load.
2. If the gas turbine unit is at or near full load operation, evaporative inlet cooling systems come into operation if the ambient conditions allow switching on of the inlet cooling system.
3. After operating at or near full load for at least approximately 30 minutes, the over-fogging system can come into operation.

In case a plant is not equipped with an evaporative inlet cooling systems, the above step 2 can be skipped.

For shutdown, the above procedure can be carried out in the reverse order.

## LIST OF REFERENCE NUMERALS

- 1 first stage of compressor (low-pressure)
- 2 second stage of compressor (high-pressure)
- 3h high-pressure turbine
- 3l low-pressure turbine
- 4 steam turbine
- 5 generator
- 6 common shaft
- 7 clutch
- 8h high-pressure combustion chamber
- 8l low-pressure combustion chamber
- 9 fuel duct, fuel
- 10 intake air
- 11 partially compressed intake air
- 12h compressed air
- 12l expanded combustion air from high-pressure combustion chamber
- 13h hot combustion air (high-pressure)
- 13l hot combustion air (low-pressure)
- 14 exhaust air
- 15 heat recovery steam generator
- 16 chimney
- 17 duct to the heat recovery steam generator
- 18 duct from the heat recovery steam generator (steam)
- 19 outlet of the steam turbine
- 20 cooling/fogging of intake air
- 21 cooling of partially compressed air
- 22 cooling air duct, compressed cooling air
- 23 droplet separator
- 24 cooling unit

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- 25 cooled cooling air
- 26 intake manifold
- 27 intake duct
- 28 filter
- 29 fogging liquid droplet injection device
- 31 expansion joint
- 32a/b/c over-fogging liquid droplet injection device
- 33 evaporative cooler
- 34 droplet catcher
- 35 water duct
- 36 atomisation spray nozzle
- 37 carrying frame

## CLAIMS

1. A gas turbine unit, essentially comprising: a compressor (1,2) for compressing intake air (10) supplied thereto and discharging the compressed air (12); a high-pressure combustion chamber (8h) in which fuel (9) is combusted with the compressed air (12) discharged from said compressor (1,2); a high-pressure turbine (3h) driven by the hot combustion air (13h) discharged from said high-pressure combustion chamber (8h); a low-pressure combustion chamber (8l) in which fuel (9) is combusted with the expanded combustion air (12l) discharged from said high-pressure turbine (3h); and a low-pressure turbine (3l) driven by the hot combustion air (13l) discharged from said low-pressure combustion chamber (8l),  
characterised in that  
said compressor (1,2) is working with a pressure ratio larger than 15 bar, and  
that at least one liquid droplet injection device (32) is provided on the upstream side of said compressor (1,2) for injecting liquid into the stream of intake air (10) in order to increase the shaft power generated by the gas turbine unit, said injection device (32) allowing the addition of liquid mass flow in the form of liquid droplets corresponding to a desired increase of shaft power of the gas turbine unit,  
wherein the amount of liquid mass flow corresponding to the desired increase of shaft power output of the gas turbine unit is added in the form of liquid droplets in a substantially stepless manner and immediately within a time interval that is substantially determined by the design characteristics of the liquid droplet injection device (32) to increase the liquid droplet mass flow.

2. A gas turbine unit, essentially comprising: a compressor (1,2) for compressing intake air (10) supplied thereto and discharging the compressed air (12); a high-pressure combustion chamber (8h) in which fuel (9) is combusted with the compressed air (12) discharged from said compressor (1,2); a high-pressure turbine (3h) driven by the hot combustion air (13h) discharged from said high-pressure combustion chamber (8h); a low-pressure combustion chamber (8l) in which fuel (9) is combusted with the expanded combustion air (12l) discharged from said high-pressure turbine (3h); and a low-pressure turbine (3l) driven by the hot combustion air (13l) discharged from said low-pressure combustion chamber (8l),  
characterised in that  
said compressor (1,2) is working with a pressure ratio larger than 15 bar, and  
that at least one liquid droplet injection device (32) is provided on the upstream side of said compressor (1,2) injecting liquid into the stream of intake air (10) to increase the shaft power generated by the gas turbine unit, said injection device (32) adding liquid mass flow in the form of liquid droplets,  
wherein the amount of liquid mass flow corresponding to a desired decrease of shaft power output of the gas turbine unit is reduced in a substantially stepless manner and immediately within a time interval that is substantially determined by the design characteristics of the liquid droplet injection device (32) to decrease the liquid droplet mass flow.
3. A gas turbine unit according to one of the preceding claims, characterised in that said liquid mass flow is substantially injected across the entire cross-section of the intake air guide (26, 27).
4. A gas turbine unit according to one of the preceding claims, characterised in that it additionally comprises a cooling system (22-25) which uses compressed or partially compressed air (22) discharged from the compressor (1,2) for cooling components of the gas turbine unit, wherein the cooling system (22 - 25) comprises a cooling unit (24) which is controlled such as to ensure constant quality of the cooled cooling air (25).
5. A gas turbine unit according to claim 4, characterised in that the at least one injection device (29, 32) can be controlled by a single feed forward signal

synchronised to the on/off-control of the liquid droplet injection device (32), wherein preferentially the signals of the two systems can be slightly displaced relative to each other in order to take account of hysteresis-effects of the gas turbine unit.

6. A gas turbine unit according to one of the preceding claims, characterised in that it comprises a fuel control valve that adjusts the fuel mass flow in order to maintain the desired firing temperatures of the gas turbine unit, wherein preferentially the control of said fuel valve is carried out by a single feed forward signal synchronised to the on/off-control of the liquid droplet injection device (32).
7. A gas turbine unit according to one of the preceding claims, characterised in that it comprises the ability to rapidly increase or decrease the shaft power output available via a generator (5) of the unit using the immediate and stepless increase or decrease of liquid mass flow.
8. A gas turbine unit according to claim 7, characterised in that the rapid increase or decrease can be generated in response to a fast increase or decrease in power demand of the electricity grid in particular in response to a drop of the frequency of the electricity grid or in response to an rise of the frequency of the electricity grid, respectively.
9. A gas turbine unit according to claim 8, characterised in that the drop or rise of the frequency of the electricity grid results in a reduction or an increase of the shaft speed of the gas turbine unit and that by immediate and stepless increase or decrease of the amount of liquid, the surge margin of the compressor of the gas turbine unit is increased or decreased by cooling the compressor blade section.
10. A gas turbine unit according to one of the claims 1 to 9, characterised in that it additionally comprises an intake manifold (26) situated upstream of said compressor (1,2) and an intake duct (27) situated upstream of said intake manifold (26) connected to said intake manifold (26) by means of an expansion joint (32), and that the liquid droplet injection device (32) is essentially situated at the expansion joint (31) between the intake duct (27) and the intake manifold (26), wherein preferably the intake duct (27) additionally comprises a silencer located

upstream of said liquid droplet injection device (32) and a filter (28) located essentially at the intake opening of the intake duct (27), and wherein preferably additional cooling means (29, 33) for cooling the intake air (10) are situated downstream of the filter (28).

11. A gas turbine unit according to one of the claims 1 to 10, characterised in that the liquid droplet injection device (32) consists of a grid (32) of fogging water ducts (35), preferably arranged in an essentially parallel manner on a carrying rack (37), on the downstream side of which fogging water ducts (35) fogging nozzles (36) are mounted for injecting droplets into the stream of intake air (10), wherein preferably the liquid droplet size injected by the liquid droplet injection device (32) is in the range of 2 to 40  $\mu\text{m}$ , preferably around 10  $\mu\text{m}$ .
12. A gas turbine unit according to claim 11, characterised in that the fogging nozzles (36) are binary nozzles fed with gas or quasi-gas and with liquid.
13. A gas turbine unit according to one of the claims 11 or 12, characterised in that the spacing of the fogging water ducts (35) as well as the spacing of the fogging nozzles (36) mounted on said fogging water ducts (35) is adapted to the flow of intake air (10) to achieve even droplet distribution in the stream of intake air (10).
14. A gas turbine unit according to one of the preceding claims, characterised in that the liquid droplet injection device is located close to the compressor inlet, preferably at the compressor bellmouth.
15. A gas turbine unit according to claim 4 and one of the claims 1 to 14, characterised in that the cooled cooling air (25) is being controlled to have a temperature around 300 and 400 degree Celsius, preferably between 330 and 380 degree Celsius allowing a tolerance of less than +/-10 degree Celsius, wherein a pressure in the range of 15 to 40 bar, preferably in the range of 20 to 30 bar is maintained.

16. Process for achieving an increase in shaft power production from a gas turbine unit, which gas turbine unit essentially comprises: a compressor (1,2) for compressing intake air (10) supplied thereto and discharging the compressed air (12); a high-pressure combustion chamber (8h) in which fuel (9) is combusted with the compressed air (12) discharged from said compressor (1,2); a high-pressure turbine (3h) driven by the hot combustion air (13h) discharged from said high-pressure combustion chamber (8h); a low-pressure combustion chamber (8l) in which fuel (9) is combusted with the expanded combustion air (12l) discharged from said high-pressure turbine (3h); and a low-pressure turbine (3l) driven by the hot combustion air (13l) discharged from said low-pressure combustion chamber (8l),

characterised in that

said compressor (1,2) is working with a pressure ratio larger than 15 bar, and that by means of at least one liquid droplet injection device (32) provided on the upstream side of said compressor (1,2) liquid is injected into the stream of intake air (10) in order to increase the shaft power generated by the gas turbine unit, wherein the amount of liquid mass flow corresponding to a desired increase of shaft power output of the gas turbine unit is added in the form of liquid droplets in a substantially stepless manner and immediately within a time interval that is substantially determined by the design characteristics of the liquid droplet injection device (32) to increase the liquid droplet mass flow.

17. Process for achieving a decrease in shaft power production from a gas turbine unit, which gas turbine unit essentially comprises: a compressor (1,2) for compressing intake air (10) supplied thereto and discharging the compressed air (12); a high-pressure combustion chamber (8h) in which fuel (9) is combusted with the compressed air (12) discharged from said compressor (1,2); a high-pressure turbine (3h) driven by the hot combustion air (13h) discharged from said high-pressure combustion chamber (8h); a low-pressure combustion chamber (8l) in which fuel (9) is combusted with the expanded combustion air (12l) discharged from said high-pressure turbine (3h); and a low-pressure turbine (3l) driven by the hot combustion air (13l) discharged from said low-pressure combustion chamber (8l),  
characterised in that  
said compressor (1,2) is working with a pressure ratio larger than 15 bar, and  
that at least one liquid droplet injection device (32) is provided on the upstream side of said compressor (1,2) injecting liquid into the stream of intake air (10) to increase the shaft power generated by the gas turbine unit, said injection device (32) adding liquid mass flow in the form of liquid droplets,  
wherein the amount of liquid mass flow corresponding to a desired decrease of shaft power output of the gas turbine unit is reduced in a substantially stepless manner and immediately within a time interval that is substantially determined by the design characteristics of the liquid droplet injection device (32) to decrease the liquid droplet mass flow.
18. Process according to one of the claims 16 or 17, characterised in that said liquid mass flow is substantially injected across the entire cross-section of the intake air guide (26, 27).
19. Process according to one of the claims 16 to 18, characterised in that the gas turbine unit additionally comprises a cooling system (22-25) which uses compressed or partially compressed air (22) discharged from the compressor (1,2) for cooling components of the gas turbine unit, wherein the cooling system (22 - 25) comprises a cooling unit (24) which is being controlled depending on the quality of the cooled cooling air (25).

20. Process according to one of the claims 16 to 19 characterised in that it is being carried out using a gas turbine unit according to one of the claims 1 to 15.
21. Process according to one of the claims 16 to 20, characterised in that between 0.5 and 5 mass%, preferably between 1.0 and 3.0 mass% of liquid are injected into the intake air (10) to achieve an increase of shaft power of the gas turbine unit of up to 10%, and that the same amounts are reduced for a reduction of the shaft power.
22. Process according to one of the claims 16 to 21, characterised in that water is injected at pressure of up to 250 bar, preferably of between 100 to 180 bar, and most preferably at a pressure of around 140 bar water pressure, and that this pressure is built up immediately within 1s and 60s, preferably within in the range of 10 – 30 s.
23. Process according to one of the claims 16 to 22, characterised in that the power output of the unit available on a generator (5) of the unit is rapidly increased or decreased using the immediate and stepless addition or reduction of liquid mass flow.
24. Process according to claim 23, characterised in that the rapid increase or decrease is generated in response to a fast increase or decrease in power demand of the electricity grid in particular in response to a drop or rise of the frequency of the electricity grid, and that more particularly the drop or rise of the frequency of the electricity grid results in a reduction or augmentation of the shaft speed of the gas turbine unit, respectively and that by immediate and stepless addition or reduction of water, the surge margin of the compressor of the gas turbine unit is increased or decreased by cooling the compressor blade section.
25. Process according to one of the claims 16 to 22, characterised in that for start-up of the gas turbine unit in a first step the gas turbine is started up to full load, in case of additional cooling systems (29, 33) these cooling systems are switched on, and that after a delay of in the range of 15 to 45 minutes, preferably in the range of 30 minutes, water is added by means of the liquid droplet injection device (32).

26. Use of a gas turbine unit according to one of the claims 1 to 16 for the production of electrical power, for the production of mechanical power or in the field of propulsion of aircrafts.

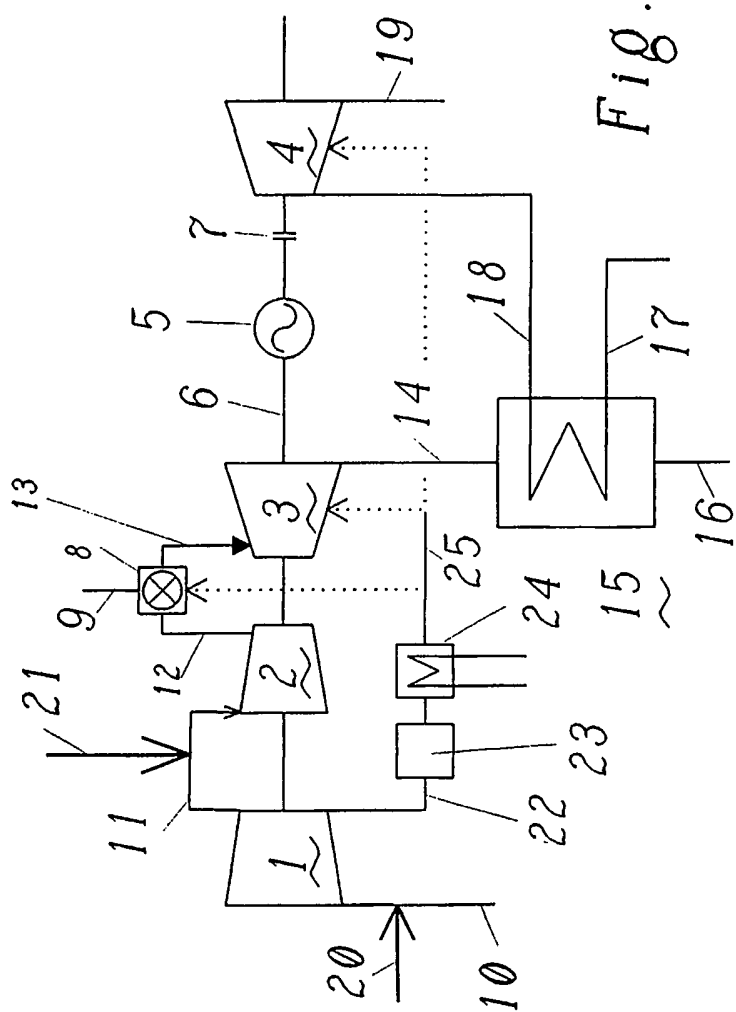


Fig. 1

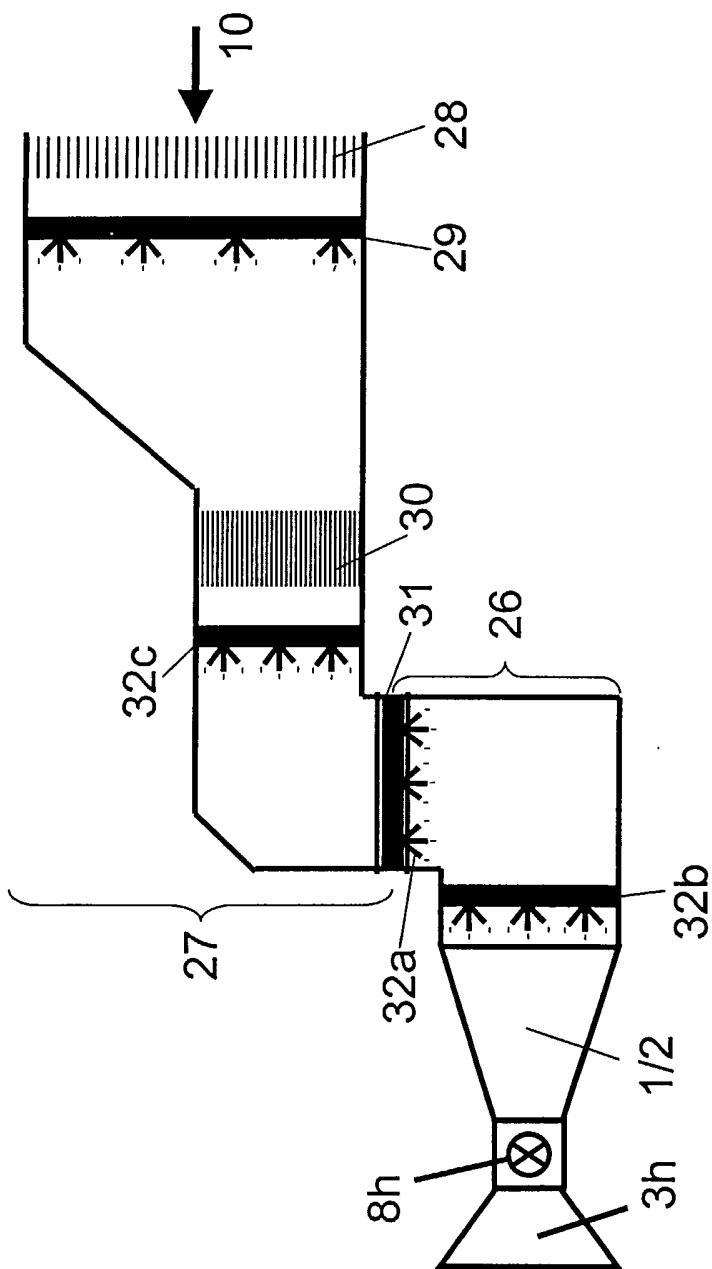


Fig. 2a

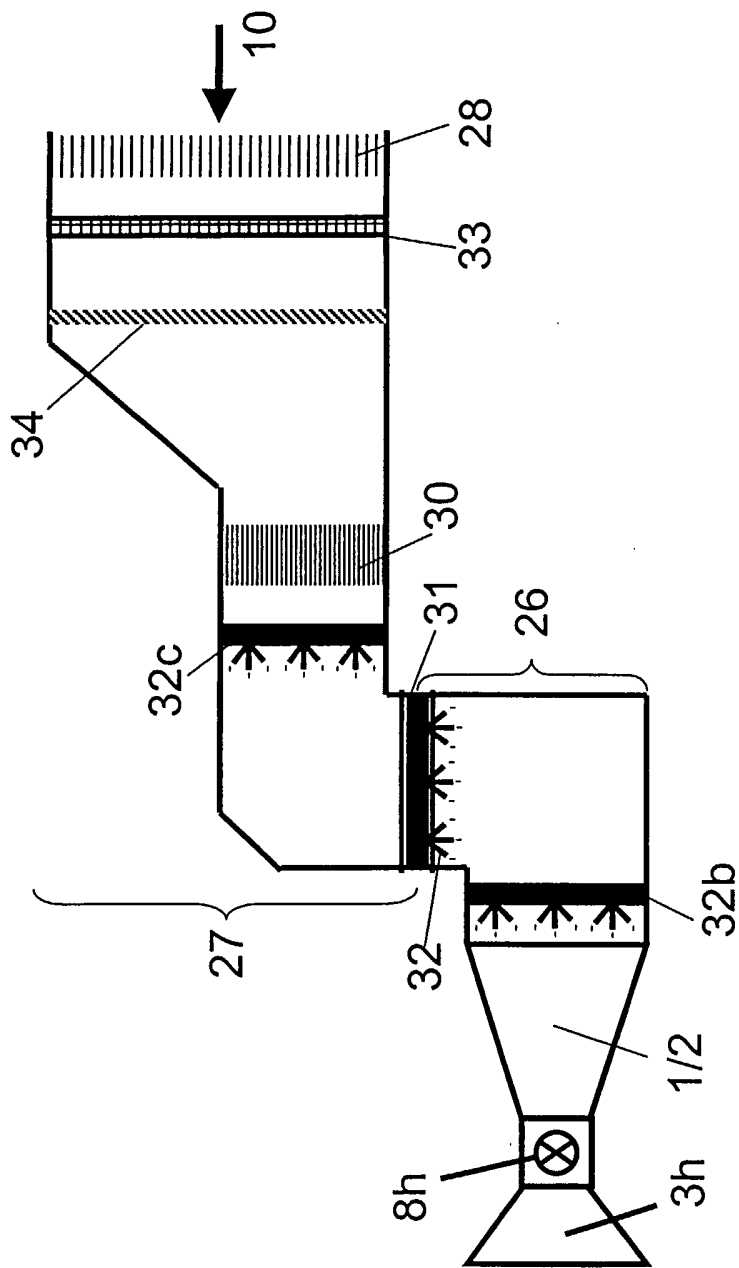


Fig. 2b

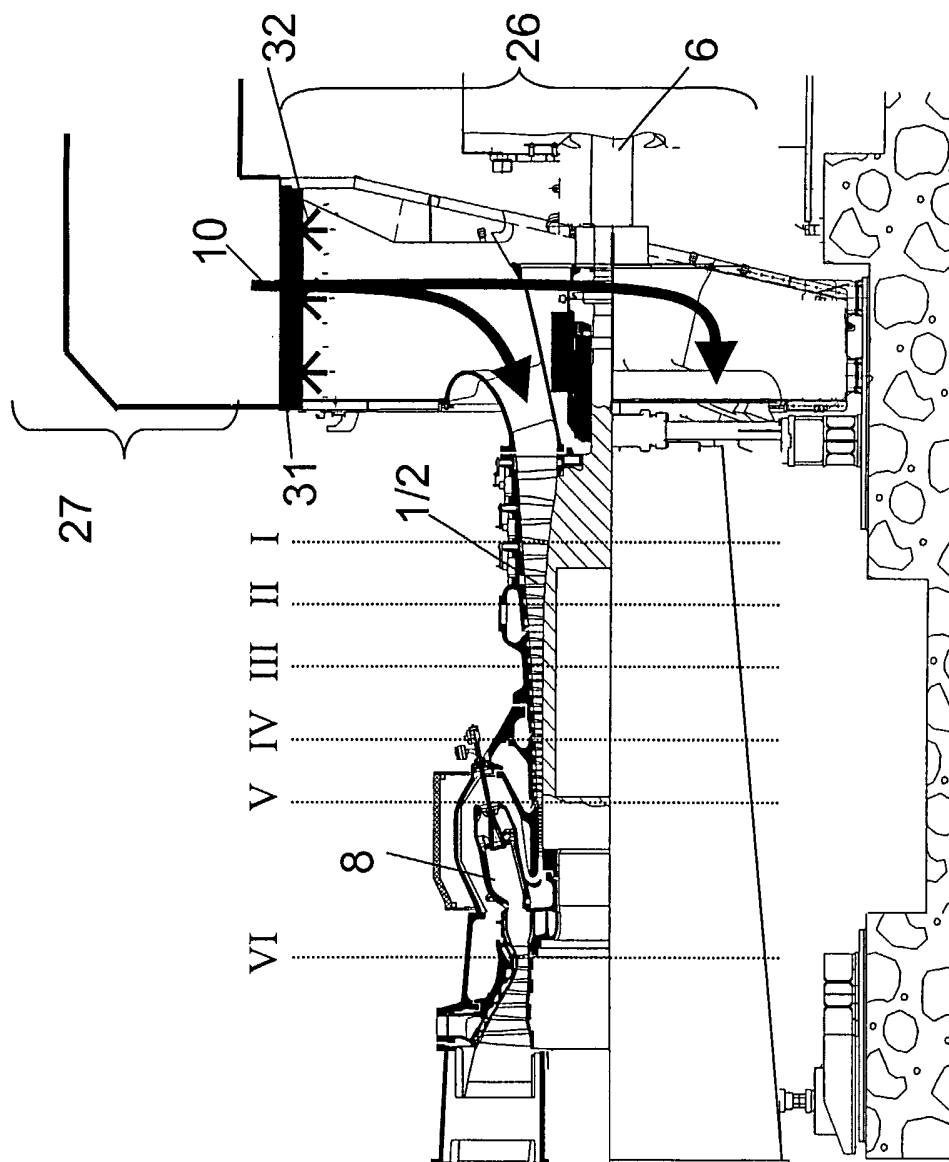


Fig. 3

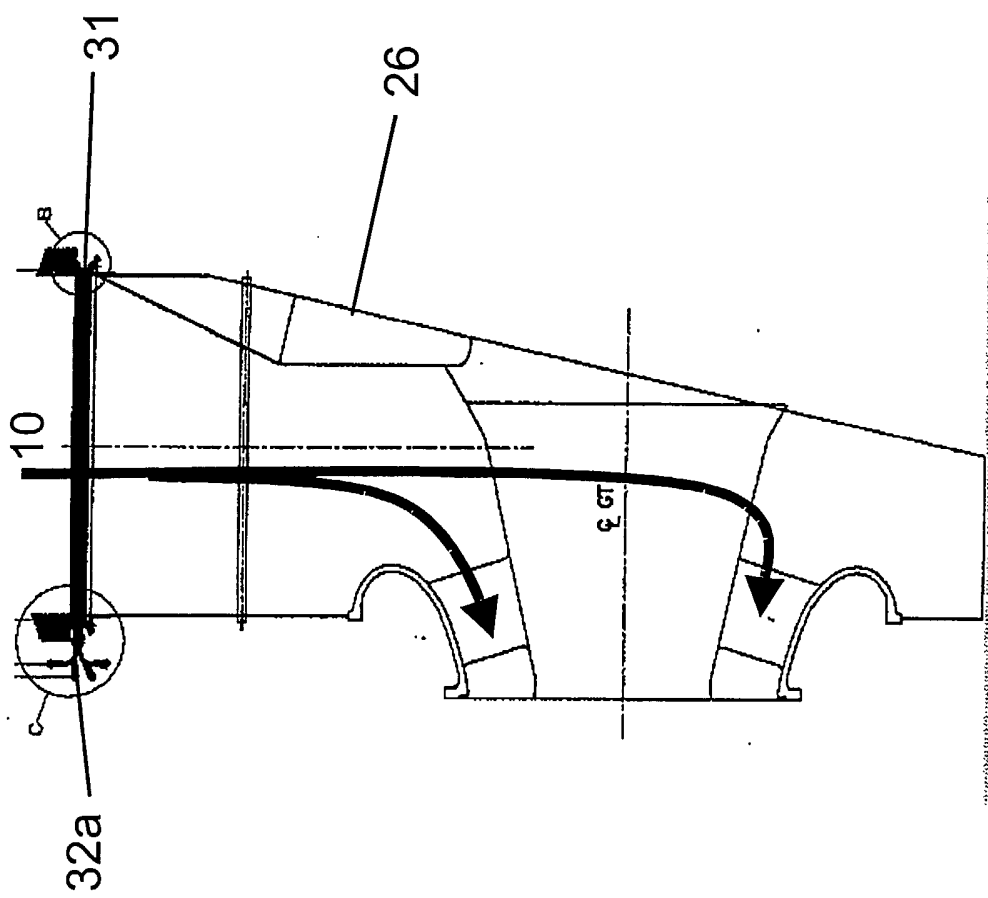


Fig. 4

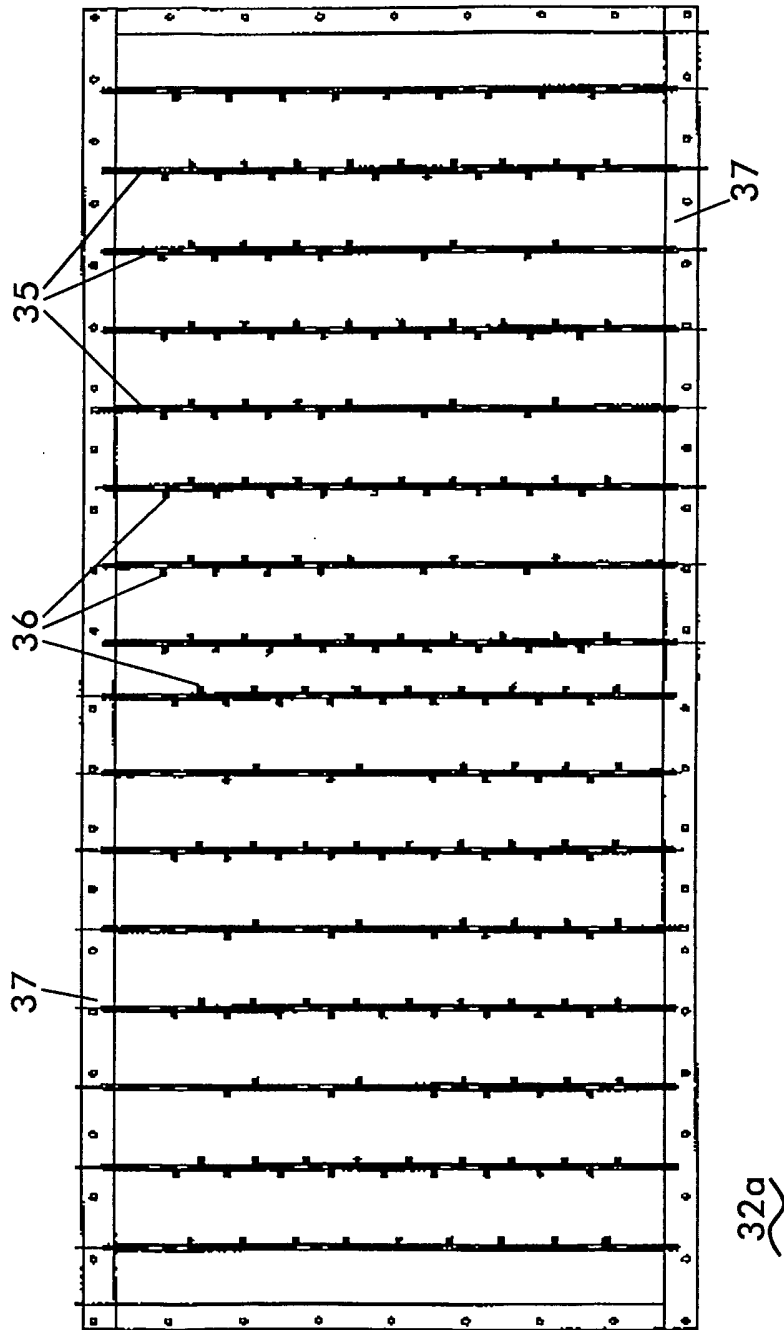


Fig. 5

## INTERNATIONAL SEARCH REPORT

Intern Application No

PCT/CH 02/00663

**A. CLASSIFICATION OF SUBJECT MATTER**  
 IPC 7 F02C7/143 F02C3/30

According to International Patent Classification (IPC) or to both national classification and IPC

**B. FIELDS SEARCHED**

Minimum documentation searched (classification system followed by classification symbols)

IPC 7 F02C F01K

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

Electronic data base consulted during the international search (name of data base and, where practical, search terms used)

EPO-Internal, WPI Data, PAJ

**C. DOCUMENTS CONSIDERED TO BE RELEVANT**

Category °	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
X	US 6 260 350 B1 (SASADA TETSUO ET AL) 17 July 2001 (2001-07-17)	1-3, 16-18, 20,23, 25,26
Y	the whole document	3,6-10, 12,13,24
X	EP 0 781 909 A (HITACHI LTD) 2 July 1997 (1997-07-02)	1-3, 16-18, 20,21,26
Y	the whole document	6,11,12, 14
X	DE 198 52 060 A (STEAG AG) 25 May 2000 (2000-05-25)	1-3, 16-18, 23,26
Y	the whole document	7,8,11, 22
	-/--	

Further documents are listed in the continuation of box C.

Patent family members are listed in annex.

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Date of the actual completion of the international search

30 April 2003

Date of mailing of the international search report

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## INTERNATIONAL SEARCH REPORT

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PCT/CH 02/00663

C.(Continuation) DOCUMENTS CONSIDERED TO BE RELEVANT		
Category °	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
Y	US 5 867 977 A (ZACHARY RICHARD E ET AL) 9 February 1999 (1999-02-09) cited in the application the whole document ---	3,4,13, 14,19
Y	US 5 313 783 A (ALTHAUS ROLF) 24 May 1994 (1994-05-24) the whole document ---	4,19
Y	US 2001/039794 A1 (COLEMAN ROBERT PAUL ET AL) 15 November 2001 (2001-11-15) the whole document ---	5,22
Y	US 6 250 064 B1 (GEORGE ANTHONY JAMES ET AL) 26 June 2001 (2001-06-26) the whole document ---	5,10
Y	EP 0 887 530 A (HITACHI LTD) 30 December 1998 (1998-12-30) the whole document -----	9,24

## INTERNATIONAL SEARCH REPORT

Information on patent family members

Intern Application No  
PCT/CH 02/00663

Patent document cited in search report	Publication date	Patent family member(s)	Publication date	
US 6260350	B1	17-07-2001	EP 0889212 A2	07-01-1999
			JP 11072029 A	16-03-1999
			US 2002139105 A1	03-10-2002
			US 2003014978 A1	23-01-2003
			US 2003019202 A1	30-01-2003
			US 2003014958 A1	23-01-2003
			US 2001022078 A1	20-09-2001
EP 0781909	A	02-07-1997	JP 2877098 B2	31-03-1999
			JP 9236024 A	09-09-1997
			EP 0781909 A2	02-07-1997
			JP 11287132 A	19-10-1999
			JP 2980095 B2	22-11-1999
			JP 10246127 A	14-09-1998
			JP 11190228 A	13-07-1999
			US 6216443 B1	17-04-2001
			US 6286301 B1	11-09-2001
			US 6357236 B1	19-03-2002
			US 2003029157 A1	13-02-2003
			US 6378284 B1	30-04-2002
			US 2002026783 A1	07-03-2002
DE 19852060	A	25-05-2000	DE 19852060 A1	25-05-2000
			WO 0028220 A1	18-05-2000
US 5867977	A	09-02-1999	US 5930990 A	03-08-1999
			AT 225906 T	15-10-2002
			AU 725901 B2	26-10-2000
			AU 3211097 A	05-12-1997
			BR 9709317 A	09-05-2000
			CA 2254922 A1	20-11-1997
			CN 1219217 A	09-06-1999
			DE 69716259 D1	14-11-2002
			DK 898645 T3	10-02-2003
			EP 1108870 A2	20-06-2001
			EP 0898645 A1	03-03-1999
			ES 2181003 T3	16-02-2003
			JP 2000510544 T	15-08-2000
			RU 2178532 C2	20-01-2002
			TW 385352 B	21-03-2000
WO 9743530 A1	20-11-1997			
US 5313783	A	24-05-1994	DE 4210544 A1	07-10-1993
			DE 59305921 D1	30-04-1997
			EP 0563520 A2	06-10-1993
			JP 6010698 A	18-01-1994
US 2001039794	A1	15-11-2001	US 6470667 B1	29-10-2002
			AU 1439700 A	28-02-2000
			EP 1108129 A2	20-06-2001
			NO 20001524 A	24-05-2000
			NO 20010393 A	23-01-2001
			TR 200100194 T2	21-05-2001
			US 2002112465 A1	22-08-2002
			WO 0008327 A2	17-02-2000
			AU 1439600 A	06-03-2000
			AU 1595500 A	28-02-2000
			BR 9906626 A	01-08-2000

## INTERNATIONAL SEARCH REPORT

Information on patent family members

Intern application No

PCT/CH 02/00663

Patent document cited in search report	Publication date	Patent family member(s)	Publication date
US 2001039794	A1	BR 9906627 A	01-08-2000
		BR 9912403 A	24-04-2001
		CA 2303316 A1	24-02-2000
		CA 2306487 A1	17-02-2000
		CA 2336584 A1	17-02-2000
		EP 1062415 A2	27-12-2000
		EP 1040261 A2	04-10-2000
		JP 2002522692 T	23-07-2002
		JP 2002522705 T	23-07-2002
		NO 20001525 A	22-05-2000
		TR 200000777 T1	21-01-2002
		TR 200001047 T1	21-09-2001
		WO 0008326 A2	17-02-2000
		WO 0009875 A2	24-02-2000
		US 6467252 B1	22-10-2002
US 6250064	B1	26-06-2001	JP 2001055929 A
			27-02-2001
EP 0887530	A	30-12-1998	CN 1207453 A
			10-02-1999
			EP 0887530 A2
			30-12-1998
			JP 11072027 A
			16-03-1999
			US 6256976 B1
			10-07-2001
			US 2001000049 A1
			22-03-2001
			US 2002043063 A1
			18-04-2002