

(19) World Intellectual Property Organization
International Bureau(43) International Publication Date
16 July 2009 (16.07.2009)

PCT

(10) International Publication Number
WO 2009/088699 A1(51) International Patent Classification:
B29C 70/34 (2006.01) B29D 31/00 (2006.01)

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(21) International Application Number:
PCT/US2008/087529

(81) Designated States (unless otherwise indicated, for every kind of national protection available): AE, AG, AL, AM, AO, AT, AU, AZ, BA, BB, BG, BH, BR, BW, BY, BZ, CA, CH, CN, CO, CR, CU, CZ, DE, DK, DM, DO, DZ, EC, EE, EG, ES, FI, GB, GD, GE, GH, GM, GT, HN, HR, HU, ID, IL, IN, IS, JP, KE, KG, KM, KN, KP, KR, KZ, LA, LC, LK, LR, LS, LT, LU, LY, MA, MD, ME, MG, MK, MN, MW, MX, MY, MZ, NA, NG, NI, NO, NZ, OM, PG, PH, PL, PT, RO, RS, RU, SC, SD, SE, SG, SK, SL, SM, ST, SV, SY, TJ, TM, TN, TR, TT, TZ, UA, UG, US, UZ, VC, VN, ZA, ZM, ZW.

(22) International Filing Date:
18 December 2008 (18.12.2008)

(84) Designated States (unless otherwise indicated, for every kind of regional protection available): ARIPO (BW, GH, GM, KE, LS, MW, MZ, NA, SD, SL, SZ, TZ, UG, ZM, ZW), Eurasian (AM, AZ, BY, KG, KZ, MD, RU, TJ, TM), European (AT, BE, BG, CH, CY, CZ, DE, DK, EE, ES, FI, FR, GB, GR, HR, HU, IE, IS, IT, LT, LU, LV, MC, MT, NL, NO, PL, PT, RO, SE, SI, SK, TR), OAPI (BF, BJ, CF, CG, CI, CM, GA, GN, GQ, GW, ML, MR, NE, SN, TD, TG).

(25) Filing Language: English

Published:

(26) Publication Language: English

- with international search report
- before the expiration of the time limit for amending the claims and to be republished in the event of receipt of amendments

(30) Priority Data:
11/971,766 9 January 2008 (09.01.2008) US

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(54) Title: CONTOURED COMPOSITE PARTS

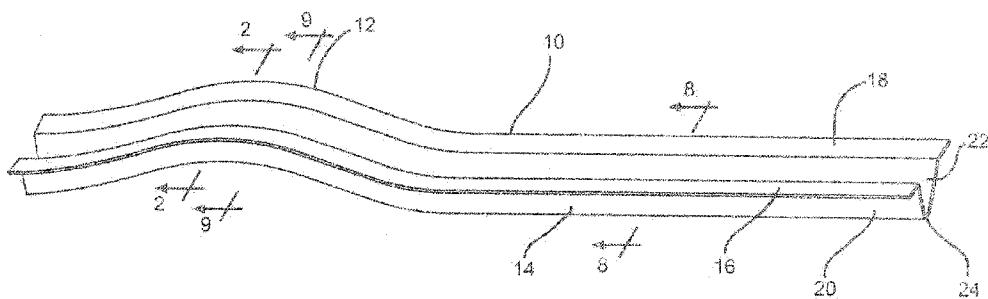


FIG. 1

(57) Abstract: One embodiment of a composite part comprises a curved elongated section comprising a plurality of stacked reinforcement layers. The stacked reinforcement layers in the curved elongated section comprise at least one of non-continuous woven fabric and non-continuous reinforcement.

CONTOURED COMPOSITE PARTS

Conventional stiffeners in aircraft and other devices often comprise hat-section stiffeners which are three flanged beams. However, these beams often may not be placed in highly

contoured areas without wrinkling and/or buckling type issues. Costly repairs are often needed

5 to provide added reinforcement to carry the required loads. Moreover, additional material such as repair plies may be needed which may increase the weight. Further, additional processing, inspection, and/or production steps may be required which may increase both the cost and time of production.

A composite part and/or method of manufacture is needed to decrease one or more problems

10 associated with one or more of the existing composite parts and/or methods of manufacture.

SUMMARY

In one aspect of the disclosure, a composite part comprises a curved elongated section comprising a plurality of stacked reinforcement layers. The stacked reinforcement layers in the

15 curved elongated section comprise at least one of non-continuous woven fabric and non-

continuous reinforcement.

In another aspect of the disclosure, a method of manufacturing a composite part is disclosed. In one step, an elongated composite part is provided. The elongated composite part comprises stacked reinforcement layers comprising at least one of non-continuous woven fabric and non-continuous reinforcement. In another step, the elongated composite part is formed into a curved elongated section comprising at least one of the non-continuous woven fabric and the non-continuous reinforcement.

These and other features, aspects and advantages of the disclosure will become better understood with reference to the following drawings, description and claims.

BRIEF DESCRIPTION OF THE DRAWINGS

Figure 1 shows a perspective view of one embodiment of a composite part which comprises a curved elongated section, which is curved in a Z plane, and a straight elongated section;

5 Figure 2 shows a cross-section view through line 2-2 of the curved elongated section of the composite part embodiment of Figure 1;

Figure 3 shows a cross-section view through line 3-3 of 45° woven fabric of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view;

10 Figure 4 shows a cross-section view through line 4-4 of 0/90° woven fabric of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view;

Figure 5 shows a cross-section view through line 5-5 of 90° reinforcement of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view;

Figure 6 shows a cross-section view through line 6-6 of 0° reinforcement of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view;

15 Figure 7 shows a bottom plan view of the composite part of Figure 1, with dashed lines used to show windows in the reinforcement layers of the curved elongated section;

Figure 8 shows a cross-section view through line 8-8 of the straight elongated section of the composite part embodiment of Figure 1;

20 Figure 9 shows another embodiment of a cross-section view through line 9-9 of the curved elongated section of the composite part embodiment of Figure 1;

Figure 10 shows a perspective view of one embodiment of a composite part which comprises a curved elongated section, which is curved in an X-Y plane, and a straight elongated section attached to a member;

25 Figure 11 shows a cross-section view through line 11-11 of the curved elongated section of the composite part embodiment of Figure 10;

Figure 12 shows a bottom plan view of the composite part of Figure 10, with dashed lines used to show windows in the reinforcement layers of the curved elongated section;

Figure 13 shows a cross-section view through line 13-13 of the straight elongated section of the composite part embodiment of Figure 10; and

5 Figure 14 shows one embodiment of a flowchart of a method of manufacturing a composite part.

DETAILED DESCRIPTION

The following detailed description is of the best currently contemplated modes of 10 carrying out the disclosure. The description is not to be taken in a limiting sense, but is made merely for the purpose of illustrating the general principles of the disclosure, since the scope of the disclosure is best defined by the appended claims.

Figure 1 shows a perspective view of one embodiment of a composite part 10 which comprises a curved elongated section 12, which is curved in a Z plane, and a straight elongated 15 section 14. The composite part 10 may be made of stacked layers of reinforcement and may comprise an aircraft part, an aircraft frame, an aircraft stringer, or a non-aircraft part, device, or structure. The curved elongated section 12 may comprise flange members 16 and 18, web members 20 and 22, and cap member 24.

Figure 2 shows a cross-section view through line 2-2 of the curved elongated section 12 20 of the composite part 10 embodiment of Figure 1. As shown, the curved elongated section 12 may comprise a plurality of stacked reinforcement layers 26 having different types/orientations of reinforcement. In other embodiments, the stacked reinforcement layers 26 may be made of varying materials. Some layers 28 of 45° (PX) woven fabric 29 may extend continuously from flange member 16 through web member 20, cap member 24, web member 22, and flange 25 member 18. Other layers 31 of 45° (PX) woven fabric 29 may extend continuously from the

web member 20, through the cap member 24, and to the web member 22, and may extend within the flange members 16 and 18. Still other layers 35 of 45° (PX) woven fabric may extend continuously around a perimeter 37 of the web member 20, the cap member 24, and the web member 22 without extending through the flange members 16 and 18. Figure 3 shows a cross-section view through line 3-3 of some of the 45° woven fabric 29 of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view. The 45° woven fabric 29 may comprise layers of reinforcement 30 at a 45° angle 32 relative to layers of reinforcement 33.

As shown in Figure 2, layers 39 of 0/90° (PW “plain weave”) woven fabric 34 may extend non-continuously along the curved elongated section 12. The 0/90° woven fabric 34 may 10 extend from flange member 16 only part-way through web member 20. The 0/90° woven fabric 34 may also extend from flange member 18 only part-way through web member 22. The 0/90° woven fabric 34 may not extend in cap member 24. There may be one or more windows 36 in the web member 20, the cap member 24, and the web member 22 where the 0/90° woven fabric 34 does not extend. These windows 36 may lessen and/or prevent the curved elongated section 15 12 from wrinkling. Figure 4 shows a cross-section view through line 4-4 of the 0/90° woven fabric 34 of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view. The 0/90° woven fabric 34 may comprise layers of reinforcement 38 which are perpendicular to layers of reinforcement 40.

As shown in Figure 2, layers 41 of 90° reinforcement 42 may extend only part-way from 20 web member 20 through cap member 24 and only part-way through web member 22. The 90° reinforcement 42 may not extend through flange members 16 or 18, and may not extend through portions 44 and 46 of web members 20 and 22. There may be one or more windows 48 in the web member 20, the cap member 24, and the web member 22 where the 90° reinforcement 42 does not extend. These windows 48 may lessen and/or prevent the curved elongated section 25 12 from wrinkling. Figure 5 shows a cross-section view through line 5-5 of the 90° reinforcement

42 of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view. The 90° reinforcement 42 may comprise parallel layers of reinforcement 50.

As shown in Figure 2, layers 43 of 0° reinforcement 52 may extend only part-way through the cap member 24. The layers of 0°reinforcement 52 may extend in the web members 5 20 and 22 but not the flange member 16 or 18. There may be one or more windows 54 in the web members 20 and 22 and the cap member 24 where the reinforcement does not extend.

These windows 54 may lessen and/or prevent the curved elongated section 12 from wrinkling.

Figure 6 shows a cross-section view through line 6-6 of the 0° reinforcement 52 of the cross-section view of the embodiment of Figure 2 converted to a flat-pattern view. The 0°

10 reinforcement 52 may comprise parallel layers of reinforcement 56 which are oriented perpendicularly to the reinforcement 50 of Figure 5.

As shown in Figure 2, the curved elongated section 12 may be attached to a member 58 which may comprise an aircraft skin and/or other type of aircraft or non-aircraft part. A plurality of filler members 60 may be attached for extra support between the curved elongated section 12 15 and the member 58.

Figure 7 shows a bottom plan view of the composite part 10 of Figure 1, with dashed lines used to show the respective windows 36, 48, and 54 in the reinforcement layers 26 of the curved elongated section 12. The windows 36, 48, and 54 may lessen and/or prevent the curved elongated section 12 from wrinkling.

20 Figure 8 shows a cross-section view through line 8-8 of the straight elongated section 14 of the composite part 10 embodiment of Figure 1. As shown, the straight elongated section 14 may comprise a second plurality of stacked reinforcement layers 64. The second plurality of stacked reinforcement layers 64 may comprise layers of 45° (PX) woven fabric 29 which may extend continuously from flange member 16 through web member 20, cap member 24, web 25 member 22, and flange member 18. The second plurality of stacked reinforcement layers 64

may further comprise layers of 0/90° (PW “plain weave”) woven fabric 34 which may extend continuously from flange member 16 through web member 20, cap member 24, web member 22, and flange member 18. Wrinkling may not be an issue in the straight elongated section 14, which may allow the layers of 45° (PX) woven fabric 29 and the layers of 0/90° (PW “plain weave”) woven fabric 34 to extend continuously through the flange members 16 and 18, web members 20 and 22, and cap member 24 without using any windows in the reinforcement layers 64.

Figure 9 shows another embodiment of a cross-section view through line 9-9 of the curved elongated section 12 of the composite part 10 embodiment of Figure 1. The cross-section view of Figure 9 may be identical to the cross-section view of Figure 2 with the exception that the 0/90° woven fabric 34 may extend within the flange members 16 and 18 and into either the web members 20 and 22 but not into the cap member 24. As a result, the windows 36 where the 0/90° reinforcement does not extend may be expanded.

Figure 10 shows a perspective view of one embodiment of a composite part 110 which comprises a curved elongated section 112, which is curved in an X-Y plane, and a straight elongated section 114 attached to a member 158 which may comprise an aircraft skin and/or other type of aircraft or non-aircraft part. The composite part 110 may be made of stacked layers of reinforcement and may comprise an aircraft part, an aircraft frame, an aircraft stringer, or a non-aircraft part, device, or structure. The curved elongated section 112 may comprise flange members 116 and 118, web members 120 and 122, and cap member 124.

Figure 11 shows a cross-section view through line 11-11 of the curved elongated section 112 of the composite part 110 embodiment of Figure 10. As shown, the curved elongated section 112 may comprise a plurality of stacked reinforcement layers 126 having different types/orientations of reinforcement. The stacked reinforcement layers 126 may be made of one or more types of reinforcement. In other embodiments, the stacked reinforcement layers 126

may be made of varying materials. Some layers 128 of 45° (PX) woven fabric 129 may extend continuously from flange member 116 through web member 120, cap member 124, web member 122, and flange member 118.

As shown in Figure 11, layers of 0/90° (PW “plain weave”) woven fabric 134 may 5 extend non-continuously along the curved elongated section 112. The 0/90° woven fabric 134 may extend only in the web members 120, and not extend in the flange members 116 and 118, or the cap member 124. In other embodiments, the 0/90° woven fabric 134 may extend only part-way within the flange members 116 and 118 without extending into the cap member 124. There 10 may be one or more windows 136 in the flange members 116 and 118, and the cap member 124 where the 0/90° woven fabric 134 does not extend. These windows 136 may lessen and/or prevent the curved elongated section 112 from wrinkling.

As shown in Figure 11, layers of non-continuous narrow strips or tows of 0° reinforcement 152 may extend from flange members 116 and 118 through the web members 120 and into the cap member 124 with a nominal gap between each strip or tow. These windows 154 15 may lessen and/or prevent the curved elongated section 112 from wrinkling.

As shown in Figure 11, the curved elongated section 112 may be attached to a member 158 which may comprise an aircraft skin and/or other type of aircraft or non-aircraft part. A plurality of filler members 160 may be attached for extra support between the curved elongated section 112 and the member 158.

20 Figure 12 shows a bottom plan view of the composite part 110 of Figure 10, with dashed lines used to show the non-continuous narrow strips or tows of 0° reinforcement within the windows 136, 148, and 154 in the 0/90° (PW “plain weave”) woven fabric layers 134 of the curved elongated section 112. The windows 136, 148, and 154 may lessen and/or prevent the curved elongated section 112 from wrinkling.

25 Figure 13 shows a cross-section view through line 13-13 of the straight elongated section

114 of the composite part 110 embodiment of Figure 10. As shown, the straight elongated section 114 may comprise a second plurality of stacked reinforcement layers 164. The second plurality of stacked reinforcement layers 164 may comprise layers of 45° (PX) woven fabric 129 which may extend continuously from flange member 116 through web member 120, cap member 5 124, web member 122, and flange member 118. The second plurality of stacked reinforcement layers 164 may further comprise layers of 0/90° (PW “plain weave”) woven fabric 134 which may extend continuously from flange member 116 through web member 120, cap member 124, web member 122, and flange member 118. Wrinkling may not be an issue in the straight elongated section 114, which may allow the layers of 45° (PX) woven fabric 129 and the layers 10 of 0/90° (PW “plain weave”) woven fabric 134 to extend continuously through the flange members 116 and 118, web members 120 and 122, and cap member 124 without using any windows in the 0/90° (PW “plain weave”) woven fabric layers 164.

Figure 14 shows one embodiment of a flowchart 266 of a method of manufacturing a composite part 10/110. In one step 268, an elongated composite part 10/110 may be 15 prepared/stacked. The prepared/stacked elongated composite part 10/110 may comprise stacking reinforcement layers 26/126 having at least one of non-continuous plain weave (PW) 0/90° woven fabric 34/134 and non-continuous 0° reinforcement 52/152, placing window plies around highly-contoured areas of the windows 36/48/54/136/148/154 and adding structural reinforcement plies into the windows 36/48/54/136/148/154. The stacked reinforcement layers 20 26/126 may be made of various reinforcements. In other embodiments, the stacked reinforcement layers 26/126 may be made of varying materials. The preparing step 268 may comprise preparing/stacking the plies of reinforcement into a flat charge the will comprise the elongated composite part 10/110 with straight/flat stacked reinforcement layers 26/126. The preparing step 268 may further comprise the straight elongated section 14/114 continuous plain 25 weave (PW) 0/90° woven fabric 34/134 extending throughout the straight elongated section

14/114 and/or continuous 45° (PX) woven fabric 29/129 extending throughout the straight elongated section 14/114.

The preparing step 268 may utilize a cutting machine, a lay-up machine, and/or another type of machines to lay-up and/or cut the straight/flat stacked reinforcement layers 26/126. The 5 preparing step 268 may additionally comprise preparing/stacking the elongated composite part 10/110 by stacking reinforcement layers 26/126 further comprising 45° (PX) woven fabric 29/129 and 90° reinforcement 42/142. The preparing step 268 may include programming a cutting machine, lay-up machine, and/or another type of machine to include a flat pattern for each of the stacked reinforcement layers 26/126 being stacked to include one or more windows 10 36/48/54/136/148/154 in the stacked reinforcement layers 26/126. The preparing step may further include interleaving a separator film between the reinforcement plies in the windows 36/48/54/136/148/154 manually and/or by using a machine.

In another step 270, the uncured, straight stacked reinforcement layers 26/126 of the prepared elongated composite part 10/110 may be formed into a curved elongated section 15 12/112. The forming step 270 may utilize a forming machine, a mold, and/or another type of forming device, structure, and/or process. In one embodiment, the forming step 270 may comprise one or more of the following: machine forming the elongated section 12/112 of the prepared elongated composite part 10/110 using the uncured, straight stacked reinforcement layers 26/126; placing the formed elongated section 12/112 into a cure mold; straightening 20 flange members 16/18/116/118, and web members 20/22/120/122, as needed; locating and forming the window plies in the contoured sections; removing the separator film; and curing the curved elongated section 12/112 without wrinkles. The formed curved elongated section 12/112 may comprise one or more plies of the non-continuous plain weave (PW) 0/90° woven fabric 34/134 and the non-continuous 0° reinforcement 52/152. The forming step 270 may further 25 comprise forming the uncured, straight stacked reinforcement layers 26/126 of the prepared

elongated composite part 10/110 into a straight elongated section 14/114. The formed straight elongated section 14/114 may comprise continuous plain weave (PW) 0/90° woven fabric 34/134 extending throughout the straight elongated section 14/114 and/or continuous 45° (PX) woven fabric 29/129 extending throughout the straight elongated section 14/114.

5 In one embodiment, the preparing and forming steps 268 and 270 may comprise preparing and forming a composite part 10 comprising at least one of an aircraft part, an aircraft frame, and an aircraft stringer. In other embodiments, varying non-aircraft structures, devices, or parts may be prepared and formed.

In still other embodiments, the composite parts and the methods disclosed in the various 10 embodiments of the instant disclosure may comprise and/or utilize woven fabric of any angular orientation, and/or reinforcement of any angular orientation. For instance, anywhere in this disclosure where a certain degree woven fabric, and/or a certain degree reinforcement is referred to, the degree of the woven fabric and/or the degree of the reinforcement may be varied.

One or more embodiments of the disclosure may reduce one or more problems of one or 15 more of the existing composite parts and/or methods of manufacture. For instance, one or more embodiments of the disclosure may provide a composite part that can be placed in a highly contoured area with reduced and/or eliminated wrinkling and/or buckling type problems; may reduce costly repairs needed to add reinforcement to carry required loads; may reduce the need for additional reinforcing material thereby reducing cost and weight; may reduce the need for 20 additional processing, inspection, and production steps; and/or may reduce one or more other types of problems.

It should be understood, of course, that the foregoing relates to exemplary embodiments of the disclosure and that modifications may be made without departing from the spirit and scope of the disclosure as set forth in the following claims.

CLAIMS

1. A composite part comprising:

a curved elongated section comprising a plurality of stacked reinforcement layers,

wherein the stacked reinforcement layers in the curved elongated section collectively comprise at

5 least one of non-continuous woven fabric and non-continuous reinforcement.

2. The composite part of claim 1 wherein the composite part comprises at least one of an aircraft part, an aircraft frame, and an aircraft stringer.

3. The composite part of claim 1 wherein the stacked reinforcement layers in the curved elongated section further comprise at least one of 45 degree woven fabric (PX) and 90 degree reinforcement.

10 4. The composite part of claim 1 wherein the curved elongated section comprises a plurality of flange members, a plurality of web members, and a cap member.

5. The composite part of claim 4 wherein the curved elongated section is curved in a Z plane and the web members and the cap member each comprise at least one of: continuous 15 plain weave (PX) 45 degree woven fabric, non-continuous plain weave (PW) 0/90 degree woven fabric, and non-continuous 0 degree reinforcement.

6. The composite part of claim 4 wherein the curved elongated section is curved in an X-Y plane and the flange members and the cap member each comprise at least one of: plain weave (PX) 45 degree woven fabric, non-continuous plain weave (PW) 0/90 degree woven 20 fabric, and non-continuous 0 degree reinforcement.

7. The composite part of claim 1 wherein the composite part further comprises a straight elongated section.

8. The composite part of claim 1 wherein the stacked reinforcement layers in the curved elongated section collectively comprise at least one of non-continuous plain weave (PW) 25 0/90 degree woven fabric and non-continuous 0 degree reinforcement.

9. A method of manufacturing a composite part comprising:
preparing an elongated composite part comprising stacked reinforcement layers
comprising at least one of non-continuous woven fabric and non-continuous reinforcement;
forming the elongated composite part into a curved elongated section comprising at least
5 one of the non-continuous woven fabric and the non-continuous reinforcement.

10. The method of claim 9 wherein the preparing step and forming step comprise
preparing and forming at least one of an aircraft part, an aircraft frame, and an aircraft stringer.

11. The method of claim 9 wherein the preparing step comprises preparing the
elongated composite part comprising stacked reinforcement layers comprising at least one of
10 non-continuous plain weave (PW) 0/90 woven fabric and non-continuous 0 degree
reinforcement, and wherein the forming step comprises forming the elongated composite part
into a curved elongated section comprising at least one of the non-continuous plain weave (PW)
0/90 woven fabric and the non-continuous 0 degree reinforcement.

12. The method of claim 9 wherein the preparing step comprises preparing the
15 elongated composite part comprising straight stacked reinforcement layers.

13. The method of claim 18 wherein the preparing step further comprises preparing
the elongated composite part comprising the stacked reinforcement layers further comprising 45
degree woven fabric (PX) and 90 degree reinforcement and non-continuous 0 degree
reinforcement.

20 14. The method of claim 9 wherein the forming step further comprises forming the
elongated composite part into a straight elongated section comprising at least one of continuous
plain weave (PW) 0/90 woven fabric extending throughout the straight elongated section,
continuous 45 degree woven fabric (PX) extending throughout the straight elongated section, and
continuous 0 degree reinforcement extending throughout the straight elongated section.

15. The method of claim 9 wherein the forming step utilizes at least one of a forming machine and a mold.

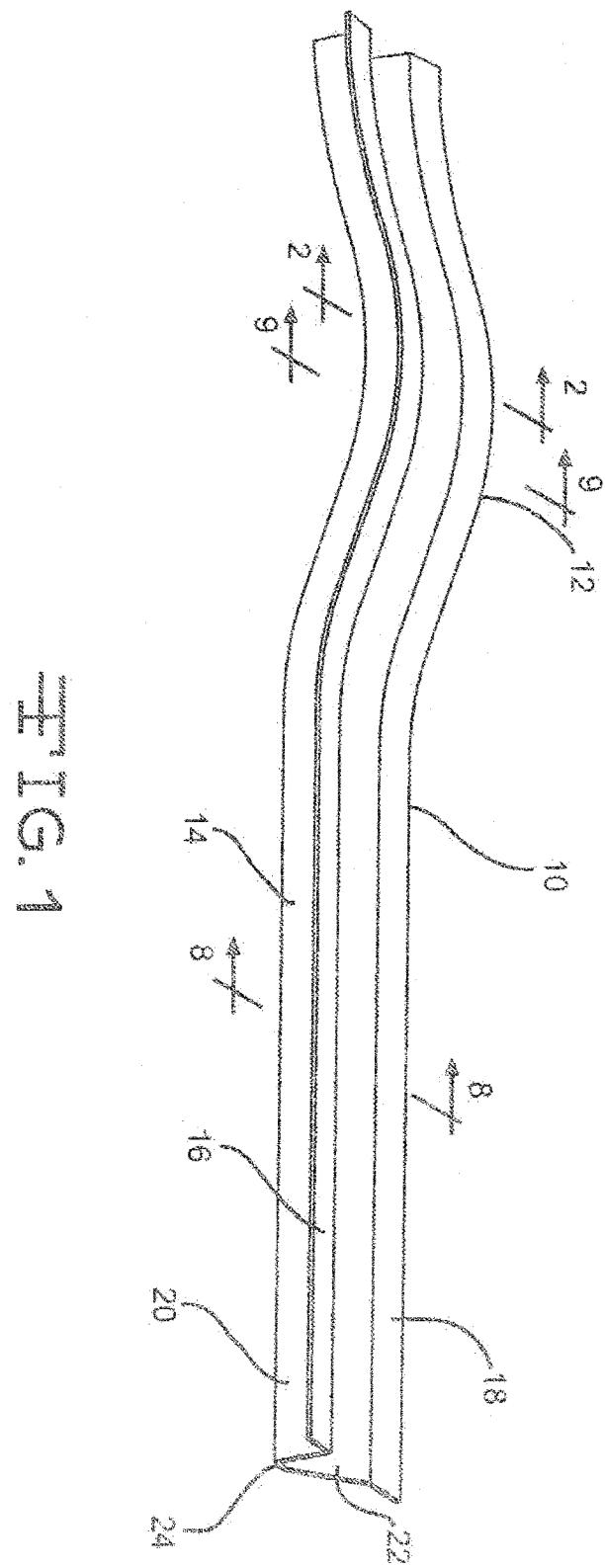
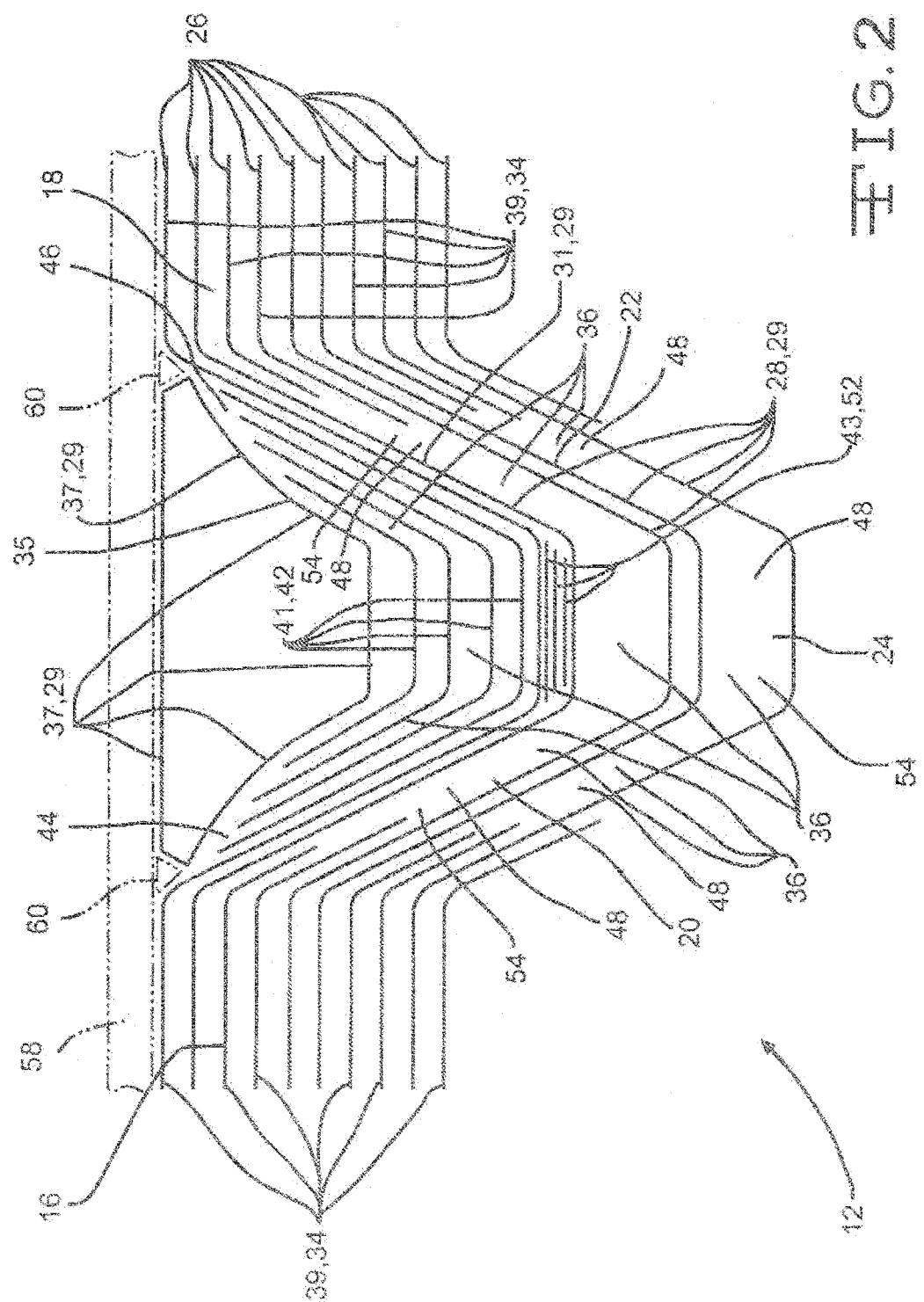


FIG. 1



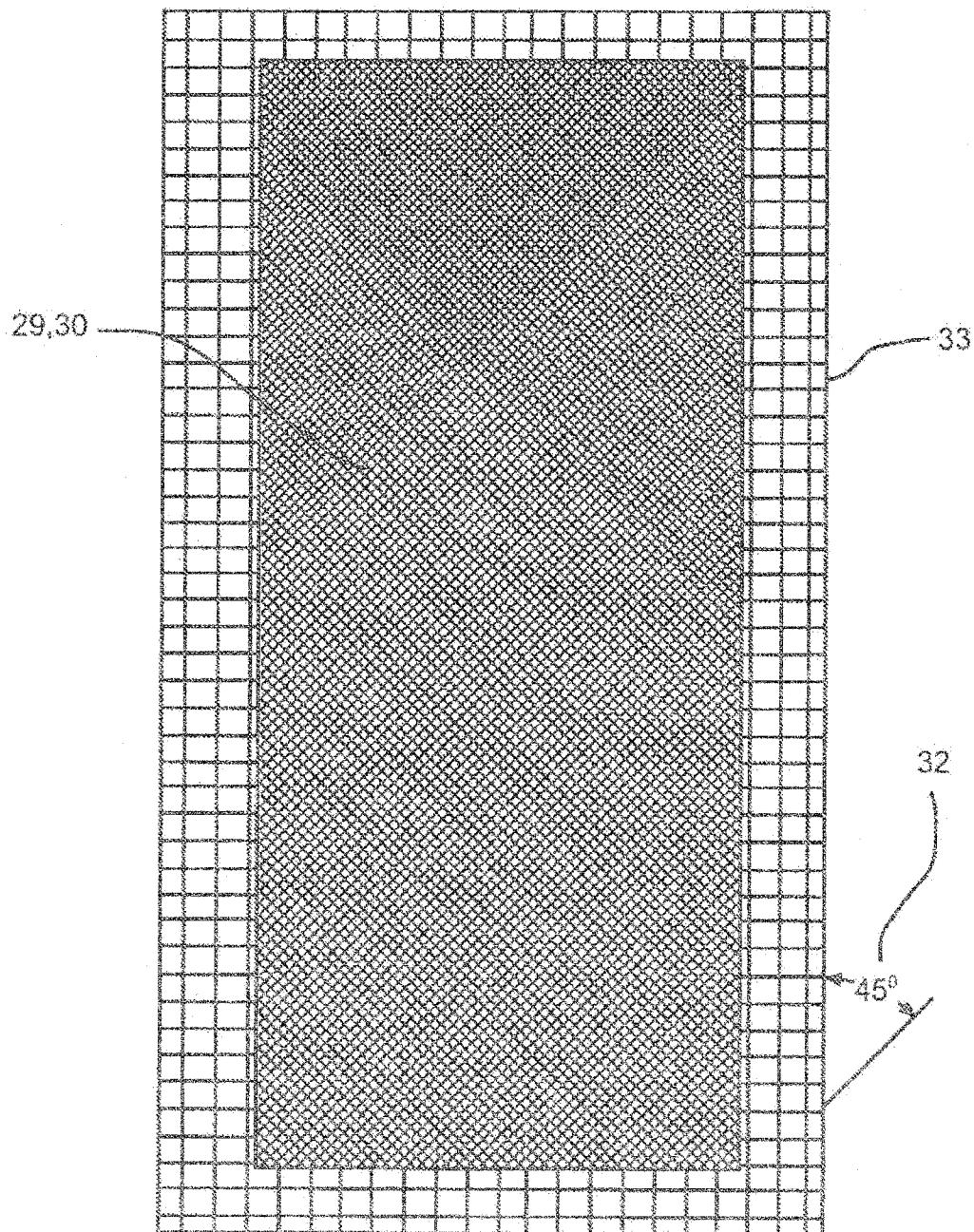


FIG. 3

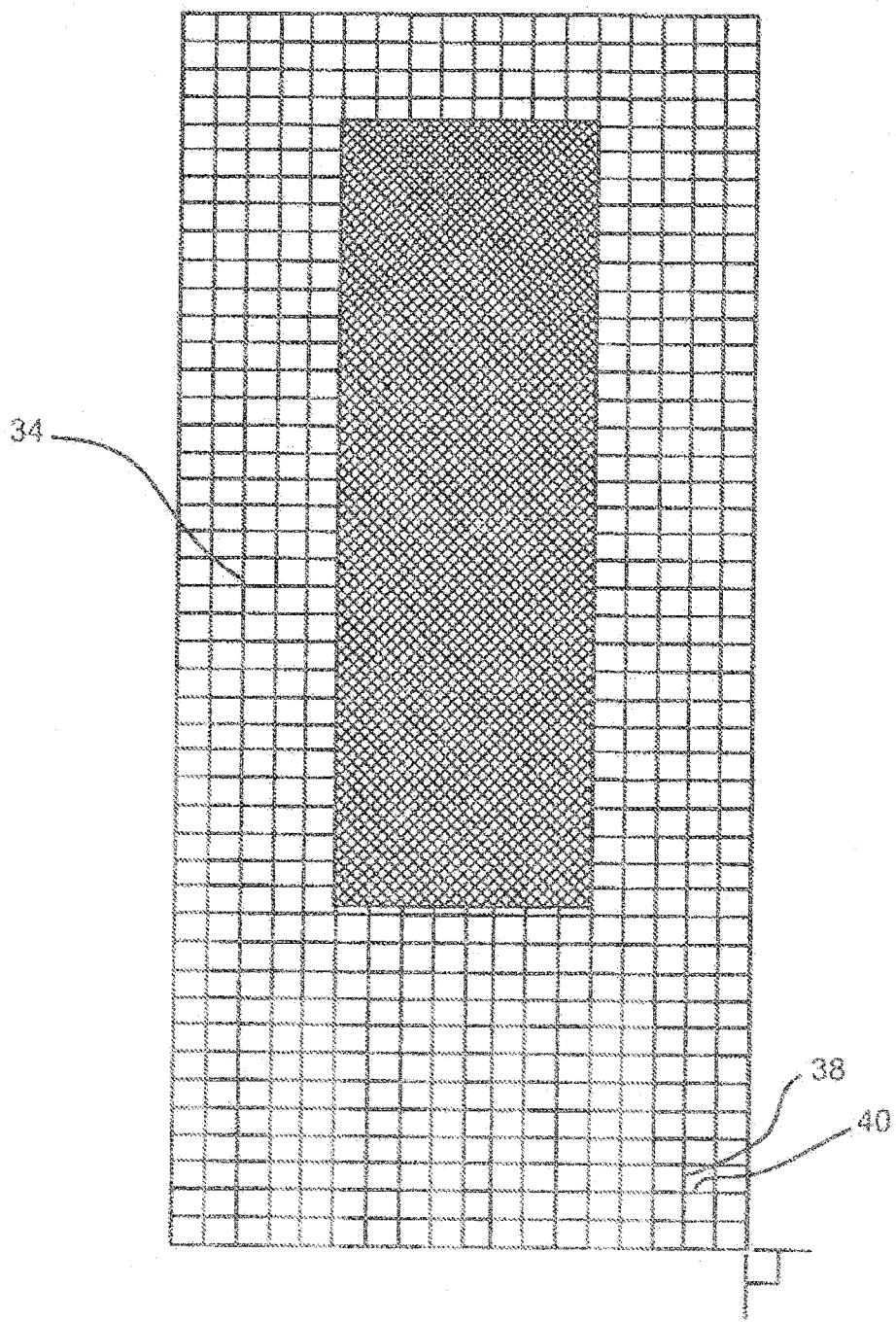


FIG. 4

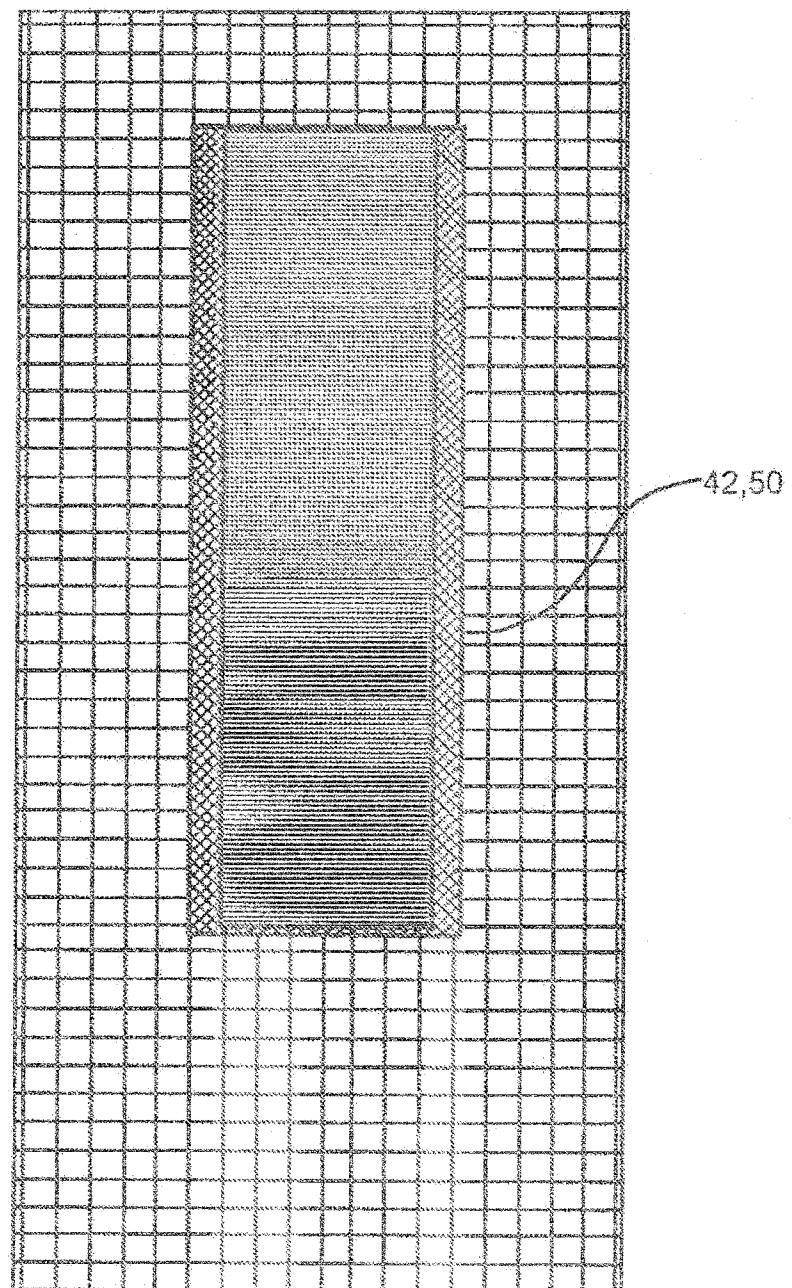


FIG. 5

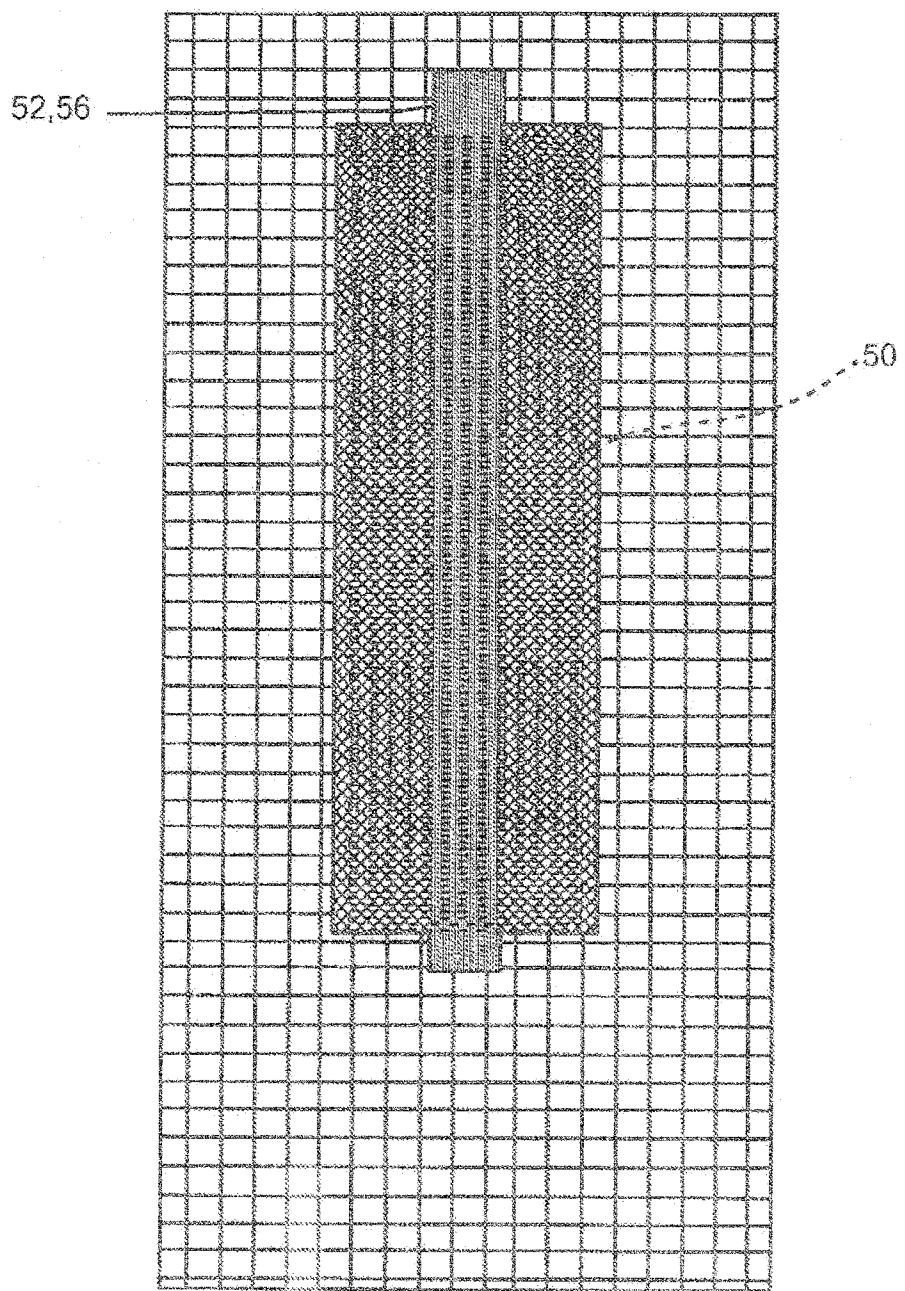


FIG. 6

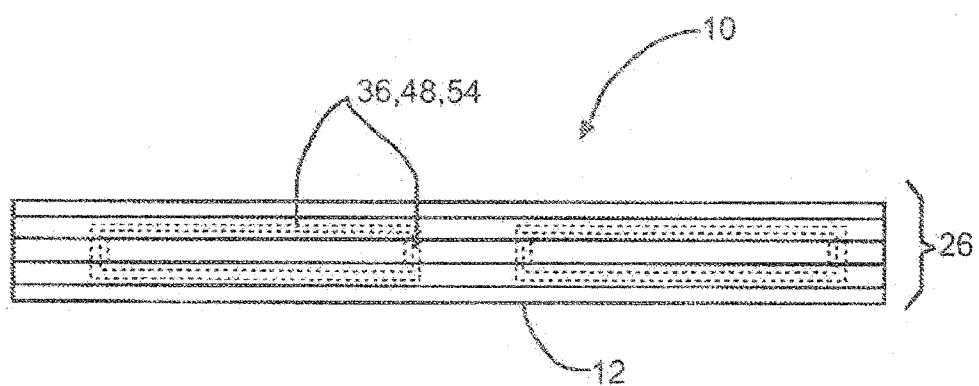


FIG. 7

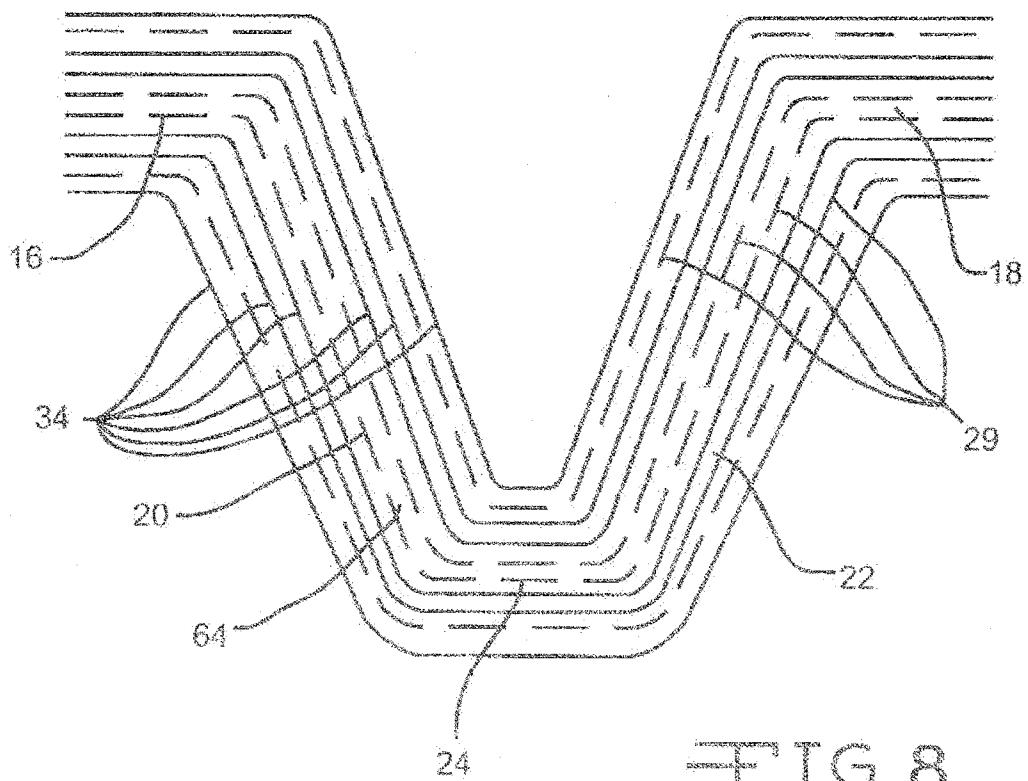


FIG. 8

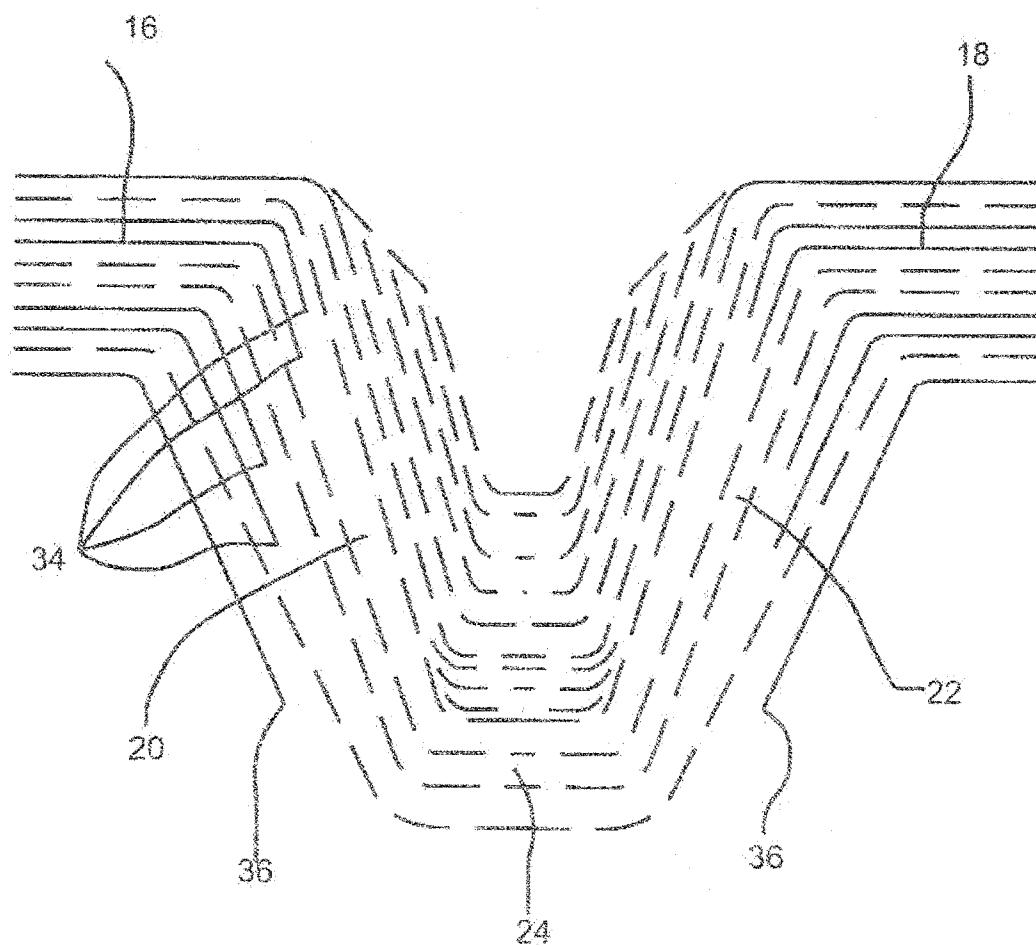
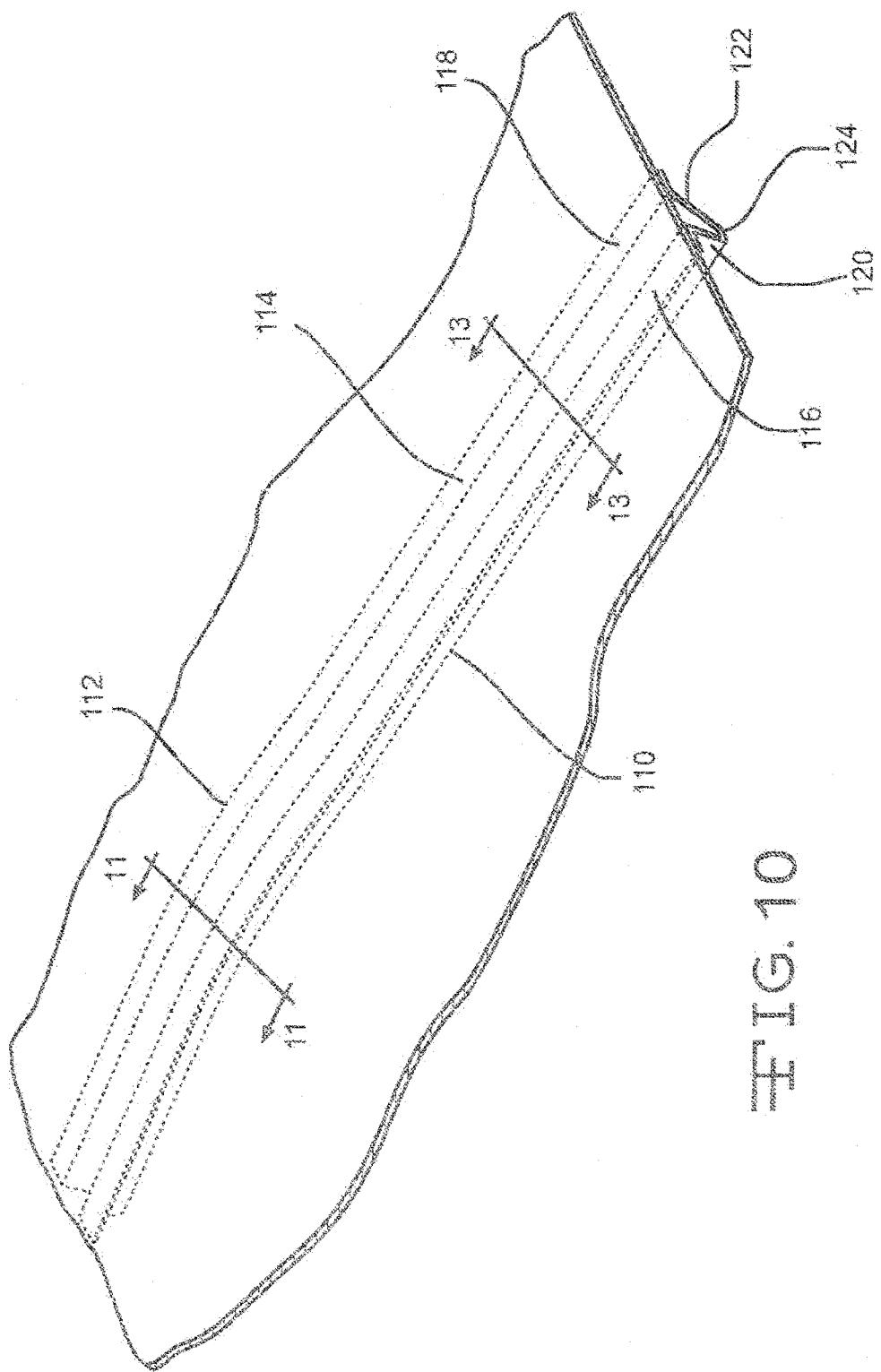


FIG. 9



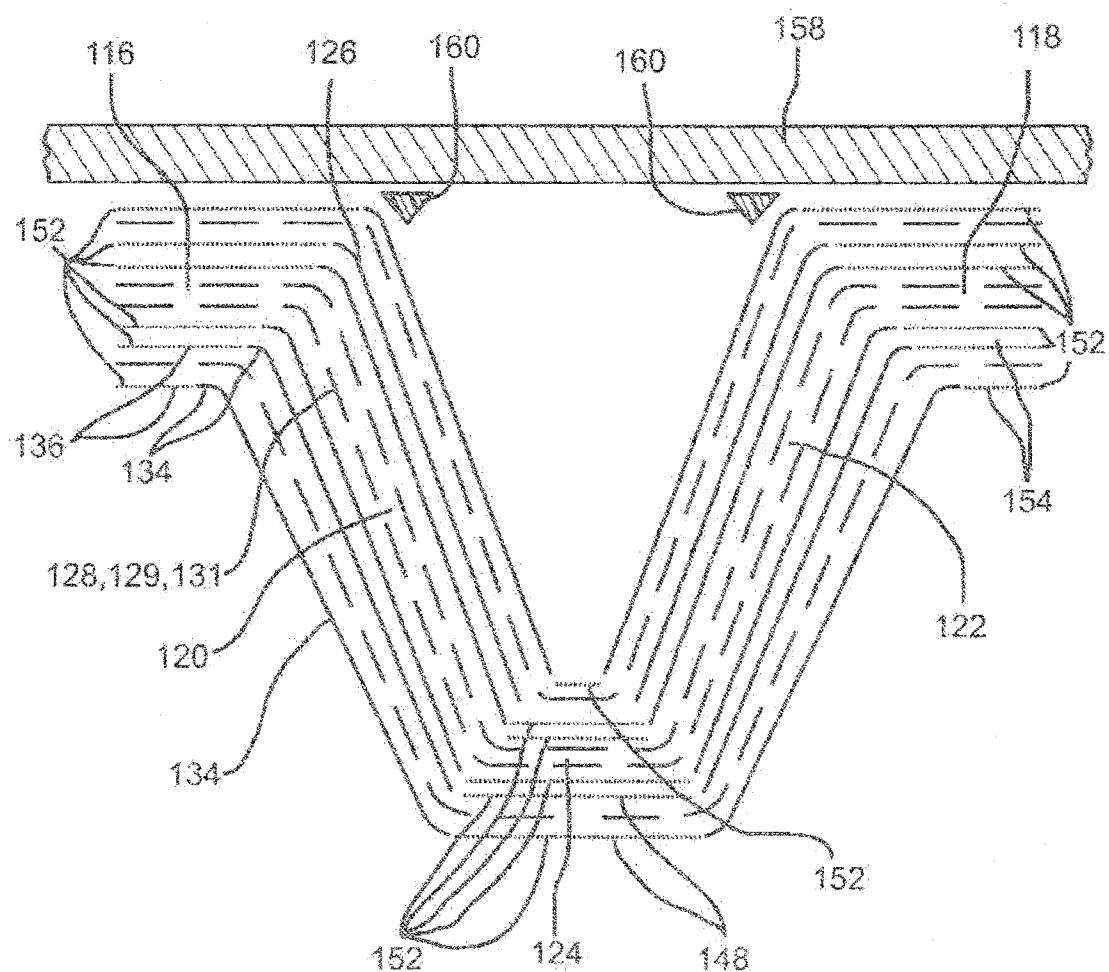


FIG. 11

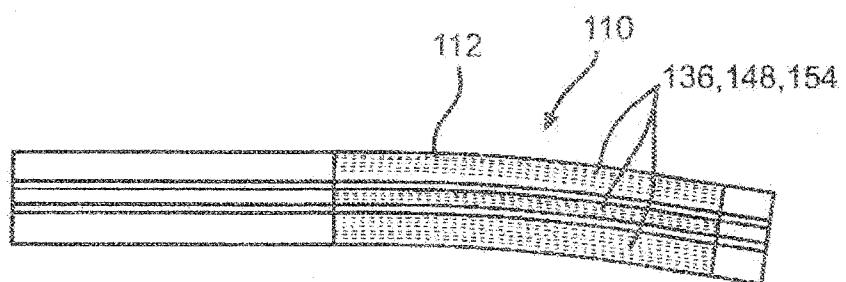


FIG. 12

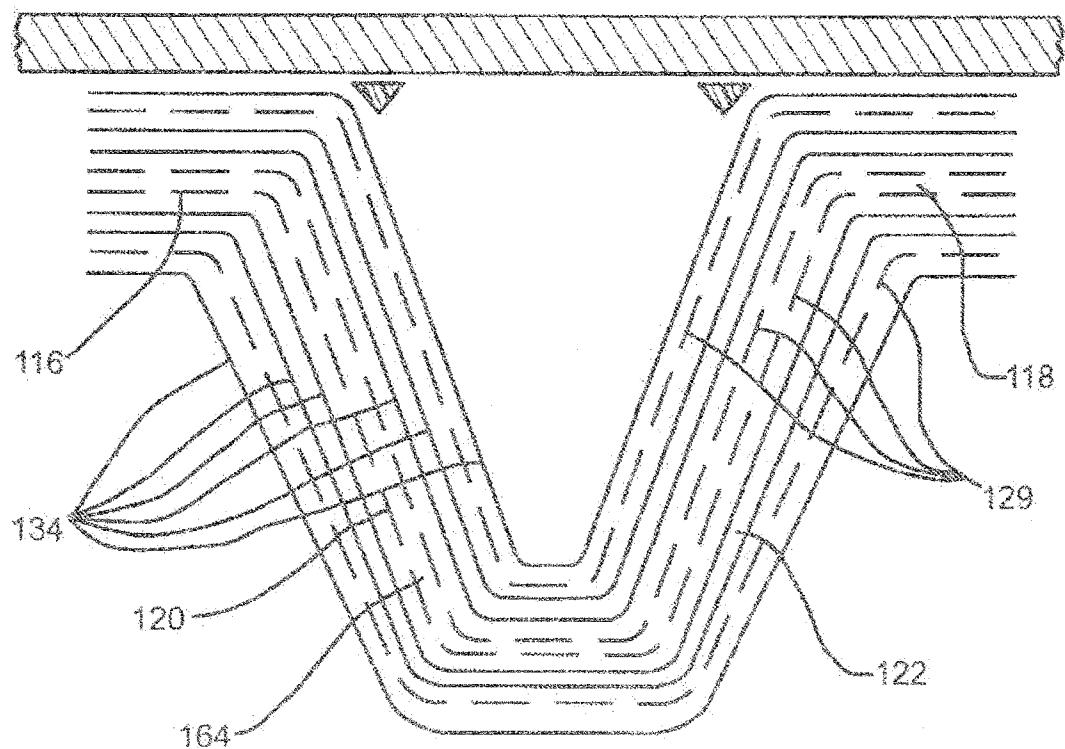


FIG. 13

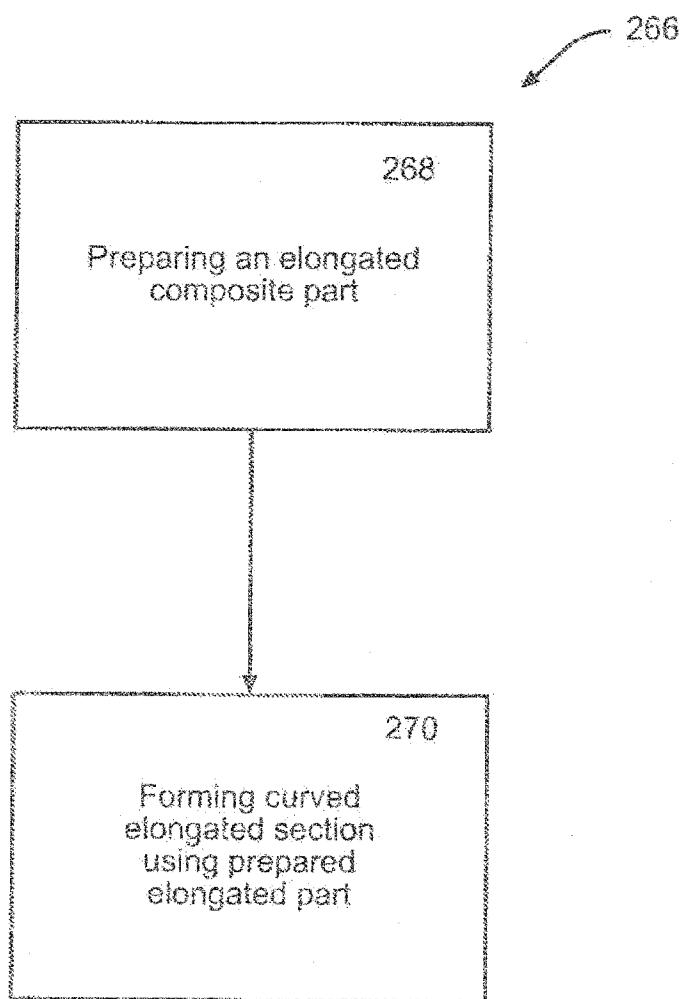


FIG. 14

INTERNATIONAL SEARCH REPORT

International application No
PCT/US2008/087529

A. CLASSIFICATION OF SUBJECT MATTER
INV. B29C70/34 B29D31/00

According to International Patent Classification (IPC) or to both national classification and IPC

B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)
B29D B29C

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

Electronic data base consulted during the international search (name of data base and, where practical, search terms used)

EPO-Internal

C. DOCUMENTS CONSIDERED TO BE RELEVANT

Category*	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
X	EP 1 800 840 A (SAAB AB [SE]) 27 June 2007 (2007-06-27) abstract figures 2b, 7a, 7b paragraph [0001] paragraph [0002] paragraph [0008] paragraph [0020] paragraph [0050] paragraph [0051]	1-4, 6, 8-11, 15
A	EP 0 373 729 A (SHELL INT RESEARCH [NL]) 20 June 1990 (1990-06-20) abstract figures 5-7 column 5, line 13 - line 53	1-15

Further documents are listed in the continuation of Box C.

See patent family annex.

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INTERNATIONAL SEARCH REPORT

Information on patent family members

International application No
PCT/US2008/087529

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