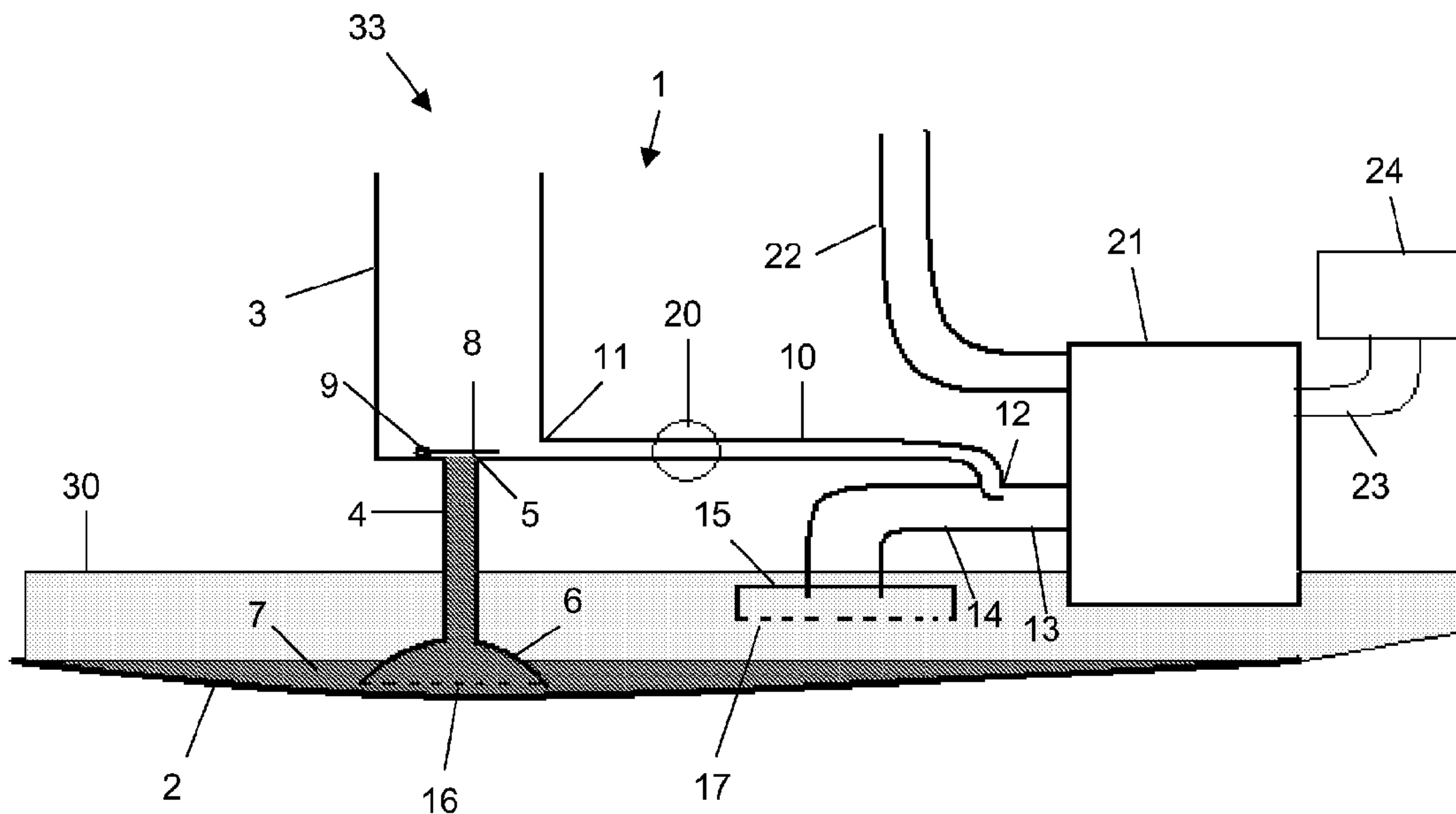




(86) **Date de dépôt PCT/PCT Filing Date:** 2008/03/05
 (87) **Date publication PCT/PCT Publication Date:** 2008/09/18
 (45) **Date de délivrance/Issue Date:** 2015/12/22
 (85) **Entrée phase nationale/National Entry:** 2009/08/20
 (86) **N° demande PCT/PCT Application No.:** GB 2008/050155
 (87) **N° publication PCT/PCT Publication No.:** 2008/110837
 (30) **Priorité/Priority:** 2007/03/12 (GB0704726.9)

(51) **Cl.Int./Int.Cl. B64D 37/14** (2006.01),
B64D 37/22 (2006.01), **B64D 37/34** (2006.01)
 (72) **Inventeur/Inventor:**
 MINTY, ANDREW, GB
 (73) **Propriétaire/Owner:**
 AIRBUS OPERATIONS LIMITED, GB
 (74) **Agent:** ADE & COMPANY INC.

(54) **Titre : SYSTEME DE RECUPERATION D'EAU**
 (54) **Title: WATER SCAVENGING SYSTEM**



(57) **Abrégé/Abstract:**

A water scavenging system (1) for removing water from an aircraft fuel tank (2), the water scavenging system comprising: a water line (10); and a valve (20) which is configured to control the flow of water in the water line in response to a change in an effective angle of the valve. In use, the valve opens automatically in response to an increase in the effective pitch of the aircraft during take off and climb. The valve comprises: a pendulum (40); and a valve closure member (46) which is coupled to the pendulum such that a change in the angular position of the pendulum relative to the water line causes the valve closure member to move from a closed position in which the valve closure member impedes the flow of water in the water line to an open position in which water can flow through the valve.

ABSTRACT

A water scavenging system (1) for removing water from an aircraft fuel tank (2), the water scavenging system comprising: a water line (10); and a valve (20) which is configured to control the flow of water in the water line in response to a change in an effective angle of the valve. In use, the valve opens automatically in response to an increase in the effective pitch of the aircraft during take off and climb.

The valve comprises: a pendulum (40); and a valve closure member (46) which is coupled to the pendulum such that a change in the angular position of the pendulum relative to the water line causes the valve closure member to move from a closed position in which the valve closure member impedes the flow of water in the water line to an open position in which water can flow through the valve.

WATER SCAVENGING SYSTEM**FIELD OF THE INVENTION**

The present invention relates to a water scavenging system.

BACKGROUND OF THE INVENTION

5 A method of removing water from an aircraft fuel tank is described in
US4809934. A water scavenge pipe collects water which is dispersed in the fuel
immediately upstream of a booster pump. A first problem with this system is that it
requires a complex arrangement of Venturis and jet pumps. A further problem is that
water may be fed to the engine when the engine is operating at a relatively low rate.
10 This may reduce the operating efficiency of the engine. A further problem is that the
system may not remove water entirely from the fuel tank during a single flight.

A method of removing water from a fuel tank is described in
US6170470. A water line feeds water into an engine from the fuel tank. A valve in a
water line is opened when the speed of the engine is above a minimum threshold,
15 because water flowing into the engine when the engine is operating at relatively high
speeds will not have a significantly adverse effect on the operational characteristics
of the engine. A problem with the method of US6170470 is that an engine speed
measuring device must be provided, along with a communication line to the valve.
Both of these elements are at risk of failing, and the communication line must be
20 routed out of the fuel tank in a fluid-tight and inherently safe manner.

A particular difficulty associated with an aircraft fuel system is that once
the aircraft has taken off the ambient air temperature drops to -30 or -40deg C and

the water in the fuel freezes after approximately 20 minutes. After the water has frozen it is trapped in the tank until the tank thaws on landing. Therefore if the water is not removed in the first twenty minutes it will gradually accumulate over several flights until it causes problems or is manually drained.

5 SUMMARY OF THE INVENTION

A first aspect of the invention provides a water scavenging system for removing water from an aircraft fuel tank, the water scavenging system comprising:

a water line; and

a valve which is configured to control the flow of water in the water line,

10 characterised in that the valve is configured to control the flow of water in the water line in response to a change in an effective angle of the valve, and in that the valve comprises a pendulum and a valve closure member which is coupled to the pendulum such that a change in the angular position of the pendulum relative to the water line causes the valve closure member to move from a closed position in
15 which the valve closure member impedes the flow of water in the water line to an open position in which water can flow through the valve.

Instead of using an engine speed measuring device, the invention provides a valve which opens automatically in response to an increase in the
20 effective pitch of the aircraft during take off and climb. Since the engine speed will be relatively high at take off and climb, this means that the water will be fed into the engine without significantly affecting its operational characteristics. A further

advantage of feeding the water into the engine during take off and climb is that it gets rid of the water before it freezes.

Typically the valve is positioned in the water line - either at its inlet, its outlet or an intermediate position along its length.

5 In the preferred embodiments described below, the water is fed into the engine from a water tank which has been filled by the action of hydrostatic pressure. However the valve may also be employed in a system of the kind described in US 6170470, in which the water is fed directly to the engine without an intermediate step of storing in a water tank. In this case, the valve can act as a substitute for the
10 engine speed measuring device used in US 6170470.

Preferably the valve comprises: a pendulum; and a valve closure member which is coupled to the pendulum such that a change in the angular position of the pendulum relative to the water line causes the valve closure member to move from a closed position in which the valve closure member impedes the flow
15 of water in the water line to an open position in which water can flow through the valve.

In a preferred embodiment the pendulum is coupled to a cam; and the valve closure member comprises a cam follower and a biasing member (such as a coil spring) for biasing the cam follower into engagement with the cam. This enables
20 the opening and closing characteristics of the valve to be easily adjusted by adjusting the shape and/or size and/or position of the cam. The cam may be an

integral part of the pendulum axle or pendulum shaft, or may be coupled indirectly to the pendulum.

In its closed configuration the valve may permit some flow, or may substantially prevent the flow of water in the water line.

5 According to a second aspect of the invention there is provided a method of removing water from an aircraft fuel tank, the method comprising opening a valve in response to an increase in an effective pitch of the aircraft to enable water to flow in a water line; and feeding water from the water line into an engine of the aircraft.

10

BRIEF DESCRIPTION OF THE DRAWINGS

Embodiments of the invention will now be described with reference to the accompanying drawings, in which:

15 Figure 1 shows an aircraft;

Figures 2-5 are schematic views of a fuel tank incorporating a water scavenging system;

Figures 6a-6c show the pendulum valve in detail;

Figures 7a-7c show a first alternative pendulum valve; and

20 Figures 8a-8c show a second alternative pendulum valve.

DETAILED DESCRIPTION OF EMBODIMENT(S)

Referring to Figure 1, an aircraft 100 comprises a fuselage 102 carrying a pair of wings, the left wing being labelled 103. Each wing carries an engine, the left hand engine being labelled 104 in Figure 1. Fuel for each engine is stored in a centre tank and one or more wing tanks.

5 A water scavenging system 1 is shown in Figure 2, installed in one of the wing tanks 2.

The water scavenging system comprises a water tank 3; and a water scavenging line 4 which has an outlet 5 in the bottom of the water tank 3, and an inlet 6 immersed in a pool of water 7 at the bottom of the fuel tank 2. The water tank
10 3 is typically a 60mm diameter vertical pipe with a length of 0.4m and an open top.

A non-return valve 8 is mounted to the water tank on a pivot 9 and is shown in its closed position in Figure 2, in which it closes the outlet 5 of the water scavenging line 4. It will be appreciated that many other types of non-return valve may be used, and the design shown in Figures 2-5 is for illustrative purposes only.
15 For instance the non-return valve may comprise a cone closure member mounted on a conical seat, in which the cone translates between its open and closed positions.

Figures 2-4 show fuel 30 in the tank 2 at relatively low, medium and high levels respectively. As the fuel tank is filled, the fuel exerts gravitational hydrostatic pressure on the water 7 pooled at the bottom of the fuel tank. This
20 hydrostatic pressure drives the water up the water scavenging line 4, through the non-return valve 8, and into the water tank 3 against the force of gravity, until the water has been almost completely removed from the bottom of the fuel tank as

shown in Figure 3. As the fuel tank continues to fill, fuel bubbles through the water (three of such fuel bubbles being shown at 3 1 in Figure 3) and collects in a layer 32 at the top of the water tank 3.

When the level of fuel reaches the top of the water tank 3, fuel flows
5 into the water tank through the opening 33 at the top of the water tank, filling it as shown in Figure 4.

Returning to Figure 2, a water output line 10 extends from the bottom of the water tank 3.

The water output line has an inlet 11 in the water tank 3, and an outlet
10 12 coupled to an pump inlet line 13 in parallel with a fuel uptake line 14. The fuel uptake line 14 has an inlet 15 in the fuel tank above the inlet 6 of the water scavenging line. The inlets 6,15 have filters or strainers 16,17 respectively.

The pump inlet line 13 leads to a pump 21. A pump output line 22 leads from the pump to the engine 104, and a recirculation line 23 leads from the pump to
15 a valve 24 in the fuel tank. The valve 24 can be opened to "de-rate" the pump 21, allowing the pump 2 1 to continue operating whilst bypassing the pump output line 22.

A valve 20 in the water output line 10 is shown schematically in Figures 3-5, and in further detail in Figures 6a-c.

20 The valve 20 comprises a pendulum 40 which is pivoted to a support structure (not shown) by an axle 41. The axle 4 1 carries a cam 42. A chamber 43 in the water output line 10 has an inlet 44 and an outlet 45. A gate 46 is mounted in the

chamber 43 and can translate between its closed position shown in Figure 6a to its open position shown in Figures 6b and 6c. In its closed position the gate 46 blocks the inlet 44 and/or the outlet 45 of the chamber 43, and in its open position the gate 46 permits fluid to flow through the valve as shown in Figures 6b and 6c.

5 The gate 46 has a cam follower 47 which is biased against the cam 42 by a coil compression spring 48.

When the aircraft is level, the valve 20 is in its closed configuration shown in Figures 3, 4 and 6a. When the "effective pitch" of the aircraft increases above 7 degrees, the valve 20 changes from its closed configuration to its positive
10 open configuration shown in Figures 5 and 6b. "Effective pitch" is defined as the combination of physical pitch (that is, the physical pitch angle of the aircraft) combined with the acceleration vector due to forward acceleration.

With the valve 20 open as shown in Figure 5, the water is driven along the water output line 10 and supplied to the pump 21 by hydrostatic pressure
15 resulting from the gravity head of the water in the water tank. The blockage at the pump inlet would be negligible. Note that the gravity head driving the water along the water output line 10 is relatively large due to the greater density of water compared with fuel. In other words the gravity head is greater than would be the case if the water tank 3 was filled with fuel. Thus the gravity head tends to drive water towards
20 the fuel pump in preference to fuel from the fuel uptake line 14, at least as long as the level of the water in the water tank is above the outlet 12 of the water output line 10. Optionally an inductor pipe (not shown) protruding into the water output line 10

may also be provided. This provides a slight restriction in the water output line 10 and tends to enhance the flow due to the Venturi effect.

Note that the arrangement shown in Figures 2-5 is schematic, and the water tank 3 may be adjusted to be directly above the outlet 12 of the water output
5 line 10, if required to give sufficient gravity head at all desired angles of pitch, and all levels of water in the water tank.

Note also that the water tank 3 is designed to be relatively tall and narrow so as to maximise the gravity head. More specifically, the inverse aspect ratio (which we define herein as H/vA , where H is the height of the water tank and A
10 is its mean cross-sectional area) is relatively high. For instance in the case of a cylindrical water tank with a 60mm diameter and a length of 0.4m, the inverse aspect ratio of the water tank is approximately 7.5.

If the water tank 3 is subjected to negative gravity forces, then the water may empty out of the opening 33 at the top of the water tank 3 into the fuel
15 tank 2. However, since the water tank 3 is relatively small (having a capacity of the order of one litre compared with the fuel tank 2 which has a capacity of the order of 8,000 litres) this should not affect performance.

Also, any water which empties in this manner will be picked up at the next refuel.

20 The water tank 3 is sized to give a 0.5 litre flow in the two to three minutes during take off and climb. The resulting water concentration to the engine is approximately 2500ppm during this phase.

After take off and climb, the valve 24 is opened to "de-rate" the pump 21, and the centre tank pumps take over. At the same time, as the pitch of the aircraft drops below 7 degrees, the valve 20 closes to its closed position shown in Figure 6b and fuel is input to the pump exclusively through the fuel uptake line 14.

5 As the aircraft goes "nose down" during descent, the effective pitch of the aircraft drops below -5 degrees, and the valve 20 changes from its closed configuration to its negative open configuration shown in Figure 6c. At this stage, the water tank 3 may be empty, or may contain a head of fuel above the fuel level in the rest of the fuel tank. Note that there will be little or no water in the water tank 3 (the
10 majority of the water having been fed to the engine previously during take off and climb) but the water tank 3 may contain some fuel. Note that if there was a significant amount of water present in the water tank it would have frozen and may jam the drain valve 20. If the water tank 3 contains fuel, then this fuel will be fed to the engine during descent, emptying the water tank ready for the next refuelling.

15 A first alternative pendulum valve 20a (which can be used instead of the valve 20) is shown in Figures 7a-7c. In this case, instead of employing a cam and cam-follower, the valve comprises a pendulum 50 which is coupled to a gate 51 by an articulated link 52.

The pendulum 50 has a vertical arm 55 and a horizontal arm 56 which
20 are fixed with respect to each other, and pivoted to a support structure (not shown) by an axle 57. The link 52 is rotatably coupled to the gate 51 at one end by a pivot 53 and to the horizontal arm 56 of the pendulum 50 at the other end by a pivot 54.

The gate 51 is slidably mounted in a chamber 55 which protrudes from the water output line on both upper and lower sides, and is driven by the link 52 between its three operating positions as shown in Figures 7a-7c.

A second alternative pendulum valve 20b (which can be used instead
5 of the valve 20 or the valve 20a) is shown in Figures 8a-8c. In this case, instead of employing a translating gate, the valve 20b has a closure member which rotates between its open and closed positions.

Specifically, the valve 20b comprises a pendulum 60 mounted on a cylindrical closure member 61 which is housed in a cylindrical chamber 62 in the
10 water output line 10. The closure member 61 has a pair of channels 63,64 which line up with the inlet and outlet of the chamber 62 when the angle of the pendulum relative to the water output line reaches +7 degrees or -5 degrees, as shown in Figures 8b and 8c respectively. Note that for ease of illustration the water output line
15 10, chamber 62 and closure member 61 are shown lying vertically and parallel with the pendulum 60: in practice they will be oriented horizontally and at right angles to the pendulum 60. Also, the angular deviation between the channels 63, 64 is exaggerated in Figures 8a-8c for ease of illustration. In practice the channels will extend at a narrower angle (and the chamber 62 and closure member 61 will be larger relative to the water output line 10) to give the required operation.

20 The advantages of the valve 20 shown in Figures 6a-6c, compared with the valves shown in Figures 7a-7c and 8a-8c are: 1. the opening and closing characteristics of the valve can be easily adjusted by adjusting the shape and/or size

and/or position of the cam 42; 2. because the gate 46 has only two operating positions, the chamber 43 is relatively compact; 3. the pendulum can be easily separated from the rest of the valve for maintenance purposes; and 4. if the pitch of the aircraft overshoots (that is, if the pitch increases substantially above +7 degrees, or decreases substantially below -5 degrees or if there is a high acceleration or deceleration) then the operation of the valve is not affected.

In an alternative embodiment (not shown) several water tanks, each with its own respective scavenging line and water output line, may all be coupled in parallel to the engine via a single shared pendulum valve 20.

In a further alternative embodiment (not shown) a thermostatically controlled drain valve may be installed in a line in parallel with the pendulum valve 20, 20a or 2b. The thermostatically controlled valve opens when the temperature drops below 2 deg C (say), which will occur as the aircraft cruises at altitude. This ensures that all water is removed from the water tank. Optionally the thermostatically controlled valve could replace the "nose down" operation of the pendulum valve 20, 20a, 20b: that is, the pendulum valve 20, 20a, 20b may be replaced by a similar pendulum valve with only a positive open position. The thermostatically controlled valve could for example be a wax-type valve as used in car-engine thermostats or a bimetallic spring that opens a valve.

CLAIMS

1. A water scavenging system for removing water from an aircraft fuel tank, the water scavenging system comprising:

a water line; and

a valve which is configured to control the flow of water in the water line,

characterised in that the valve is configured to control the flow of water in the water line in response to a change in an effective angle of the valve, and in that the valve comprises a pendulum and a valve closure member which is coupled to the pendulum such that a change in the angular position of the pendulum relative to the water line causes the valve closure member to move from a closed position in which the valve closure member impedes the flow of water in the water line to an open position in which water can flow through the valve.

2. The system of claim 1 wherein the valve closure member translates between its open and closed positions.

3. The system of claim 1 or 2 wherein the pendulum is coupled to a cam; and wherein the valve closure member comprises a cam follower and a biasing member for biasing the cam follower into engagement with the cam.

4. The system of claim 3 wherein the biasing member comprises a coil spring.

5. The system of any one of claims 1 to 4 wherein the valve closure member is coupled to the pendulum by a link which is rotatably coupled to the valve closure member at one end and rotatably coupled to the pendulum at another end.

6. The system of any one of claims 1 to 5 wherein the valve closure member rotates between its open and closed positions.

7. The system of any one of claims 1 to 6 wherein the valve has a closed configuration in which it substantially prevents the flow of water in the water line.

8. A method of removing water from an aircraft fuel tank, the method comprising opening a valve in response to an increase in an effective pitch of the aircraft to enable water to flow in a water line; and feeding water from the water line into an engine of the aircraft.

9. The method of claim 8 further comprising closing the valve in response to a decrease in the effective pitch of the aircraft.

Figure 1

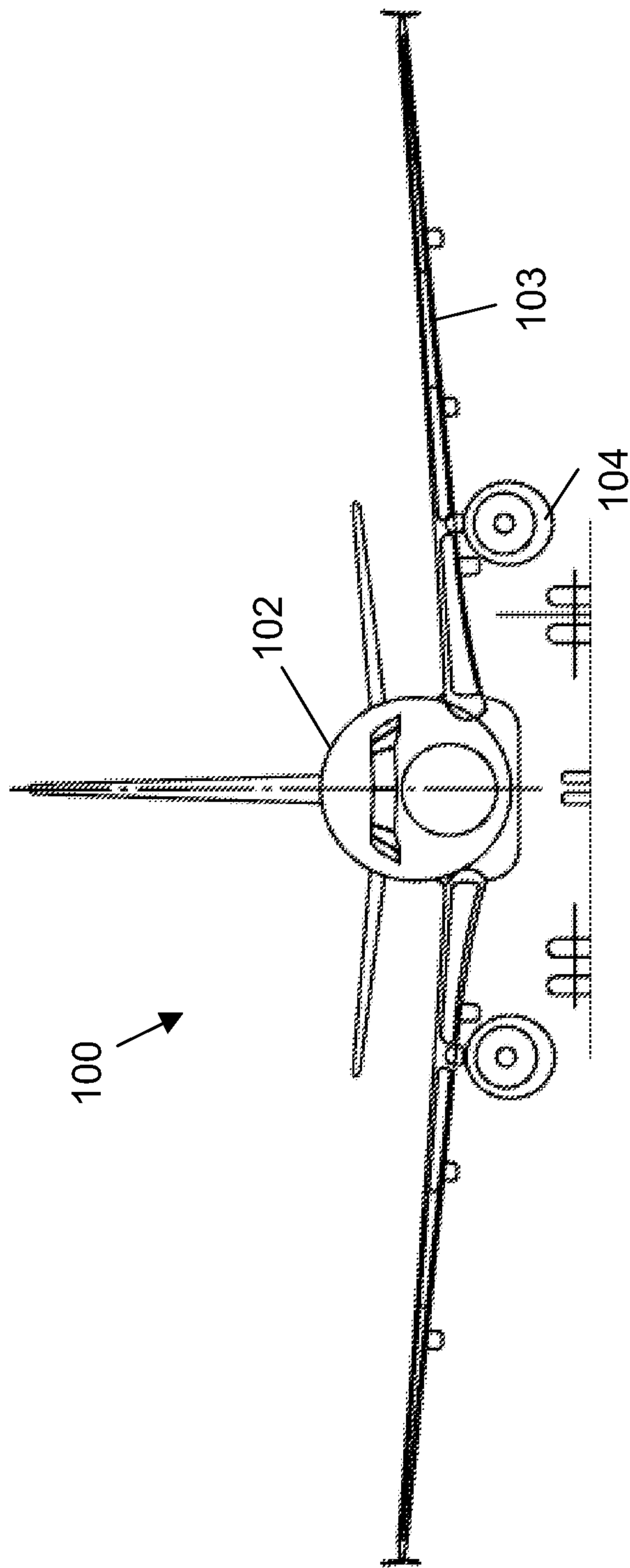


Figure 2

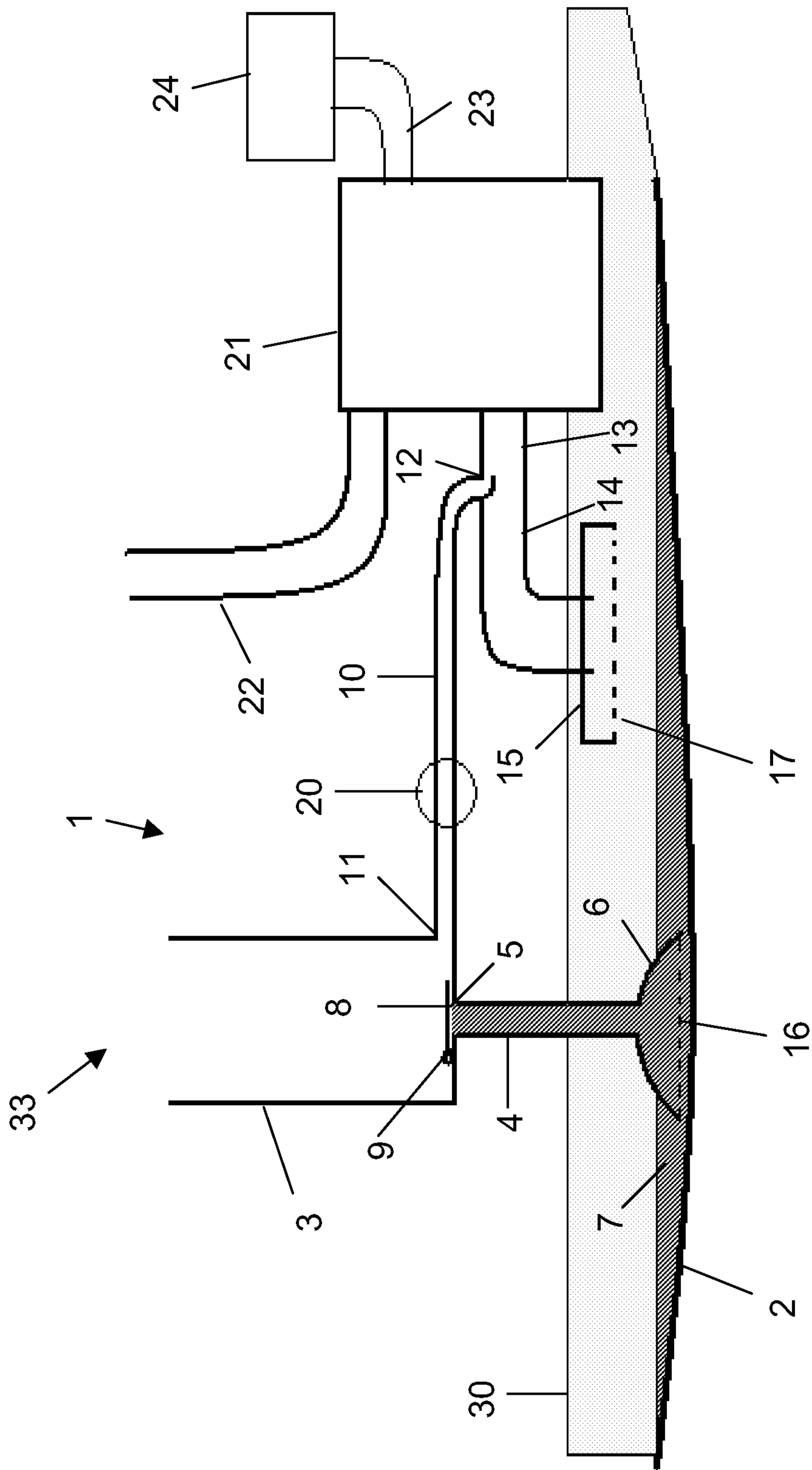


Figure 3

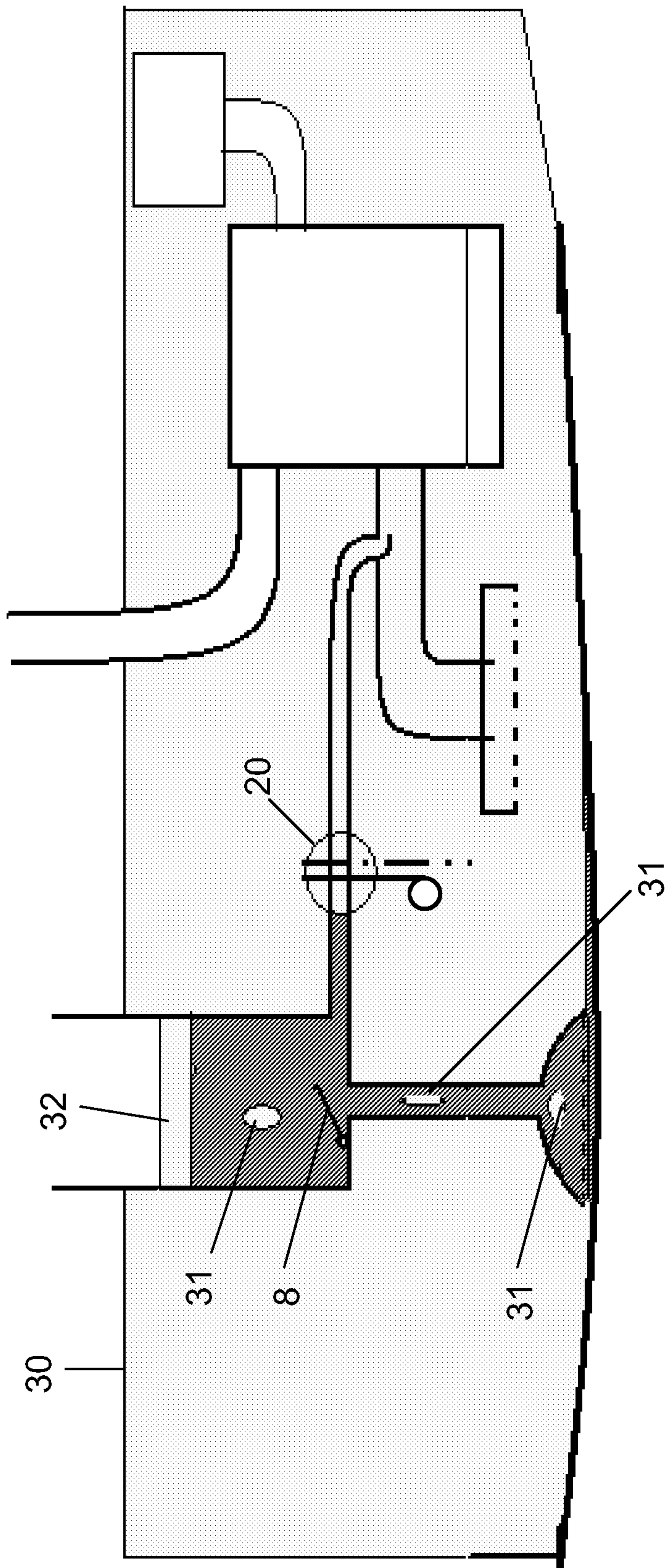


Figure 4

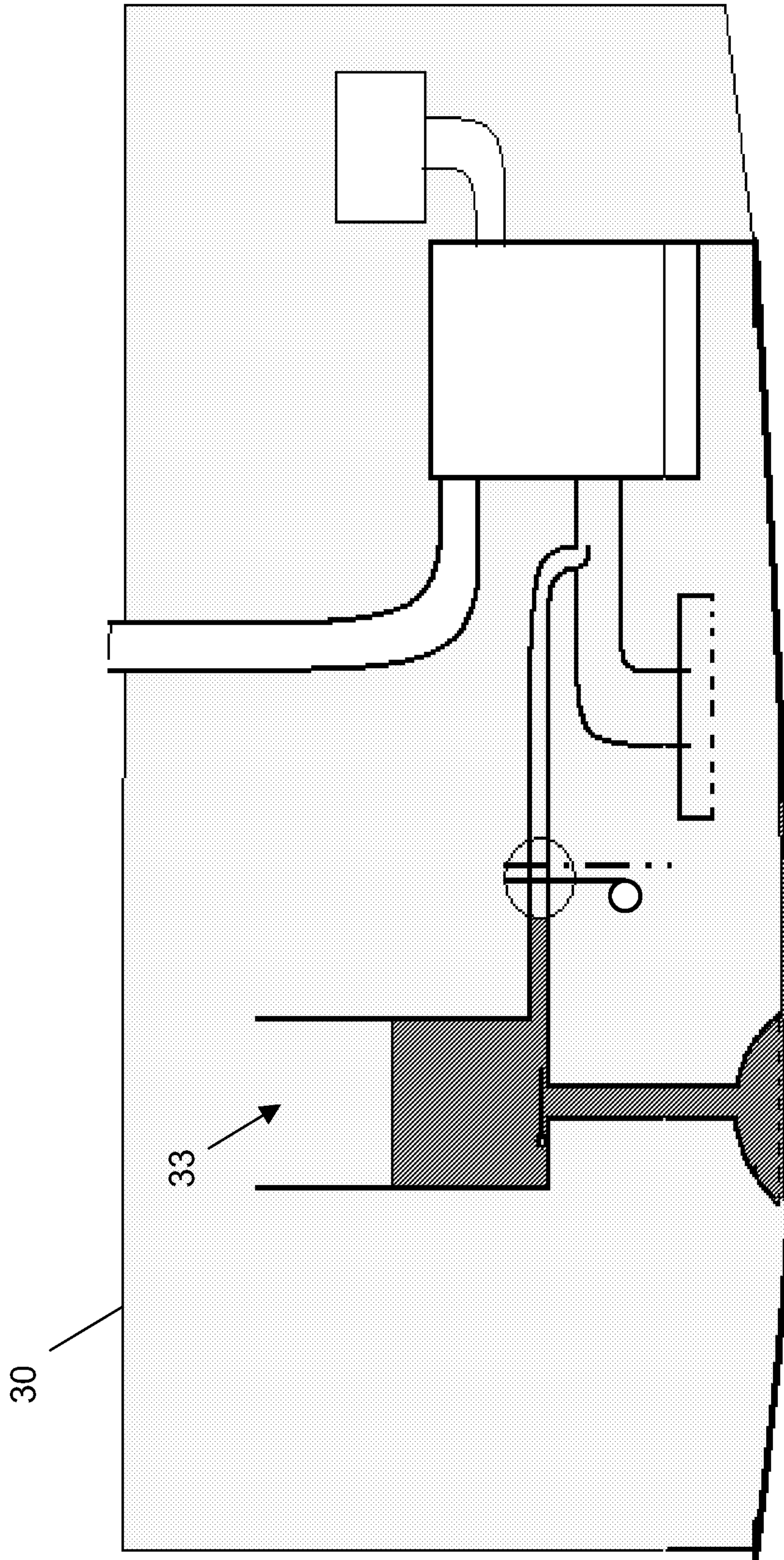


Figure 5

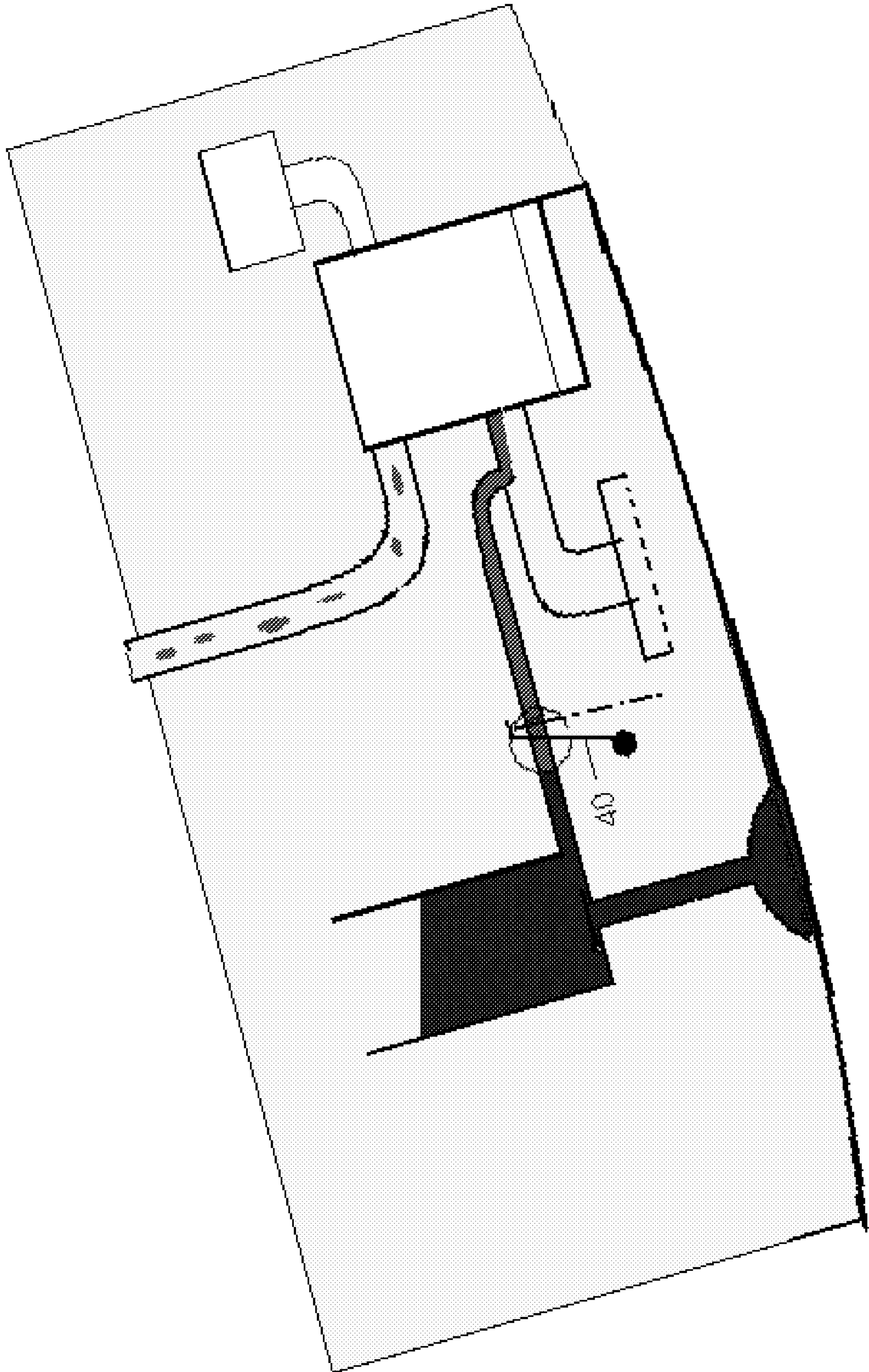


Figure 6a Figure 6b Figure 6c

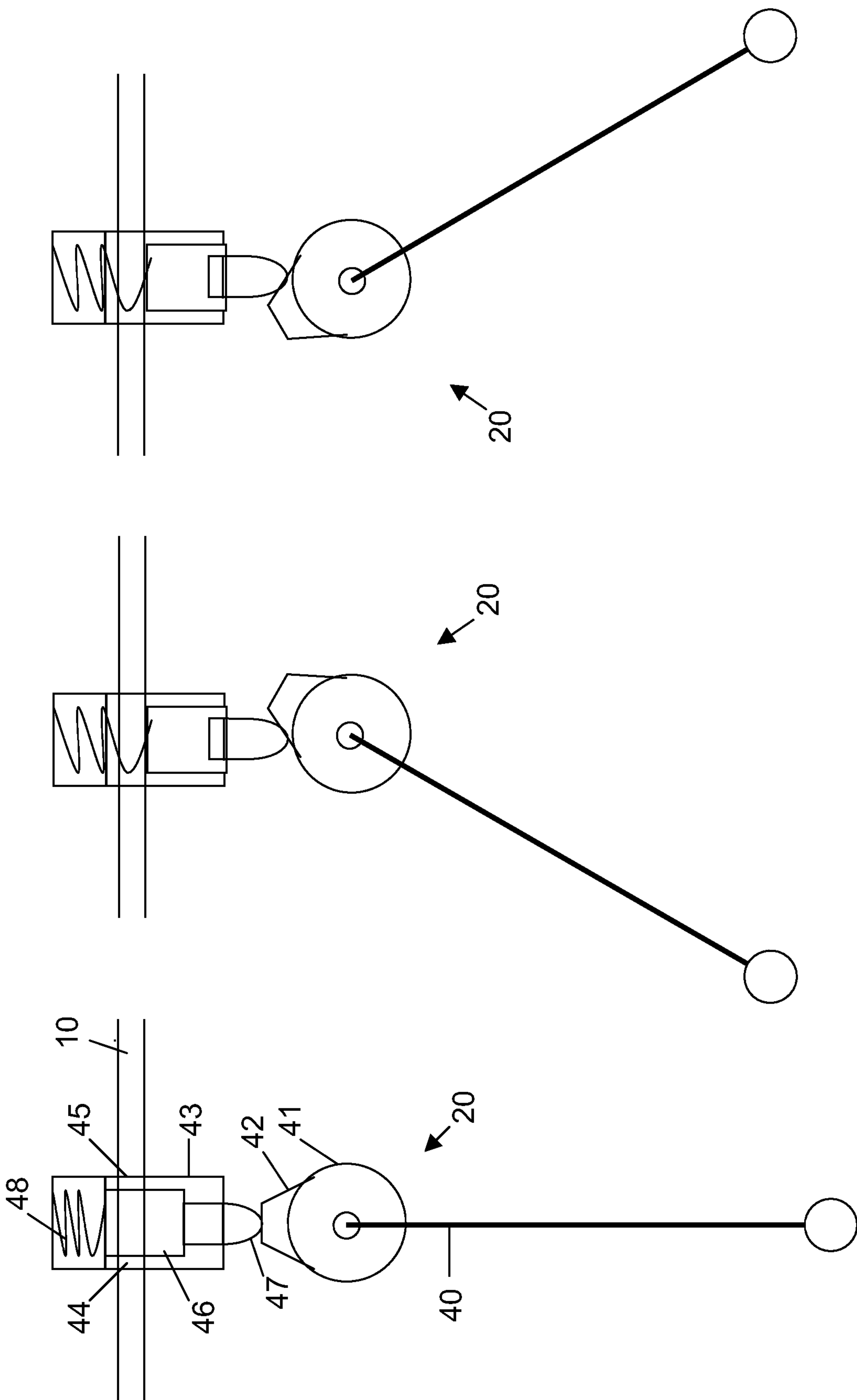


Figure 7a Figure 7b Figure 7c

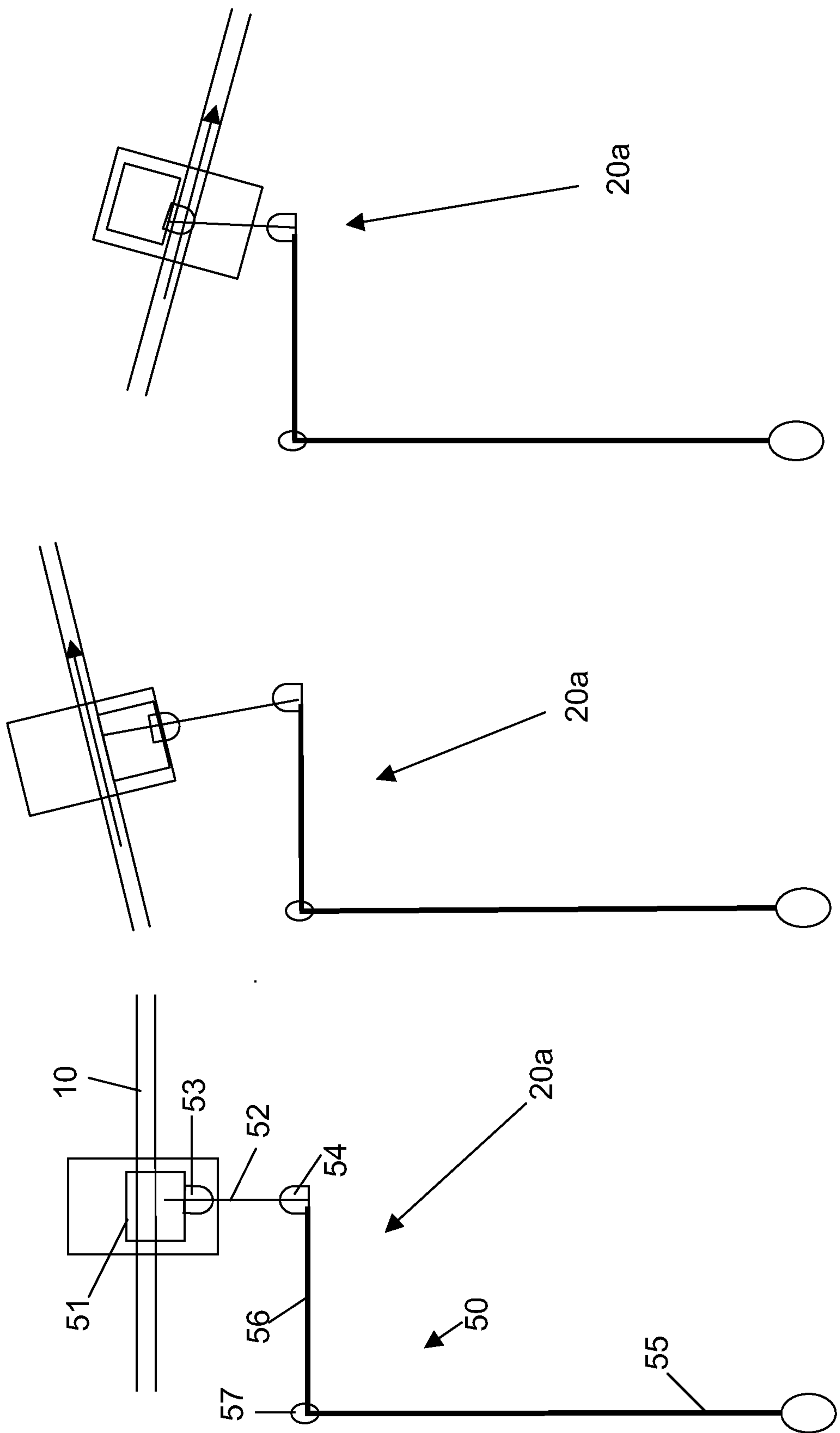


Figure 8a Figure 8b Figure 8c

