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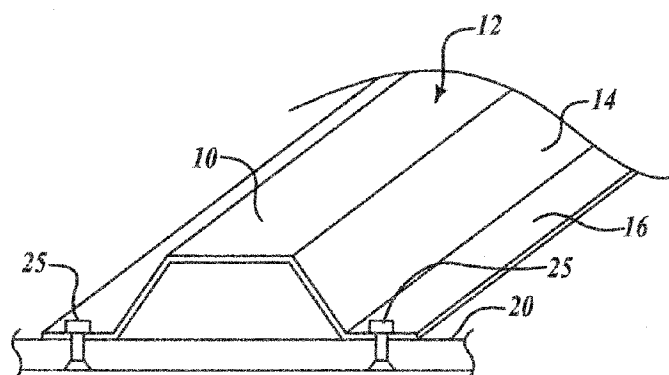


FIG. 1 (PRIOR ART)

(57) Abstract: A stiffened skin panel of an aircraft that includes an aircraft skin panel of composite material and a composite stringer consolidated with the aircraft skin panel. The stringer has a pair of stringer sides, each extending from a stringer top to a stringer leg. The stringer sides may each extend through a wide radius, smooth continuous curve to a stringer leg. The wide radius, smooth continuous curve may be proximate the base of the stringer side in a region where the stringer side transitions to the stringer leg.

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SHAPED COMPOSITE STRINGERS AND METHODS OF MAKING

TECHNICAL FIELD

The embodiments described herein generally relate to aircraft stringers, and more particularly relate to shaped composite stringers and methods of making these.

5

BACKGROUND

Aircraft generally include an airframe, which may be regarded as an underlying skeleton, to which skin panels are attached to form a smooth aerodynamic outer surface. The wings also include underlying structure covered with skin panels. Typically, skin panels are light and thin to minimize the weight of the aircraft and increase its payload and range. Since skin panels are thin, they are generally flexible and require stiffening to prevent undesired movement, flexing and vibration during flight.

Hat stringers have been used for decades in the aerospace industry for stiffening metal fuselage sections and metal wing skins on both commercial and military aircraft. These stringers are formed of thin metal panel with acute angles that result in a trapezoidal shape. Relatively simple metal-forming techniques are used to bend the metal into the acute angles required for this shape. These metal forming techniques include brake forming or rolling the metal into the hat stringer shape. These techniques allow the production of hat stringers with tight, constant angular bends and straight or flat legs. In FIG. 1, for example, a hat stringer 10 is attached to a fuselage skin panel 20 by a series of fasteners 25 (two shown) that are attached at intervals along the leg 16 of the hat stringer 10. The hat stringer 10 is trapezoidal having a flat upper surface 12 and sloping sides 14 that are at an angle to the legs 16, which are substantially aligned with the flat upper surface 12. The intersection between sides 14 and legs 16 may be characterized as sharp, tight or having a small radius of curvature.

Composite materials that include a resin matrix and filler have found increasing application in the aerospace industry because of their relatively light weight and favorable physical properties. In general, the fillers may be reinforcing or non-reinforcing in nature and may be in a variety of shapes, for example, powders, particulates, fibers tapes of unidirectional fibers, woven fabrics, and the like. The resins are organic polymeric materials that may include for example the commonly used epoxy resins.

As composite materials (e.g., carbon fiber and epoxy matrix resin) became more prevalent in the industry for fuselage skin and wing skin panels, hat stringers remained popular for a variety

of reasons. For example, while hat stringers can be made from a single stack of material, other less common shapes such as I-shaped, J-shaped, or T-shaped stiffeners require the combination of multiple stacks and radius fillers.

As stringers transitioned from metallic materials to composites, modifying the stringer
5 shape from the straight legs and tight bends of hat stringers was not a high priority. This is largely because the structural performance of traditional hat shapes is well understood and because when creating tools in which composite hats can be cured, straight or flat shapes with relatively small radii details are simpler to manufacture.

Composite hat stringers are now used extensively in certain commercial transport aircraft.
10 An example of a process for attaching composite stringers to a fuselage section is illustrated in FIG. 2. A hat stringer 10 rests on fuselage skin which is backed by a semi-rigid caul sheet 35. The interior of the hollow hat stringer 10 has an inflatable rubber bladder 36, and the exterior of the hat stringer 10 is covered by an inside mold line (IML) tool. The bladder 36 is inflated while the IML tool provides outside pressure on the hat stringer 10. Thus, the hat stringer 10 is
15 consolidated to the fuselage skin 20 by pressure of the IML tool 30 (and the bladder 36) and curing of resin, while the caul sheet 35 controls the contour of the fuselage skin 20.

This IML fabrication method requires flawless performance of several steps. For example, the composite material must be precisely located in the IML tool cavities to avoid over or under stuffing gaps. The bladders must inflate during cure to apply compaction and cure
20 pressure. The caul sheet must be flexible enough to account for variations in material thickness and/or ply mislocation while at the same time being sufficiently rigid to create a smooth aerodynamic surface.

An OML (outside mould line) cure process is shown in FIG. 3. These cauls panels 42, shaped like the hat stringers 10, assist in forcing the composite material into the radius during the
25 cure cycle and in smoothing out irregularities. Thus, the fuselage skin 20 is backed by an outside mold line tool (OML) 32 and the hat stringer 10 rests on the fuselage 20, with a bladder 36 in its interior. A caul panel 42 conforming to the exterior shape of the hat stringer 10 rests on the stringer 10. A bag 40 envelopes at least the caul panel 42 and the upper surface of the fuselage skin 20. A vacuum is pulled on the bag 40 while bladder 36 is inflated and pressure is applied via
30 IML tool 32 to consolidate the hat stringer 10 to the fuselage as the resin cures to bond the composite fuselage skin to the composite stringer 10. This caul panel-assisted co-cure method requires the fabrication and use of hundreds of uniquely designed, high maintenance, cauls and associated master tools which imposes a significant cost burden.

Accordingly, it is desirable to develop composite stringers and methods of making these that simplify processes and reduce costs. In addition, it is desirable that the composite stringers have large radii to facilitate forming the stringer out of a composite blank. Further, the stringer shapes should perform as well, or better, structurally (e.g., resist column buckling and four-point bending) than traditional trapezoidal shaped hats, when made with composite materials. In addition, there is a need for methods of making composite stringers that have smooth and gentle (larger) radii of curvature near the stringer base and that eliminate the need to use caul sheets to assist the co-cure of stringer to panel. Furthermore, other desirable features and characteristics of the composite stringers will become apparent from the subsequent detailed description and the appended claims, taken in conjunction with the accompanying drawings and the foregoing technical field and background.

BRIEF SUMMARY

An exemplary embodiment provides a stiffened skin panel of an aircraft that includes an aircraft skin panel of composite material and a composite stringer consolidated with the aircraft skin panel. The stringer has a pair of stringer sides that each extend from a stringer top to a stringer leg. The stringer sides each curve smoothly and continuously through a wide radius curve to the stringer leg.

Another exemplary embodiment provides a stiffened panel of composite material and a composite stringer consolidated with the panel. The stringer has a pair of stringer sides, each stringer side extending from a stringer top to a stringer leg. An attachment zone includes a portion of the stringer and a portion of the panel proximate stringer point of attachment to the panel. The attachment zone is substantially free of broken or separated filler of the stringer composite material.

A further example of an embodiment provides a method of stiffening a composite panel with a stringer. The method includes the steps of determining the location and spacing of stringers on a composite panel to be stiffened and positioning the composite panel on a mold tool. It further includes laying down forms conforming to the shapes of the stringers to be formed on the composite panel, at determined locations and intervals. Further the method includes preparing lay ups of composite material for forming into stringers; laying the prepared lay up of composite material over each of the forms; and enclosing at least the composite panel and lay up-covered forms with a vacuum bag. The stringers are consolidated to the composite panel under heat and pressure to produce a stiffened composite panel. The produced panel is characterized in

that the stringers each have a pair of stringer sides that extend smoothly through a wide radius, continuous curve to a stringer leg.

BRIEF DESCRIPTION OF THE DRAWINGS

Various embodiments will hereinafter be described in conjunction with the following
5 drawing figures, which are not to scale and which may be schematic in nature, wherein like numerals denote like elements, and

FIG. 1 is a perspective view of a prior art hat stringer attached to an aircraft skin panel;

FIG. 2 is a schematic end view depicting a prior art inside mold line lay up using an inflatable bladder and caul sheet;

10 FIG. 3 is a schematic end view of a prior art outside mold line lay up using a caul panel and an inside bladder;

FIG. 4 is a schematic end view of an exemplary embodiment of a lay up using an OML tool to fabricate a shaped composite stringer;

15 FIGS. 5-9 illustrate a non-limiting list of exemplary embodiments of composite stringer shapes in end view; and

FIG. 10 illustrates an exemplary embodiment of a method of making a stiffened panel.

DETAILED DESCRIPTION

The following detailed description is merely exemplary in nature and is not intended to limit the described embodiments or the application and uses of the described embodiments.
20 Furthermore, there is no intention to be bound by any expressed or implied theory presented in the preceding technical field, background, brief summary or the following detailed description.

Exemplary embodiments provide improved stringer structure and consolidation with a panel to be stiffened. By reconfiguring the stringer shape so that the tight radii are eliminated, embodiments provide a smooth shape with large radius curvature, at least where the stringer sides
25 merge into the stringer legs. The sides of the stringer may have an inflexion point that is a smooth transition point. The composite material used to form the stringer drapes naturally to conform to the smooth shape. The contouring is sufficiently smooth and gradual (i.e., without sharp transitions and tight radii) that it permits the use of tape plies instead of fabric, resulting in significant weight savings. Structural integrity is maintained without impairment of stringer
30 performance.

In general, some of the exemplary embodiments of the stringers have sides that include an intermediate zone that in turn includes an inflexion point where the curvature of the stringer side

transitions smoothly from a convex curve to a concave curve or from a concave curve to a convex curve. The term “smooth” as it relates to the curvature of the stringer means transitioning from a straight line to a curve at a tangent point to the curve or from one curve to another curve at a mutual tangent point. The term “continuous as it pertains to curves, means there are no
5 discontinuities in the smooth curvature and no sharp bends or kinks. The intermediate zone may be located along the stringer side proximate the center of the stringer (with reference to stringer height) or proximate the lower end or upper end of the stringer side. Exemplary embodiments of the stringers are symmetrical on either side of a vertical center line, as shown in FIGS. 5-9, for example, and as described below.

10 Exemplary embodiments allow the fabric to drape to shape via its natural path based on its own inherent flexibility. This simplifies the fabrication process. Tape or fabric is applied over a shaped cure-mandrel without need for a stringer pre-forming operation, as is required for the tight radii of the traditional trapezoid hat stringer. The cure mandrel used with these special shaped stringers may be selected from a variety of materials and may include for example a
15 mandrel of a flyaway material (i.e., low density foam which eliminates the need for inflation bladders), an inflatable rubber bladder, or a removable solid or hollow mandrel. Foam mandrels may be selected based on specific performance properties such as, acoustical attenuation, flame retardation and structural load carrying capability. By increasing the load carrying capability of the stringer through unique shaping, the cross sectional area may be reduced and weight savings
20 may be realized

FIG. 4 is an exemplary embodiment of a lay up 100 that includes an OML tool 200 which supports the underside of an aircraft skin panel 150, such as a fuselage or wing panel. A mandrel or inflatable bladder 120 rests on the upper surface of the aircraft skin panel 150, and flexible composite plies 110 are draped over the mandrel or bladder 120. A vacuum bag 130 envelopes
25 at least the ply-draped mandrel or bladder 120 to apply pressure to the press the plies 110 to the mandrel or bladder 120 when a vacuum is pulled on the bag 130. Because of the gradual smooth curves formed by draping of plies 110, the exemplary embodiment has no sharp or tight angles and has an inflexion point 102 along sides of the stringer to be formed when the plies 110 are cured. When the plies 110 are consolidated to the aircraft skin panel 150 by pressure and co-
30 curing of plies 110 and skin panel 150, the stringer 300 formed (FIG 6) from plies 110 is smoothly curved and the legs 308 of the stringer extend from the sides 304 of the stringer 300 in a smooth curve devoid of any sharp turns, “kinks” or tight angles. Stringer 300 may be characterized as parabolic or bell-shaped.

The “attachment zone” of a composite stringer to a skin panel means the most inboard (relative to the stringer) zone where the stringer attaches to the panel, which is also known as the flange. The exemplary embodiments presented here that have smooth, wide radius, continuous, curvature from stringer side to stringer leg minimize or eliminate any potential for ply breakage, separation from a main body of filler (e.g., woven fabric or tape) and also minimizing excess resin accumulation.

A non-limiting group of further exemplary embodiments of composite stringer shapes are illustrated in FIGS. 5-9. Other shapes are, of course, also possible and within the scope of the presently disclosed technology.

In FIG. 5, stringer 300 is substantially parabolic in shape and has a rounded top 302 curving downward to sides 304 in a convex curve. The sides 304 transition from the rounded top 302 at tangent points 305 to the curvature of the top 302 and then descend in a straight line before a transition at the tangent point of a wide radius curve 306 to the stringer legs 308.

Stringer 310 shown in FIG. 6 has a substantially flat top 312 with rounded curves 313 leading to sides 314 that extend smoothly downward, substantially vertically. Each of the sides 314 in this example form an inflexion point 315. The lower ends of sides 314 curve through a large radius turn 318 and extend to legs 316.

Stringer 320 shown in FIG. 7 may be characterized as substantially bell-shaped and also has a substantially flat top 322 with rounded extremities 325 extending to sides 324 that extend downward and curve through a large radius turn 328 to legs 326. As indicated, sides 324 may extend straight downward at a 90° angle to at and angle α that may range from about 90 to about 45°.

Stringer 340 shown in FIG. 8 has a rounded top 342 and is substantially parabolic or bell-shaped. The rounded top 342 extends smoothly in a convex curvature to sides 344 that spread outward as they extend downward. The sides 344 then extend through a wide angle curve 345 to form legs 346. The angle α , as indicated, may range from about 35 to about 65°.

Stringer 350 shown in FIG. 9 has a rounded top 352 and is substantially parabolic or bell-shaped and has a rounder overall shape with shorter straight sides than stringer 340 of FIG. 8. The rounded top 352 extends smoothly in a convex curvature to sides 354 that spread outward as they extend downward. The sides 354 then extend through a wide angle curve 355 to form legs 356. The angle α , as indicated, may range from about 35 to about 65°.

From the foregoing examples, it is apparent that many composite stringer shapes are possible and within the scope of the presently disclosed technology, provided that these shapes

include a smooth, wide-radius transition from the stringer sides to the stringer legs (and in some instances also from the stringer top to the stringer sides) that minimizes or eliminates the potential for excess resin accumulation at the transition point, fiber parting, delamination, fiber wrinkling, waviness or marcelling (in-plane waviness) and porosity in the matrix resin.

5 Because exemplary embodiments of the stringers have smooth shapes (at least where the stringer sides merge into the stringer legs and merge into the stringer top), it is possible to use tape plies with these embodiments rather than fabric prepregs. Tape plies have each ply of the tape in a single plane. The orientation of plies in the tape permits the use of a series of optimally oriented tapes to form stringers. The use of tape plies reduces material costs and offers potential
10 for weight reduction.

FIG. 10 illustrates an exemplary embodiment of a method 400 of making the shaped composite stringers. Of course, other methods may also be used. In this instance, an OML tool is used. In process 410, a determination is made about stringer spacing and location on the skin panel, wing, fuselage or other component to be stiffened with stringers. Once a determination
15 has been made, the component is laid down on the OML tool, in process 420. Bladders or mandrels in the shape of the stringer are then laid on the component at predetermined positions and intervals, in process 430. A tape lay up (or other composite fiber lay up) is then prepared in process 440. The tape lay up may have a series of oriented stacked layers and the tape may be pre-impregnated with a suitable resin, such as epoxy resin. The tape lay up is laid over the
20 bladder or mandrel, in process 450. At this point, the assembly of component, bladder or mandrel, and tape lay up is bagged within a vacuum bag, in process 460. A vacuum is pulled on the bag, and the tape lay up is consolidated to the component, in process 470. Typical temperature may be about 350 °F for a typical epoxy resin matrix, and pressure may be about 85 psi. Other temperature and pressure conditions may apply to other resins. Generally, but not
25 necessarily, the resin of the panel and of the stringer is the same. Accordingly, during consolidation the resin of the stringer and the resin of the panel are concurrently cured or “co-cured.” This facilitates formation of a strong bond between the stringer and the panel as resin flows and consolidates into a matrix in the attachment zone includes both panel and stringer fillers. After consolidation, the bag is removed, in process 480. The stringer is now consolidated
30 with the component. In the exemplary process, the flexible draped tape forms a smoothly curved stringer thereby minimizing the possibility of structural issues in the stringer. The stringer is free of tight radius curves or abrupt bends and has smooth, continuous, wide-radius curves. Further

in some exemplary embodiments, the use of caul panels or caul sheets is not required. This eliminates the costs associated with using these devices.

While at least one exemplary embodiment has been presented in the foregoing detailed description, it should be appreciated that a vast number of variations exist. It should also be
5 appreciated that the exemplary embodiment or exemplary embodiments are only examples, and are not intended to limit the scope, applicability, or configuration of the described embodiments in any way. Rather, the foregoing detailed description will provide those skilled in the art with a convenient road map for implementing the exemplary embodiment or exemplary embodiments. It should be understood that various changes can be made in the function and arrangement of
10 elements without departing from the scope as set forth in the appended claims and the legal equivalents thereof. For example, the stringers could be cured in a separate operation and bonded to the skin with adhesive in a secondary operation.

CLAIMS

1. A stiffened skin panel of an aircraft comprising:
a composite aircraft skin panel (150); and
5 a composite stringer (300) consolidated with the aircraft skin panel (150), the stringer (300) comprising a pair of stringer sides (304) each extending from a stringer top (302) to a stringer leg (308), the stringer sides (304) each extending through a wide radius (306), smooth continuous curve to a stringer leg (308).
2. The stiffened skin panel of Claim 1, wherein an upper portion of the composite
10 stringer (300) is substantially parabolic or elliptical in cross section.
3. The stiffened skin panel of Claim 1, wherein the composite stringer (310) is substantially bell-shaped in cross section.
4. The stiffened skin panel of Claim 1, wherein the stringer top comprises a substantially flat stringer top (322) or a convex curved shape (342).
- 15 5. The stiffened skin panel of Claim 1, wherein the stringer sides (304) each comprise an intermediate zone comprising an inflexion point (305).
6. The stiffened skin panel of Claim 5, wherein the intermediate zone is located proximate a center of each stringer side (304).
7. The stiffened skin panel of Claim 5, wherein the intermediate zone is located
20 proximate a lower portion of each stringer side (304).
8. The stiffened skin panel of Claim 1, wherein the composite stringer (300) comprises consolidated stacked tape plies, at least some of the stacked tape plies oriented in different directions.
9. The stiffened skin panel of Claim 1, wherein the composite aircraft skin panel
25 (150) and the composite stringer (300) comprise a common resin and filler material.
10. The stiffened skin panel of Claim 1, further comprising an attachment zone comprising a portion of the composite stringer (300) and a portion of the composite aircraft skin

panel (150) proximate a stringer point of attachment to the composite aircraft skin panel (150), the attachment zone substantially free of excess resin.

11. The stiffened skin panel of Claim 1, further comprising an attachment zone comprising a portion of the stringer (300) and a portion of the composite aircraft skin panel (150)
5 proximate a stringer point of attachment to the composite aircraft skin panel (150), the attachment zone substantially free of composite fibers separated from and protruding from fibrous material of the composite stringer (300).

12. A stiffened panel comprising:
a panel (150) comprising composite material;
10 a stringer (300) consolidated with the panel (150), the stringer (300) comprising composite material, the stringer comprising a pair of stringer sides (304), each stringer side (304) extending from a stringer top (302) to a stringer leg (308); and
an attachment zone comprising a portion of the stringer (300) and a portion of the panel (150) comprising a stringer point of attachment to the panel (150), the attachment zone
15 substantially free of broken plies of the composite material of the stringer (300).

13. The stiffened panel of Claim 12, wherein an upper portion of the stringer (300) is substantially parabolic or elliptical in cross section.

14. The stiffened panel of Claim 12, wherein the stringer (310) is substantially bell-shaped in cross section.

20 15. The stiffened panel of Claim 12, wherein the stringer (300) comprises consolidated stacked tape plies, at least some of the stacked tape plies oriented in different directions.

16. The stiffened panel of Claim 12, wherein the attachment zone is free of excess resin.

25 17. The stiffened panel of Claim 12, wherein the attachment zone is free of composite fibers separated from and protruding from fibrous material of the stringer.

18. A method of stiffening a composite panel with a stringer, the method comprising:
determining location and spacing of stringers on a composite panel to be stiffened;

positioning the composite panel on a mold tool;

laying down on the composite panel, at determined locations and spacing, forms conforming to a shape of the stringers to be formed;

preparing lay ups of composite material for forming into stringers;

5 laying a prepared lay up of composite material over each of the forms;

enclosing at least the composite panel and lay up-covered forms with a vacuum bag;

consolidating the stringers to the composite panel under heat and pressure; and

producing a stiffened composite panel wherein the stringers each comprise a pair of stringer sides each extending from a stringer top to a stringer leg, and each stringer side
10 extending in a continuous smooth curve to a stringer leg.

19. The method of Claim 18, wherein the step of positioning comprises positioning in an outside mold line tool.

20. The method of Claim 18, wherein the step of consolidating comprises consolidating by forming attachment zone free of composite fibers separated from and
15 protruding from fibrous material of the stringer and free of excess resin.

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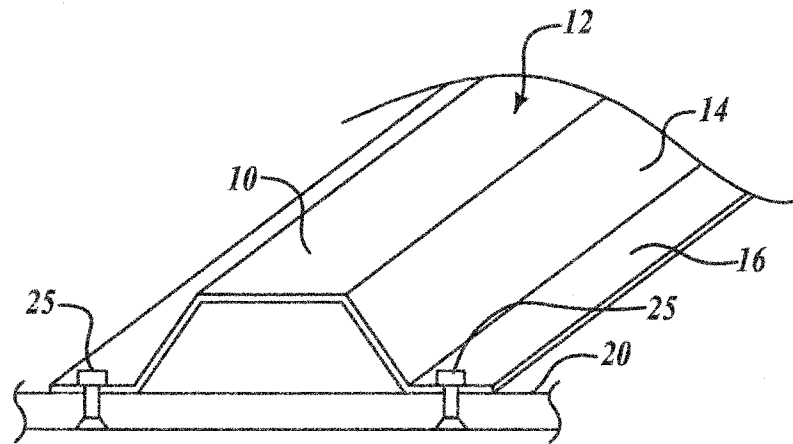


FIG. 1 (PRIOR ART)

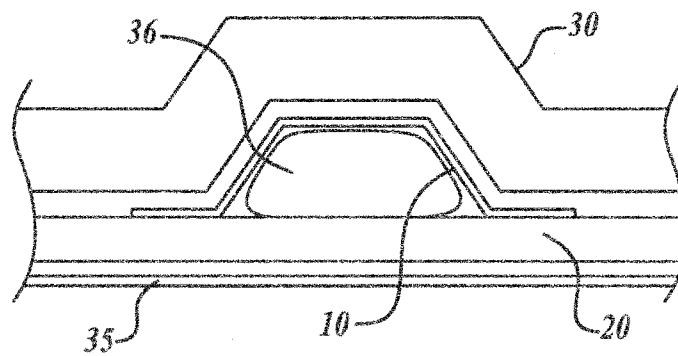


FIG. 2 (PRIOR ART)

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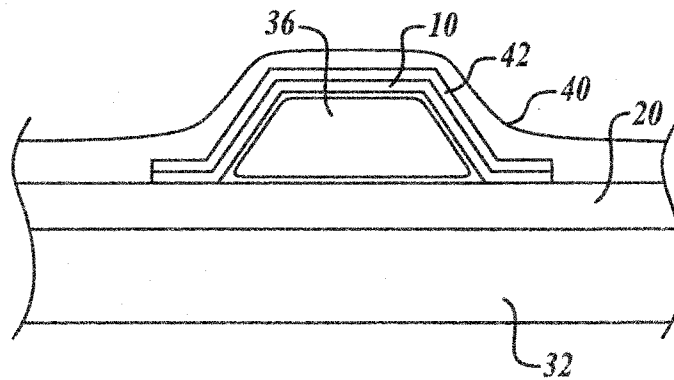


FIG. 3 (PRIOR ART)

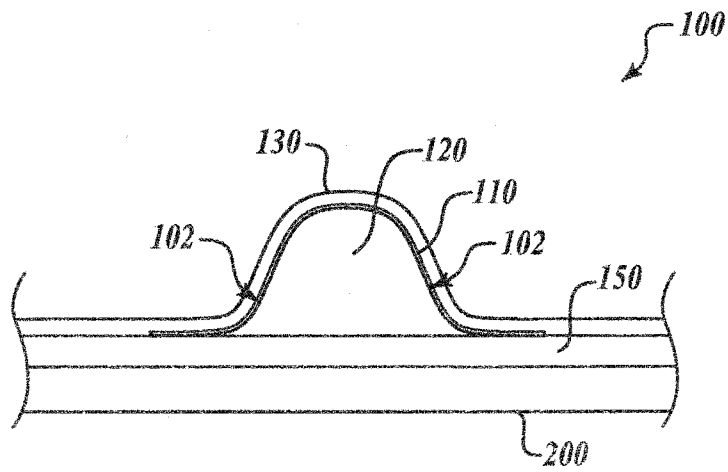


FIG. 4

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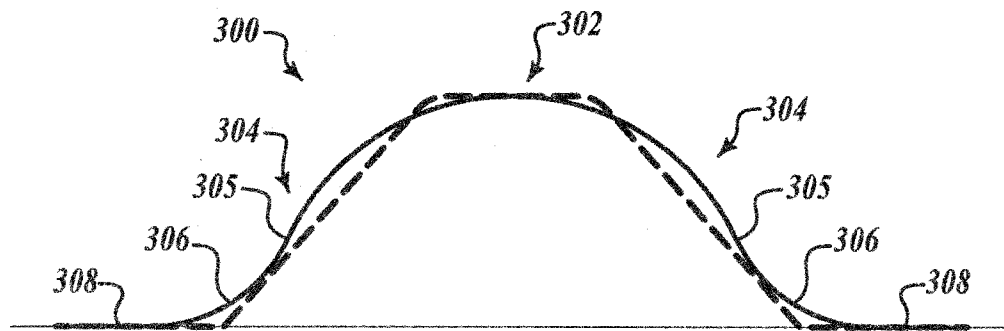


FIG. 5

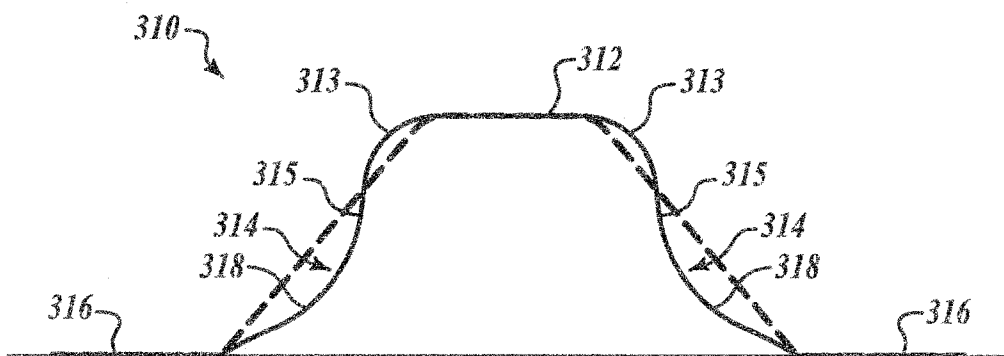
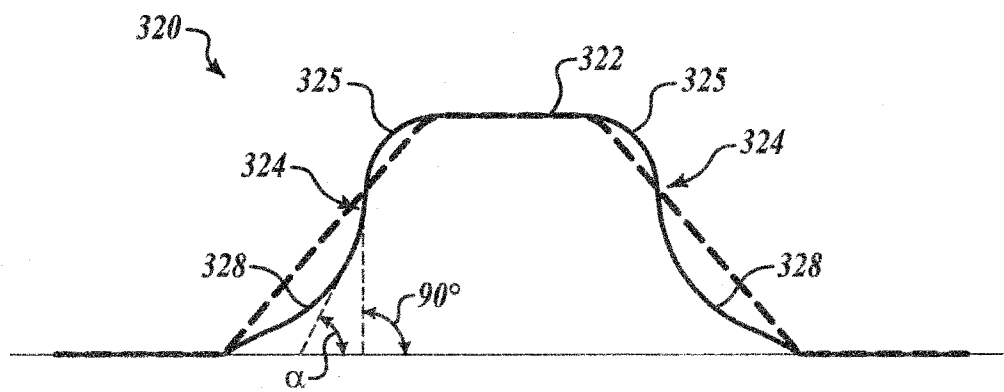
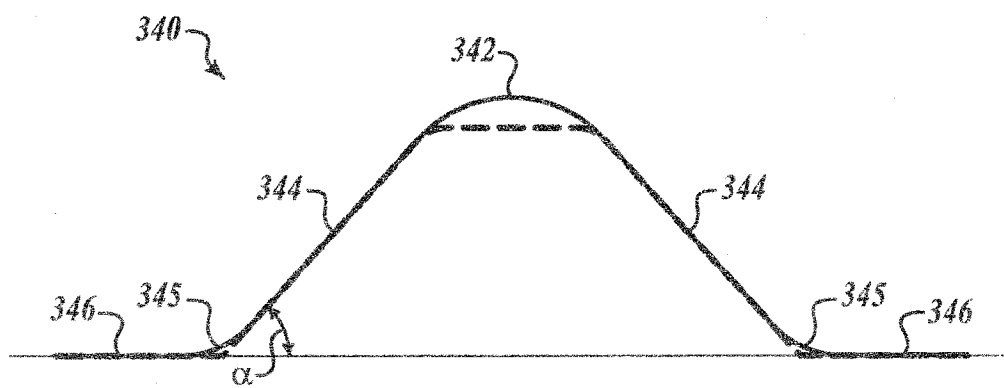
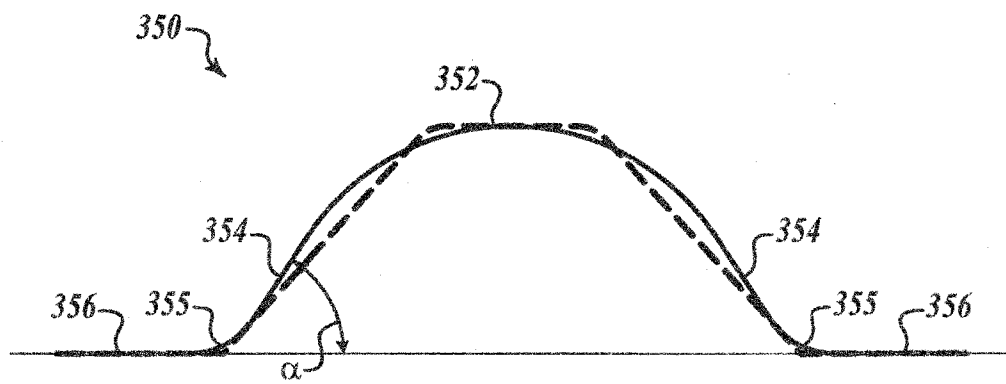


FIG. 6

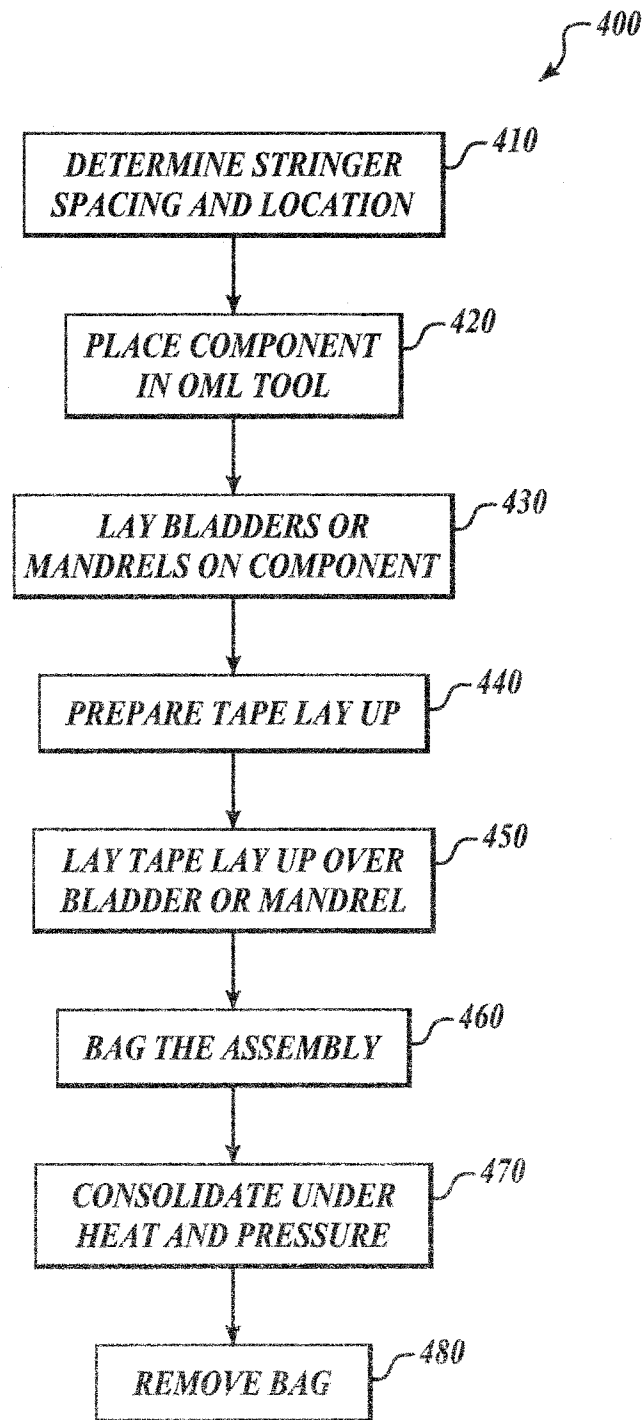
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**FIG. 7****FIG. 8**

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**FIG. 9**

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**FIG. 10**