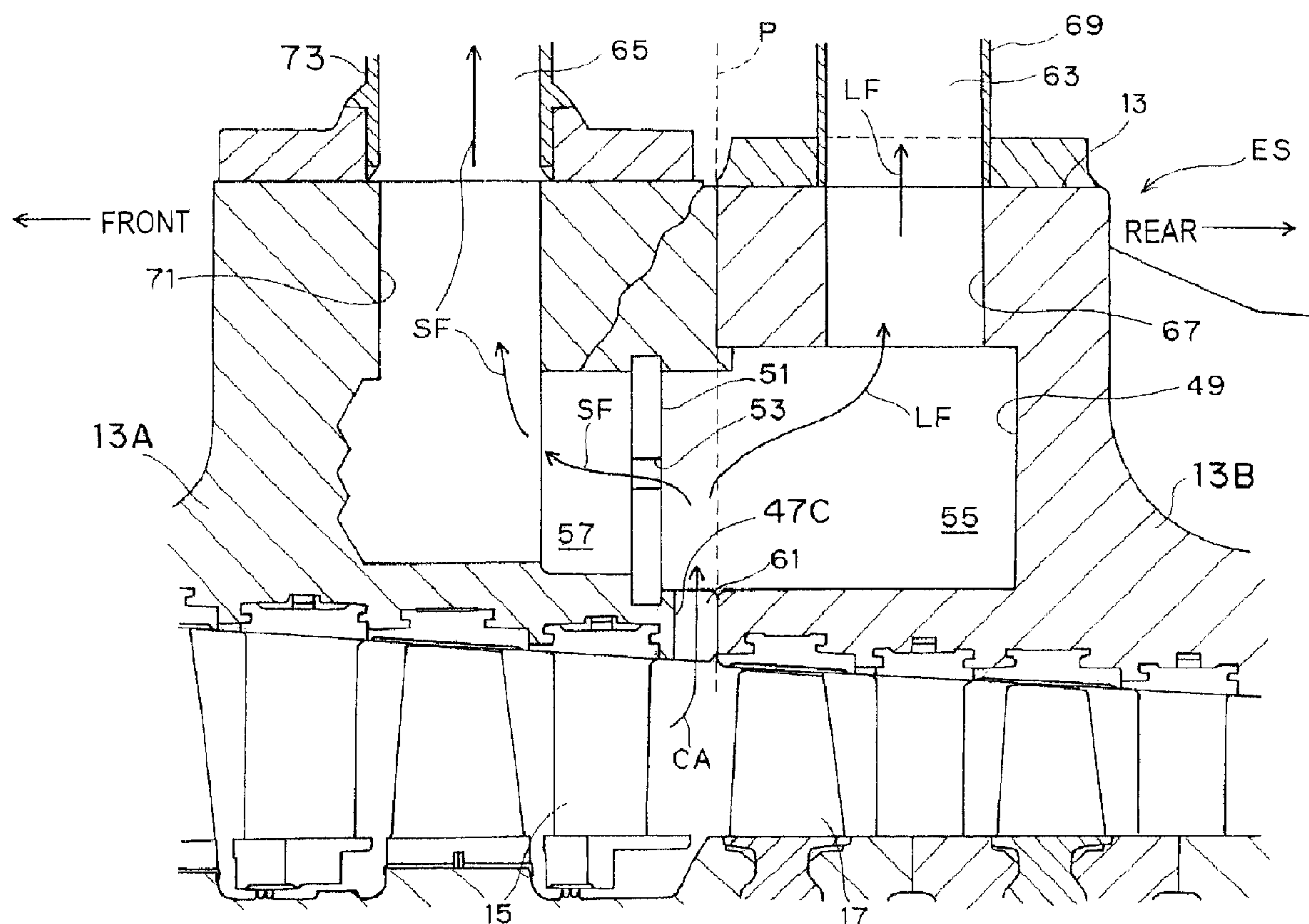




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AND GAS TURBINE ENGINE WITH THE STRUCTURE



(57) Abrégé/Abstract:

Provided is a structure of a gas turbine engine for extracting a part of compressed air generated by a compressor. The structure comprises a cylindrical housing 13 extending around an axis C and forming therein at least a part of a compressing chamber in



(57) **Abrégé(suite)/Abstract(continued):**

which air is flowed in a direction parallel to the axis as it is compressed to generate the compressed air, the housing having an annular portion 43A, 43B extending around the compressing chamber to define an annular chamber 54, an annular partition 51 dividing the annular chamber into an annular first plenum chamber 55 and an annular second plenum chamber 57 positioned on an upstream side of the first plenum chamber with respect to the direction and separated from the annular first plenum chamber. The cylindrical housing has first apertures 61 defined therethrough to communicate between the compressing chamber and said first plenum chamber. The partition has second apertures 53 defined therethrough to communicate between the first and second plenum chambers. The first and second apertures are configured so that that the first apertures have a first total cross-sectional area and the second apertures have a second total cross-sectional area which is smaller than the first total cross-sectional area.

ABSTRACT

Provided is a structure of a gas turbine engine for extracting a part of compressed air generated by a compressor. The structure comprises a cylindrical housing 13 extending around an axis C and forming therein at least a part of a compressing chamber in which air is flowed in a direction parallel to the axis as it is compressed to generate the compressed air, the housing having an annular portion 43A, 43B extending around the compressing chamber to define an annular chamber 54, an annular partition 51 dividing the annular chamber into an annular first plenum chamber 55 and an annular second plenum chamber 57 positioned on an upstream side of the first plenum chamber with respect to the direction and separated from the annular first plenum chamber. The cylindrical housing has first apertures 61 defined therethrough to communicate between the compressing chamber and said first plenum chamber. The partition has second apertures 53 defined therethrough to communicate between the first and second plenum chambers. The first and second apertures are configured so that that the first apertures have a first total cross-sectional area and the second apertures have a second total cross-sectional area which is smaller than the first total cross-sectional area.

STRUCTURE FOR EXTRACTING COMPRESSED AIR FROM COMPRESSOR OF
GAS TURBINE ENGINE AND GAS TURBINE ENGINE WITH THE
STRUCTURE

5 TECHNICAL FIELD

[0001] The present invention relates to a structure for extracting a compressed air from a compressor of gas turbine engine. Also, the present invention relates to a gas turbine engine incorporating the structure.

10

BACKGROUND OF THE INVENTION

[0002] In a gas turbine engine with a multi-stage compressor, a compressed air generated by a compressor is in part used for the cooling of a turbine, for example.

15 The compressed air is extracted from the compressor through a plurality of slits formed in a compressor housing at regular intervals in the circumferential direction so as to prevent a pressure drop in the air extracting zone and the resultant uneven pressure distribution in the compressor
20 with respect to the circumferential direction.

[0003] For this purpose, JP 2001-304194 (A) discloses to provide various sizes of slits in order to extract various amounts of compressed air depending on usages of the extracted air. This technique may be advantageous in that
25 a suitable pressure distribution is obtained for various

amounts of air to be extracted.

[0004] On the other hand, the technique requires that the various sizes of slits are formed integrally in the compressor housing, which complicates design modifications
5 of the slits according to the requirement of change in the amount of air. Also, the slits are defined by different members defining inward and outward portions of the compressor housing. This complicates a dimension control of, in particular, narrow slits for extracting smaller amount
10 of air, which needs elevated size precisions. Therefore, an even pressure distribution may not be obtained within the interior of the compressor.

[0005] An object of the present invention is to provide a structure for extracting a part of compressed air
15 generated by a compressor, which makes it easy to control a dimension of the narrow slits for extracting smaller amount of air and maintains an even pressure distribution within the interior of the compressor of the gas turbine engine by a simple arrangement defined therein.

20

SUMMARY OF THE INVENTION

[0006] To this end, a structure of a gas turbine engine for extracting a part of compressed air generated by a compressor, comprises

a cylindrical housing extending around an axis and forming therein at least a part of a compressing chamber in which air is flowed in a direction parallel to the axis as it is compressed to generate the compressed air, the
5 housing having

an annular portion extending around the compressing chamber to define an annular chamber,

an annular partition dividing the annular chamber into an annular first plenum chamber and an annular
10 second plenum chamber positioned on an upstream side of the first plenum chamber with respect to the direction and separated from the annular first plenum chamber,

the cylindrical housing having a plurality of first apertures defined therethrough to communicate between
15 the compressing chamber and said first plenum chamber;

the partition having a plurality of second apertures defined therethrough to communicate between the first and second plenum chambers, the first and second apertures being configured so that that the first apertures
20 have a first total cross-sectional area and the second apertures have a second total cross-sectional area which is smaller than the first total cross-sectional area;

the second plenum chamber being designed to be communicated to a second extracting passage for extracting
25 the compressed air from the second plenum chamber.

[0007] According to the arrangement, because the second apertures with the smaller total cross section area are provided on the downstream side of the first apertures with the larger total cross section area, a substantially even
5 pressure is established in the circumferential direction within the compressing chamber. Also, an amount of extraction air can be varied simply by changing the size of the second apertures.

[0008] In another aspect of the extraction structure for
10 the gas turbine engine according to the invention, the first plenum chamber is designed to be communicated to a first extracting passage for extracting the compressed air from the first plenum chamber.

[0009] In another aspect of the extraction structure
15 for the gas turbine engine according to the invention, the first extracting passage has a valve provided therein for closing the first extracting passage.

[0010] In another aspect of the extraction structure for the gas turbine engine according to the invention, the
20 cylindrical housing has a first housing portion and a second housing portion positioned on the upstream side of the first housing portion, the first and second housing portions having opposed first and second radially extending mating surfaces, respectively, which mate with each other
25 in a sealing engagement when the first and second housing

portions are assembled with each other, each of the first apertures being defined by a recess formed on first or second mating surface or respective recesses formed on first and second mating surfaces, the first and second
5 mating surfaces having annular first and second recesses defined therein, respectively, to form the annular chamber when the first and second housing portions are assembled with each other.

[0011] In another aspect of the extraction structure for
10 the gas turbine engine according to the invention, the second housing portion is made of a plurality of housing segments, the housing segments having respective annular chamber portions forming a part of the annular chamber when the housing segments are assembled with each other, each of
15 the annular chamber portions having inner and outer peripheral guides forming continuous annular inner and outer guides when the housing segments are assembled with each other, the annular partition being made of a plurality of partition segments, the partition segments being
20 assembled to the housing segments with inner and outer peripheral edges thereof being guided by the inner and outer guides, respectively.

[0012] As described above, according to the extraction structure for the gas turbine engine of the invention, even
25 the small amount of extraction air is controlled precisely.

Also, an even pressure is established in the circumferential direction within the compressing chamber.

BRIEF DESCRIPTIONS OF THE DRAWINGS

5 [0013] Fig. 1 is a partially broken-away side elevation of a gas turbine engine which incorporates an extraction structure according to an embodiment of the invention;

Fig. 2 is a partial longitudinal cross section of the extraction structure in Fig. 1;

10 Fig. 3 is a front view of an annular member used in the extraction structure in Fig. 1;

Fig. 4 is a longitudinal cross section of the extraction structure taken on a longitudinal plane angularly different from that in Fig. 2; and

15 Fig. 5 is s a rear view of a front housing in the extraction structure in Fig. 1.

PREFERRED EMBODIMENT OF THE INVENTION

[0014] With reference to the accompanying drawings, a preferred embodiment according to the invention will be described below.

[0015] Fig. 1 shows a partially broken-away side elevation of a gas turbine engine (hereinafter referred to as "engine") which incorporates an extracting structure
25 according to an embodiment of the invention. The engine 1

comprises compressor 3, combustors 5 and turbine 7 so that an intake air IA is compressed by the compressor 3 and then supplied to the combustors 5 where it is combusted with fuel F injected into the combustors 5 to generate a high-temperature and high-pressure combustion gas G which is used for driving the turbine 7. In the following descriptions, left and right sides of the engine 1 with respect to a longitudinal axis C in the drawing will be referred to as "front/upstream" and "rear/downstream", respectively, as necessary.

[0016] According to the embodiment, the compressor 3 is an axial-flow compressor and comprises a compressor rotor 11A which constitutes a front rotational portion of the engine. The front portion of the compressor rotor 11A is surrounded by a two-fraction compressor housing 13 to define an annular compression chamber 30 between the compressor rotor 11A and the compressor housing 13. The compressor housing 13 supports a plurality stages of compressor stator blade wheels 15 positioned in the compression chamber 30 and spaced apart from each other in the longitudinal direction. The compressor rotor 11A supports a plurality stages of compressor rotor blade wheels 17 positioned in the compressor chamber 30 and spaced apart from each other in the longitudinal direction so that each of the rotor blade wheels 17 is positioned on

the upstream side of the associated stator blade wheel 15, which allows the stator and rotor blade wheels 15 and 17 to cooperated with each other to compress the intake air IA drawn from the inlet 19. The compressed air CA is then
5 supplied into the combustors 5 through a diffuser 21 provided on the downstream side of the compressor 3.

[0017] The combustors 5 are provided at regular intervals in the circumferential direction of the engine 1, where the compressed air CA from the compressor 3 is mixed
10 and combusted with fuel F injected into the interiors of the combustors 5 to generate a high-temperature and high pressure combustion gas G which is then supplied to the turbine 7 through a turbine nozzle 23 or the first stage stator blade wheel of the turbine.

15 [0018] The compressor rotor 11A is connected to a high-pressure turbine rotor 11B of the turbine 7. A low-pressure turbine rotor 11C is located on the downstream side of the high-pressure turbine rotor 11B. The compressor rotor 11A is supported for rotation by the
20 compressor housing 13 through front and middle bearings 33A and 33B. The low-pressure turbine rotor 11C is supported for rotation through the turbine shaft 11D by the rear bearing 33C.

[0019] The compressor housing 13 is made of a front
25 housing 13A surrounding the front to middle portion of the

compressor rotor 11A (e.g., first to seventh stage stator blade wheels) and a rear housing 13B surrounding the middle to rear portion of the compressor 11A (e.g., eighth to last stage stator blades). The front and rear housings 13A and 5 13B are connected to each other by connection bolts 41. Fig. 2 shows details of the connection between the front and rear housings 13A and 13B. As shown in the drawing, an extraction structure ES for extracting a part of the compressed air CA from the compressor 3 is provided around 10 a mating surface P or contact surface of the front and rear housings 13A and 13B.

[0020] The front and rear housings 13A and 13B have outwardly projecting annular flanges 43A and 43B integrally formed at rear and front connection ends thereof, 15 respectively. For example, the front housing 13A has annular front flange 43A at its rear end and the rear housing 13B has annular rear flange 43B having the same outer diameter as the front flange 43A. The front and rear flanges 43A and 43B have through holes extending parallel 20 to the longitudinal axis and provided at regular intervals in the circumferential direction. Inserted through the holes are connection bolts 41 with external threads at distal ends thereof on which knots 45 are tightened to bring the front and rear housings 13A and 13B into close 25 connection with each other in the longitudinal direction.

[0021] As shown in Fig. 2, the front and rear flanges 43A and 43B have rear and front end surfaces 44A and 44B, respectively, extending on a plane P and opposing to each other so that, when the front and rear housings 13A and 13B
5 are assembled with each other, the rear and front surfaces 44A and 44B of the front and rear housings 13A and 13B mate with each other to form a seal therebetween.

[0022] The rear end surface 44A of the front flange 43A has an annular front recess 47 or groove defined therein
10 and the front end surface 44B of the rear flange 43B has an annular rear recess 49 or groove defined therein so that, when the front and rear housings 13A and 13B are assembled with each other as shown in the drawing, the front and rear recesses 47 and 49 cooperated with each other to form an
15 annular chamber 54 therebetween.

[0023] The front recess 47 of the front flange 43A receives an annular partition or plate 51 which is independent of the front flange 43A. As shown in Fig. 3, the plate 51 has a number of second apertures 53 extending
20 therethrough and provided at regular intervals in the circumferential direction. The plate 51 may be made of plural segments having the same size in the circumferential direction. In the embodiment, the plate 51 is made of two halves or semi-annular segments 51A and 51B. This
25 arrangement ensures that the plate 51 is easy to be

assembled in the compressor 3. The plate 51 is supported by slidably engaging its inner and outer circumferential edges in the annular engagement grooves 47A and 47B formed continuously in the opposing surfaces of the outer and inner circumferential walls 45A and 45B, defining radially opposed inner surfaces of the front recess 47. In this embodiment, the front housing 13A and the plate 51 are each divided into two halves. Therefore, when assembling, one and the other halves of the plate 51 are slidably engaged in one and the other halves of the grooves 47A and 47B of the front housing, respectively. Then, the two halves of the front housing are assembled with each other to form the front housing 13A.

[0024] The housing 13 has a number of through-holes or first apertures 61 communicating between the compressing chamber 30 and the annular chamber 54, in particular, a first plenum chamber 55 described below. In the embodiment, the compressing chamber 30 and the annular chamber 54 are communicated with each other through communication holes 61 defined by radially extending grooves 47C in a rear end surface 44A of the inner circumferential wall 45B of the front housing 13A. The communication holes may be made of either grooves 47C defined in the rear end surface 44A of the front housing 13A or grooves defined in the front end surface 44B of the rear housing 13B or both, or of through-

holes extending through the front or rear housing. The grooves are formed easier than the through-holes. In the embodiment, the communication holes 61 are provided between the seventh stator blade wheel 15 and the eighth rotor blade wheel 17, allowing a part of the compressed air to be introduced through the communication holes 61 into the first plenum chamber 55.

[0025] Once the front and rear housings 13A and 13B are assembled with each other, the annular chamber 54 positioned around the compressor housing 13 and surrounded by the flanges 43A and 43B is divided into the first plenum chamber 55 positioned on the rear side of the plate 51 and the second plenum chamber 57 positioned on the front side of the plate, the first and second plenum chambers 55 and 57 being communicated with each other through apertures 53 and the first plenum chamber 55 and the compressing chamber 30 being communicated through communication holes 61.

[0026] A total cross section area or total flow passage area of the apertures 53 is designed to be smaller than that of the communication holes 61. In the embodiment, the number of the apertures 53 is designed to be less than that of the communication holes 61. Also, in the embodiment, the number of the apertures 53 is designed to be about one half of the communication holes 61 and the total cross section area of the apertures 53 is designed to be about

10% of the total cross section area of the communication holes 61.

[0027] Fig. 4 shows a cross section of the extraction structure ES taken along a longitudinal plane which is angularly different from that in Fig. 2. As shown in the drawing, the first plenum chamber 55 is communicated with one or more first extraction passages 63 for extracting the compressed air from the first plenum chamber 55. The second plenum chamber 57 is communicated with one or more second extraction passages 65 for extracting the compressed air from the second plenum chamber 57. The first extraction passages 63 are each made of a first extraction hole 67 defined in the rear housing 13B and a first extraction tube 69 connected to the first extraction hole 67. The second extraction passages 65 are each made of a second extraction hole 71 defined in the front housing 13A and a second extraction tube 73 connected to the second extraction hole 71. Preferably, the first and second extraction holes 67 and 71 are formed on angularly different longitudinal planes.

[0028] As described above, because the total cross section area of the communication holes 61 is set to be larger than that of the apertures 53, the first extraction passages 63 work as a major extraction passage for extracting a larger amount of compressed air through

communication holes 61 and the first plenum chamber 55. The second extraction passages 65, on the other hand, work as a minor extraction passage for extracting a smaller amount of compressed air through the apertures 53 and the
5 second plenum chamber 57. The number of the extraction holes 63 and 65 are determined to obtain the necessary amount of extraction air. In this embodiment, the compressor housing 13 has two first extraction passages 63 provided at a regular interval in the circumferential
10 direction and one extraction passage 65.

[0029] In the embodiment shown in Fig. 1, the major first extraction passage 63 is connected to an exhaust gas passage 31. The exhaust gas passage 31 has a valve mechanism 77 so that the first extraction passage 63 can be
15 closed by the valve mechanism 77 except that a large amount of compressed air is needed to be extracted through the first extraction passage 63. The minor second extraction passage 65 is connected to a high-temperature portion of the engine, for example, a bearing chamber 81 which
20 accommodates the middle bearing 33B, so that the compressed air CA is guided to the high-temperature portion for its cooling.

[0030] Operations of the extraction mechanism according to the embodiment will be described below. When starting
25 the engine 1 in Fig. 1, a large amount of compressed air CA

is preferably extracted to reduce a start-up torque. Then, the valve mechanism 77 on the first extraction passage 63 is opened. This allows that the compressed air CA is guided through the communication holes 61 (Fig. 5) into the
5 first plenum chamber 55 (Fig. 4). In this instance, because the total cross section area of the first extraction passages 63 is sufficiently larger than that of the communication holes 61, the communication holes 61 work as a reducer between the compressing chamber 30 and the
10 first plenum chamber 55 to prevent an uneven pressure distribution in the circumferential direction within an interior of the compressor 3.

[0031] Also, an amount of compressed air CA flowing through the communication holes 63 into the first plenum
15 chamber 55 is sufficiently larger than a maximum amount of air passing through the apertures 53, calculated on the basis of the total cross section area thereof. In addition, the central axes of the apertures 53 are oriented parallel to the longitudinal axis and substantially orthogonal to
20 the central axes of the communication holes 61 and the first extraction passages 63 oriented in the radial directions. Therefore, a substantial part of the compressed air CA entered through the communication passages 61 into the first plenum chamber 55 are guided as
25 the major extraction compressed air LF into the first

extraction passages 63 and then the exhaust gas passage 31, rather than being guided through the apertures 53 into the second plenum chamber 57.

[0032] In the normal operation of the engine1, the valve
5 mechanism 77 of the first extraction passage 63 is closed so that only a small amount of air is extracted. In this operation, as shown in Fig. 2, the compressed air CA in the compressor 3 is introduced as a small amount of compressed air SF through the communication holes 61 into the first
10 plenum chamber 55. The communication holes 61, however, do not work as reducer because the total cross section of the communication holes 61 is set to be larger than that corresponding to an amount of air being extracted. The total cross section area of the apertures 53 in the plate
15 51 sufficiently smaller than that corresponding to the amount of air being extracted, which ensures that an even pressure distribution is established in the first plenum chamber 55 and the compressing chamber 30. Also, the plate 51 is formed independently of the compressor housing 15.
20 Therefore, the apertures 53 for controlling small amount of air are formed precisely. Also, the dimension, the number, the positions of the apertures can readily be modified for the specifications.

[0033] Although in the previous embodiment the major
25 first extraction passage 63 is connected to the exhaust gas

passage 31 in the starting operation of the engine 1 and the minor second extraction passage 65 is connected to the middle bearing chamber 81 for cooling, the extraction passage 63 and 65 may be connected to other internal and/or
5 external portions of the engine 1 where different amounts of compressed air are needed, irrespective of the intended use.

[0034] In view of the foregoing, although preferred embodiments of the invention have been described with
10 reference to the accompanying drawings, various additions, eliminations and/or modifications can be made without departing from the gist of the invention and they are within the scope of the invention. For example, although in the previous embodiment the plate 51 is mounted to the
15 front housing 13A by the slidably engagement in the circumferential grooves 47A and 47B of the front housing 13A, it may be mounted in different ways such as using fixing rings. In this instance, the plate or partition may be fixed by nipping it between the ring and the annular
20 front or rear end surface of the front or rear housing. Also, although the plate 51 is made of plural segments, it may be made of a single annular member. In this instance, the annular member is fixed using the fixing member or ring. Further, although in the previous embodiment the plate 51
25 is mounted in the front housing 13A, it may be mounted in

the rear housing 13B.

PARTS LIST

[0035]

- 5 1: gas turbine engine
- 3: compressor
- 13: compressor housing
- 51: plate (annular member)
- 53: aperture
- 10 55: first plenum chamber
- 57: second plenum chamber
- 61: slit (through hole)
- 63: first extraction passage (major extraction passage)
- 65: second extraction passage (minor extraction passage)
- 15 CA: compressed air
- ES: extraction structure

20

CLAIMS

1. A structure of a gas turbine engine for extracting a part of compressed air generated by a compressor,
5 comprising:

a cylindrical housing extending around an axis and forming therein at least a part of a compressing chamber in which air is flowed in a direction parallel to the axis as it is compressed to generate the compressed air, the
10 housing having

an annular portion extending around the compressing chamber to define an annular chamber,

an annular partition dividing the annular chamber into an annular first plenum chamber and an annular
15 second plenum chamber positioned on an upstream side of the first plenum chamber with respect to the direction and separated from the annular first plenum chamber,

the cylindrical housing having a plurality of first apertures defined therethrough to communicate between
20 the compressing chamber and said first plenum chamber;

the partition having a plurality of second apertures defined therethrough to communicate between the first and second plenum chambers, the first and second apertures being configured so that that the first apertures
25 have a first total cross-sectional area and the second

apertures have a second total cross-sectional area which is smaller than the first total cross-sectional area;

the second plenum chamber being designed to be communicated to a second extracting passage for extracting
5 the compressed air from the second plenum chamber.

2. The structure of claim 1, wherein the first plenum chamber is designed to be communicated to a first extracting passage for extracting the compressed air from the first plenum chamber.

10 3. The structure of claim 1 or 2, wherein the first extracting passage has a valve provided therein for closing the first extracting passage.

4. The structure in any of claims 1-3, wherein the cylindrical housing has a first housing portion and a
15 second housing portion positioned on the upstream side of the first housing portion, the first and second housing portions having opposed first and second radially extending mating surfaces, respectively, which mate with each other in a sealing engagement when the first and second housing
20 portions are assembled with each other, each of the first apertures being defined by a recess formed on first or second mating surface or respective recesses formed on first and second mating surfaces, the first and second mating surfaces having annular first and second recesses
25 defined therein, respectively, to form the annular chamber

when the first and second housing portions are assembled with each other.

5 5. The structure of claim 4, wherein the second housing portion is made of a plurality of housing segments, the housing segments having respective annular chamber portions forming a part of the annular chamber when the housing segments are assembled with each other, each of the annular chamber portions having inner and outer peripheral guides forming continuous annular inner and outer guides when the
10 housing segments are assembled with each other, the annular partition being made of a plurality of partition segments, the partition segments being assembled to the housing segments with inner and outer peripheral edges thereof being guided by the inner and outer guides, respectively.

15 6. A gas turbine engine, comprising:

 a compressor for compressing air to generate a compressed air;

 a combustor for combusting the compressed air with fuel to generate a high pressure combustion gas; and

20 a turbine for rotating a rotor by using the combustion gas;

 the compressor having a cylindrical housing extending around an axis and forming therein at least a part of a compressing chamber in which air is flowed in a

direction parallel to the axis as it is compressed to generate the compressed air, the housing having

an annular portion extending around the compressing chamber to define an annular chamber,

5 an annular partition dividing the annular chamber into an annular first plenum chamber and an annular second plenum chamber positioned on an upstream side with respect to the direction and separated from the annular first plenum chamber,

10 the cylindrical housing having a plurality of first apertures defined therethrough to communicate between the compressing chamber and said first plenum chamber;

the partition having a plurality of second apertures defined therethrough to communicate between the
15 first and second plenum chambers, the first and second apertures being configured so that that the first apertures have a first total cross-sectional area and the second apertures have a second total cross-sectional area which is smaller than the first total cross-sectional area;

20 the second plenum chamber being designed to be communicated to a second extracting passage for extracting the compressed air from the second plenum chamber.

7. The gas turbine engine of claim 6, wherein the first
25 plenum chamber is designed to be communicated to a first

extracting passage for extracting the compressed air from the first plenum chamber.

8. The gas turbine engine of claim 6 or 7, wherein the first extracting passage has an on-off valve provided
5 therein for closing the first extracting passage.

9. The gas turbine engine in any of claims 6-8, wherein the cylindrical housing has a first housing portion and a second housing portion positioned on the upstream side of the first housing portion, the first and second housing
10 portions having opposed first and second radially extending mating surfaces, respectively, which mate with each other in a sealing engagement when the first and second housing portions are assembled with each other, each of the first apertures being defined by a recess formed on first or
15 second mating surface or respective recesses formed on first and second mating surfaces, the first and second mating surfaces having annular first and second recesses defined therein, respectively, to form the annular chamber when the first and second housing portions are assembled
20 with each other.

10. The gas turbine engine of claim 9, wherein the second housing portion is made of a plurality of housing segments, the housing segments having respective annular chamber portions forming a part of the annular chamber when
25 the housing segments are assembled with each other, each of

the annular chamber portions having inner and outer peripheral guides forming continuous annular inner and outer guides when the housing segments are assembled with each other, the annular partition being made of a plurality
5 of partition segments, the partition segments being assembled to the housing segments with inner and outer peripheral edges thereof being guided by the inner and outer guides, respectively.

10

Fig. 1

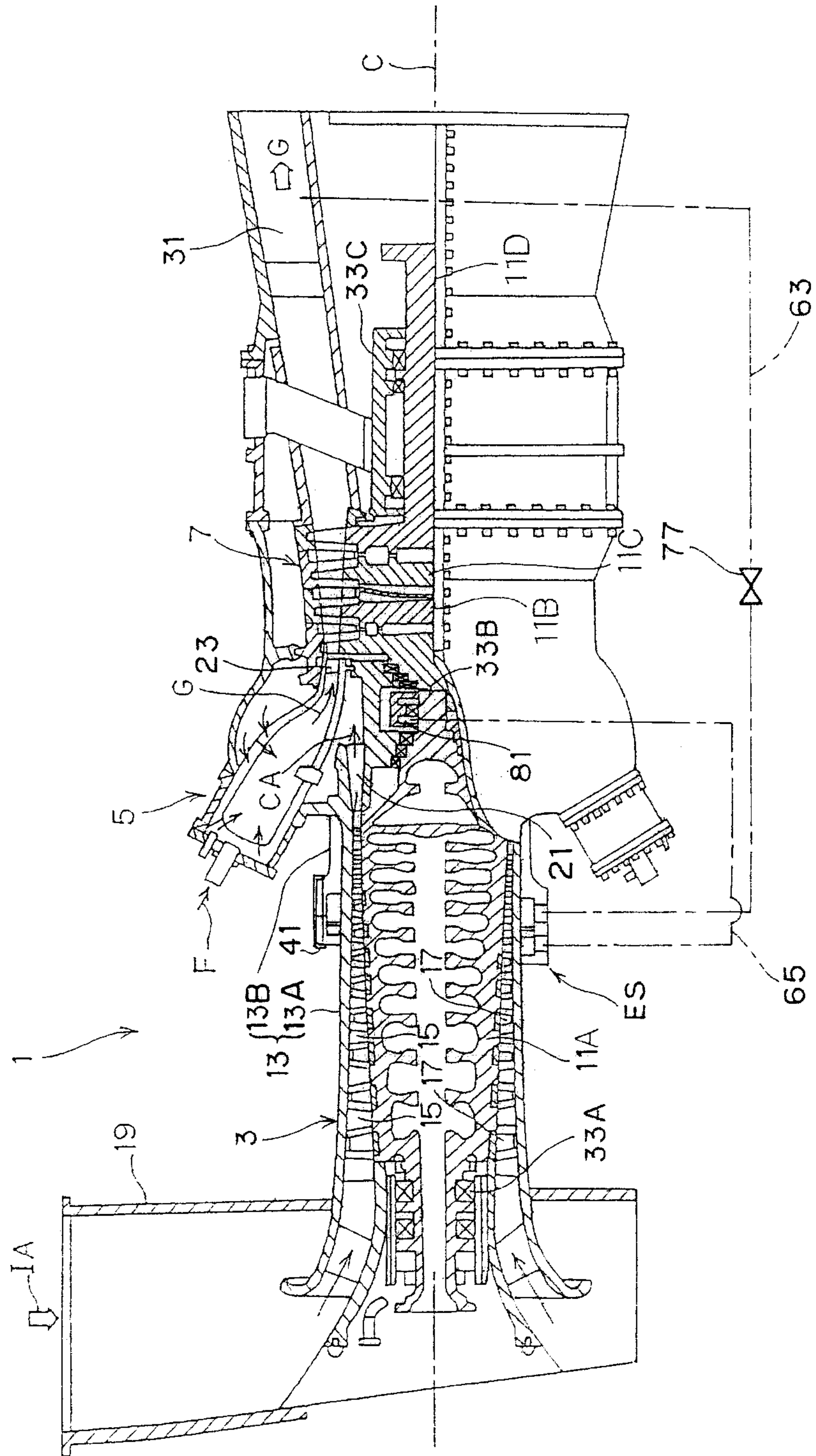
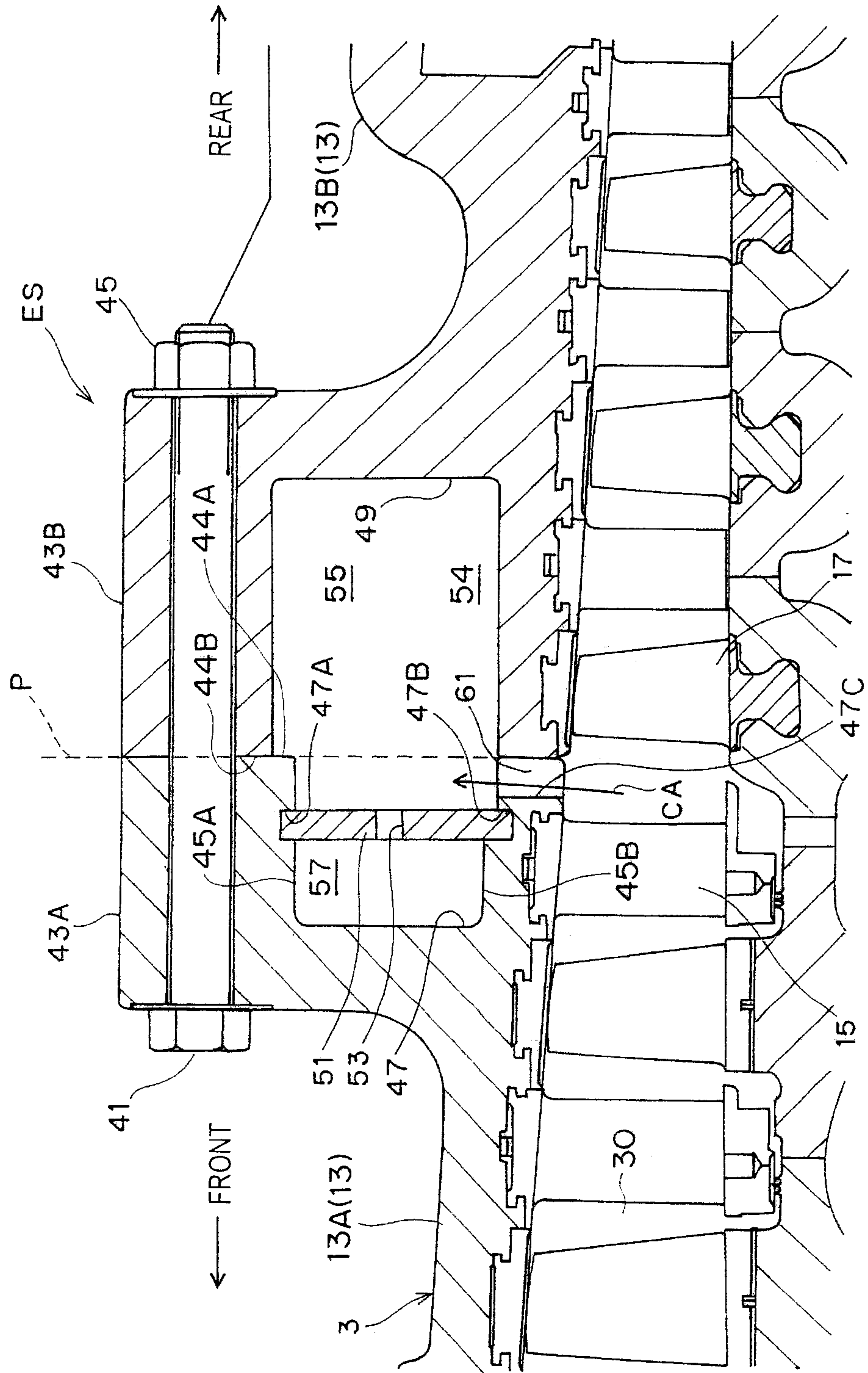


Fig.2



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Fig.3

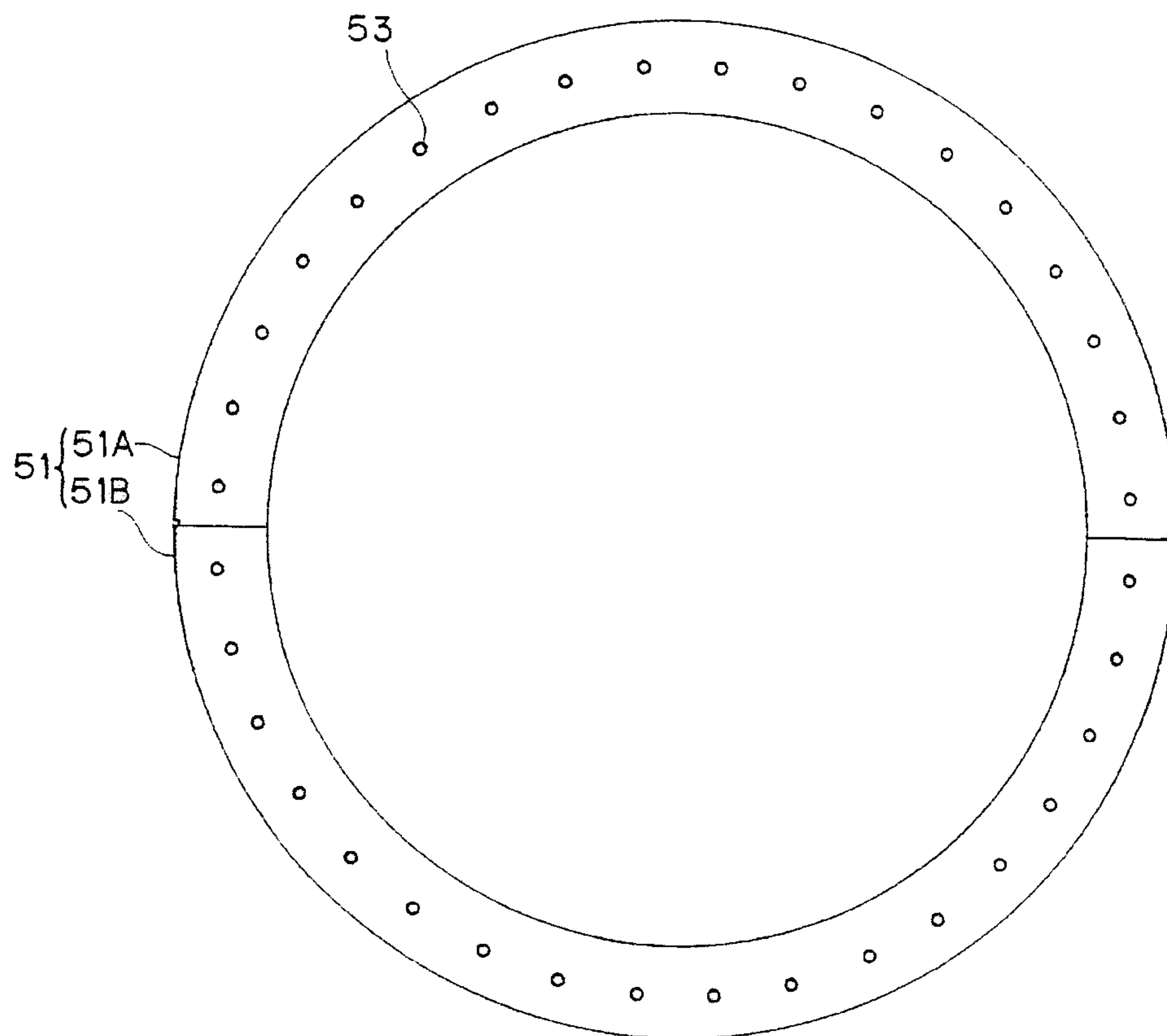
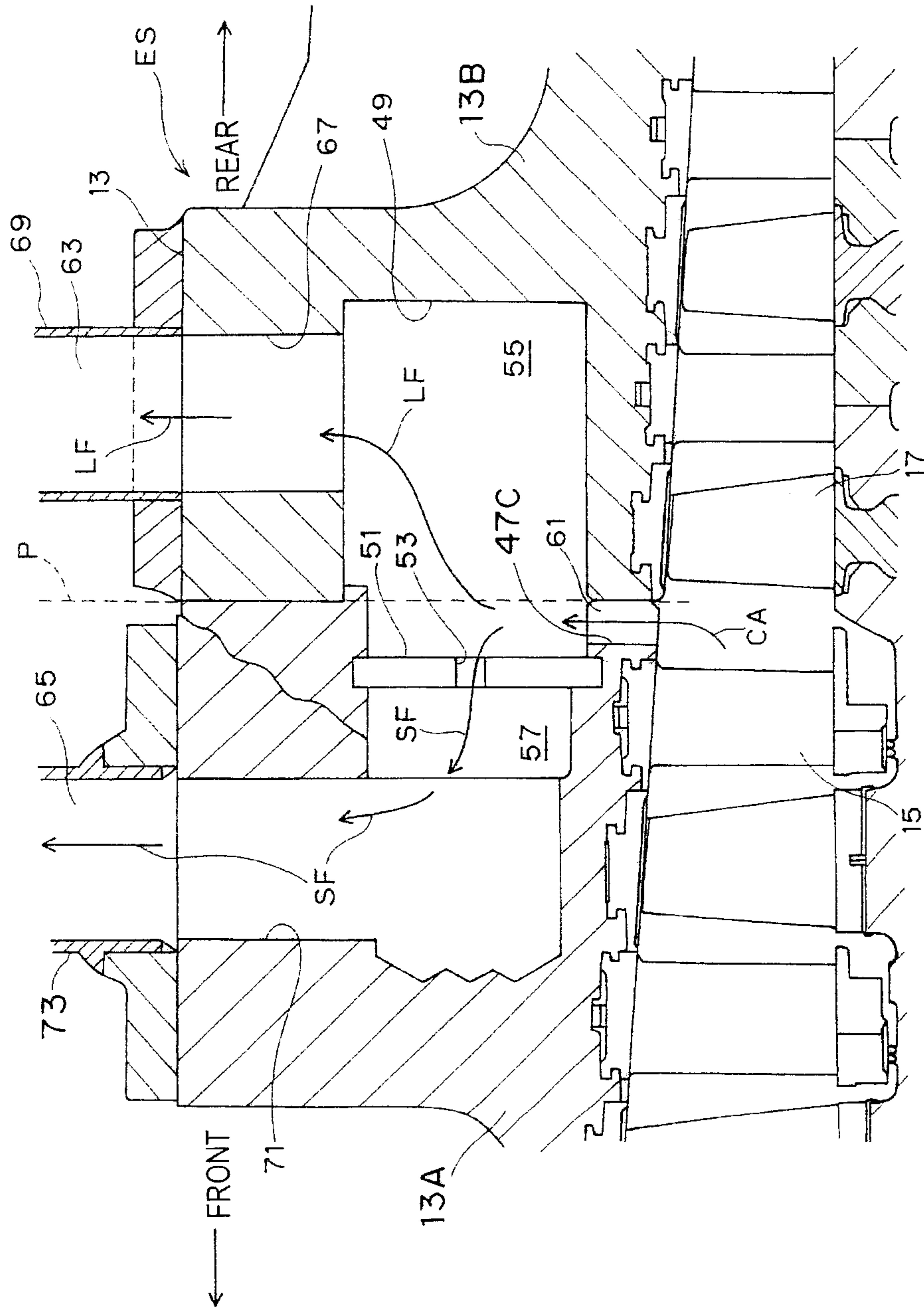


Fig. 4



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Fig.5

