



(22) Date de dépôt/Filing Date: 2017/09/08

(41) Mise à la disp. pub./Open to Public Insp.: 2018/05/01

(45) Date de délivrance/Issue Date: 2021/03/16

(30) Priorité/Priority: 2016/11/01 (US15/340371)

(51) Cl.Int./Int.Cl. *G01C 25/00* (2006.01)

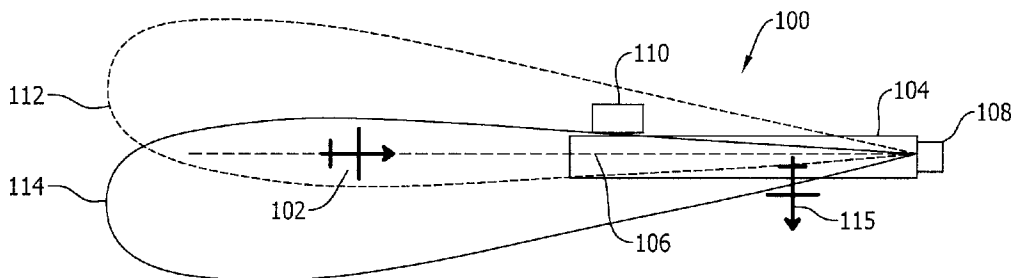
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(54) Titre : SYSTEME DE COMMANDE DE VOL A DETECTION D'ANOMALIE DE LOCALISATEUR D'INSTRUMENT
BASSE FREQUENCE DE SYSTEME D'ATTERRISSAGE ET METHODE D'UTILISATION

(54) Title: FLIGHT CONTROL SYSTEM WITH LOW-FREQUENCY INSTRUMENT LANDING SYSTEM LOCALIZER
ANOMALY DETECTION AND METHOD OF USE



(57) **Abrégé/Abstract:**

A flight control module for detecting anomalies in ILS localizer signals during landing of an aircraft is provided. The flight control module includes a communication interface and a processor coupled thereto. The communication interface is configured to receive inertial data, GPS data, and an ILS localizer deviation for the aircraft. The processor is configured to compute an inertial localizer deviation based on the inertial data and a GPS localizer deviation based on the GPS data. The processor is configured to compare the ILS localizer deviation to an average of the inertial localizer deviation and the GPS localizer deviation to detect a low-frequency anomaly in the ILS localizer deviation. The processor is configured to initiate a transition from controlling the aircraft based on the ILS localizer deviation to controlling the aircraft based on the inertial localizer deviation when the low-frequency anomaly is detected.

ABSTRACT OF THE DISCLOSURE

A flight control module for detecting anomalies in ILS localizer signals during landing of an aircraft is provided. The flight control module includes a communication interface and a processor coupled thereto. The communication interface is
5 configured to receive inertial data, GPS data, and an ILS localizer deviation for the aircraft. The processor is configured to compute an inertial localizer deviation based on the inertial data and a GPS localizer deviation based on the GPS data. The processor is configured to compare the ILS localizer deviation to an average of the inertial localizer deviation and the GPS localizer deviation to detect a low-frequency
10 anomaly in the ILS localizer deviation. The processor is configured to initiate a transition from controlling the aircraft based on the ILS localizer deviation to controlling the aircraft based on the inertial localizer deviation when the low-frequency anomaly is detected.

**FLIGHT CONTROL SYSTEM WITH LOW-FREQUENCY
INSTRUMENT LANDING SYSTEM LOCALIZER ANOMALY
DETECTION AND METHOD OF USE**

BACKGROUND

The field of the disclosure relates generally to flight control systems and, more specifically, to a flight control module utilizing a synthetic inertial localizer deviation for detecting low-frequency instrument landing system (ILS) localizer anomalies.

5 Many known aircraft feature an automated landing system that controls the aircraft during landing. Automated landing systems have become increasingly more common and are frequently relied on for both instrument landings under instrument flight rules (IFR) and landings performed under visual flight rules (VFR). Known automated landing systems utilize various receivers, such as multi-mode receivers
10 (MMRs), for example, to receive guidance signals transmitted from the ground. Such guidance signals may include, for example, ILS signals, global positioning service (GPS) landing system (GLS) signals, and/or microwave landing system (MLS) signals. The guidance signals inform the aircraft of its position relative to a desired vertical and lateral path to the runway and through roll-out after touchdown. The
15 desired vertical path is referred to as the glideslope and the lateral path is referred to as the localizer. The glideslope is typically defined as a **3°** descent with a desired intercept with the ground at **1000** feet beyond the runway threshold. The localizer guides the aircraft to the runway centerline.

The guidance signals transmitted from the ground are received by an on-
20 board antenna and routed to redundant MMRs. Each MMR computes a localizer deviation and a glideslope deviation that are routed to a flight control module that includes the automated landing system. The localizer deviation is an indication of the aircraft's position relative to the desired path to the runway centerline. For example, the localizer deviation may indicate the aircraft is approximately **2°** left of the runway
25 centerline. The glideslope deviation is an indication of the aircraft's position relative to the desired glideslope to the runway. For example, the glideslope deviation may

indicate the aircraft is 1° below the desired glideslope. The flight control module uses the localizer deviation and the glideslope deviation to command the automated landing system and to command control surfaces of the aircraft.

5 Under IFR conditions, the runway is typically kept clear of potential disruptions in the localizer signals transmitted from the ground. This is sometimes referred to as protecting the critical area. For example, taxiing aircraft and ground equipment are steered clear of the runway while a landing aircraft is on approach. Similarly, airborne aircraft are prohibited from flying across the airfield. In these situations, the throughput of the runway and surrounding airfield is reduced. Under 10 VFR conditions, restrictions on operation of the runway and surrounding airfield are more relaxed, i.e., the critical area is not protected, allowing the runway and surrounding airfield to operate with greater throughput. More frequent use of automated landing systems under VFR conditions increases the likelihood of disruptions in the localizer signals transmitted from the ground. Such disruptions are 15 referred to as anomalies in the localizer signals. Anomalies may result in drifting of the localizer input to automated landing systems, which may further result in misguiding the aircraft during landing and ultimately transitioning from automated landing to a manual landing.

20 **BRIEF DESCRIPTION**

According to one aspect of the present disclosure, a flight control module for detecting anomalies in instrument landing system (ILS) localizer signals during landing of an aircraft is provided. The flight control module includes a communication interface and a processor coupled thereto. The communication interface is 25 configured to receive inertial data, global position system (GPS) data, and an ILS localizer deviation for the aircraft. The processor is configured to compute an inertial localizer deviation based on the inertial data and a GPS localizer deviation based on the GPS data. The processor is configured to compare the ILS localizer deviation to

an average of the inertial localizer deviation and the GPS localizer deviation so as to detect a low-frequency anomaly in the ILS localizer deviation. The processor is configured to initiate a transition from controlling the aircraft based on the ILS localizer deviation to controlling the aircraft based on the inertial localizer deviation
5 when the low-frequency anomaly is detected.

According to another aspect of the present disclosure, a flight control system for landing an aircraft is provided. The flight control system includes a communication bus, a GPS system, a multi-mode receiver (MMR), and a flight control module. The communication bus is coupled to the GPS system, the MMR,
10 and the flight control module. The GPS system is configured to transmit a GPS position of the aircraft onto the communication bus. The MMR is configured to transmit an ILS localizer deviation onto the communication bus. The ILS localizer deviation is generated according to received ILS localizer signals. The flight control module is configured to receive inertial data for the aircraft, the GPS position, and
15 the ILS localizer deviation on the communication bus. The flight control module is further configured to compute a GPS localizer deviation and an inertial localizer deviation. The flight control module is further configured to detect a low-frequency anomaly in the received ILS localizer signals based on variances in the ILS localizer deviation relative to an average localizer deviation computed for the GPS localizer
20 deviation and the inertial localizer deviation.

According to yet another aspect of the present disclosure, a method of detecting a low-frequency anomaly in an instrument landing system (ILS) localizer signal received by an aircraft during landing is provided. The method includes receiving an ILS localizer signal. The method includes computing an ILS localizer
25 deviation based on the ILS localizer signal. The method includes controlling the aircraft according to the ILS localizer deviation. The method includes computing a GPS localizer deviation. The method includes computing an inertial localizer deviation. The method includes computing an average localizer deviation of the GPS localizer deviation and the inertial localizer deviation. The method includes detecting

the low-frequency anomaly when a difference between the average localizer deviation and the ILS localizer deviation exceeds a threshold. The method includes controlling the aircraft according to the inertial localizer deviation after detecting the low-frequency anomaly.

5 The features, functions, and advantages that have been discussed can be achieved independently in various embodiments or may be combined in yet other embodiments further details of which can be seen with reference to the following description and drawings.

BRIEF DESCRIPTION OF THE DRAWINGS

10 FIG. 1 is a top perspective diagram of an exemplary instrument landing system (ILS) for use in landing an aircraft;

 FIG. 2 is a side perspective diagram of the exemplary ILS shown in FIG. 1;

 FIG. 3 is a top perspective diagram of the aircraft shown in FIGS. 1 and 2 during landing;

15 FIG. 4 is block diagram of an exemplary flight control system for the aircraft shown in FIG.s 1-3;

 FIG. 5 is a functional block diagram of an exemplary flight control module for use in the flight control system shown in FIG. 4;

 FIG. 6 is a functional block diagram of another exemplary flight control module for use in the flight control system shown in FIG. 4; and

20 FIG. 7 is a flow diagram of an exemplary method of detecting a low-frequency anomaly in an ILS localizer signal received by an aircraft during landing.

DETAILED DESCRIPTION

As used herein, an element or step recited in the singular and proceeded with the word “a” or “an” should be understood as not excluding plural elements or steps unless such exclusion is explicitly recited. Furthermore, references to “one embodiment” of the present invention or the “exemplary embodiment” are not
5 intended to be interpreted as excluding the existence of additional embodiments that also incorporate the recited features.

FIG. 1 is a perspective diagram, from a top-view, of an exemplary instrument landing system (ILS) **100** for use in landing an aircraft **102**. FIG. 2 is another
10 perspective diagram, from a side view, of ILS **100**. Aircraft **102** is illustrated during approach for landing on a runway **104**. Runway **104** is characterized by a runway centerline **106** that extends towards and beyond aircraft **102** for illustrative purposes.

Referring to FIG. 1, in the exemplary embodiment, ILS **100** includes a localizer transmitter **108** and a glideslope transmitter **110**. Localizer transmitter **108**
15 transmits a first localizer beam **112** and a second localizer beam **114** towards aircraft **102**. First localizer beam **112** and second localizer beam **114** cooperate to define an ILS localizer signal that is received by aircraft **102** and processed to generate localizer deviations indicative of the lateral position of aircraft **102** relative to runway centerline **106**. The localizer signals are typically used to control aircraft **102** through
20 landing and rollout, i.e., deceleration along runway prior to taxi.

Under certain circumstances, another aircraft **115** or ground equipment (not shown) may travel through the path of first localizer beam **112** or second localizer beam **114** while aircraft **102** is on approach. Such activity may disrupt first localizer beam **112** or second localizer beam **114**, introducing multi-path effects in the ILS
25 localizer signals that are received by aircraft **102**. The multi-path effects typically manifest as low-frequency oscillations or high-frequency oscillations in the ILS localizer deviations. High-frequency oscillations typically result from an airborne aircraft or other fast moving object moving quickly through first localizer beam **112** or

second localizer beam **114**. Such oscillations typically have a period of **1** second or fewer, i.e., one Hertz or greater. Some known systems for aircraft **102** are configured to detect these high-frequency anomalies in the ILS localizer signals. Low-frequency oscillations typically result from slow moving vehicles on the ground, such as, for example, taxiing aircraft and ground equipment. Such oscillations typically have a period greater than **1** second, i.e., below one Hertz.

Referring to FIG. **2**, aircraft **102** approaches runway **104** along a predefined glideslope **116** that intercepts runway **104** at a predefined distance beyond a runway threshold for runway **104**. The predefined distance is typically, for example, at least **1000** feet from the runway threshold, which is typically the site of glideslope transmitter **110**. Glideslope **116** is defined by a glideslope angle **118** measured between glideslope **116** and runway **104**. A typical glideslope, for example, is defined by glideslope angle **118** being equal to approximately **3°**. Glideslope transmitter **110** transmits a first glideslope beam **120** and a second glideslope beam **122** towards aircraft **102**. First glideslope beam **120** and second glideslope beam **122** define an ILS glideslope signal that is received by aircraft **102** and that is processed to generate glideslope deviations indicative of the vertical position of aircraft **102** relative to glideslope **116**. The glideslope signals are typically used to control aircraft **102** until it reaches a flare altitude, i.e., an altitude when the nose of aircraft **102** pitches up prior to landing, which typically occurs between **50** and **75** feet. When the flare altitude is reached, aircraft **102** typically switches to a radio altimeter to guide aircraft **102** to runway **104** for landing. In alternative embodiments, flare altitude may be greater than **75** feet or, in some embodiments, below **50** feet.

FIG. **3** is a perspective diagram of aircraft **102** during landing. FIG. **3** illustrates a top-view of aircraft **102** landing on runway **104**. Aircraft **102** includes a guidance control point (GCP) **302** to which all flight control commands for aircraft **102** are referenced. GCP **302** is typically located at the nose of aircraft **102**. Aircraft **102** also includes an inertial reference unit (IRU) **304** that includes various sensors for detecting linear and angular accelerations of aircraft **102**, which are translatable

to accelerations, velocities, and attitude of aircraft **102** along three axis, i.e., pitch, roll, and yaw. IRU **304** is typically located at or near the center of aircraft **102**, which is illustrated in FIG. **3** as the intersection of the wings **305** and fuselage **307** of aircraft **102**. Accordingly, GCP **302** and IRU **304** are typically separated by a
5 distance **306** extending along a portion of the length of fuselage **307** of aircraft **102**.

Runway **104** includes runway centerline **106** characterized by a runway heading **308** relative to magnetic North (N). Runway heading **308** is generally known by aircraft **102** and its navigations systems, and is sometimes referred to as a magnetic runway heading. During landing, aircraft **102** travels at a ground speed **310**
10 along a track angle, or simply track **312**, relative to North. Ground speed **310** and track **312** are measurable by IRU **304**. Moreover, aircraft **102** travels with an aircraft heading **314** relative to North, which is generally defined as the direction the nose of aircraft **102** is pointing. Aircraft heading **314** is also measurable by IRU **304**. Notably, under certain circumstances, such as cross-winds, for example, track **312** and
15 aircraft heading **314** may be different.

FIG. **3** illustrates aircraft **102** at an orientation with a localizer deviation **316** measured from runway centerline **106** to IRU **304**. Given inertial accelerations measured by IRU **304** and runway heading **308**, aircraft **102** may compute an inertial cross-runway velocity **318** that, over time, increases or decreases localizer deviation
20 **316** and that can be translated to GCP **302**.

FIG. **4** is block diagram of an exemplary flight control system **400** for aircraft **102**, shown in FIGS. **1-3**. Flight control system **400** includes a flight control module **402** that controls aircraft **102** by transmitting commands to an actuator control module **404**. Flight control module **402** communicates with actuator control module
25 **404** over a communication bus **406**. Actuator control module **404** controls one or more actuators **408** that are attached to various flight control surfaces of aircraft **102**. Actuator control module **404** communicates with actuators **408** over a communication bus **410**.

Aircraft **102** includes various sensors **412** that measure flight parameters and generate data that is transmitted onto a communication bus **414**. Flight control module **402** is communicably coupled to communication bus **414** through communication interface **415** and gains access to the data.

5 Sensors **412** include various accelerometers and gyroscopes located at IRU **304** that provide cross-runway acceleration **416**, ground speed **310**, track angle **312**, and aircraft heading **314**. Communication bus **414** is configured to be coupled to IRU **304**, which provides the inertial data. Communication bus **414** is further coupled to various other data sources, such as a navigation system (not shown) that provides
10 runway heading **308** and a radar altimeter **418** that provides altitude above ground level for aircraft **102**. In alternative embodiments, other measurements of altitude may be utilized and may be available on aircraft **102**, such as, for example, barometric altitude or GPS ellipsoid altitude; however, altitude above ground level is most relevant to landing aircraft **102**. Communication bus **414** is further coupled to a
15 left MMR **420** and a right MMR **422**. Left MMR **420** provides a left MMR localizer deviation **424**. Right MMR **422** provides a right MMR localizer deviation **426**. Communication bus **414** is further coupled to a GPS **430** that provides a GPS position **432** for aircraft **102**.

 Communication interface **415** receives first and second localizer deviation
20 signals indicative of respective localizer deviations, such as left MMR localizer deviation **424** and right MMR deviation **426**, computed based on the localizer transmission received by aircraft **102**.

 Communication interface **415** is further configured to receive GPS position **432** over communication bus **414**. GPS position **432** is typically expressed as a
25 latitude and longitude.

 Flight control module **402** gains access to MMR localizer deviations **424** and **426** on communication bus **414** through communication interface **415**, which may include an ARINC-**429** interface circuit card configured to communicate within flight

control module **402** using peripheral component interconnect (PCI), PCI Express, PC/**104**, Ethernet, compact PCI, or other suitable protocol.

Flight control module **402** includes an automated landing system **428**. Flight control module **402** receives and processes data from communication bus **414** to
5 detect low-frequency ILS anomalies in ILS localizer signals. Flight control module **402** includes a complementary filter **434** and a lag filter **436** for filtering MMR localizer deviations **424** and **426**. Flight control module **402** includes an enabling processor **438** that enables or disables a low-frequency ILS localizer anomaly detection processor **440** based on inertial data, MMR localizer deviations **424** and
10 **426**, and GPS position **430**.

Enabling processor **438** operates, for example, when aircraft **102** is between altitudes of **600** feet and **200** feet. In the event a low-frequency ILS anomaly exists as aircraft **102** descends through **200** feet, enabling processor **438** prevents low-frequency ILS localizer anomaly detection processor **440** from engaging and
15 synchronizing to the anomaly. Enabling processor **438** generates a disable signal **442** that controls whether low-frequency ILS localizer anomaly detection processor **440** will engage when aircraft **102** descends below, for example, **200** feet.

Low-frequency ILS localizer anomaly detection processor **440** detects anomalies based on inertial data, MMR localizer deviations **424** and **426**, and GPS
20 position **430**. Low-frequency ILS localizer anomaly detection processor **440** generates an anomaly detected signal **444** that is used by flight control module **402** and automated landing system **428** to control aircraft **102**. For example, when an anomaly is detected in MMR localizer deviations **424** or **426**, automated landing system **428** transitions from control based on MMR localizer deviations **424** and **426**
25 to control based on an inertial localizer deviation.

FIG. **5** is a functional block diagram of flight control module **402** and, more specifically, a synthetic inertial localizer deviation (SILD) processor **500**. SILD processor **500** includes a complementary filter **502**, a runway heading correction

block **504**, an inertial cross-runway velocity block **506**, and a translation-to-GCP block **508**.

Complementary filter **502** blends high-frequency content of inertial data from IRU **304** with low-frequency content of an ILS localizer deviation **501**, such as MMR localizer deviations **424** and **426**, to produce a smooth, complementary-filtered localizer deviation **503**. Complementary-filtered localizer deviation **503** is fed back and subtracted **510** from ILS localizer deviation **501** to produce a localizer deviation error value. The localizer deviation error value is gained by K_3 and integrated **512**. The result of integration **512** is added **514** to cross-runway acceleration **416**, and then added **516** to the localizer deviation error value gained **518** by K_2 . The result of summing **516** is integrated **520** and added **522** to the localizer deviation error value gained **524** by K_1 . The result of summation **522** is a localizer deviation rate **526** that is integrated **528** to generate complementary-filtered localizer deviation **503**. Complementary filter **502** is further characterized by, but not limited to the following equation, where D_{CF} is complementary-filtered localizer deviation **503** as a function of time, t , and expressed in radians, D_{sel} is ILS localizer deviation **501** as a function of time, t , and expressed in radians, and A_{CR} is cross-runway acceleration **416** as a function of time, t .

$$D_{CF} = \left[\frac{K_1 S^2 + K_2 S + K_3}{S^3 + K_1 S^2 + K_2 S + K_3} \right] \times D_{sel} + \left[\frac{S}{S^3 + K_1 S^2 + K_2 S + K_3} \right] \times A_{CR} \quad \text{EQ. 1}$$

When aircraft **102** descends, as measured by radar altimeter **418**, below an altitude threshold **530**, complementary-filtered localizer deviation **503** is latched **532** as an initial condition for integration **534** of inertial cross-runway velocity **318**. Aircraft altitude from radar altimeter **418** is compared **536** to altitude threshold **530**, e.g., **200** feet, to trigger latch **532**.

Inertial cross-runway velocity block **506** computes inertial cross-runway velocity **318** as a function of an adjusted runway heading **538**, ground speed **310**, and track **312**. Inertial cross-runway velocity **318** is a projection of ground speed **310** along track **312** onto a cross-runway vector; computed as ground speed **310**

multiplied **540** by the sine **542** of the difference **544** between track **312** and adjusted runway heading **538**. Inertial cross-runway velocity block **506** is further characterized by, but not limited to, the following equation, where V_{CR} is cross-runway velocity **318** as a function of time, t , S_{GND} is ground speed **310** as a function of time, t , T is track **312** as a function of time, t , and $H_{run,adj}$ is adjusted runway heading **538** as a function of time, t , and expressed in radians.

$$V_{CR}(t) = S_{GND}(t) \times \sin(T(t) - H_{run,adj}(t)) \quad \text{EQ. 2}$$

Runway heading correction block **504** computes adjusted runway heading **538** to correct for errors in magnetic runway heading **308** available on aircraft **102**. Adjusted runway heading **538** represents actual runway azimuth relative to North for use in inertial cross-runway velocity block **506** and translation-to-GCP block **508**. Adjusted runway heading **538** is computed as a function of magnetic runway heading **308**, ground speed **310**, track **312**, and localizer deviation rate **526** from complementary filter **502**. Localizer deviation rate **526** is derived from EQ. 1, above, and is a component of complementary-filtered localizer deviation **503**. Localizer deviation rate **526** is a derivative of complementary-filtered localizer deviation **503**, i.e., $S \times D_{CF}$, and is represented by the following equation, where D_{rate} is localizer deviation rate **526** as a function of time, t , and expressed in radians per second.

$$D_{rate} = \left[\frac{K_1 S^3 + K_2 S^2 + K_3 S}{S^3 + K_1 S^2 + K_2 S + K_3} \right] \times D_{sel} + \left[\frac{S^2}{S^3 + K_1 S^2 + K_2 S + K_3} \right] \times A_{CR} \quad \text{EQ. 3}$$

In runway heading correction block **504**, localizer deviation rate **526** is divided **546** by ground speed **310** and subtracted **548**, along with magnetic runway heading **308**, from track **312**. The result of subtraction **548** is lag-filtered **550**, i.e., low-pass filtered, with a time constant, τ . The result of lag-filtering **550** is added **552** to magnetic runway heading **308** to generate adjusted runway heading **538**. Runway heading correction block **504** is further characterized, but not limited to, the following equation, where H_{run} is magnetic runway heading **308** and is expressed in radians.

$$H_{run,adj} = \frac{\tau S}{\tau S + 1} \times H_{run} + \frac{1}{\tau S + 1} \times T - \frac{1}{\tau S + 1} \times \frac{D_{rate}}{S_{GND}} \quad \text{EQ. 4}$$

Integration **534** integrates cross-runway velocity **318** from an initial condition at altitude threshold **530**, which is latched at complementary-filtered localizer deviation **503** with respect to GCP **302**. Integration **534** produces an inertial localizer deviation **535** with respect to IRU **304**. A compensation **554** is added **556** to inertial localizer deviation **535** to correct for the difference between inertial localizer deviation **535** at IRU **304** and inertial localizer deviation **535** at GCP **302**. For example, GCP **302**, at the nose of aircraft **102**, is at a different cross-runway position than IRU **304** during “crabbed” approaches for cross-wind landings. Compensation **554** accounts for changes in aircraft heading **314** that occur below altitude threshold **530**, because integration **534** is initialized, at altitude threshold **530**, to complimentary-filtered localizer deviation **503**, which is computed with respect to GCP **302**.

Translation-to-GCP block **508** computes compensation **554** as a function of aircraft heading **314**, adjusted runway heading **538**, and distance **306** along the fuselage of aircraft **102** between IRU **304** and GCP **302**. Distance **306** is multiplied **558** by the sine **560** of a difference **562** between aircraft heading **314** and adjusted runway heading **538**, yielding compensation **554** as a function of time, t . When aircraft **102** descends to altitude threshold **530**, a hold value **564** of compensation **554** is latched **532**. Hold value **564** represents the portion of compensation **554** already incorporated into inertial localizer deviation **535** via the initial condition of complementary-filtered localizer deviation **503** latched **532** at altitude threshold **530**. Hold value **564** is subtracted **566** from compensation **554** to capture only the changes in aircraft heading **314** that occur below altitude threshold **530**. Translation-to-GCP block **508** is further characterized, but not limited to, the following equation, where C is compensation **554** as a function of time, t , $L_{IRU-GCP}$ is distance **306** along the fuselage of aircraft **102** between IRU **304** and GCP **302**, H_{ac} is aircraft heading **314** as a function of time, t , and t_{200} is the time at which aircraft **102** descends to altitude threshold **530**.

$$C(t) = L_{IRU-GCP} \left[\sin \left(H_{ac}(t) - H_{run,adj}(t) \right) - \sin \left(H_{ac}(t_{200}) - H_{run,adj}(t_{200}) \right) \right]$$

EQ. 5

FIG. 6 is a functional block diagram of flight control module **402** for use in flight control system **400** shown in FIG. 4. Flight control module **402** includes enabling processor **438** and low-frequency ILS localizer anomaly detection processor **440**. Low-frequency ILS localizer anomaly detection processor **440** detects anomalies once aircraft has descended below a low approach altitude threshold, for example, **200** feet. Enabling processor **438** is operable when aircraft has descended below a high approach altitude threshold and either enables or disables low-frequency ILS localizer anomaly detection processor **440** based on anomaly detection between the low approach altitude threshold and the high approach altitude threshold. In one embodiment, for example, the low approach altitude threshold is **200** feet and the high approach altitude threshold is **600** feet. In certain embodiments, enabling processor is operable between the high approach altitude threshold and just below the low approach altitude threshold. For example, where the low approach altitude threshold is **200** feet and the high approach altitude threshold is **600** feet, enabling processor **438** is operable from **600** feet to **190** feet.

Enabling processor **438** includes a SILD processor **602**, such as SILD processor **500** shown in FIG. 5, and a GPS localizer deviation processor **604**. SILD processor **602** computes inertial localizer deviation **535** as a function of complementary-filtered localizer deviation **503**, ground speed **310**, track **312**, aircraft heading **314**, and runway heading **308**. GPS localizer deviation processor **604** computes a GPS localizer deviation **606** as a function of runway heading **308**, complementary-filtered localizer deviation **503**, and GPS position **432**, shown in FIG. 4. GPS position **432** is represented by a GPS latitude **608** and a GPS longitude **610**. SILD processor **602** and GPS localizer deviation processor **604** are initialized when aircraft **102** descends to the high approach altitude threshold to the value of complementary-filtered localizer deviation **503** at the high approach altitude. For example, SILD processor **602** and GPS localizer deviation processor **604** are

initialized to the value of complementary-filtered localizer deviation **503** when aircraft **102** reaches **600** feet altitude above ground level.

Enabling processor **438** includes an average module **612** that computes an average localizer deviation **614** as an average of inertial localizer deviation **535** and
5 GPS localizer deviation **606**. Average localizer deviation **614** is compared to a lag-
filtered localizer deviation **616** to detect low-frequency ILS localizer anomalies. Lag-
filtered localizer deviation **616** is a product of passing ILS localizer deviation **501**
through lag filter **436**. Enabling processor **438** compares **618** an absolute value, or
10 magnitude **620**, of a difference **622** between lag-filtered localizer deviation **616** and
average localizer deviation **614** to a threshold **624**. Generally, if magnitude **620**
exceeds threshold **624**, enabling processor **438** declares an anomaly. Threshold **624**
is generally in the range of **25** to **100** feet. For example, in one embodiment,
threshold **624** is **50** feet. Enabling processor **438** utilizes lag filter **436** to smooth ILS
15 localizer deviation **501** rather than complementary filter **434**, because
complementary filter **434** blends inertial data with ILS localizer deviation **501** and
would slow divergence of the ILS localizer deviation **501**, and further would impede
detection of low-frequency anomalies.

Enabling processor **438** includes a comparator **626** that compares inertial
localizer deviation **535** and GPS localizer deviation **606**. Comparator **626** checks that
20 inertial localizer deviation **535** and GPS localizer deviation **606** agree within a
threshold. If inertial localizer deviation **535** and GPS localizer deviation **606** disagree
enough, enabling processor **438** disables low-frequency ILS localizer anomaly
detection processor **440** via disable signal **442**.

Enabling processor **438** includes various logic to generate disable signal **442**
25 and an anomaly detected signal **628**. Anomaly detected signal **628** is logically true
when inertial localizer deviation **535** and GPS localizer deviation **606** agree within
the threshold, as determined by comparator **626**, and when comparison **618** of
difference **622** to threshold **624** is true. Anomaly detected signal **628** is generated by
an and-gate **630**. Enabling processor **438** includes a time delay **632** to ensure

difference **622** between lag-filtered localizer deviation **616** and average localizer deviation **614** exceeds threshold **624** for at least a minimum duration of time before declaring an anomaly. Enabling processor **438** includes an or-gate **634** to control disabling signal **442**. Disabling signal **442** is logically true when an anomaly is detected, as indicated by anomaly detected signal **628**, or when comparator **626** indicates inertial localizer deviation **535** and GPS localizer deviation **606** disagree.

Low-frequency ILS localizer anomaly detection processor **440** includes a SILD processor **636**, such as SILD processor **500** shown in FIG. **5**, and a GPS localizer deviation processor **638**. SILD processor **636** computes inertial localizer deviation **535** as a function of complementary-filtered localizer deviation **503**, ground speed **310**, track **312**, aircraft heading **314**, and runway heading **308**. GPS localizer deviation processor **604** computes GPS localizer deviation **606** as a function of runway heading **308**, complementary-filtered localizer deviation **503**, and GPS position **432**, shown in FIG. **4**. GPS position **432** is represented by a GPS latitude **608** and a GPS longitude **610**. SILD processor **636** and GPS localizer deviation processor **638** are initialized when aircraft **102** descends to the low approach altitude threshold to the value of complementary-filtered localizer deviation **503** at the low approach altitude. For example, SILD processor **636** and GPS localizer deviation processor **638** are initialized to the value of complementary-filtered localizer deviation **503** when aircraft **102** reaches **200** feet altitude above ground level.

Low-frequency ILS localizer anomaly detection processor **440** includes an average module **640** that computes average localizer deviation **614** as an average of inertial localizer deviation **535** and GPS localizer deviation **606**. Average localizer deviation **614** is compared to lag-filtered localizer deviation **616** to detect low-frequency ILS localizer anomalies. Low-frequency ILS localizer anomaly detection processor **440** compares **642** an absolute value **644** of a difference **646** between lag-filtered localizer deviation **616** and average localizer deviation **614** to a threshold **648**. Generally, if difference **646** exceeds threshold **648**, Low-frequency ILS localizer anomaly detection processor **440** declares an anomaly. Threshold **648** is generally

in the range of **1** to **50** feet. For example, in one embodiment, threshold **648** is **25** feet. Low-frequency ILS localizer anomaly detection processor **440**, as in enabling processor **438**, utilizes lag filter **436** to smooth ILS localizer deviation **501** rather than complementary filter **434**.

5 Low-frequency ILS localizer anomaly detection processor **440** includes a comparator **650** that compares inertial localizer deviation **535** and GPS localizer deviation **606**. Comparator **650** checks that inertial localizer deviation **535** and GPS localizer deviation **606** agree within a threshold. If inertial localizer deviation **535** and GPS localizer deviation **606** disagree enough, low-frequency ILS localizer anomaly
10 detection processor **440** cannot use comparison **642** to detect low-frequency ILS localizer anomalies.

 Low-frequency ILS localizer anomaly detection processor **440** includes an and-gate **652** to enforce several necessary conditions for detection of low-frequency ILS localizer anomalies. The first condition is that disable signal **442** is false,
15 generally indicating that enabling processor **438** has not disabled low-frequency ILS localizer anomaly detection processor **440**. More specifically, disabling signal **442** should indicate enabling processor **438** has not detected a low-frequency ILS localizer anomaly in its operative range of altitudes, i.e., between a high approach altitude threshold and a low approach altitude threshold, e.g., below **600** feet and
20 above **200** feet. The second condition is that comparator **650** indicates inertial localizer deviation **535** and GPS localizer deviation **606** agree within the appropriate threshold.

 A third condition is that aircraft **102** has not completed performance of landing and rollout. Low-frequency ILS localizer anomaly detection processor **440** includes a
25 time delay **654** that is triggered when aircraft **102** touches down. Touch down of aircraft **102** is indicated by a touch down signal **656** that is received, for example, through communication interface **415** and communication bus **414**. Touch down signal **656** indicates touch down of aircraft **102**, for example, by a logic-high signal. Time delay **654** ensures low-frequency ILS localizer anomaly detection processor

440 remains enabled for a predetermined duration after touch down and during rollout. When the predetermined duration elapses, time delay **654** negates the third condition for detection of low-frequency ILS localizer anomalies.

Given that (1) low-frequency ILS localizer anomaly detection processor **440** is not disabled by enabling processor **438**, (2) inertial localizer deviation **535** and GPS localizer deviation **606** agree, and (3) aircraft **102** has not completed landing and rollout, low-frequency ILS localizer anomaly detection processor **440** is free to carry out anomaly detection. When comparison **642** detects an anomaly, the detection is latched **658** and anomaly detected signal **444** indicates a low-frequency ILS localizer anomaly has been detected. In response, flight control module **402** may carry out one or more remedial steps. In certain embodiments, flight control module **402** transitions control of aircraft **102**, by automated landing system **428**, from ILS localizer-based control, i.e., based on complementary-filtered localizer deviation **503**, to inertial localizer-based control, i.e., based on inertial localizer deviation **535**. In certain embodiments, flight control module **402** may transmit an indicator signal to a pilot interface system (not shown) to notify a pilot of aircraft **102** of the detected anomaly. In such embodiments, control of aircraft **102** may ultimately be transitioned from automated landing system **428** to the pilot. The pilot interface system may relay anomaly detected signal **444** to the pilot using one or more of visual indicators and aural indicators.

Generally, low-frequency ILS localizer anomalies are temporary, presenting and resolving themselves within the time frame of the approach by aircraft **102**. Low-frequency ILS localizer anomaly detection processor **440** includes the ability to recover from the detected low-frequency ILS localizer anomaly through a “healing” procedure. Low-frequency ILS localizer anomaly detection processor **440** includes a comparison **660** of the difference **646** between average localizer deviation **614** and lag-filtered ILS localizer **616** to a healing threshold **662**. When a low-frequency ILS localizer anomaly is latched **658**, the anomaly can be reset **662** if difference **646** falls below healing threshold **662**. Low-frequency ILS localizer anomaly detection

processor **440** includes a time delay **664** to ensure difference **646** remains below healing threshold **662** for a predetermined duration before negating anomaly detected signal **444**. Healing threshold **662** is generally in the range of **1** to **50** feet, and is typically less than threshold **648** for initially declaring the low-frequency ILS localizer anomaly. For example, difference **646** must rise above threshold **648**, e.g., **25** feet, to declare the low-frequency ILS localizer anomaly, and difference **646** must fall below healing threshold **662**, e.g., **15** feet, to negate that low-frequency ILS localizer anomaly. Moreover, difference **646** must fall below healing threshold **662** for the predetermined duration defined by time delay **664** before negating the low-frequency ILS localizer anomaly.

FIG. **7** is a flow diagram of a method **700** of detecting a low-frequency anomaly in an ILS localizer signal received by aircraft **102** during landing. Method **700** may be embodied in or carried out by flight control system **400** and flight control module **402**, shown in FIG. **4**. Method **700** begins when an ILS localizer signal is received **710** by aircraft **102**. Aircraft **102** routes the ILS localizer signal to left MMR **420** and right MMR **422**, which compute **720** ILS localizer deviations, i.e., left MMR localizer deviation **424** and right MMR localizer deviation **426**, based on the received ILS localizer signal. Flight control module **402** selects complementary-filtered localizer deviation **503** for use in controlling **730** aircraft **102** using automated landing system **428** during landing.

Flight control module **402** and, more specifically, low-frequency ILS localizer anomaly detection processor **440** and GPS localizer deviation processor **638** compute **740** GPS localizer deviation **606** based on GPS position **432** of aircraft **102** with respect to runway centerline **106**. Flight control module **402** and, more specifically, low-frequency ILS localizer anomaly detection processor **440** and SILD processor **636** compute **750** inertial localizer deviation **535** based on inertial data from IRU **304**. In alternative embodiments, GPS localizer deviation processor **638** and SILD processor **636** are embodied in another device other than flight control module **402**, where automated landing system **428** resides. For example, in one

embodiment, low-frequency ILS localizer anomaly detection processor **440** is embodied in left MMR **420**, right MMR **422**, or both. Similarly, enabling processor **438** may be embodied in left MMR **420**, right MMR **422**, or both, for example.

Low-frequency ILS localizer anomaly detection processor **440** averages **760**
5 GPS localizer deviation **606** and inertial localizer deviation **535** for comparison **642**
to lag-filtered localizer deviation **616**. A low-frequency anomaly is detected **770** when
difference **646** between lag-filtered localizer deviation **616** and average localizer
deviation **614** exceeds threshold **648**. When the low-frequency anomaly is detected,
10 control of aircraft **102** is transitioned from being ILS localizer-based to inertial
localizer-based. Aircraft **102** is then controlled **780** according to inertial localizer
deviation **535** until manual control of aircraft **102** is assumed by a pilot.

In certain embodiments, method **700** includes transmitting an indicator, based
on anomaly detected signal **628** and/or anomaly detected signal **444** to a pilot
interface system, the indicator indicative of detection of the low-frequency anomaly.
15 The pilot interface system may, in certain embodiments, relay the indicator to a pilot
in the form of a visual indicator, an aural indicator, or both.

In certain embodiments, computing **750** inertial localizer deviation **535**
includes initializing inertial localizer deviation **535** to complementary-filtered localizer
deviation **503** when aircraft **102** descends below a low approach altitude threshold,
20 e.g., **200** feet. Computing **750**, in such embodiments, further includes integrating
inertial data, such as, for example, ground speed **310**, track angle **312**, cross-runway
acceleration **416**, and aircraft heading **314** on communication bus **414**.

In certain embodiments, method **700** includes initializing SILD processor **602**
to complementary-filtered localizer deviation **503** when aircraft **102** descends below
25 a high approach altitude threshold, e.g., **600** feet. In such embodiments, method **700**
includes initializing GPS localizer deviation processor **604** to complementary-filtered
localizer deviation **503** when aircraft **102** descends below the high approach altitude
threshold. In such embodiments, method **700** further includes detecting, by enabling

processor **438**, an earlier low-frequency anomaly based on variances between lag-filtered localizer deviation **616** and average localizer deviation **614**, which is averaged **612** based on inertial localizer deviation **535** and GPS localizer deviation **606**. When the earlier low-frequency anomaly is detected, enabling processor **438**
5 disables low-frequency ILS localizer anomaly detection processor **440** before aircraft **102** descends beneath the low approach altitude threshold.

In certain embodiments, method **700** includes computing a difference between GPS localizer deviation **606** and inertial localizer deviation **535**, and disabling detection of the low-frequency anomaly when the difference exceeds a
10 threshold. Such embodiments carry out the comparison in comparator **626** and/or comparator **650**.

The above described embodiments of flight control systems for use by an aircraft during landing provide low-frequency ILS localizer anomaly detection. More specifically, embodiments described herein provide a flight control module that
15 utilizes a combination of an inertial localizer deviation and a GPS localizer deviation to compare with an ILS localizer deviation to detect low-frequency anomalies. For example, if the ILS localizer deviation differs from an average of the GPS localizer deviation and the inertial localizer deviation, an anomaly is detected. Embodiments described herein further provide a monitoring function by an enabling processor that
20 detects low-frequency anomalies below a high approach altitude threshold, e.g., **600** feet down to below a low approach altitude threshold, e.g., **190** feet. When a low-frequency anomaly is detected by the enabling processor, detection below the low approach altitude threshold, e.g., **200** feet, is disabled to prevent synchronization of the computations to the anomaly itself rather than to the actual distance from runway
25 centerline. Detection of low-frequency anomalies below the low approach altitude threshold is carried out similarly to the monitoring from **600** feet to **190** feet, and generates an anomaly detected signal that can be used to transition control of the aircraft from ILS localizer-based to inertial localizer-based. Further, embodiments described herein provide visual indicators, aural indicators, or both to a pilot of the

aircraft to alert them of the low-frequency anomaly such that the landing can be performed manually.

Exemplary embodiments of methods, systems, and apparatus for flight control systems are not limited to the specific embodiments described herein, but rather, components of systems and/or steps of the methods may be utilized independently and separately from other components and/or steps described herein. For example, the methods may also be used in combination with other non-conventional flight control systems, and are not limited to practice with only the systems and methods as described herein. Rather, the exemplary embodiment can be implemented and utilized in connection with many other applications, equipment, and systems that may benefit from increased efficiency, reduced operational cost, and reduced capital expenditure.

An exemplary technical effect of the methods, systems, and apparatus described herein includes at least one of: (a) detecting low-frequency ILS localizer anomalies during landing of an aircraft; (b) transitioning from ILS localizer deviation-based control of the aircraft during an automated landing to a SILD-based control of the aircraft when low-frequency ILS localizer anomalies are detected; (c) improving availability of automated landing systems through low-frequency ILS localizer anomaly detection; (d) reducing false positive detection of low-frequency ILS localizer anomalies through verification of SILD data with GPS localizer deviation data; (e) improving localizer guidance below **200** feet through monitoring ILS localizer signals for low-frequency anomalies from **600** to below **200** feet; and (f) providing an indicator of detected low-frequency ILS localizer anomalies via one or more visual or aural indicators.

Some embodiments involve the use of one or more electronic or computing devices. Such devices typically include a processor, processing device, or controller, such as a general purpose central processing unit (CPU), a graphics processing unit (GPU), a microcontroller, a reduced instruction set computer (RISC) processor, an application specific integrated circuit (ASIC), a programmable logic

circuit (PLC), a field programmable gate array (FPGA), a digital signal processing (DSP) device, and/or any other circuit or processing device capable of executing the functions described herein. The methods described herein may be encoded as executable instructions embodied in a computer readable medium, including, without
5 limitation, a storage device and/or a memory device. Such instructions, when executed by a processing device, cause the processing device to perform at least a portion of the methods described herein. The above examples are exemplary only, and thus are not intended to limit in any way the definition and/or meaning of the terms processor, processing device, and controller.

10 In the embodiments described herein, memory may include, but is not limited to, a computer-readable medium, such as a random access memory (RAM), and a computer-readable non-volatile medium, such as flash memory. Alternatively, a floppy disk, a compact disc – read only memory (CD-ROM), a magneto-optical disk (MOD), and/or a digital versatile disc (DVD) may also be used. Also, in the
15 embodiments described herein, additional input channels may be, but are not limited to, computer peripherals associated with an operator interface such as a mouse and a keyboard. Alternatively, other computer peripherals may also be used that may include, for example, but not be limited to, a scanner. Furthermore, in the exemplary embodiment, additional output channels may include, but not be limited to, an
20 operator interface monitor.

This written description uses examples to disclose various embodiments, which include the best mode, to enable any person skilled in the art to practice those embodiments, including making and using any devices or systems and performing any incorporated methods. The patentable scope is defined by the claims, and may
25 include other examples that occur to those skilled in the art. Such other examples are intended to be within the scope of the claims if they have structural elements that do not differ from the literal language of the claims, or if they include equivalent structural elements with insubstantial differences from the literal languages of the claims.

EMBODIMENTS IN WHICH AN EXCLUSIVE PROPERTY OR PRIVILEGE IS CLAIMED ARE DEFINED AS FOLLOWS:

1. A flight control module for detecting anomalies in instrument landing system (ILS) localizer signals during landing of an aircraft, comprising:

a communication interface configured to receive inertial data for said aircraft, global positioning system (GPS) data for said aircraft, and an ILS localizer deviation; and

a processor coupled to said communication interface and configured to:

compute an inertial localizer deviation based on the inertial data,

compute a GPS localizer deviation based on the GPS data,

compare the ILS localizer deviation to an average of the inertial localizer deviation and the GPS localizer deviation to detect a low-frequency anomaly in the ILS localizer deviation, and

initiate a transition from controlling said aircraft based on the ILS localizer deviation to controlling said aircraft based on the inertial localizer deviation when the low-frequency anomaly is detected.

2. The flight control module of Claim 1, wherein said communication interface is further coupled to a radar altimeter configured to measure altitude of said aircraft above ground level, and wherein said processor is further configured to initialize computation of the inertial localizer deviation and the GPS localizer deviation to the ILS localizer deviation when said aircraft descends beneath a low approach altitude threshold.

3. The flight control module of Claim 2, wherein said processor is further configured to:

initialize a second computation of the inertial localizer deviation and the GPS localizer deviation when said aircraft descends beneath a high approach altitude threshold;

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compare the second computations to the ILS localizer deviation to detect an earlier low-frequency anomaly when said aircraft is located between the high approach altitude threshold and the low approach altitude threshold; and

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disable detection of the low-frequency anomaly below the low approach altitude threshold when the earlier low-frequency anomaly is detected between the high approach altitude threshold and the low approach altitude threshold.

4. The flight control module of Claim 3, wherein the low approach altitude threshold is **200** feet and the high approach altitude threshold is **600** feet.

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5. The flight control module of any one of Claims 1-4, wherein said processor is further configured to:

compare the inertial localizer deviation to the GPS localizer deviation;
and

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disable detection of the low-frequency anomaly when a difference between the inertial localizer deviation and the GPS localizer deviation exceeds a threshold.

6. The flight control module of any one of Claims 1-5, wherein the processor is further configured to:

compute the average of the inertial localizer deviation and the GPS localizer deviation;

compute an absolute value difference between the average and the ILS localizer deviation; and

latch an anomaly indicator when the absolute value difference exceeds a trip threshold.

5 7. The flight control module of Claim 6, wherein the processor is further configured to reset said anomaly indicator after the absolute value difference falls below a healing threshold for a healing duration.

8. The flight control module of any one of Claims 1-7, wherein the processor is further configured to initiate an indication to a pilot of said aircraft that the low-
10 frequency anomaly has been detected.

9. A flight control system for landing an aircraft, said flight control system comprising:

a communication bus;

15 a global positioning system (GPS) coupled to said communication bus and configured to transmit a GPS position of said aircraft onto said communication bus;

20 a multi-mode receivers (MMR) coupled to said communication bus and configured to transmit an instrument landing system (ILS) localizer deviation onto said communication bus, the ILS localizer deviation generated according to received ILS localizer signals; and

a flight control module coupled to said communication bus and configured to:

receive inertial data for said aircraft, the GPS position, and the ILS localizer deviation on said communication bus,

compute a GPS localizer deviation and an inertial localizer deviation, and

detect a low-frequency anomaly in the received ILS localizer signals based on variances in the ILS localizer deviation relative to an average localizer deviation computed for the GPS localizer deviation and the inertial localizer deviation.

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10. The flight control system of Claim **9** further comprising an automated landing system configured to transition from controlling said aircraft based on the ILS localizer deviation to controlling said aircraft based on the inertial localizer deviation when the low-frequency anomaly is detected.

11. The flight control system of any one of Claims **9-10**, wherein said flight control module is further configured to transmit an indicator to a pilot interface system, the indicator indicative of detection of the low-frequency anomaly.

12. The flight control system of Claim **11**, wherein said pilot interface system is configured to present a visual indicator to a pilot in response to the indicator received from said flight control module.

13. The flight control system of any one of Claims **11-12**, wherein said pilot interface system is configured to present an aural indicator to a pilot in response to the indicator received from said flight control module.

14. The flight control system of any one of Claims **9-13**, wherein said flight control module is further configured to:

enable detection of the low-frequency anomaly when said aircraft descends below a low approach altitude threshold; and

disable detection of the low-frequency anomaly below the low approach altitude threshold when the low-frequency anomaly is otherwise detected above the low approach altitude threshold.

15. The flight control system of any one of Claims **9-14**, wherein said flight control module is further configured to disable detection of the low-frequency anomaly when a difference between the GPS localizer deviation and the inertial localizer deviation exceeds a threshold.

5 **16.** The flight control system of any one of Claims **9-15**, wherein said flight control module is further configured to disable detection of the low-frequency anomaly after a rollout duration beginning when said aircraft touches down.

17. The flight control system of any one of Claims **9-16**, wherein said flight control module is further configured to:

10 initiate computation of the GPS localizer deviation and the inertial localizer deviation based on a complementary-filtered ILS localizer deviation; and

 compare the average localizer deviation to a lag-filtered ILS localizer deviation to detect the low-frequency anomaly.

15 **18.** The flight control system of any one of Claims **9-17**, wherein said flight control module is further configured to receive inertial data including ground speed, track angle, cross-runway acceleration, and aircraft heading on said communication bus.

20 **19.** A method of detecting a low-frequency anomaly in an instrument landing system (ILS) localizer signal received by an aircraft during landing, said method comprising:

 receiving an instrument landing system (ILS) localizer signal;

 computing a ILS localizer deviation based on the ILS localizer signal;

 controlling the aircraft according to the ILS localizer deviation;

25 computing a global position system (GPS) localizer deviation;

computing an inertial localizer deviation;

computing an average localizer deviation of the GPS localizer deviation and the inertial localizer deviation;

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detecting the low-frequency anomaly when a difference between the average localizer deviation and the ILS localizer deviation exceeds a threshold; and

controlling the aircraft according to the inertial localizer deviation after detecting the low-frequency anomaly.

10 **20.** The method of Claim **19**, wherein computing the inertial localizer deviation comprises:

initializing the inertial localizer deviation to the ILS localizer deviation when the aircraft descends below a low approach altitude threshold; and

15

integrating inertial data for the aircraft to continuously update the inertial localizer deviation below the low approach altitude threshold and through rollout.

21. The method of Claim **20** further comprising:

20

initializing a second inertial localizer deviation to the ILS localizer deviation when the aircraft descends below a high approach altitude threshold;

initializing a second GPS localizer deviation to the ILS localizer deviation when the aircraft descends below the high approach altitude threshold;

25

detecting an earlier low-frequency anomaly based on variances between the ILS localizer deviation and a second average localizer

deviation of the second inertial localizer deviation and the second GPS localizer deviation; and

disabling detection of the low-frequency anomaly before the aircraft descends beneath the low approach altitude threshold.

5 **22.** The method of any one of Claims **19-21** further comprising complementary-filtering the ILS localizer deviation for initializing the inertial localizer deviation and for controlling the aircraft.

23. The method of any one of Claims **19-22** further comprising:

10 computing a difference between the GPS localizer deviation and the inertial localizer deviation; and

disabling detection of the low-frequency anomaly when the difference exceeds a threshold.

24. The method of any one of Claims **19-23** further comprising:

lag-filtering the ILS localizer deviation; and

15 computing a difference between the average localizer deviation and the lag-filtered ILS localizer deviation for detection of the low-frequency anomaly.

20 **25.** The method of any one of Claims **19-24** further comprising transmitting an indicator to a pilot interface system, the indicator indicative of detection of the low-frequency anomaly.

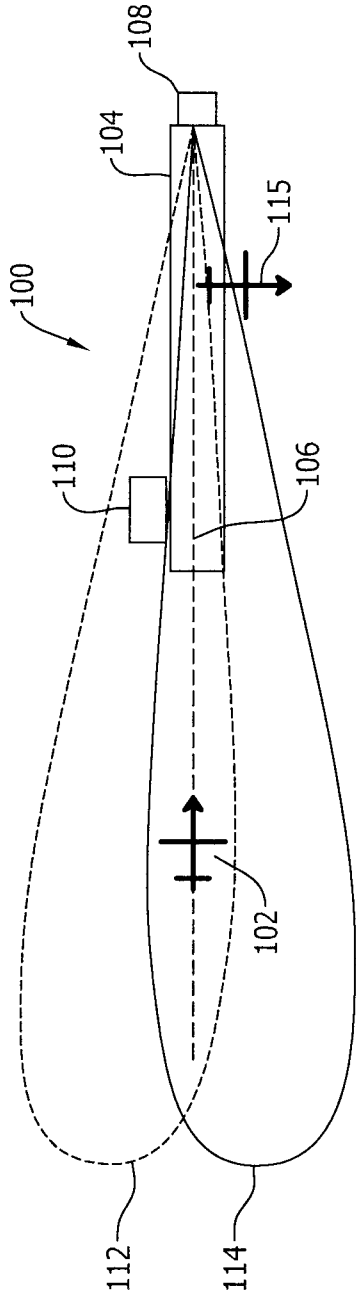


FIG. 1

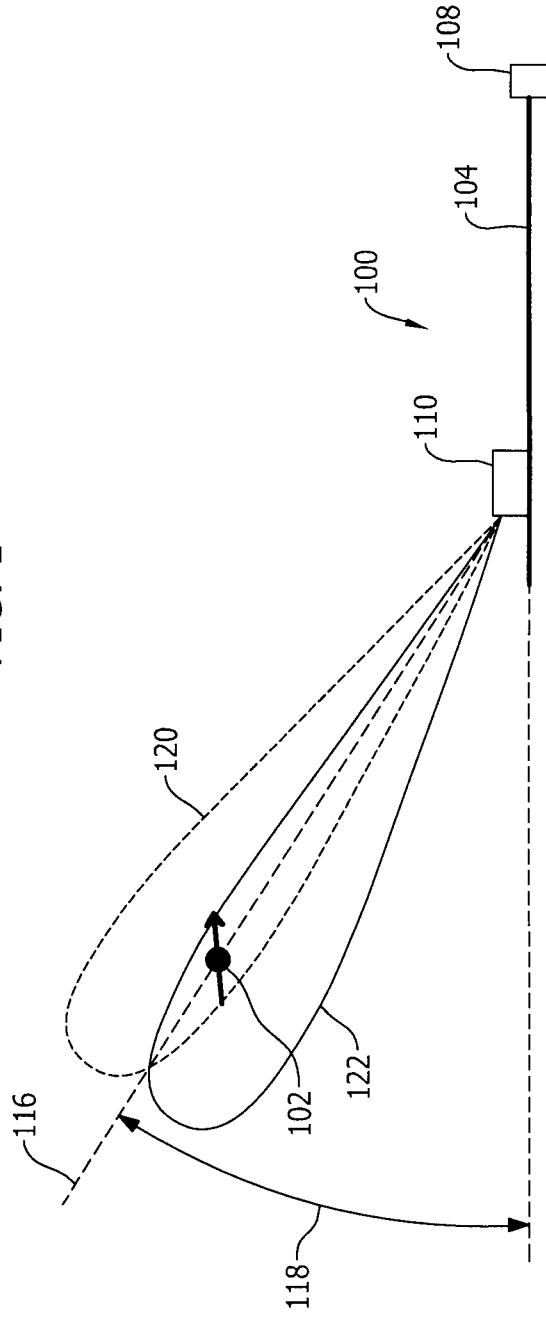


FIG. 2

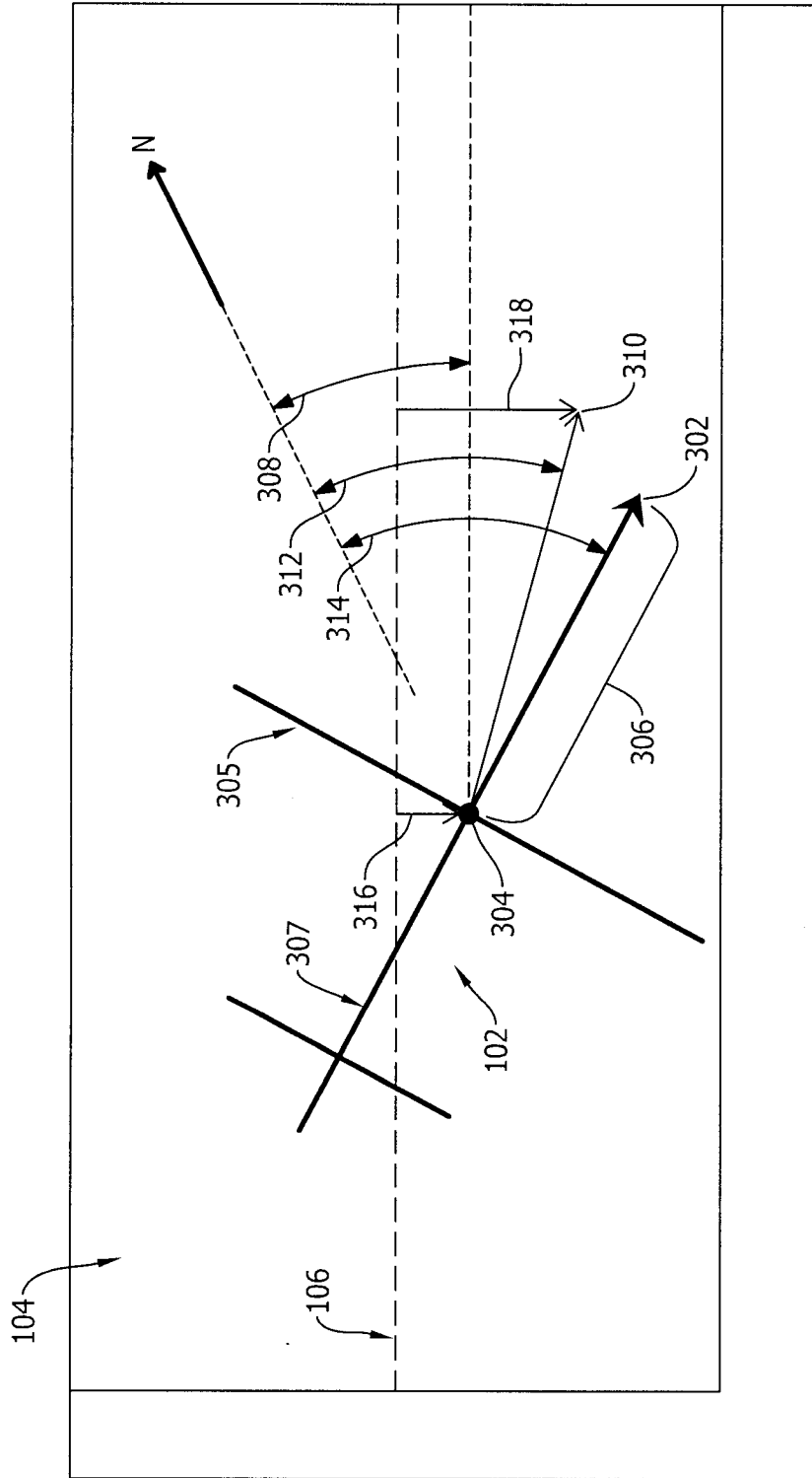


FIG. 3

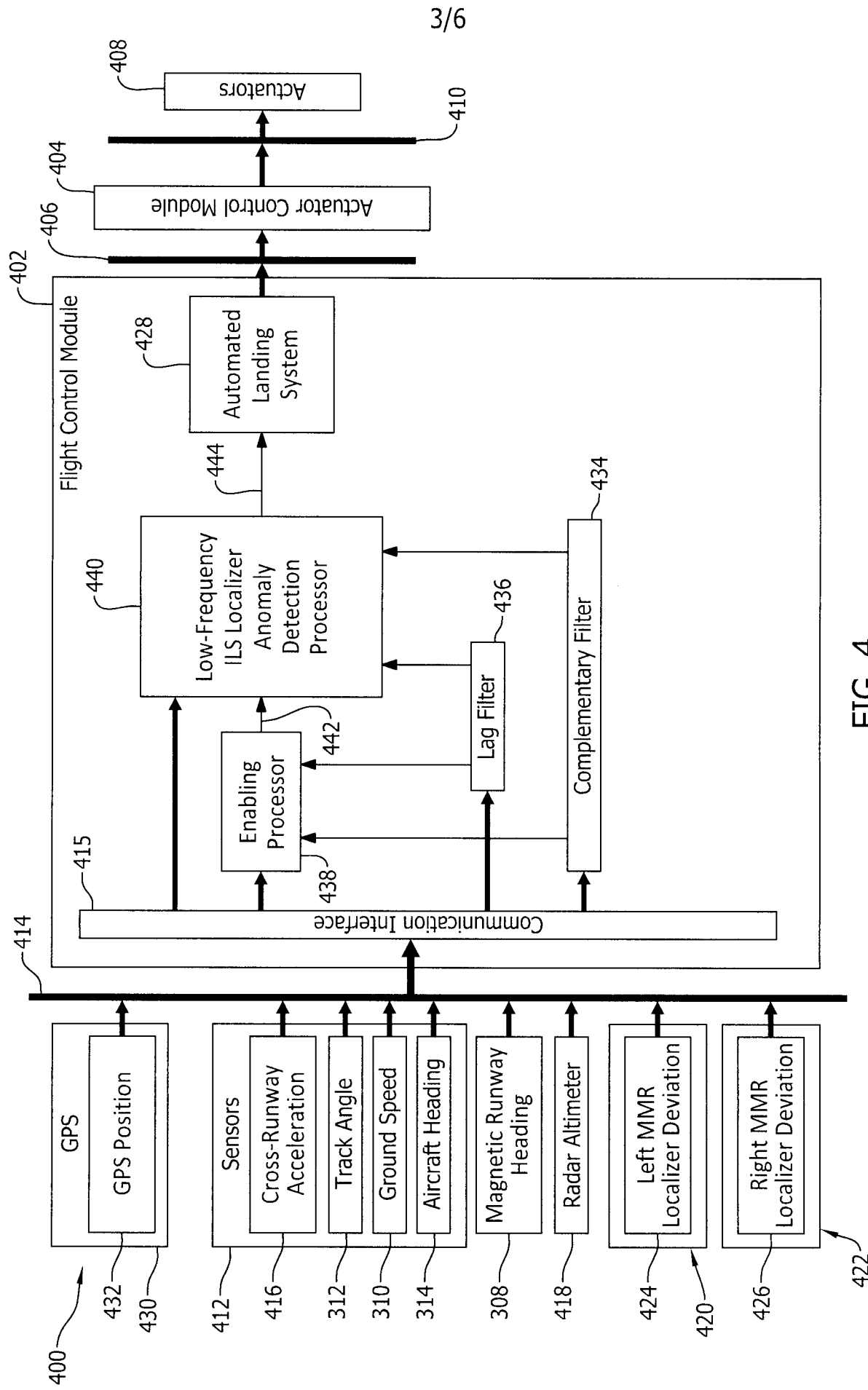


FIG. 4

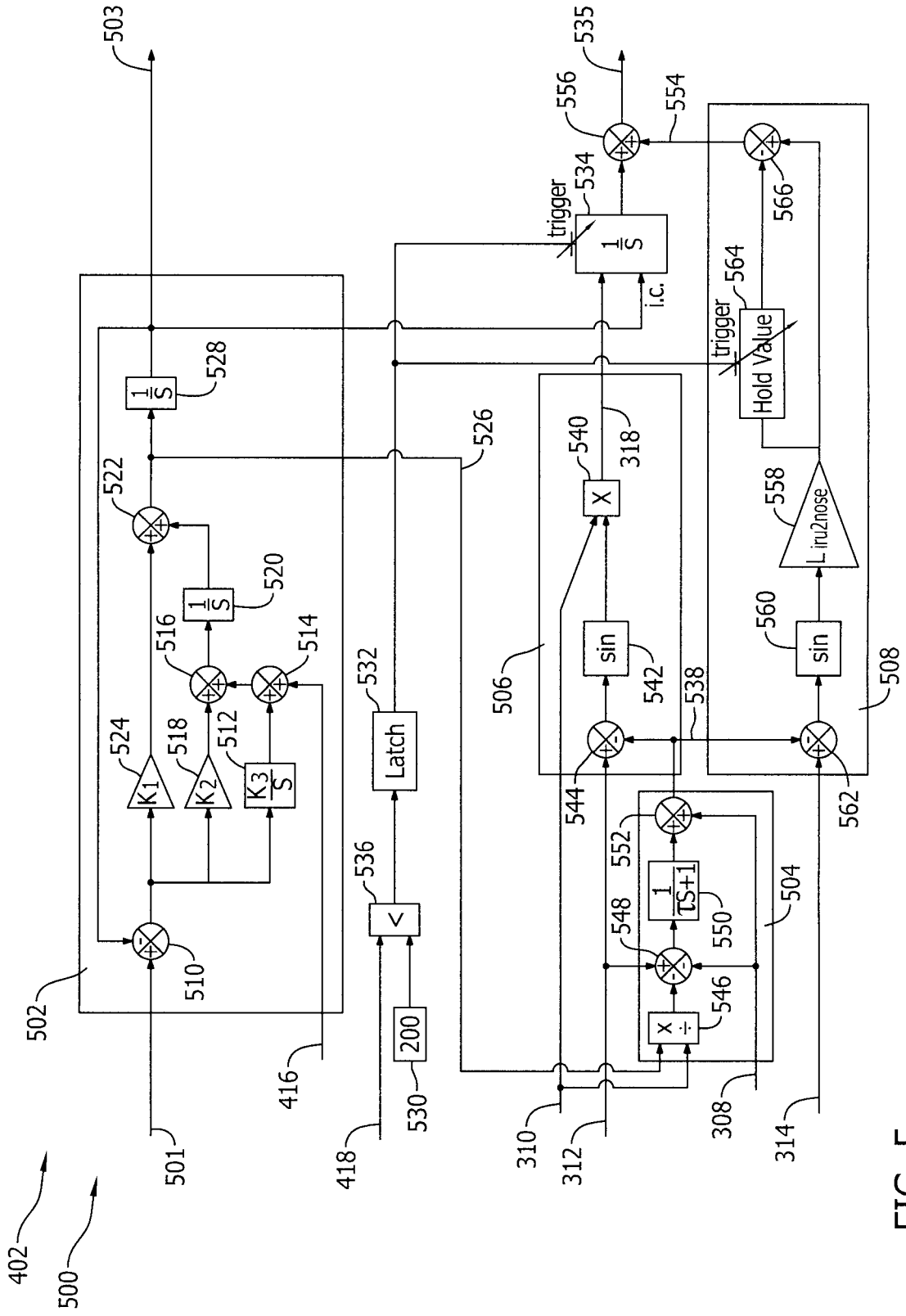
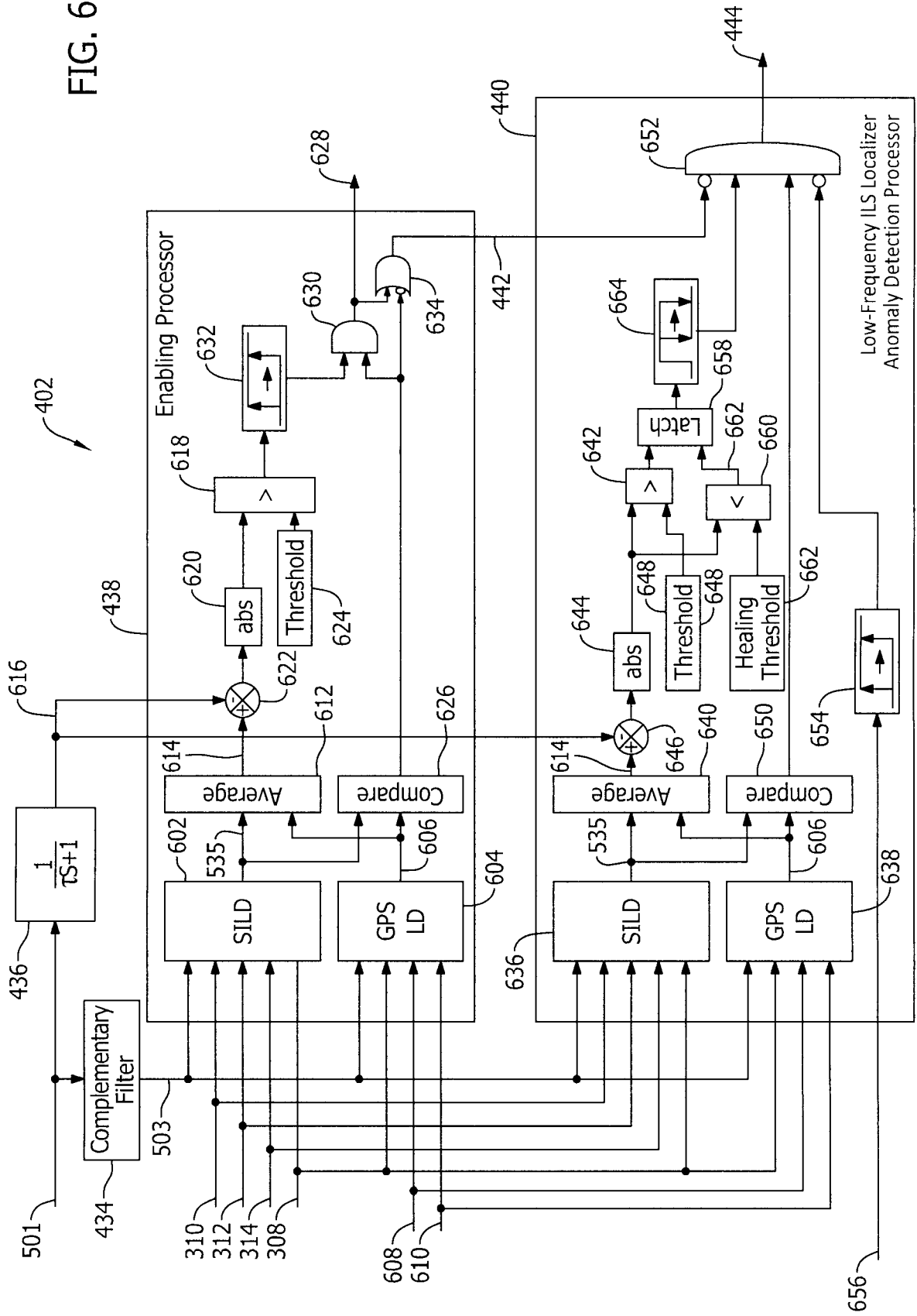


FIG. 5

FIG. 6



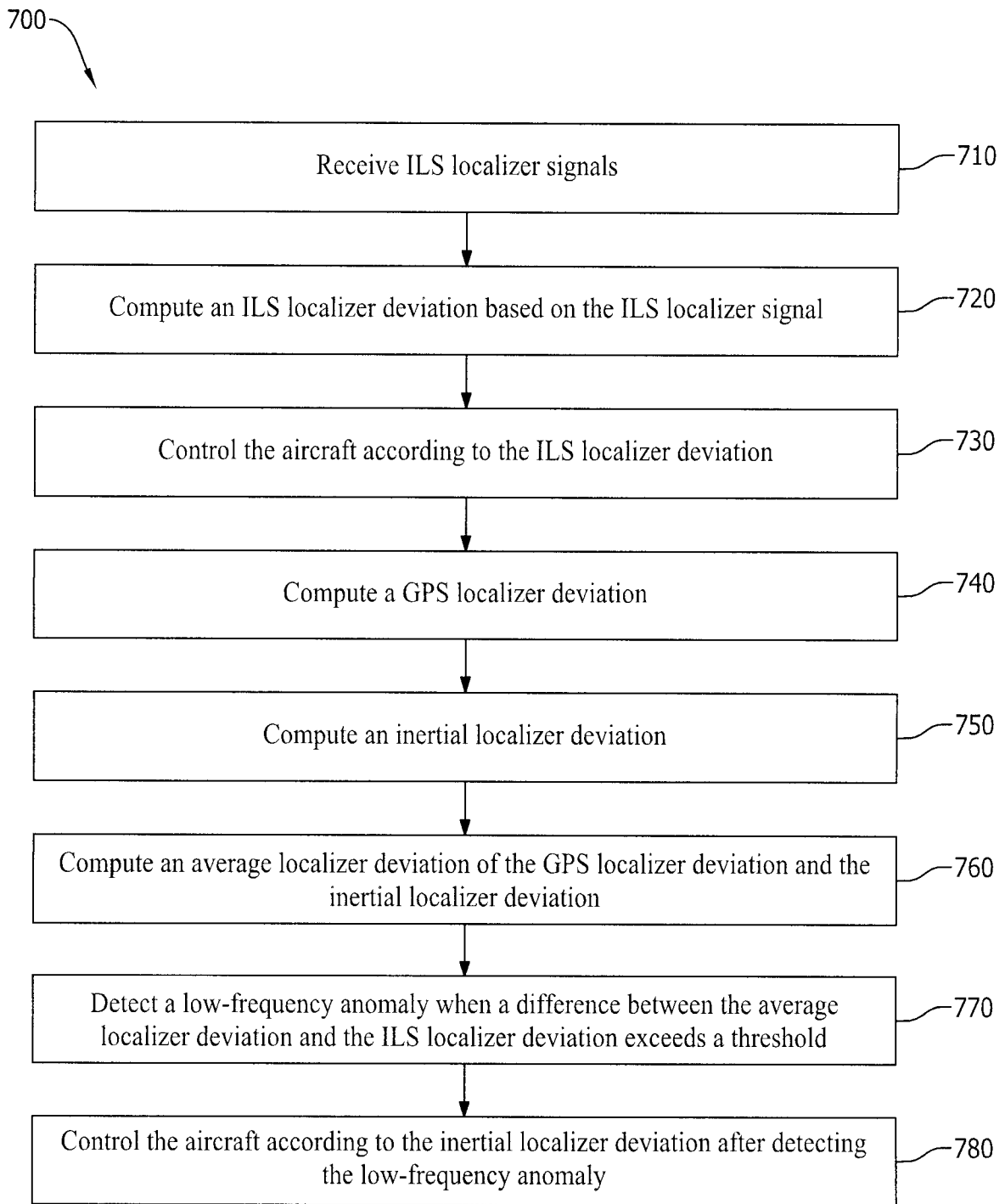


FIG. 7

