A COMBUSTOR RESONATOR SECTION WITH AN INTERNAL THERMAL BARRIER COATING AND METHOD OF FABRICATING THE SAME

Abstract: A combustor (100) for a gas turbine engine (30) has a circumferentially extending liner (109) defining at least a portion of an interior combustion chamber (107) and a hot gas path (115). The liner includes a resonator section (112) including at least one resonator (116A, 116B) having a resonator chamber (125A, 125B) formed on an exterior of the liner. A thermal barrier coating (118) is disposed along an inner surface of the liner including an inner surface (130) of the resonator section. The resonator further includes a plurality of apertures (117) and each aperture extends through the liner and the thermal barrier coating at the resonator section and there is fluid flow communication between the combustion chamber and the resonator chamber.
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see Notice of 17 October 2013
**INTERNATIONAL SEARCH REPORT**

**International application No**

PCT/US2012/056801

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### A. CLASSIFICATION OF SUBJECT MATTER

INV. F23M5/00 F23M99/00 B21D53/84 F23R3/06 F23R3/42

F02C7/24 F23R3/06

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### ADD.

According to International Patent Classification (IPC) or to both national classification and IPC

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### B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)

B21D F02C F23M F23R

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Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

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Electronic data base consulted during the international search (name of data base and, where practicable, search terms used)

EPO-Internal , WPI Data

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### C. DOCUMENTS CONSIDERED TO BE RELEVANT

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<td>Y, P</td>
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**Date of the actual completion of the international search**

11 November 2013

**Date of mailing of the international search report**

18/11/2013

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