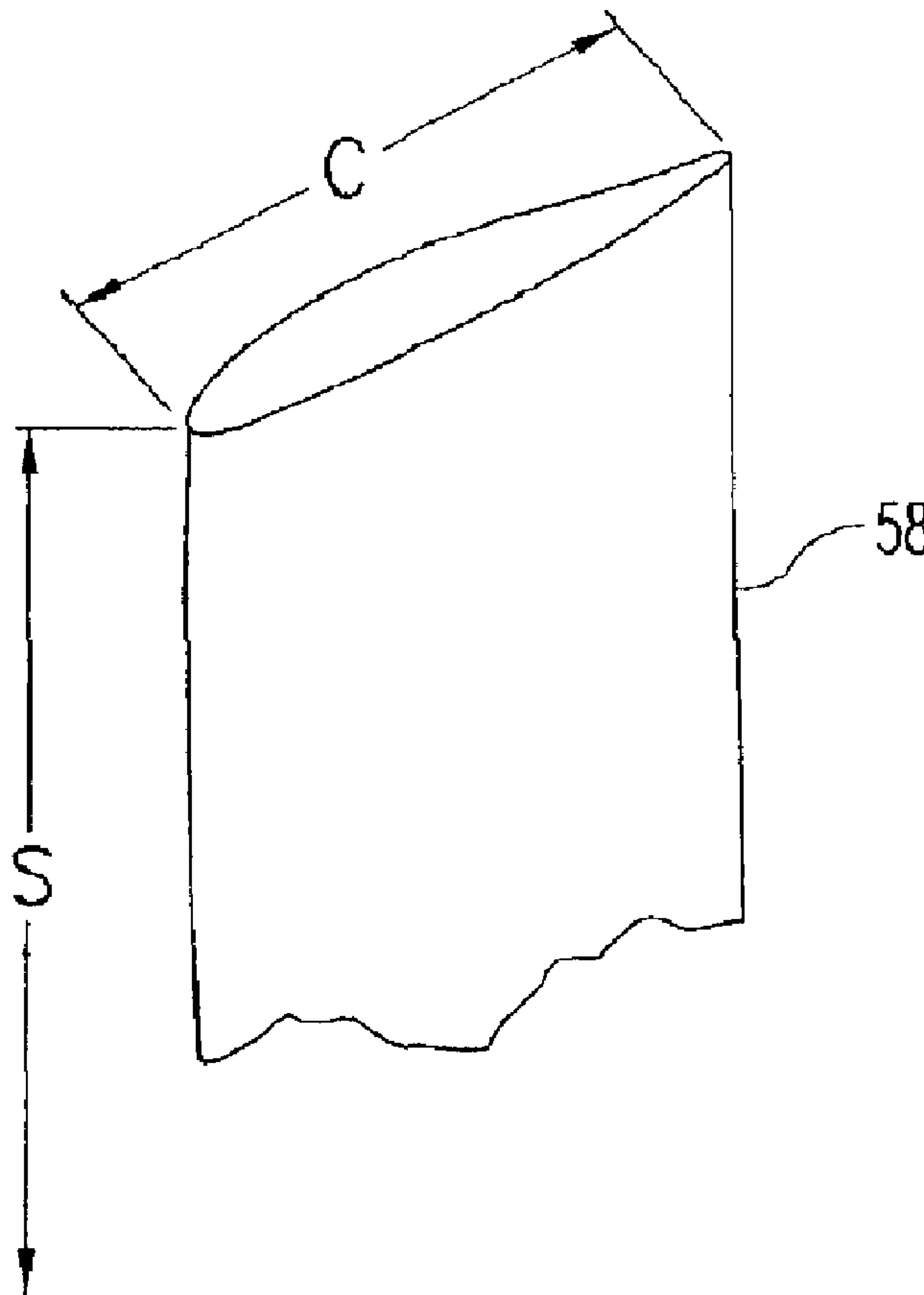




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(54) **Titre :** COMPOSANT DE MOTEUR A TURBINE A GAZ A PROFIL AERODYNAMIQUE
(54) **Title:** GAS TURBINE ENGINE AIRFOIL SHAPED COMPONENT



(57) **Abrégé/Abstract:**

A gas turbine engine airfoil shaped component is disclosed including in one form a compressor blade or vane having a tip portion capable of producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow.

(57) Abrégé(suite)/Abstract(continued):

In one form the tip portion includes a camber angle distribution that produces relatively large change in camber angle near the leading edge but relatively fiat distribution for remainder of blade. In one form the tip portion includes relatively large change in camber angle near trailing edge.

ABSTRACT

A gas turbine engine airfoil shaped component is disclosed including in one form a compressor blade or vane having a tip portion capable of producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow. In one form the tip portion includes a camber angle distribution that produces relatively large change in camber angle near the leading edge but relatively flat distribution for remainder of blade. In one form the tip portion includes relatively large change in camber angle near trailing edge.

GAS TURBINE ENGINE AIRFOIL SHAPED COMPONENT

This application is a divisional of Canadian Patent Application No. 2,762,785 filed December 28, 2011.

TECHNICAL FIELD

The present invention generally relates to gas turbine engine airfoil shaped components such as vanes and rotatable blades, and more particularly, but not exclusively, to compressor blades and vanes.

BACKGROUND

Improving the performance of gas turbine engines remains an area of interest. Some existing systems have various shortcomings relative to certain applications. Accordingly, there remains a need for further contributions in this area of technology.

SUMMARY

In one aspect of the present invention there is provided an apparatus comprising: a compressor blade of a gas turbine engine capable of being rotated to produce a pressure rise in a working fluid during operation of the gas turbine engine, the compressor blade including a main blade portion having a first camber shape that is blended with a tip portion having a second camber shape, the second camber shape producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow, wherein the second camber shape is blended into the first camber shape starting at about 2/3 of a height of the compressor blade, wherein the second camber shape includes a normalized camber angle defined as $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$ where β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location, and wherein the normalized camber angle reaches at least 0.4 in the first 10% of normalized chord.

In another aspect of the present invention there is provided an apparatus comprising: a compressor blade of a gas turbine engine capable of being rotated to produce a pressure rise in a working fluid during operation of the gas turbine engine, the compressor blade including a main blade portion having a first camber shape that is blended with a tip portion having a second camber shape, the second camber shape producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow, wherein the second camber shape includes a normalized camber angle defined as $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$ where β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location; and wherein normalized camber angle reaches at least 0.4 in the first 10% of normalized chord.

In a further aspect of the present invention there is provided an apparatus comprising: a compressor blade of a gas turbine engine capable of being rotated to produce a pressure rise in a working fluid during operation of the gas turbine engine, the compressor blade including a main blade portion having a first camber shape that is blended with a tip portion having a second camber shape, the second camber shape producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow, wherein the

second camber shape provides a cumulative integral according to the relationship $0.4 < \frac{\int_0^1 [\int_0^x \beta' dx] dx}{\int_0^1 \beta' dx} \leq 0.5$ where x is chord location, $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$, β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location.

In another aspect of the present invention there is provided an apparatus comprising: a gas turbine engine having a compressor; a compressor airfoil member disposed in the compressor and having a camber angle variation along a chord and a concave pressure side surface at a tip portion of the compressor airfoil member from a leading edge to an intermediate-chord location that results in a normalized camber angle that reaches at least 0.4 in the first 10% of normalized chord; and wherein the normalized camber angle is defined as $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$ where β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location.

In a further aspect of the present invention there is provided an apparatus comprising: a gas turbine engine including a compressor for increasing a total pressure of a working fluid prior to delivery to a combustor for mixing with fuel, the compressor having a plurality of airfoil members that extend between a first flow surface and a second flow surface, the plurality of airfoil members having a first camber shape at a tip portion that provides a cumulative integral according to the relationship $0.4 < \frac{\int_0^1 [\int_0^x \beta' dx] dx}{\int_0^1 \beta' dx} \leq 0.5$ where x is chord location, $\frac{\beta_i - \beta}{\beta_i - \beta_e}$, β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location.

BRIEF DESCRIPTION OF THE FIGURES

Fig. 1 depicts an embodiment of a gas turbine engine.

Fig. 2 depicts an embodiment of a compressor airflow member.

Fig. 3 depicts CFD analysis of an embodiment of a compressor airflow member.

Fig. 4 depicts CFD analysis of an embodiment of a compressor airflow member.

Fig. 5 depicts CFD analysis of an embodiment of a compressor airflow member.

Fig. 6 depicts a camber angle.

Fig. 7 depicts a chart comparing an embodiment of an airflow member against other designs.

Fig. 8 depicts a weighting distribution.

Fig. 9 depicts a performance of an embodiment of a compressor airflow member.

DETAILED DESCRIPTION OF THE ILLUSTRATIVE EMBODIMENTS

For the purposes of promoting an understanding of the principles of the invention, reference will now be made to the embodiments illustrated in the drawings and specific language will be used to describe the same. It will nevertheless be understood that no limitation of the scope of the invention is thereby intended. Any alterations and further modifications in the described embodiments, and any further applications of the principles of the invention as described herein are contemplated as would normally occur to one skilled in the art to which the invention relates.

With reference to Fig. 1, one embodiment of a gas turbine engine 50 is illustrated and includes a compressor 52, combustor 54, and turbine 56. In one form the gas turbine engine 50 can be coupled with an aircraft to provide propulsive power. As used herein, the term "aircraft" includes, but is not limited to, helicopters, airplanes, unmanned space vehicles, fixed wing vehicles, variable wing vehicles, rotary wing vehicles, unmanned combat aerial vehicles, tailless aircraft, hover crafts, and other airborne and/or extraterrestrial (spacecraft) vehicles. Further, the present inventions are contemplated for utilization in other applications that may not be coupled with an aircraft such as, for example, industrial applications, power generation, pumping sets, naval propulsion, weapon systems, security systems, perimeter defense/security systems, and the like known to one of ordinary skill in the art.

In operation of the gas turbine engine 50 the compressor 52 receives a working fluid such as air and compresses the working fluid before being delivered to the combustor 54. The combustor 54 mixes the compressed working fluid with a fuel and burns the mixture to produce a core flow having products of combustion. The turbine 56 receives and expands the core flow to produce power to drive the compressor 52. The gas turbine engine 50 can take a variety of forms including turbojet, turbofan, turboprop, and turboshaft. In addition, the gas turbine engine 50 can be a variable or adaptive cycle engine, can have centrifugal flow components alternatively and/or additionally to axial flow components, and can include additional components than those depicted in the illustrative embodiment, among other possible variations. In short, the gas turbine engine 50 can take on a variety of different embodiments.

Turning now to Fig. 2, the compressor 52 of Fig. 1 includes an airflow member 58 disposed within the flowpath and operable to alter a direction of a flow of working fluid passing through the compressor 52. In some embodiments the airflow member 58 can be a rotating compressor blade or a relatively fixed stator vane. The airflow member 58 can have an airfoil-like aerodynamic shape that includes characteristics such as chord, span, camber, mean camber line, and camber angle.

The shape of the airflow member 58 in the present application produces a number of unique design-point and off-design-point characteristics of efficiency, pressure ratio, and operating range. The shape also reduces tip-gap flow, among other possible characteristics. As will be described further below, the

shape generally includes a relatively large change in camber angle for the first portion of the blade chord with a relatively flat camber angle distribution for the remaining majority of the blade. To set forth one non-limiting example, the airflow member 58 can have a relatively large change in camber angle near the leading edge that occurs within about the first 5% of blade chord. Near the leading edge and tip of the airflow member 58 the pressure surface includes a relatively large concave region that serves to produce a relatively low increase in static pressure rise on the pressure surface. The relatively flat camber angle distribution for the majority of the blade at the tip results in a reduced suction-peak Mach number. The combination of reduced suction-surface suction peak and the local static pressure on the pressure-surface tends to reduce the overall driving pressure difference across the rotor tip. The effect also serves to produce a localized jet of stream-wise momentum flow. In addition to the shape of the camber angle just described near the leading-edge and tip of the airflow member 58, in some forms the airflow member 58 can also have a relatively large change in camber angle near its trailing edge.

Figs. 3, 4, and 5 depict one form of the airflow member 58 and its effects on static pressure and velocities. Computational fluid dynamics (CFD) analysis was performed on the airflow member 58 and shows a reduced static pressure region, and resultant increased velocity region, near the leading edge of the tip region of the airflow member 58. The dashed circle in Fig. 3 highlights the pressure surface of the airflow member 58 in the tip region, and in particular highlights the local decrease in static pressure relative to the rest of the airflow

member 58. Fig. 4, a view looking down the span of the airflow member 58, also shows the local decrease in static pressure of the pressure-side within the dashed circle. In addition, the view of Fig. 4 shows that the pressure-surface low pressure region occurs at approximately the same true chord fraction as the suction-surface pressure suction peak. The pressure-surface static pressure is initially high near the leading edge, it reduces along the chord of the airflow member 58 before increasing again at approximately the 30%-40% true chord. Fig. 5 depicts rotor tip velocity contour. The dashed circle is again positioned near the leading edge of the airflow member 58 and it encircles a region of the pressure side where the relatively low pressure causes a local stream-wise acceleration of the flow which is roughly depicted by the arrow. The induced flow acts like a localized fluid jet. The increase in stream-wise momentum, coupled with a reduction in the driving circumferential pressure gradient that induces the leakage flow across the tip, reduces the magnitude and penetration of the leakage flow into the flow passage. Such an effect can help to delay the onset of tip stall.

Turning now to a more detailed discussion of the properties of the airflow member 58, Fig. 6 depicts a camber line 60 extending from the point i, representing the inlet to the airfoil section of the airflow member 58, to a point e representing the exit of the airfoil section. The points i and e are arranged along the line 62. In one form the camber line 60 is the mean camber line and the line 62 is the chord line. The angle β_i represents the inlet metal camber angle and the angle β_e represents the exit metal camber angle. As will be appreciated, any

point along the camber line 60 will have a camber angle β just as the points i and e have a camber angle. In one form of the airflow member 58, the camber angle distribution of the airfoil section follows according to the relationships

$$\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e} \quad (1)$$

$$\beta' = \left[\frac{1}{2} + \frac{1}{2^n} \left\{ \ln(1 - k + 2kx') - \ln(1 + k - 2kx') \right\} \right]^n \quad (2)$$

where β' is normalized camber angle, β_i is inlet metal angle, β_e is exit metal angle, β is the camber angle at a given normalized chord location, x' is normalized chord, and n and k are constants. In one form $k=0.96$ and $n=0.25$.

Fig. 7 depicts the distribution of normalized camber angle against normalized chord for one embodiment of the present application plotted against distributions for other designs. As will be apparent from the figure, the depicted embodiment of the present application experiences a relatively high level of turning in the first 5% of true chord and a relatively flat camber-angle distribution for the majority of the chord length downstream of the leading-edge region. In other embodiments of the airflow member 58, the normalized camber angle is at least 0.5 in the first 10% of normalized chord. In still other embodiments the normalized camber angle is at least 0.5 in the first 5% of normalized chord. In yet other embodiments the normalized beta is at least 0.6 in the first 10% of normalized chord.

In one form the camber angle distribution of the present application can be expressed as a cumulative integral as shown with the equation

$$0.4 < \frac{\int_0^1 \left[\int_0^z \beta' dx \right] dz}{\int_0^1 \beta' dx} \leq 0.5 \quad (3)$$

where x represents chord. The lower bound of the cumulative integral can range anywhere between 0.4 and 0.4695. To set forth just a few non-limiting examples, in some forms the lower bound of the cumulative integral is 0.42. In still other forms the lower bound of the cumulative integral is 0.45. In still further forms the lower bound of the cumulative integral is 0.46.

The relationships described above regarding the camber angle are with respect to a camber angle at a given span location of the airflow member 58. In one non-limiting form the camber angle relationships and embodiments described above relate to the camber angle distribution at the tip of the airflow member 58. In some applications the camber angle distribution described above can be blended into a camber angle distribution of a main part of the airflow member 58. Such a blending can start at a variety of span height locations and proceed to the tip. To set forth just one non-limiting example, the blending can start at approximately 2/3 span of the airflow member 58. Various techniques can be used to blend the main part of the airflow member to the camber angle distribution discussed above. For example, the blending can be accomplished by a weighted average from a minimum at the blend point to a maximum at the tip. In some embodiments the blending can occur in a cubic weighted distribution.

Other embodiments can include other distributions such as quadratic or hyperbolic tangent distributions, to set forth just two non-limiting alternatives. Fig. 8 discloses a non-limiting weight function of a cubic blending. In the embodiment disclosed, the streamline number 21 corresponds to full span height. The blending begins in the embodiment of Fig. 8 at about streamline 15.

The maximum thickness location of the tip camber style can occur at a number of chord locations, but in one non-limiting embodiment the location is about the half true chord location. The location of the maximum thickness can be blended into the main blade maximum thickness location with a weighting distribution following that of the camber blending, to set forth just one non-limiting example.

Fig. 9 illustrates various performance characteristics of one embodiment of the airflow member 58. As can be seen from the figure, greater efficiencies and pressure ratios are provided across a range of pressure ratios and inlet corrected flows.

Some features of the airflow member 58 can be adjusted when incorporating a tip camber style into a main airflow member 58 style. For example, the trailing edge metal angle may need to be adjusted to counter the effect of a rapidly varying setting/stagger angle in the tip region which may be caused by the nature of the camber style at the tip and also the manner in which the tip camber style is blended into the main portion of the airflow member 58.

One aspect of the present application provides an apparatus comprising a gas turbine engine having a compressor, a compressor airfoil member disposed in the compressor and having a gap between a tip of the compressor airfoil member and a flow surface of the compressor, the compressor airfoil member having a camber angle variation along a chord and a concave pressure side surface at the tip of the compressor airfoil member from a leading edge to an intermediate-chord location that results in a normalized beta of at least 0.4 in the first 10% of normalized chord, and wherein normalized beta is defined as

$$\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e} \text{ where } \beta_i \text{ is inlet metal angle, } \beta_e \text{ is exit metal angle, and } \beta \text{ is the}$$

camber angle at a given chord location.

Yet another aspect of the present application provides an apparatus comprising a gas turbine engine having a compressor, a compressor airfoil member disposed in the compressor and having a camber angle variation along a chord and a concave pressure side surface at the tip of the compressor airfoil member from a leading edge to an intermediate-chord location that results in a normalized beta of at least 0.4 in the first 10% of normalized chord, and wherein normalized beta is defined as $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$ where β_i is inlet metal angle, β_e is exit

metal angle, and β is the camber angle at a given chord location.

One feature of the present application provides wherein the normalized beta is at least 0.5 in the first 10% of normalized chord.

Another feature of the present application provides wherein the normalized beta is at least 0.5 in the first 5% of normalized chord.

A still further feature of the present application provides wherein the normalized beta is at least 0.6 in the first 10% of normalized chord.

Yet still a further feature of the present application provides herein the normalized beta follows the equation

$$\beta' = \left[\frac{1}{2} + \frac{1}{2^3} \left(\ln(1 - k + 2kx') - \ln(1 + k - 2kx') \right) \right]^n \text{ where } x' \text{ is normalized chord and } n$$

and k are constants.

Yet still another feature of the present application provides wherein $k=0.96$ and $n=0.25$.

Still yet another feature of the present application provides wherein the compressor airfoil member is operable to increase a pressure of a working fluid flowing through the compressor during operation of the gas turbine engine.

Still a further feature of the present application provides means for blending the camber angle variation at the tip of the compressor airfoil member to a camber angle variation at an intermediate height of the compressor airfoil member.

Yet another feature of the present application provides wherein the distribution of normalized beta along the normalized chord includes an inflection point at a mid-chord location.

Another aspect of the present application provides an apparatus comprising a gas turbine engine including a compressor for increasing the total pressure of a working fluid prior to delivery to a combustor for mixing with fuel, the compressor having a plurality of airfoil members that extend between a first flow surface and a second flow surface and form gaps between ends of the airfoil

members and the second wall, the plurality of airfoil members having a camber shape at their tips that provides a cumulative integral according to the

$$\text{relationship } 0.4 < \frac{\int_0^1 \left[\int_0^x \beta' dx \right] dx}{\int_0^1 \beta' dx} \leq 0.5 \text{ where } x \text{ is chord location, } \beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}, \beta_i \text{ is}$$

inlet metal angle, β_e is exit metal angle, and β is the camber angle at a given chord location.

Still another aspect of the present application provides an apparatus comprising a gas turbine engine including a compressor for increasing the total pressure of a working fluid prior to delivery to a combustor for mixing with fuel, the compressor having a plurality of airfoil members that extend between a first flow surface and a second flow surface, the plurality of airfoil members having a camber shape at their tips that provides a cumulative integral according to the

$$\text{relationship } 0.4 < \frac{\int_0^1 \left[\int_0^x \beta' dx \right] dx}{\int_0^1 \beta' dx} \leq 0.5 \text{ where } x \text{ is chord location, } \beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}, \beta_i \text{ is}$$

inlet metal angle, β_e is exit metal angle, and β is the camber angle at a given chord location.

A feature of the present application provides wherein the lower bound of the cumulative integral is 0.42.

Another feature of the present application provides wherein the camber shape at the tips is blended into a main portion of the plurality of airfoil members at an intermediate blade height.

Still yet another feature of the present application provides wherein the camber shape at the tips is blended into a main portion of the plurality of airfoil members at about 2/3 of the height of the plurality of airfoil members.

Still a further feature of the present application provides wherein the camber shape at the tips is blended into a camber shape in the main portion of the plurality of airfoil members with a weighted average between the two shapes.

Yet still a further feature of the present application provides wherein the camber style at the tips is blended into the camber style in the main body of the plurality of blades using a cubic weighting distribution.

A yet still further feature of the present application provides wherein the camber shape at the tips includes a maximum thickness-to-chord at about 50% chord.

Another feature of the present application provides wherein the normalized beta follows the equation

$$\beta' = \left[\frac{1}{2} + \frac{1}{2^3} \left\{ \ln(1 - k + 2kx') - \ln(1 + k - 2kx') \right\} \right]^n \text{ where } x' \text{ is normalized chord,}$$

$k=0.96$, and $n=0.25$.

Yet another aspect of the present application provides an apparatus comprising a compressor blade of a gas turbine engine capable of being rotated to produce a pressure rise in a working fluid during operation of the gas turbine engine, the compressor blade including a main blade portion having a first

camber shape that is blended with a tip portion having a second camber shape, the second camber shape producing a relatively low pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow.

A feature of the present application provides wherein the second camber shape is blended into the first camber shape at about 2/3 of the height of the compressor blade.

Another feature of the present application provides wherein the second camber shape is blended into the first camber shape with a weighted average.

A further feature of the present application provides wherein the maximum thickness-to-chord of the second camber shape is at about the mid-chord location.

A still further feature of the present application provides wherein the second camber shape includes a normalized camber angle defined as

$\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$ where β_i is inlet metal angle, β_e is exit metal angle, and β is the camber angle at a given chord location, and wherein normalized camber angle is at least 0.6 in the first 10% of normalized chord.

Still yet another feature of the present application provides wherein the second camber shape provides a cumulative integral according to the

$$\text{relationship } 0.4 < \frac{\int_0^1 \left[\int_0^x \beta' dx \right] dx}{\int_0^1 \beta' dx} \leq 0.5 \text{ where } x \text{ is chord location, } \beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}, \beta_i \text{ is}$$

inlet metal angle, β_e is exit metal angle, and β is the camber angle at a given chord location.

Still another feature of the present application provides wherein the lower bound of the cumulative integral is 0.42.

While the invention has been illustrated and described in detail in the drawings and foregoing description, the same is to be considered as illustrative and not restrictive in character, it being understood that only the preferred embodiments have been shown and described and that all changes and modifications that come within the spirit of the inventions are desired to be protected. It should be understood that while the use of words such as preferable, preferably, preferred or more preferred utilized in the description above indicate that the feature so described may be more desirable, it nonetheless may not be necessary and embodiments lacking the same may be contemplated as within the scope of the invention, the scope being defined by the claims that follow. In reading the claims, it is intended that when words such as "a," "an," "at least one," or "at least one portion" are used there is no intention to limit the claim to only one item unless specifically stated to the contrary in the claim. When the language "at least a portion" and/or "a portion" is used the item can include a portion and/or the entire item unless specifically stated to the contrary.

Claims:

1. An apparatus comprising:

a compressor blade of a gas turbine engine capable of being rotated to produce a pressure rise in a working fluid during operation of the gas turbine engine, the compressor blade including a main blade portion having a first camber shape that is blended with a tip portion having a second camber shape, the second camber shape producing a lower pressure near the leading edge of the tip to induce a local jet of stream-wise momentum flow, wherein the second camber shape

provides a cumulative integral according to the relationship $0.4 < \frac{\int_0^1 [\int_0^x \beta' dx] dx}{\int_0^1 \beta' dx} \leq 0.5$

where x is chord location, $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$, β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location.

2. The apparatus of claim 1, wherein the lower bound of the cumulative integral is 0.42.

3. The apparatus of claim 1, wherein the camber shape at the tip portion is blended into the main blade portion of the compressor blade at an intermediate blade height.

4. The apparatus of claim 3, wherein the second camber shape is blended into the main blade portion of the compressor blade at about 2/3 of a height of the compressor blade.

5. The apparatus of claim 4, wherein the second camber shape is blended into the first camber shape in the main blade portion of the compressor blade using a cubic weighting distribution.

6. The apparatus of claim 3, wherein the second camber shape is blended into the first camber shape in the main blade portion of the compressor blade with a weighted average between the two camber shapes.

7. The apparatus of claim 1, wherein the second camber shape includes a maximum thickness-to-chord at about 50% chord.
8. The apparatus of claim 1, wherein a normalized camber angle follows the equation $\beta' = \left[\frac{1}{2} + \frac{1}{2^3} \{ \ln(1 - k + 2kx') - \ln(1 + k - 2kx') \} \right]^n$ where x' is normalized chord, $k = 0.96$, and $n = 0.25$.
9. An apparatus comprising:
a gas turbine engine including a compressor for increasing a total pressure of a working fluid prior to delivery to a combustor for mixing with fuel, the compressor having a plurality of airfoil members that extend between a first flow surface and a second flow surface, the plurality of airfoil members having a first camber shape at a tip portion that provides a cumulative integral according to the relationship $0.4 < \frac{\int_0^1 [\int_0^x \beta' dx] dx}{\int_0^1 \beta' dx} \leq 0.5$ where x is chord location, $\beta' = \frac{\beta_i - \beta}{\beta_i - \beta_e}$, β_i is inlet metal angle, β_e is exit metal angle, and β is a camber angle at a given chord location.
10. The apparatus of claim 9, wherein the lower bound of the cumulative integral is 0.42.
11. The apparatus of claim 9, wherein the first camber shape is blended into a main portion of the plurality of airfoil members at an intermediate blade height.
12. The apparatus of claim 11, wherein the first camber shape is blended into a second camber shape at the main portion of the plurality of airfoil members at about 2/3 of a height of the plurality of airfoil members.
13. The apparatus of claim 11, wherein the first camber shape is blended into a second camber shape in the main portion of the plurality of airfoil members with a weighted average between the two shapes.

14. The apparatus of claim 12, wherein the first camber shape is blended into the second camber shape in the main body of the plurality of blades using a cubic weighting distribution.
15. The apparatus of claim 9, wherein the first camber shape includes a maximum thickness-to-chord at about 50% chord.
16. The apparatus of claim 9, wherein the normalized camber angle follows the equation $\beta' = \left[\frac{1}{2} + \frac{1}{2^3} \{ \ln(1 - k + 2kx') - \ln(1 + k - 2kx') \} \right]^n$ where x' is normalized chord, $k=0.96$, and $n=0.25$.

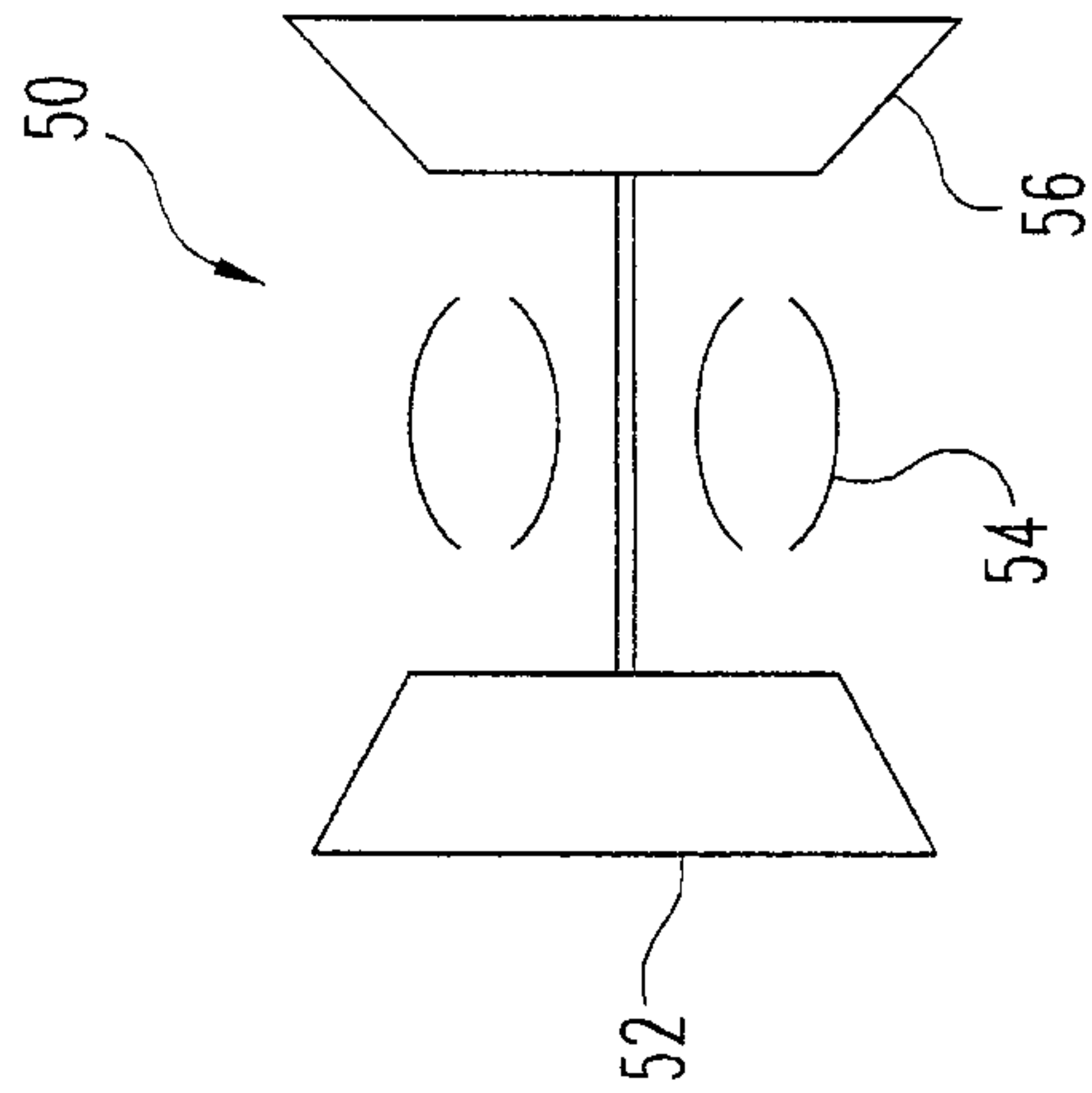


Fig. 1

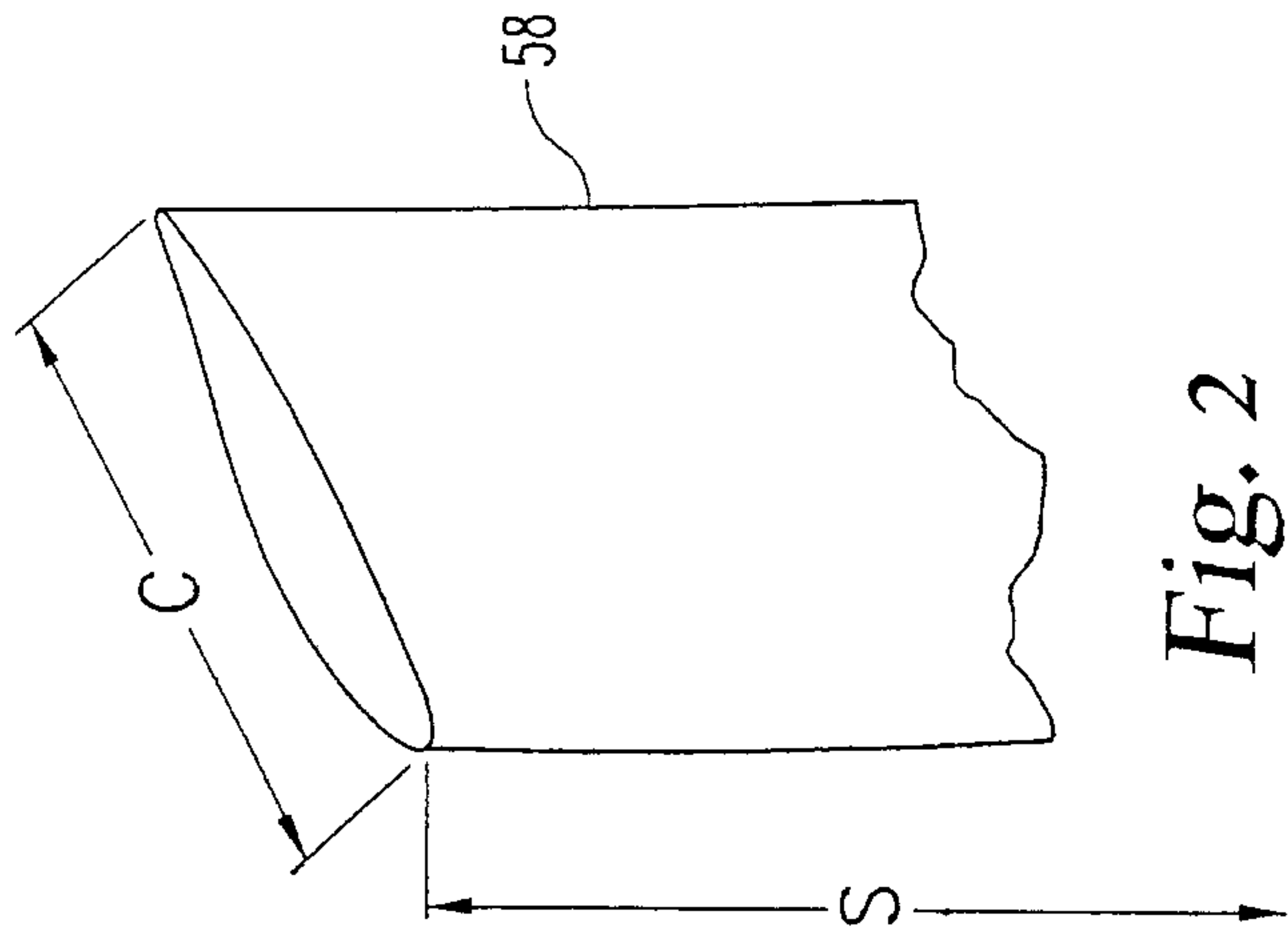


Fig. 2

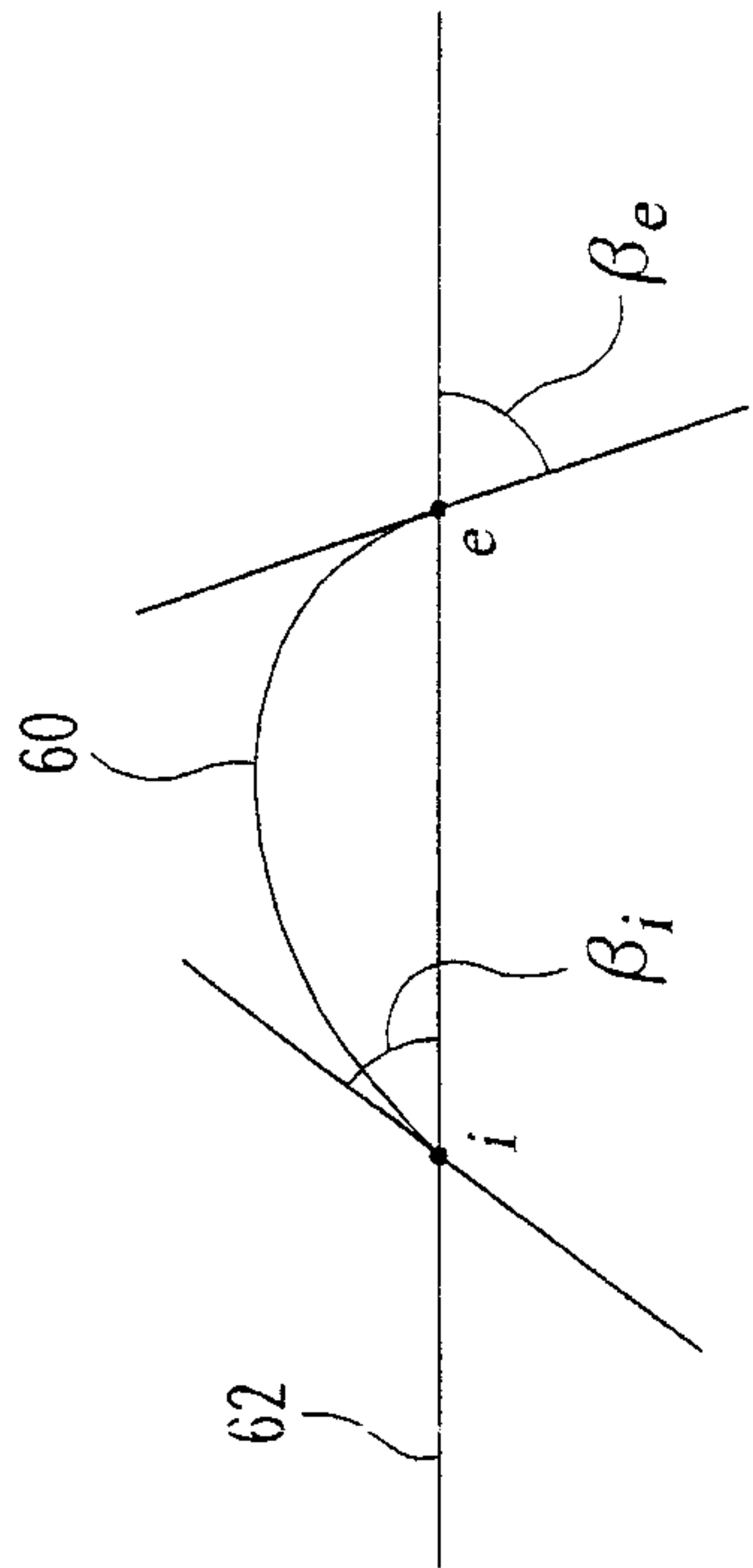


Fig. 6

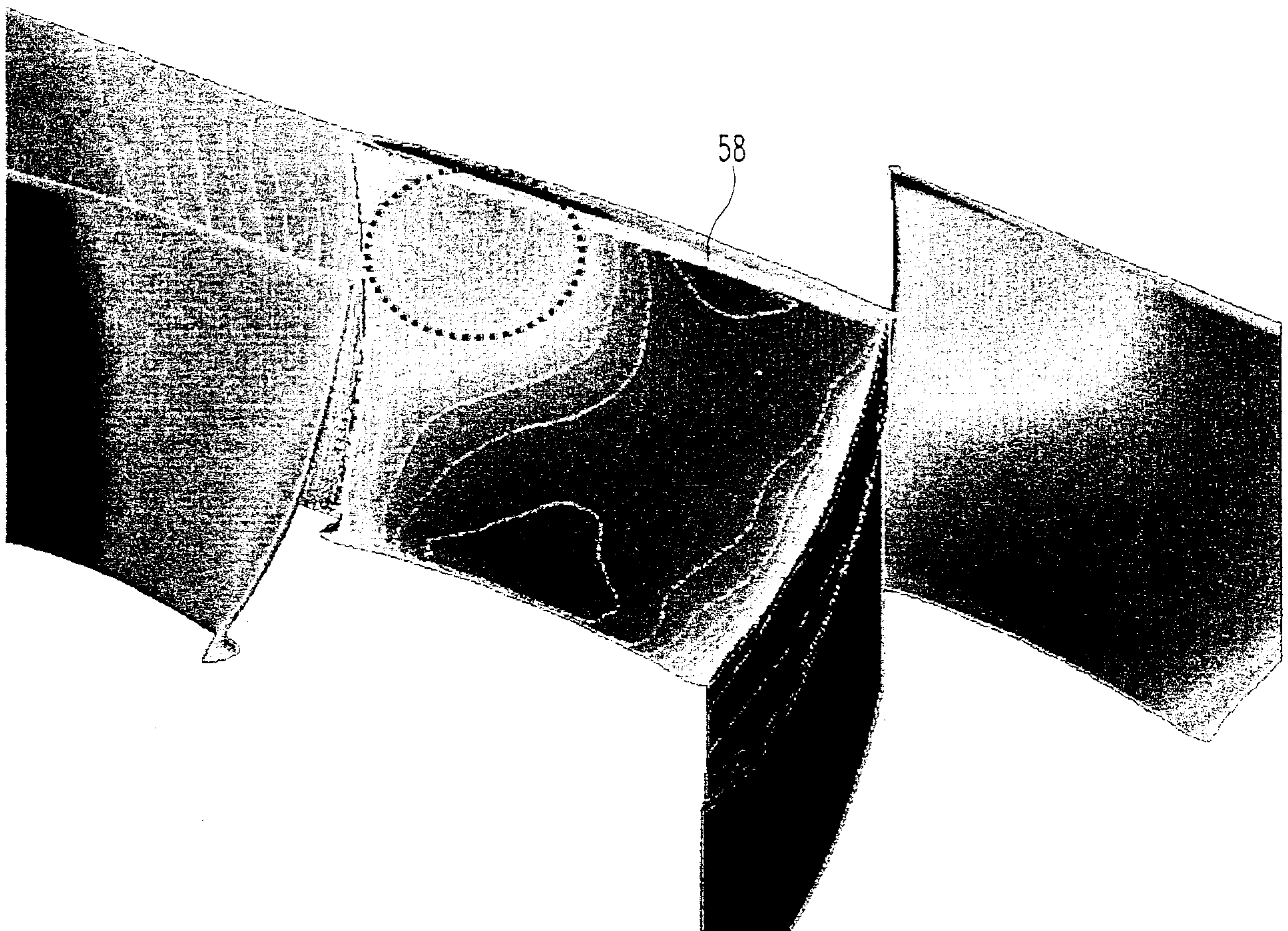
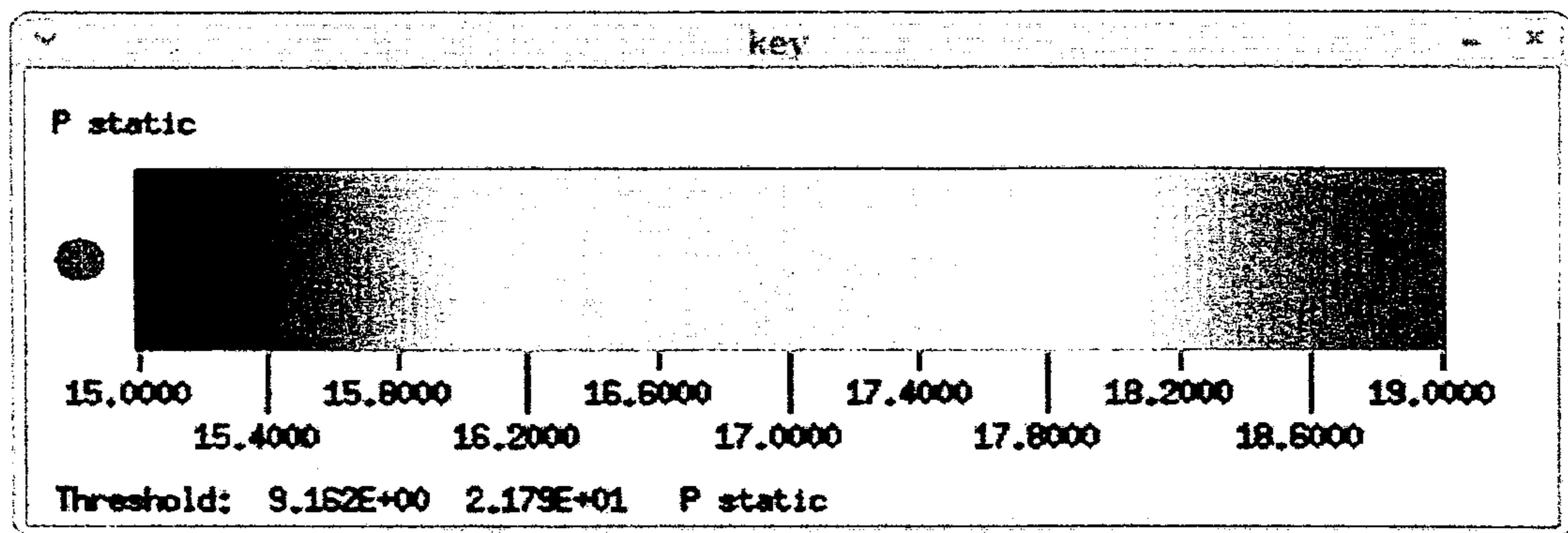
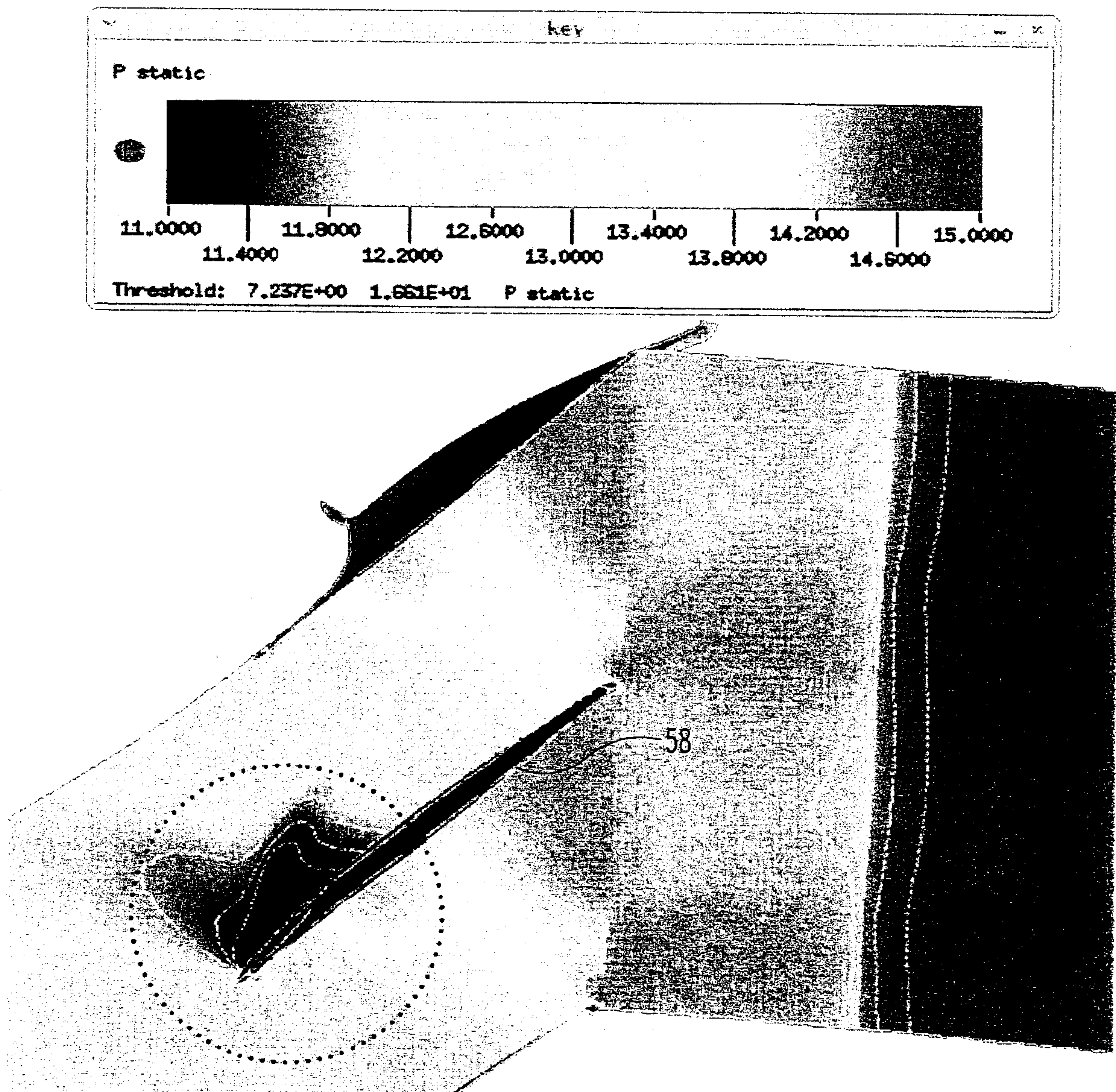


Fig. 3

**Fig. 4**

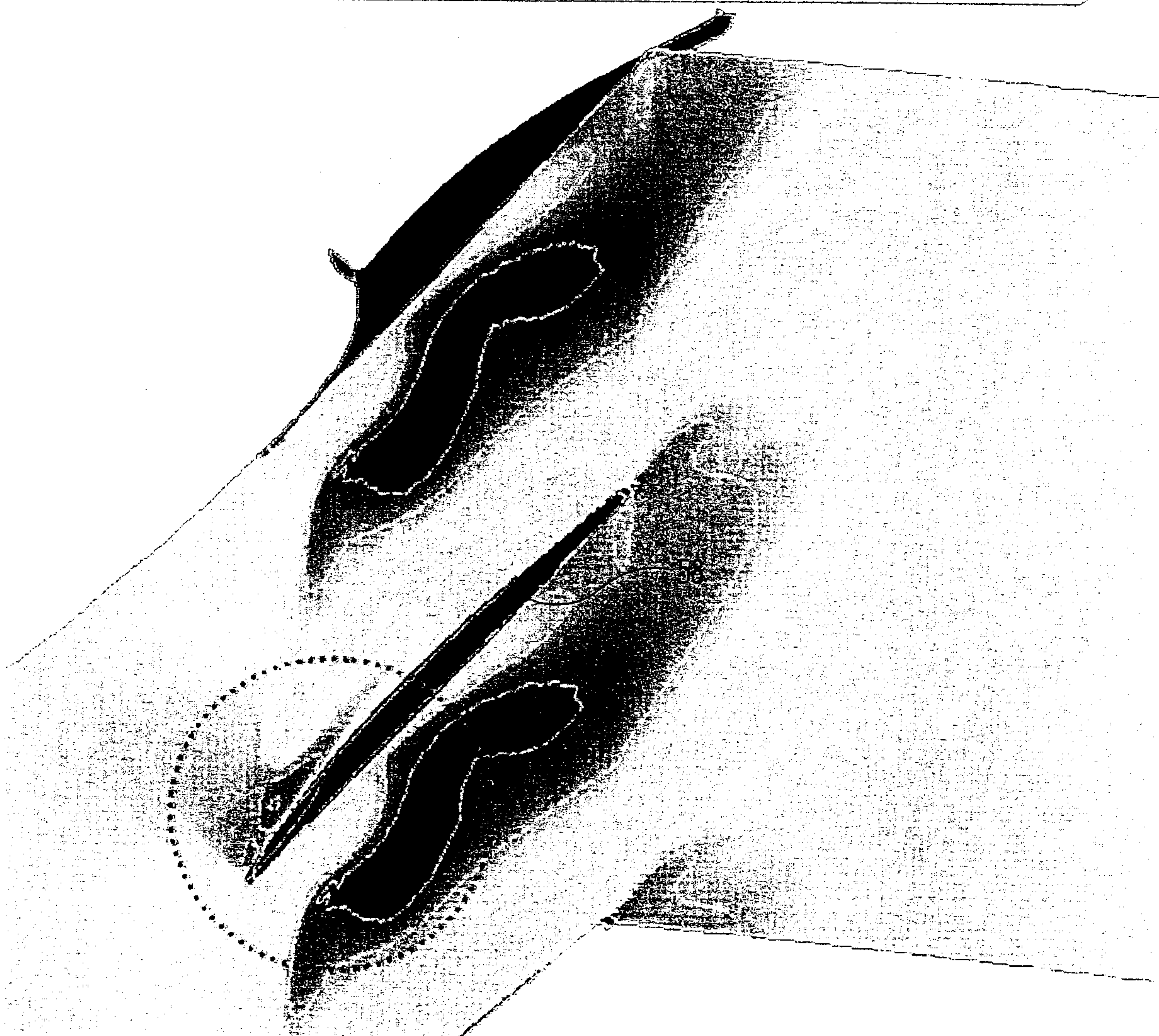
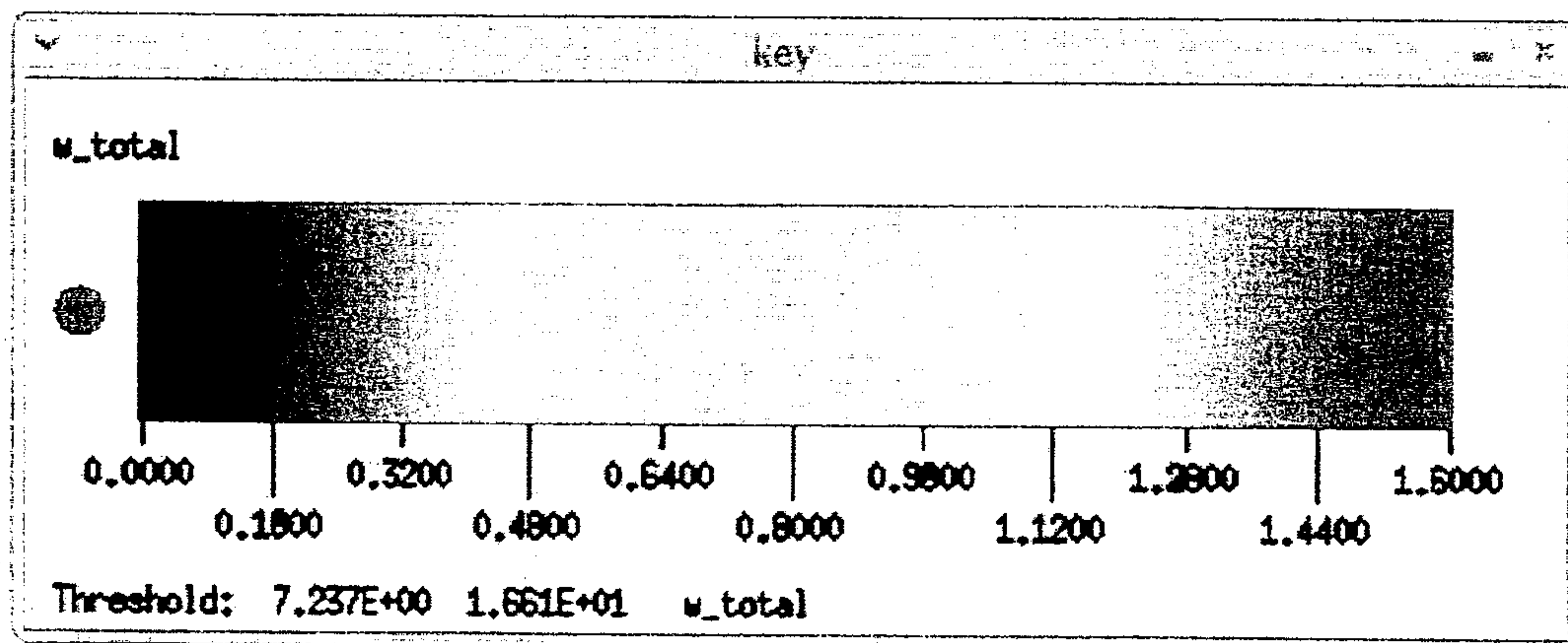


Fig. 5

Normalized Beta Distributions vs. Normalized Chord

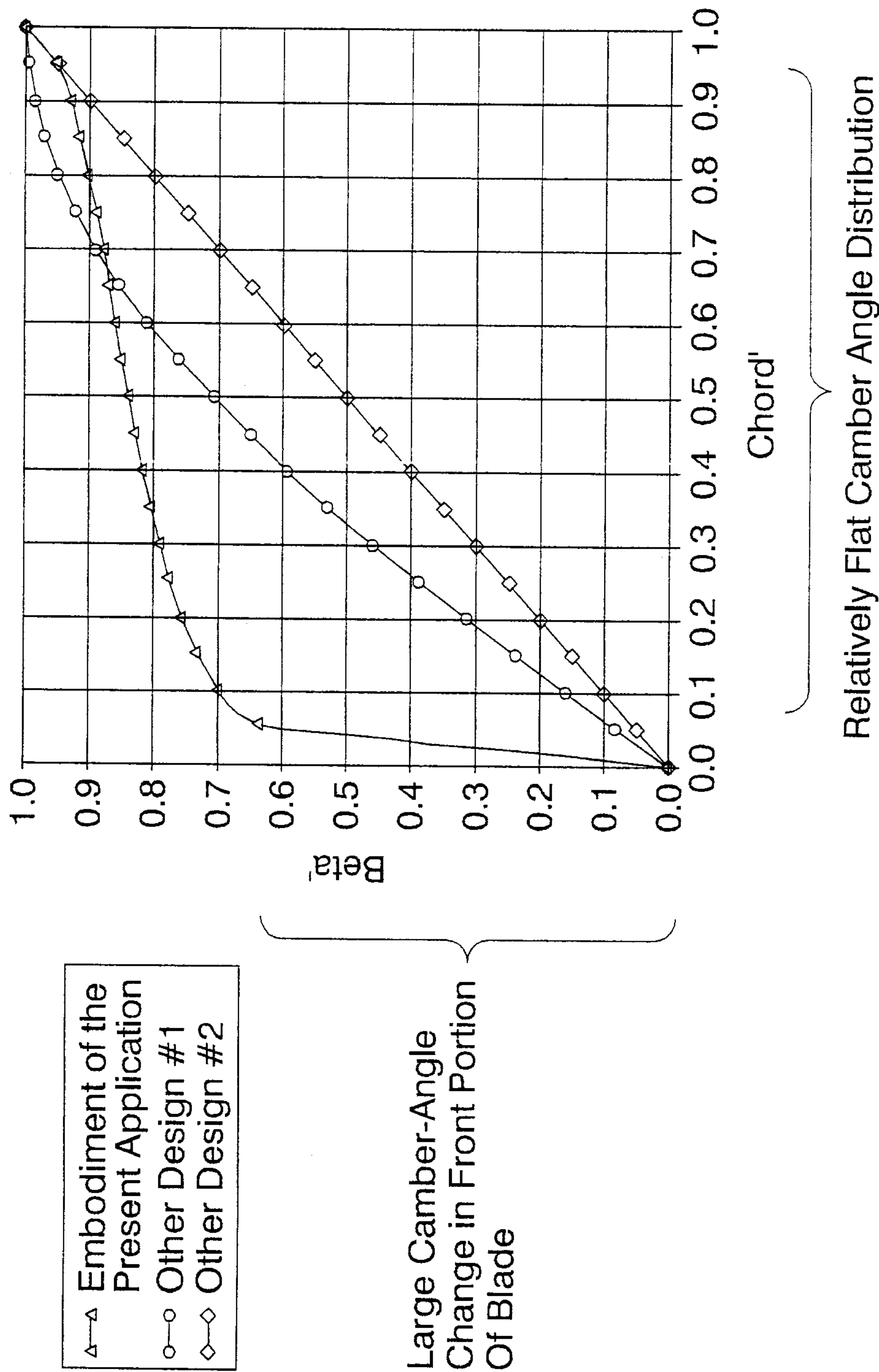


Fig. 7

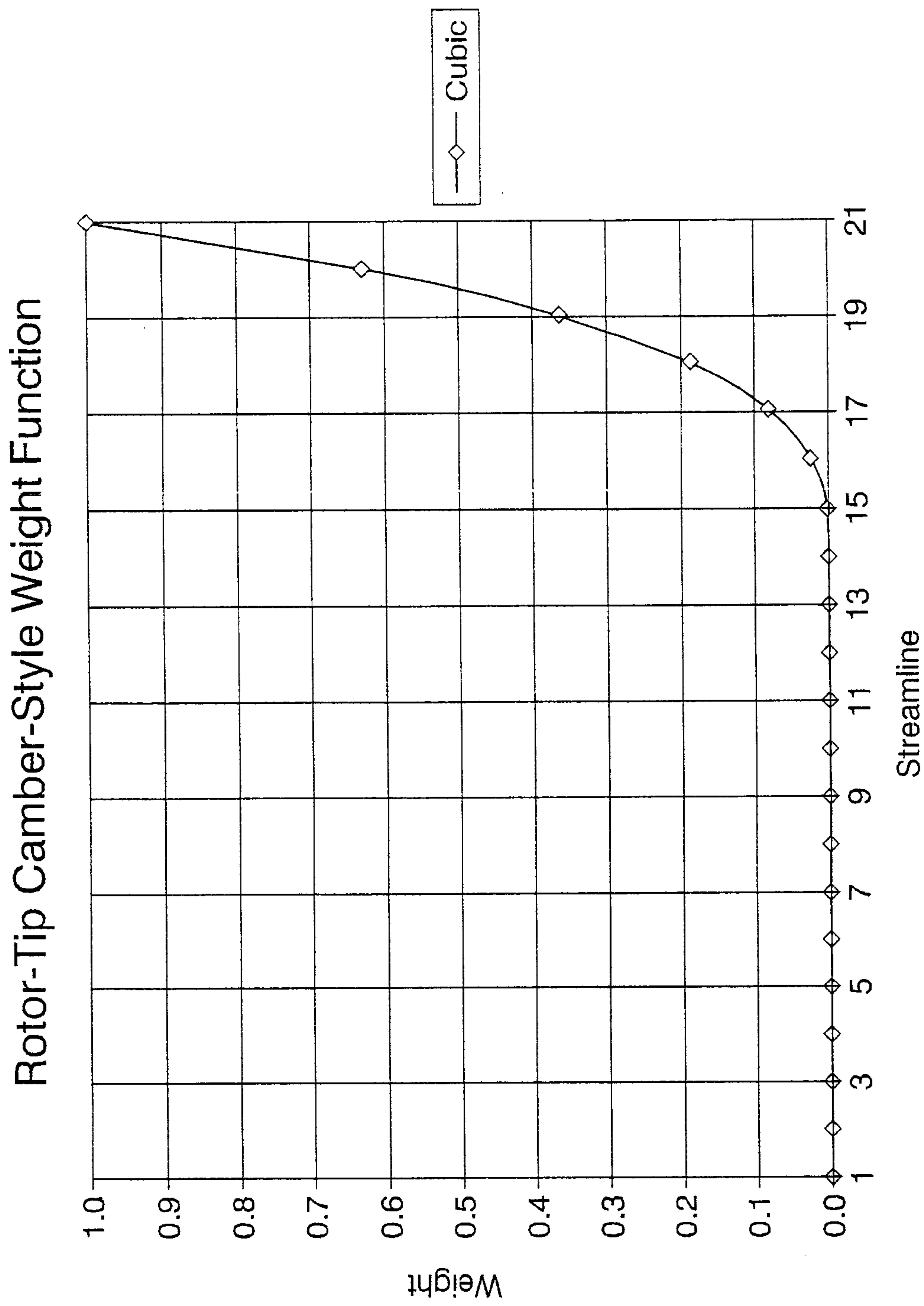


Fig. 8

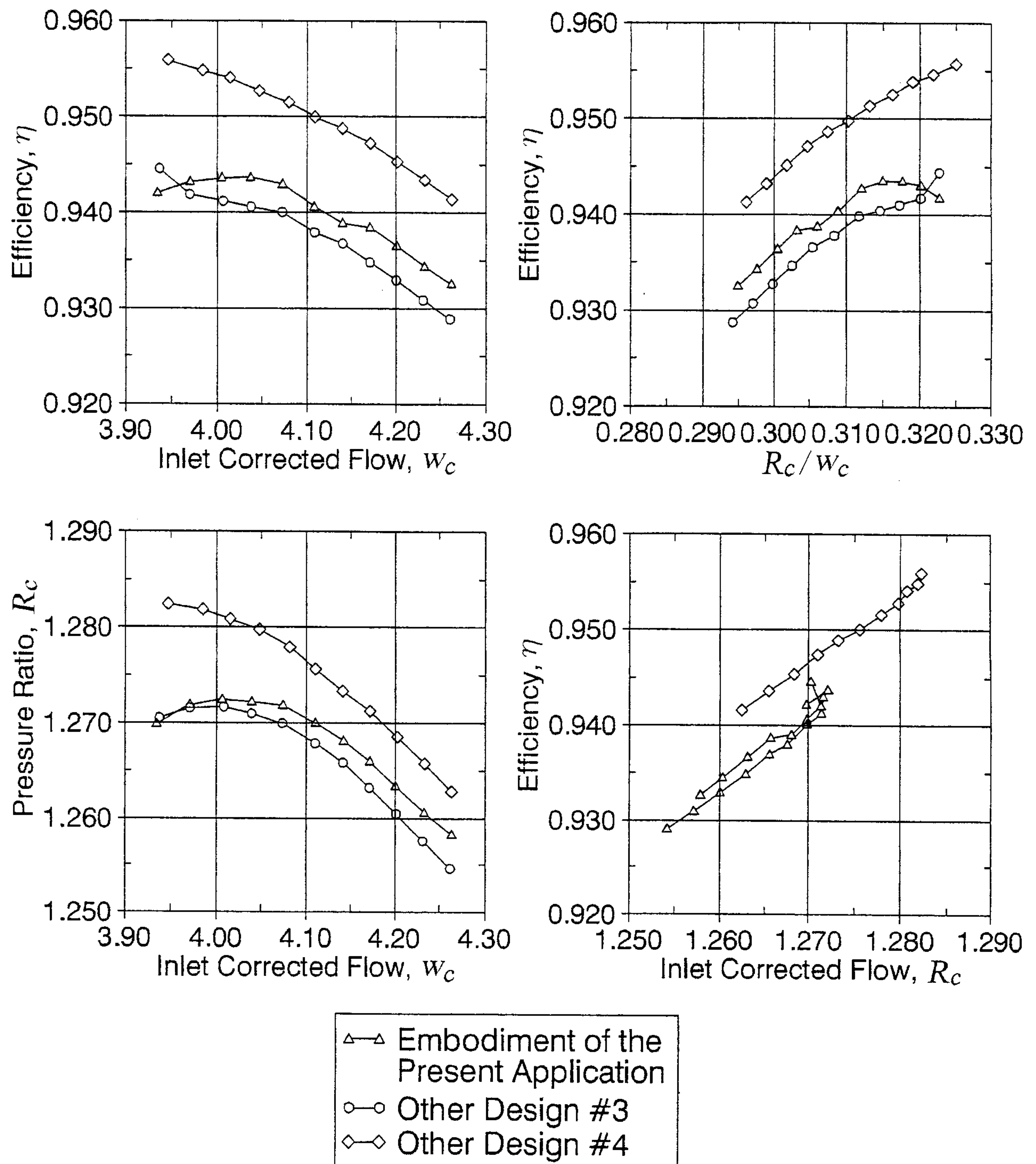


Fig. 9

