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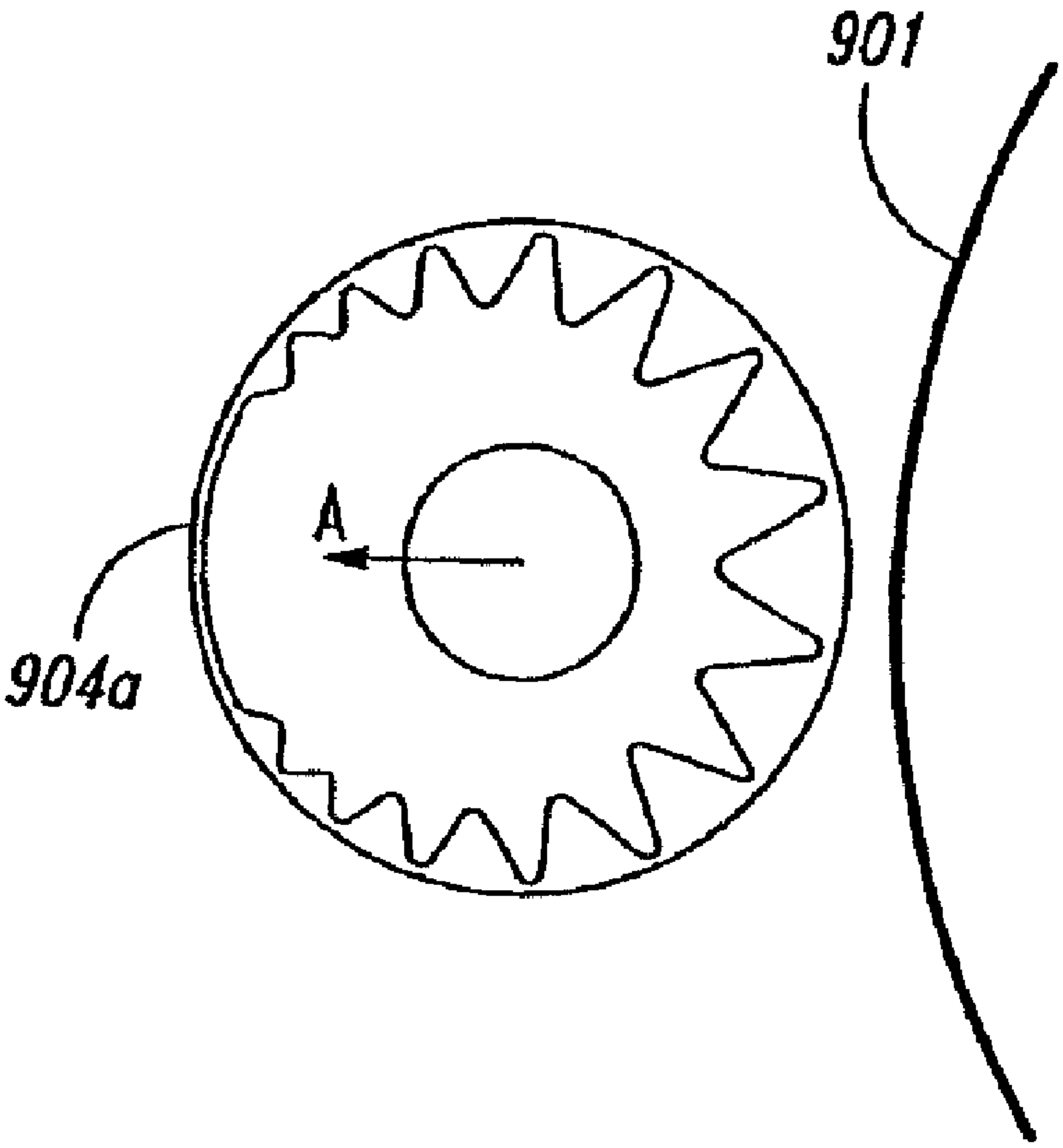

(57) **Abrégé/Abstract:**
A method for controlling aircraft noise. The method involves directing gas through a jet engine nozzle. The method also involves controlling a total thrust vector of the gas to be non-parallel to an acoustic intensity vector at one or more one acoustic frequencies by passing the gas adjacent to multiple nozzle projections having different geometric features, and mixing the gas with adjacent freestream air at the nozzle projections.

ABSTRACT

A method for controlling aircraft noise. The method involves directing gas through a jet engine nozzle. The method also involves controlling a total thrust vector of the gas to be non-parallel to an acoustic intensity vector at one or more one acoustic frequencies by passing the gas adjacent to multiple nozzle projections having different geometric features, and mixing the gas with adjacent freestream air at the nozzle projections.

JET ENGINE NOZZLE EXIT CONFIGURATIONS AND ASSOCIATED SYSTEMS AND METHODS

TECHNICAL FIELD

The present disclosure is directed to jet engine nozzle exit configurations and associated systems and methods, including nozzles having chevrons or other projections that vary in a circumferential or azimuthal manner around an exit perimeter of the nozzle.

BACKGROUND

Aircraft manufacturers are under continual pressure to reduce the noise produced by aircraft in order to satisfy increasingly stringent noise certification rules. Aircraft engines are a major contributor to overall aircraft noise. Accordingly, aircraft engines in particular have been the target of manufacturers' noise reduction efforts. Aircraft engines have been made significantly quieter as a result of advanced high bypass ratio engines. These engines derive a significant fraction of their total thrust not directly from jet exhaust, but from bypass air which is propelled around the core of the engine by an engine-driven forwardly mounted fan. While this approach has significantly reduced aircraft noise when compared with pure turbojet engines and low bypass ratio engines, engine and aircraft federal regulations nevertheless continue to require further engine noise reductions.

One approach to reducing engine noise is to increase the amount of mixing between the high velocity gases exiting the engine, and the surrounding freestream air. Figure **1** illustrates a nozzle **20** having "chevrons" that are designed to produce this effect. Chevrons generally include certain types of serrations on the nozzle lip, typically, triangular in shape having some curvature in the lengthwise cross-section, which slightly immerses them in the adjacent flow. The chevron can project either inwardly or outwardly, by an amount that is on the order of the upstream boundary layer thickness on the inner or outer surface, respectively. In general, the chevron planform shape can also be trapezoidal or rectangular. The nozzle **20** includes a core flow duct **40** through which the engine core flow is directed, and a fan flow duct **30** arranged annularly around the core flow duct **40**, through which the fan air passes. The exit aperture of the fan flow duct **30** can include fan flow chevrons **35**, and the exit aperture of the core flow duct **40** can include core flow chevrons **45**. The chevrons typically reduce the low-frequency noise by increasing the rate at which the engine flow streams mix with the surrounding freestream air at the length scale of the nozzle diameter. While this approach has resulted in noise reduction compared with nozzles that do not include chevrons, further noise reduction is desired to meet community noise standards.

SUMMARY

The following summary is provided for the benefit of the reader only, and is not intended to limit in any way the invention as set forth by the claims. Particular aspects of the disclosure are directed to an aircraft system that includes a jet engine exhaust nozzle having an internal flow surface and an exit aperture. The exit aperture has a perimeter that includes multiple projections extending in an aft direction. The projections can be circumferentially spaced about the perimeter, and a geometric feature of the multiple projections can change in a monotonic manner along at least a portion of the perimeter. For example, successive projections can have a length that decreases in a direction away from a wing of the aircraft along the perimeter. In other aspects, the geometric feature can include an angular deflection of the projection, a shape of the projection, and/or a density of the projections around the perimeter. The manner in which the geometric feature is varied can reduce engine noise.

In further particular embodiments, the engine can include a turbofan engine, and the exhaust nozzle can include a first internal flow surface positioned to receive a fan flow and a second internal flow surface positioned to receive an engine core flow. Each flow surface can terminate at an exit aperture, and each exit aperture can include multiple projections. A geometric feature of the projections at the fan flow internal surface can vary in a manner that is different from the manner in which the geometric projections of the core flow surface vary.

In still further particular embodiments, the manner in which the geometric feature of the projections varies can depend upon the particular installation of the nozzle. For example, when the nozzle is positioned near an aircraft wing, the projections can be longer at the portion of the nozzle close to the wing, and shorter at the portion of the nozzle distant from the wing. When the nozzle is positioned proximate to an aircraft fuselage, the projections can be longer toward the fuselage and shorter at a portion of the nozzle positioned away from the fuselage. The variation of the projection geometric feature can be selected to reduce the acoustic signature on the ground and/or in the aircraft cabin.

Other aspects of the disclosure are directed to methods for manufacturing an aircraft. One method includes selecting a fuselage configuration and a wing configuration. The method can further include selecting a turbofan nozzle configuration to include a fan flow duct having a first internal surface positioned to receive a fan flow, and a core flow duct having a second internal flow surface positioned to receive an engine core flow. The method can still further include selecting an exit aperture of at least one of the ducts to have a perimeter that includes multiple projections extending in an aft direction, with a portion of individual neighboring projections spaced apart from each other by a gap. A geometric feature of at least some of the projections is selected in a manner that depends at least in part on a location of the engine nozzle relative to the fuselage, the wing, or both the fuselage and the wing.

In one embodiment, there is provided a method for controlling aircraft noise due to gas flow through a jet engine of an aircraft. The method involves directing the gas flow from the jet engine through a jet engine nozzle, and causing an acoustic intensity vector of gas flow from the nozzle, at one or more one acoustic frequencies to be non-parallel to a total thrust vector of the gas flow from the nozzle by causing the gas flow from the nozzle to pass adjacent to multiple nozzle projections, and mixing the gas with adjacent freestream air at the nozzle projections. The multiple nozzle projections extend aft from a perimeter of an exit of the jet engine nozzle, and the multiple nozzle projections have different lengths, and are non-moveable relative to the gas flow, or moveable relative to the perimeter of the exit of the nozzle. A length of the multiple nozzle projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreases successively over at least three projections from one projection to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane.

The nozzle may be pre-configured such that the total thrust vector is parallel to a longitudinal axis of the nozzle.

Causing the gas flow from the nozzle to pass adjacent to the multiple nozzle projections may involve arranging the multiple nozzle projections so that projections on opposite sides of the longitudinal axis of the nozzle are not the same length to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle.

Arranging the multiple nozzle projections to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle may involve arranging the multiple nozzle projections such that shorter nozzle projections are furthest away from a fuselage of the aircraft to cause the acoustic intensity vector to be directed away from a fuselage of the aircraft.

Arranging the multiple nozzle projections to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle may involve arranging the multiple nozzle projections such that shorter nozzle projections are located upwardly relative to the longitudinal axis of the nozzle to cause the acoustic intensity vector to be directed upwardly relative to the longitudinal axis of the nozzle.

In another embodiment, there is provided a nozzle apparatus for controlling aircraft noise due to gas flow through a jet engine on an aircraft. The nozzle apparatus includes a core flow duct terminating in a core exit aperture having a core aperture perimeter comprising a plurality of core flow projections. The plurality of core flow projections extend aft from the core aperture perimeter, and have different lengths. A length of the multiple nozzle projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreases successively over at least three projections from one projection to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane. The core flow projections are non-moveable relative to the gas flow from the nozzle, or moveable relative to the core aperture perimeter. The gas flow from the nozzle passes adjacent the plurality of core flow projections, and mixes with adjacent freestream air at the plurality of core flow projections thereby causing acoustic noise having an acoustic intensity vector at one or more one acoustic frequencies, in a direction non-parallel to a longitudinal axis of the nozzle apparatus.

The nozzle may be pre-configured such that the total thrust vector is parallel to a longitudinal axis of the nozzle.

At least one of the plurality of fan flow projections and the plurality of core flow projections may be arranged so that projections on opposite sides of the longitudinal axis of the nozzle are not the same length to thereby cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle.

The plurality of core flow projections may be arranged such that shorter nozzle projections are furthest away from a fuselage of the aircraft to cause the acoustic intensity vector of the gas flow from the nozzle to be directed away from a fuselage of the aircraft.

The plurality of core flow projections may be arranged such that shorter nozzle projections are located upwardly relative to the longitudinal axis of the nozzle to cause the acoustic intensity vector of the gas flow from the nozzle to be directed upwardly relative to the longitudinal axis of the nozzle apparatus.

In another embodiment, there is provided an aircraft having a fuselage and at least one jet engine having the nozzle apparatus as described above.

In another embodiment, there is provided an aircraft system. The aircraft system includes a turbofan engine exhaust nozzle that includes a fan flow duct having a first internal flow surface positioned to receive a fan flow, and a core flow duct having a second internal flow surface positioned to receive an engine core flow. At least one of the fan flow duct and the core flow duct has a varying flow area with a convergent section, a divergent section downstream of the convergent section, a throat between the convergent and divergent sections, and an exit aperture having a perimeter that includes multiple projections that are (a) non-moveable relative to the core flow duct or (b) moveable relative to the perimeter, the projections extending in an aft direction, with circumferentially adjacent projections spaced apart from each other by a gap, and wherein the gaps are positioned downstream of the throat, and wherein a length of the projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreasing successively over at least three projections from one projection to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane.

The fan flow duct may have the convergent section, the divergent section and the throat.

The fan flow duct may have the varying flow area with the convergent section, the divergent section downstream of the convergent section, and the throat between the convergent and divergent sections. The core flow duct may have the exit aperture with a perimeter that includes multiple projections extending in the aft direction, with an aft portion of individual neighboring projections spaced apart from each other by a gap.

The projections of the fan flow duct may vary in a first manner around the perimeter of the exit aperture of the fan flow duct, and the projections of the core flow duct may vary in a second manner around the perimeter of the exit aperture of the core flow duct, the second manner being different than the first manner.

The projections of the fan flow duct may be first projections at a first perimeter, and the projections of the core flow duct may be second projections at a second perimeter. The first projections may decrease in length around the first perimeter from a **12:00** position at the first perimeter to a **6:00** position at the first perimeter; and the second projections increase in length around the second perimeter from a **12:00** position at the second perimeter to a **6:00** position at the second perimeter.

The projections may have a length that varies in a monotonic manner around a portion of the perimeter.

BRIEF DESCRIPTION OF THE DRAWINGS

Figure **1** schematically illustrates a nozzle configured in accordance with the prior art.

Figure **2** illustrates an aircraft having a nozzle configured in accordance with an embodiment.

Figure **3** is a partially schematic, side elevation view of a turbofan engine nozzle having projections arranged in accordance with an embodiment.

Figure **4** is a partially schematic, rear elevation view of an embodiment of the nozzle shown in Figure **3**.

Figure **5** is a partially schematic, side elevation view of an embodiment of the nozzle shown in Figures **3** and **4**, installed beneath an aircraft wing in accordance with another embodiment.

Figures **6**A-**6**C illustrate acoustic characteristics of an existing nozzle and a nozzle configured in accordance with an embodiment.

Figure **7** is a partially schematic, side elevation view of a nozzle having projections at its exit that vary in accordance with another embodiment.

Figure **8** is a partially schematic, side elevation view of an embodiment of the nozzle shown in Figure **7** mounted to a wing in accordance with another embodiment.

Figure **9**A is a partially schematic, rear elevation view of two nozzles mounted proximate to an aircraft fuselage, each having exit projections that vary in accordance with another embodiment.

Figures **9**B-**9**D are schematic illustrations of acoustic intensity vectors corresponding to nozzles configured in accordance with still further embodiments.

Figure **10** is a schematic illustration representative of nozzle projection variations in accordance with several embodiments.

Figures **11**A-**11**D illustrate geometric characteristics of nozzle projections that may be varied in accordance with further embodiments.

Figure **12** is a graph illustrating an expected effect of nozzle projection variation on sound attenuation at a variety of frequencies.

Figures **13**A-C illustrate projections arranged in accordance with still further embodiments.

Figure **14** is a schematic illustration of a nozzle gas path flow area in accordance with an embodiment.

DETAILED DESCRIPTION

Aspects of the present disclosure are directed to nozzle exit configurations and associated systems and methods. Specific details of certain embodiments are described below with reference to Figures **2-14**. Several details of structures or processes that are well-known and often associated with such methods and systems are not set forth in the following description for purposes of brevity. Moreover, although the following disclosure sets forth several embodiments of different aspects of the invention, several other embodiments of the invention can have different configurations or different components than those described in this section. Accordingly, there may be other embodiments with additional elements and/or without several of the elements described below with reference to Figures **2-14**.

Figure **2** is an illustration of a commercial jet transport aircraft **200** having wings **202**, a fuselage **201**, and a propulsion system **203**. The illustrated propulsion system **203** includes two turbofan engines **206** carried by the wings **202**. Each engine **206** is housed in a nacelle **204**, which includes an inlet **205** and a nozzle **220**. The nozzles **220** include particular features, discussed in greater detail below, that reduce and/or direct the noise generated by the engines **206** in a selected manner. As is also discussed below, the manner in which the noise is reduced and/or directed can depend upon a particular installation of the propulsion system **203**. Accordingly, in other embodiments, the aircraft **200** can include a different number of engines and/or engines carried by different portions of the aircraft, along with nozzles **220** that are tailored to the particular installation.

Figure **3** is an enlarged side elevation view of an embodiment of the nozzle **220** as shown in Figure **2**. The nozzle **220** can include a fan flow duct **230** having a fan internal flow surface **232** that directs fan flow away from the upstream engine along a fan flow path **231**. The nozzle **220** also includes a core flow duct **240** having a core internal flow surface **242** that directs the core flow away from the engine along a core flow path **241**. The fan flow duct **230** terminates at a fan exit aperture **233** that is defined at least in part by a fan aperture perimeter **234** having multiple first or fan flow projections **235** that extend in an aft direction. Each of the fan flow projections **235** can have a generally triangular or chevron shape in a particular embodiment shown in Figure **3**, and can accordingly include aft or tip portions **219** that are spaced apart from each other by a gap **218**. The fan flow projections **235** can have other shapes (e.g., trapezoidal or irregular) in other embodiments. As is also shown in Figure **3**, at least one geometric feature of the fan flow projections **235** changes in a generally monotonic manner along at least a portion of the fan aperture perimeter **234**. For example, as shown in Figure **3**, the length of successive fan flow projections **235** changes in a circumferential direction around the fan aperture perimeter **234**. As will be discussed in greater detail below, other features of the fan flow projections **235** may be changed in addition to, or in lieu of, the length of the projections.

As is also shown in Figure **3**, the core flow path **241** terminates at a core exit aperture **243** having a perimeter **244** with second or core flow projections **245**. The core exit aperture **243** can be downstream of the fan exit aperture **233**, as shown in Figure **3**, or it can have other locations relative to the fan exit aperture **233** (e.g., upstream) in other embodiments. In a particular embodiment shown in Figure **3**, the core flow projections **245** have geometric shapes and features that remain generally uniform around the perimeter **244** of the core exit aperture **243**. In other embodiments discussed later with reference to additional Figures, the core flow projections **245** can have geometric features that vary around the perimeter **244**. The manners in which the core flow projections **245** and/or the fan flow projections **235** vary can depend upon factors which can include the manner in which the nozzle **220** is mounted to an aircraft, the frequency range over which noise reduction is desired, and/or the region of the local environment in which the noise is to be reduced (e.g., the ground beneath the aircraft and/or the aircraft interior). The nozzle **220** can have either fan flow projections **235**, core flow projections **245**, or both. In at least some embodiments, the projections may extend around only a portion of the corresponding perimeter (e.g., with no projections on the remainder of the perimeter), and/or may have irregular spacings.

Figure **4** is a forward-looking schematic view of the nozzle **220**, schematically illustrating the fan flow projections **235** and the core flow projections **245**. As shown in Figure **4**, the length of the fan flow projections **235** changes in a monotonic fashion from the **12**:**00** position to the **6**:**00** position in both clockwise and counterclockwise directions. Accordingly, the monotonic change of this geometric feature extends over **180**° of the fan aperture perimeter **234** (e.g., opposite lateral halves of the nozzle **220** are generally symmetric). In other embodiments, the change can take place over a greater or lesser circumferential range. For example, the monotonic change may in some embodiments extend over a portion of the fan exit aperture **234** occupied by three fan flow projections **235**. In still further embodiments, the monotonic variation can apply to groups or sets of fan flow projections **235**. For example, pairs of fan flow projections **235** (or core flow projections **245**) may have characteristics that vary in a monotonic manner. Further details of one such arrangement are described below with reference to Figure **11**D. In any of these embodiments, the change in the geometric feature can result in an asymmetric nozzle **220**.

Figure **5** is a partially schematic, side elevation view of the nozzle **220** and the nacelle **204** installed on the wing **202**. In this arrangement, the nacelle **204** is carried below the wing **202** and is supported by a pylon **207** relative to the wing **202**. Accordingly, the fan flow projections **235** are longer toward the wing **202** than they are away from the wing **202**, which can advantageously reduce nozzle noise without compromising thrust levels. In particular, the wing **202** can include movable trailing edge devices **208**, such as flaps. The exhaust jet flow exiting the nozzle **220** can interact with the wing **202**, and particularly with any trailing edge devices **208**. This jet-flap interaction can increase the noise above that which is generated by the nozzle **220** alone. Such interactions can also occur between the downstream wake of the pylon **207** and the exhaust flow. Accordingly, it may be advantageous to encourage additional mixing between the nozzle flow and the adjacent freestream flow near the pylon **207** and near the lower surface of the wing **202**, including near the trailing edge device **208** to reduce this jet-flap interaction.

The projections can enhance mixing between the jet flow and the ambient flow by introducing axial or streamwise vorticity generated by the pressure difference between the outwardly and inwardly facing surfaces of the fan flow projections **235**. It is expected that by encouraging additional mixing in these regions, the flow velocity gradients, and/or the flow velocity magnitudes in these regions will be reduced, compared to levels that would be present without the enhanced mixing provided by the fan flow projections **235**. The enhanced mixing that can lead to decreased turbulence intensity far away from the nozzle can also increase it near the nozzle. Accordingly, the elongated fan flow projections **235** can be concentrated in the region expected to provide an enhanced acoustic performance (e.g., toward the top of the nozzle **220**). At the same time, the fan flow projections **235** positioned toward the bottom of the nozzle **220** can be smaller than those positioned toward the top. An expected benefit of this arrangement is that the smaller projections **235** near the bottom of the nozzle **220** impinge less into the flow exiting the nozzle **220** and accordingly have a reduced impact on the mass flow exiting the nozzle **220** and the turbulence intensity downstream near the bottom sector. As a result, the potential reduction in thrust created by the presence of the fan flow projections **235** and the potential increase in the turbulence intensity overall can be mitigated by having smaller fan flow projections **235** in those regions that may not be as important for sound reduction as are other regions.

Figure **6**A schematically illustrates the effect described above. In this Figure, a thrust vector T and an acoustic intensity vector A are superimposed on a schematic illustration of the nozzle **220**. The thrust vector T represents the direction and magnitude of the thrust produced by the nozzle **220**, and the acoustic intensity vector A represents the direction and magnitude of the vector sum of far field acoustic intensities in the upper and lower hemispheres projected in the plane of the nozzle axis and the observer at a particular frequency or range of frequencies. For a nozzle having no projections, or uniform projections (such as are shown in Figure **1**), the thrust vector T and the acoustic intensity vector A are generally parallel and generally axial. By tailoring the fan flow projections **235** in the manner shown in Figures **3**-**5**, the acoustic intensity vector component directed toward the observer (assumed to be below the nozzle in Figure **6**A) can be reduced. This can be achieved by directing the acoustic intensity vector A effectively upward, thus reducing the downwardly directed component, or simply by reducing the magnitude of the acoustic intensity vector A without changing its direction. At the same time, the thrust vector T can remain axial. In fact, in a particular embodiment using this arrangement, the direction of the thrust vector T with the azimuthally varying fan flow projections **235** is identical or nearly identical to that associated with a nozzle having no projections.

Figures **6**B and **6**C compare measured acoustic test data proximate to an uninstalled baseline nozzle **20** generally similar to that shown in Figure **1**, with an uninstalled nozzle **220** generally similar to that shown in Figure **3**. At the particular frequency shown in these Figures (**1223** Hz), the peak acoustic emission level at the source is reduced by approximately **1**.**4** dB, as is indicated graphically by the contour plots of constant sound level shown in these Figures. At the same time, the overall thrust vector direction is expected to be unchanged (e.g., axial), for the configuration shown in Figure **6**C, as compared with the baseline configuration shown in **6**B. The thrust level for the configuration shown in Figure **6**C is expected to be at least very close to, if not equal to, the thrust level for the configuration shown in Figure **6**B. It is expected that the low impact of the circumferentially varying fan flow projections **235** on the thrust level may be due to the smaller projections **235** at the bottom perimeter of the nozzle **220** leading to a higher effective area of the nozzle. These projections tend not to extend into the nozzle exit flow by a great amount (e.g., they are not significantly immersed in the nozzle flow), and so have a reduced impact on nozzle mass flow rate, discharge coefficient and thrust. The foregoing results for noise reduction at the source are expected to also be significant for community noise reduction.

A comparison of acoustic data far away from the nozzle **220** (in the "far field") at low frequencies showed that the isolated nozzle **220** reduced noise compared to an isolated conventional round nozzle (with no projections) over a large sector of aft angles by about **3** to **4** dB at take-off, and by about **1**.**5** dB when compared to an isolated baseline nozzle **20** generally similar to that shown in Figure **1**. Under installed conditions, the range of observer angles and the frequencies over which the noise benefit attributed to the nozzle **220** is observed is reduced somewhat, impacting the overall noise benefit; however, embodiments with the installed nozzle **220** are still quieter than embodiments with the baseline nozzle **20** (Figure **1**).

One feature of the foregoing embodiments described above with reference to Figures **3**-**6**C is that azimuthally or circumferentially varying one or more geometric features of the fan flow projections **235** can reduce overall acoustic emissions from the engine, without an adverse or significantly adverse effect on engine thrust. In particular, relatively low frequency noise may be reduced and/or deflected away from observers on the ground. This noise is generally associated with jet-mixing interactions, for example, the type of mixing that occurs between the exhaust jet and the freestream flow, particularly adjacent to the pylon and the wing. The effect of reducing jet-wing and/or jet-pylon interaction noise can be particularly important on takeoff and approach, where community noise issues are a significant design factor. In particular, during takeoff, jet velocities are very high (although the trailing edge devices are typically not deployed by a great amount), while on landing, the trailing edge devices are deployed by a greater amount, while the jet exit velocities are not as high. In either embodiment, jet interaction noise can be a significant contributor to the overall acoustic signature of the aircraft, and can be reduced by a beneficial amount without a significant thrust penalty, as a result of projections having geometric features that vary circumferentially around the nozzle exit.

Another contributor to the overall acoustic signature of the aircraft is shockcell noise, which is typically associated with supersonic fan flow. Accordingly, shockcell noise may also be reduced by projections which diminish circumferential coherence and thereby weaken the shockcells addressed by the arrangement of the fan flow projections. In some cases, the core flow may also contribute to shockcell noise, in which case the second or core flow projections may be tailored, in addition to (or in lieu of) tailoring the fan flow projections.

Comparison of shockcell noise data between an embodiment of the nozzle **220** and a conventional round coaxial nozzle without projections (during a flight test at cruise conditions) showed a noise reduction of up to **5** dB on the exterior of the fuselage on the side where the engine was located. At the same time, the overall thrust vector direction between these two nozzles was unchanged, and the thrust level of the nozzle **220** actually increased slightly (0.65% at cruise) when compared to the conventional nozzle with no projections.

Figure **7** illustrates a nozzle **720** having first or fan flow projections **735** and second or core flow projections **745**. The fan flow projections **735** and the core flow projections **745** vary in monotonic, opposite manners. That is, the fan flow projections **735** tend to be longer toward the bottom of the nozzle **720** than toward the top of the nozzle **720**, while the core flow projections **745** vary in the opposite manner. The variation of the fan flow projections **735** is the opposite of the arrangement of fan flow projections **235** shown in Figure **3**. Accordingly, this arrangement may be suitable when the nozzle **720** is carried by a pylon extending downwardly (rather than upwardly) from the engine. Such an arrangement is shown in Figure **8**. In particular, Figure **8** illustrates the wing **202** with an upper surface mounted pylon **807** carrying a nacelle **804** housing the nozzle **720**. In this arrangement, the trailing edge devices **208** deploy downwardly (in a typical fashion) and, therefore, may not contribute significantly to the jet-flap interaction noise described above. However, the downstream wake of the pylon **807** may interact with the exhaust products and accordingly, it may be advantageous to have the fan flow projections **735** be longer in a region adjacent to the pylon **807**, than in a region distant from the pylon **807**.

Figure **9**A illustrates an aircraft **900** having two engine nacelles **904**a, **904**b that depend from or are at least proximate to the fuselage **901**. In this particular embodiment, each of the engine nacelles **904**a, **904**b is carried by the fuselage **901** via a corresponding pylon **907**. The nacelles **904**a, **904**b can include fan flow projections **935**a, **935**b that are configured to reduce the noise transmitted to the interior of the fuselage **901** (e.g., the passenger compartment). In particular, the fan flow projections **935**a, **935**b can be longer at a position close to the fuselage **901** than they are in a position distant from the fuselage **901**. As a result, the fan flow projections **935**a on the left nacelle **904**a tend to be longest near the **3**:**00** position, and shortest near the **9**:**00** position, while the fan flow projections **935**b on the second nacelle **904**b have the opposite arrangement. It is expected that the enhanced mixing provided by the longer fan flow projections **935**a, **935**b near the fuselage **901** (which may have relatively greater immersion into the flow) can reduce the acoustic signature close to the fuselage **901**, and can accordingly reduce the sound level experienced by passengers within the passenger compartment. The fan flow projections **935**a, **935**b that are more distant from the fuselage **901** can be shorter so as to reduce the overall effect of the fan flow projections **935**a, **935**b on engine thrust. Figure **9**B illustrates an acoustic intensity vector A corresponding to the sound level expected to be produced by the left nacelle **904**a at a given frequency. In particular, the net acoustic intensity vector A points outwardly away from the fuselage **901**, indicating that sound levels are expected to be lower near the fuselage **901** than distant from the fuselage **901**.

The manner in which the geometric features of the projections vary around the perimeter of the nozzle can be selected to have a wide variety of effects, and different feature changes can be superimposed so as to address different acoustic requirements simultaneously. While superimposing different feature changes may not necessarily result in an optimum level of noise reduction for each requirement, the combination may be one that results in an overall noise reduction that meets multiple design requirements. For example, the longer fan flow projections **235** positioned toward the top of the nozzle (described above with reference to Figure **3**) may be combined with the longer projections **935**a, **935**b positioned toward the inboard side of the nozzle (described above with reference to Figure **9**A). The result may be fan flow projections having an increased length toward the top of the nozzle to reduce jet-flap interaction noise, and also longer toward the fuselage to reduce cabin noise. The projections may be shorter toward the bottom of the nozzle and toward the side of the nozzle away from the fuselage, so as not to significantly impact the overall exhaust product mass flow and thrust level, in a region of the nozzle where reduced acoustic signature may not be as important as it is near the fuselage and near the wing.

Figure **9**C schematically illustrates a nacelle **904**c and nozzle **920** having projections configured to meet multiple acoustic objectives in the manner described above. In particular, longer projections **935**c toward the top of the nozzle **920** are positioned to reduce jet-mixing noise (e.g., due to an overhead wing and/or pylon), as represented by a first acoustic radiation vector A**1**. Longer projections **935**d toward the inboard side of the nozzle **920** are positioned to reduce shock-cell noise, as represented by a second acoustic vector A**2**.

Figure **9**D schematically illustrates a nozzle **920** configured in accordance with another embodiment of the invention to include two types of azimuthally varying projections: fan flow projections **935**d that are longer and/or more immersed toward the top of the nozzle (near the pylon), and core flow projection **945**d having monotonically decreasing lengths in a direction away from the fuselage **901**. It is expected that this arrangement can reduce both community noise at low frequencies and shockcell/cabin noise at higher frequencies.

In still further embodiments, the manner in which the projections vary around the nozzle perimeter (and therefore the degree of mixing between the adjacent flows) can be changed depending on flight regime of the aircraft, by changing the degree to which the projections are immersed as a function of time. This arrangement can be used to reduce different spectra of noise in different flight regimes. For example, to obtain more mixing between the fan flow and the freestream air near the pylon (e.g., to reduce low-frequency noise during take-off), the projections near the pylon can be actively bent inwardly during takeoff. If mid-frequency shockcell noise at cruise is reduced by another type of azimuthal variation, (e.g., by immersing projections near the fuselage by a greater amount than projections away from the fuselage), then this change can be made during the appropriate flight regime (e.g., during cruise). Such desired azimuthal variations in projection immersions can be obtained, for example, by using shape memory alloys inside the projections and suitable heat control elements. This arrangement can be applied to fan flow projections, and/or core flow projections. Further aspects of active systems for accomplishing this variation are included in U.S. Patent No. **6,718,752**.

As discussed above, certain aspects of the manners by which projection geometric features are varied can be combined in a wide variety of ways. Figure **10** illustrates schematically representative features that may be applied to the fan flow projections (along the horizontal axis), and/or the core flow projections (along the vertical axis). In these illustrations, R refers to regular or baseline projections that do not vary circumferentially, T refers to projections that are longer toward the top than the bottom, B refers to projections that are longer toward the bottom than the top, K refers to an arrangement in which projections are longer toward the top and the bottom, and V refers to an arrangement in which the immersion or degree to which the projections are bent inwardly toward the flow varies around the circumference of the nozzle, but the length does not. Depending upon the desired acoustic signature and the particular installation in which the nozzle is placed, these features may be combined in any of a variety of manners.

Figures **11**A-**11**D illustrate representative features of individual projections **1135** that may be varied in accordance with particular embodiments of the invention. For example, Figure **11**A illustrates multiple projections **1135** located at a perimeter **1121** of a corresponding nozzle **1120**. Geometric features of each projection **1135** that can be varied include the length **1122** of the projection **1135**, the width **1123** of the projection **1135**, and/or the apex angle **1124** of the projection **1135**. The overall shape of the projection **1135** may also be varied. For example, the projections **1135** can have a triangular or chevron shape as shown in Figure **11**A, with generally sharp vertices, or the projections **1135** may have other shapes and/or shapes with rounded or other less abrupt transitions between edges. The number of projections **1135** per unit length of the perimeter **1121** is another variable that may be selected to have the desired effect on the acoustic signature, again depending upon the particular installation. As shown in Figure **11**B, the angle **1125** between the projection **1135** and the flow surface located just upstream of the projection **1135**, or the curvature of the projection **1135** can also be varied so as to vary the immersion or degree to which the projection **1135** is deflected or bent inwardly into the nozzle flow. As shown in Figure **11**C, the density of projections **1135** (e.g., the number of projections **1135** per unit length along the nozzle exit perimeter) can also be varied. As noted above, in particular embodiments, there may be portions of the nozzle perimeter or circumference without projections, and/or the gap spacing between projections may vary in an irregular manner.

Many of the foregoing factors may be varied in combination with each other to produce a desired geometry. For example, if each projection **1135** has a fixed width **1123**, then reducing the length **1122** of the projection **1135** will change the apex angle **1124**. In at least some embodiments, the projections **1135** form part of an inwardly-sloping body of revolution around the axial centerline of the nozzle. Accordingly, longer projections **1135** will tend to be more immersed in the nozzle flow than shorter projections. In other embodiments the projections can be deflected outwardly away from the nozzle centerline, as opposed to inwardly toward the nozzle centerline. Similar considerations can be applied to determine the geometric features of such projections.

In a particular embodiment shown in Figure **11**D, at least some adjacent projections can be alternately immersed inwardly and outwardly (e.g., by the same amount or by different amounts). Accordingly, the nozzle **1120** can include pairs of inwardly deflected projections **1135**a and outwardly deflected projections **1135**b. The vortices from the adjacent edges of inwardly deflected projection **1135**a and neighboring outwardly deflection projection **1135**b tend to merge to form only one axial vortex from those adjacent edges. Thus, for all practical purposes, each pair of alternately immersed projections can act like one projection having a larger combined width and a stronger axial vorticity. The parameters described above for obtaining azimuthal variation of mixing with respect to individual neighboring projections can also apply to each pair taken as a unit. For example, in order to obtain a monotonic variation in mixing from the top of the nozzle **1120** to the bottom of the nozzle **1120** the projections **1135**a, **1135**b can have a monotonically decreasing level of immersion (inwardly for the inwardly deflected projections **1135**a and outwardly for the outwardly deflected projections **1136**b) from top to bottom. In other embodiments, other geometric characteristics of the projection pairs can be varied.

Figure **12** is a schematic illustration of four nozzles, labeled **1220**a-d, each of which has core flow projections with a different configuration, in accordance with several embodiments. For example, nozzle **1220**a has core flow projections that do not vary in a circumferential direction, nozzle **1220**b has core flow projections that are longer at the top than at the bottom, nozzle **1220**c has the opposite arrangement, and nozzle **1220**d has core flow projections that are longer at the top and bottom and shorter in an intermediate region. In this particular embodiment, the fan flow projections for each of these nozzles are uniform. The graph of Figure **12** illustrates the level of jet-flap interaction noise reduction associated with each of the nozzle configurations **1220**a-d, as a function of frequency (on a logarithmic scale) compared to a simple round coaxial nozzle with no projections. Nozzles **1220**a, b, d each reduce noise by a lesser amount a higher frequencies than at lower frequencies. By contrast, nozzle **1220**c has a greater noise reduction capability at higher frequencies than at lower frequencies. Figure **12** accordingly indicates that the manner in which the geometric feature varies around the perimeter of the nozzle may be selected based (at least in part) on the frequency of the noise that is to be reduced. If lower frequency noise is to be reduced, nozzles **1220**a, b or d may be appropriate, and if higher frequency noise is to be reduced, nozzle **1220**c may be more appropriate. Typically, community noise is a greater problem at lower frequencies than at higher frequencies, while cabin noise is typically a greater problem at higher frequencies than at lower frequencies. Accordingly, the appropriate arrangement of nozzle projections (or combination of nozzle projection arrangements) can be selected in a manner that depends on the particular noise reduction target. Similar noise reduction trends as a function of frequency were found for nozzles having varying fan flow projections and uniform core flow projections; however, in at least some of these cases, the reduction in the noise that is due to jet-flap interaction was higher than for the (baseline) nozzle **1220**a.

Figures **13**A-C and **14** illustrate still further geometric features that may be varied to achieve desired thrust and acoustic signature results in accordance with further embodiments. In particular, Figures **13**A-**13**C illustrate nozzles having different root locus lines **1326** (shown as root locus lines **1326**a-**1326**c) and tip locus lines **1327** (shown as tip locus lines **1327**a-**1327**c). The root locus lines **1326**a-**1326**b connect the root locations of successive fan flow projections **1335**, and the tip locus lines **1327**a-**1327**c connect the tip locations of the same projections **1335**. Figure **13**A illustrates a generally vertical root locus line **1326**a and an aft-canted tip locus line **1327**a. Figure **13**B illustrates a forwardly-canted root locus line **1326**b and a generally vertical tip locus line **1327**b. Figure **13**C illustrates a forwardly-canted root locus line **1326**C, an aft-canted tip locus line **1327**c, and a generally vertical centroid locus line **1328**c. The appropriate orientation of the root and tip locus lines may be selected to produce the desired acoustic vector, thrust vector, and/or other appropriate parameter. For example, canting the root locus line **1326** and/or the tip locus line **1327** may cant the thrust vector. If a particular azimuthal arrangement of projections **1335** shifts the thrust vector in an undesirable manner, canting the root locus line **1326** and/or the tip locus line **1327** can be used to correct the thrust vector back to the desired orientation. This methodology is illustrated in the context of fan flow projections, but may be applied to core flow projections in addition to or in lieu of the fan flow projections.

Figure **14** illustrates the "rolling ball" flow area through the fan flow duct of a nozzle configured in accordance with another embodiment. Figure **14** illustrates that the nozzle has a locally convergent-divergent arrangement, with a geometric throat T upstream of a corresponding root locus line **1426**. This arrangement is expected to have several beneficial effects. For example, a local convergent-divergent region of the nozzle is expected to have enhanced aerodynamic effects at particular flight regimes. By positioning the geometric throat T upstream of the root locus line **1426**, the effective exit area of the nozzle can be controlled such that it does not become susceptible to fan instability problems at low nozzle pressure ratios of the fan stream. The latter can occur when using inwardly immersed fan flow projections which can aerodynamically effectively behave like convergent nozzles. The shape of the projections that controls the local convergent-divergent behavior of the rolling ball area can be used to control the effective exit area and avoid fan instabilities. It is expected that this arrangement can reduce thrust degradation. It will be understood that in at least some cases, the nozzle can include an aerodynamic convergent section downstream of the local convergent-divergent region discussed above.

From the foregoing, it will be appreciated that specific embodiments have been described herein for purposes of illustration, but that various modifications may be made. For example, several of the embodiments described above were described in the context of nozzles having core flow paths that extend axially further aft than the corresponding fan flow paths (e.g., externally mixed nozzles). In other embodiments, the nozzles may be internally mixed and may have fan flow paths that extend further aft than the corresponding core flow paths. The nozzles may have a variety of exit perimeter shapes, including round, rectangular and elliptical.

Still further embodiments are described in the following documents: AIAA Paper **2006-2467**, entitled "Reducing Propulsion Airframe Aeroacoustic Interactions with Uniquely Tailored Chevrons: **1**. Isolated Nozzles," dated May **8-10**, **2006**; AIAA Paper **2006-2434**, entitled "Reducing Propulsion Airframe Aeroacoustic Interactions with Uniquely Tailored Chevrons: **2**. Installed Nozzles," dated May **8-10**, **2006**; AIAA Paper **2006-2435**, entitled "Reducing Propulsion Airframe Aeroacoustic Interactions with Uniquely Tailored Chevrons: **3**. Jet-Flap Interaction," dated May **8-10**, **2006**; AIAA Paper **2006-2439**, entitled "Flight Test Results for Uniquely Tailored Propulsion-Airframe Aeroacoustic Chevrons: Shockcell Noise," dated May **8-10**, **2006**; AIAA Paper **2006-2438**, entitled "Flight Test Results for Uniquely Tailored Propulsion-Airframe Aeroacoustic Chevrons: Community Noise," dated May **8-10**, **2006**; AIAA Paper **2006-2436**, entitled "Computational Analysis of a Chevron Nozzle Uniquely Tailored for Propulsion Airframe Aeroacoustics," dated May **8-10**, **2006**; AIAA Paper **2005-0996**, entitled "Relative Clocking of Enhanced Mixing Devices for Jet Noise Benefit," dated January **10-13**, **2005**; AIAA Paper **2005-2934**, entitled "Jet Noise Characteristics of Chevrons in Internally Mixed Nozzles," dated May **23-25**, **2005**; and AIAA Paper **2006-0623**, entitled "Internal Flow and Noise of Chevrons and Lobe Mixers in Mixed-Flow Nozzles," dated January **9-12**, **2006**.

Aspects described in the context of particular embodiments may be combined or eliminated in other embodiments. For example, many of the geometric features described individually above may be combined in any of a variety of manners to meet corresponding acoustic and thrust design goals, while integrating appropriately with other structures of the aircraft into which the nozzles are integrated. Further, while advantages associated with certain embodiments have been described in the context of those embodiments, other embodiments may also exhibit such advantages, and not all embodiments need necessarily exhibit such advantages.

THE EMBODIMENTS IN WHICH AN EXCLUSIVE PROPERTY OR PRIVILEGE IS CLAIMED ARE DEFINED AS FOLLOWS:

1. A method for controlling aircraft noise due to gas flow through a jet engine of an aircraft, the method comprising:

    directing the gas flow from a jet engine through a jet engine nozzle; and

    causing an acoustic intensity vector of gas flow from the nozzle, at one or more one acoustic frequencies to be non-parallel to a total thrust vector of the gas flow from the nozzle by:

        causing the gas flow from the nozzle to pass adjacent to multiple nozzle projections; and

        mixing the gas with adjacent freestream air at the nozzle projections;

    wherein the multiple nozzle projections extend aft from a perimeter of an exit of the jet engine nozzle, and wherein the multiple nozzle projections have different lengths, and are:

        non-moveable relative to the gas flow from the nozzle; or

        moveable relative to the perimeter of the exit of the nozzle; and

    wherein a length of the multiple nozzle projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreases successively over at least three projections from one projection directing the gas flow through a jet engine nozzle comprises installing the jet engine nozzle on a jet engine such that a longitudinal axis of the jet engine nozzle is parallel to the total thrust vector of the gas flow from the nozzle to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane.

2. The method of claim **1** wherein the nozzle is pre-configured such that the total thrust vector is parallel to a longitudinal axis of the nozzle.

3. The method of claim **2** wherein causing the gas flow from the nozzle to pass adjacent to the multiple nozzle projections comprises arranging said multiple nozzle projections so that projections on opposite sides of the longitudinal axis of the nozzle are not the same length to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle.

4. The method of claim **3** wherein arranging said multiple nozzle projections to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle comprises arranging said multiple nozzle projections such that shorter nozzle projections are furthest away from a fuselage of the aircraft to cause the acoustic intensity vector to be directed away from a fuselage of the aircraft.

5. The method of claim **3** wherein arranging said multiple nozzle projections to cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle comprises arranging said multiple nozzle projections such that shorter nozzle projections are located upwardly relative to the longitudinal axis of the nozzle to cause the acoustic intensity vector to be directed upwardly relative to the longitudinal axis of the nozzle.

6. A nozzle apparatus for controlling aircraft noise due to gas flow through a jet engine on an aircraft, the nozzle apparatus comprising:

a core flow duct terminating in a core exit aperture having a core aperture perimeter comprising a plurality of core flow projections, wherein the plurality of core flow projections extend aft from the core aperture perimeter and have different lengths and wherein a length of the multiple nozzle projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreases successively over at least three projections from one projection to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane and wherein the core flow projections are:

non-moveable relative to the gas flow; or

moveable relative to the core aperture perimeter; and

whereby the gas flow from the nozzle passes adjacent said plurality of core flow projections, and

mixes with adjacent freestream air at said plurality of core flow projections thereby causing acoustic noise having an acoustic intensity vector at one or more one acoustic frequencies in a direction non-parallel to a longitudinal axis of the nozzle apparatus.

7. The apparatus of claim **6** wherein the nozzle is pre-configured such that the total thrust vector is parallel to a longitudinal axis of the nozzle.

8. The apparatus of claim **7** wherein at least one of said plurality of fan flow projections and said plurality of core flow projections are arranged so that projections on opposite sides of the longitudinal axis of the nozzle are not the same length to thereby cause the acoustic intensity vector of the gas flow from the nozzle to be non-parallel to the longitudinal axis of the nozzle.

9. The apparatus of claim **8** wherein said plurality of core flow projections are arranged such that shorter nozzle projections are furthest away from a fuselage of the aircraft to cause the acoustic intensity vector of the gas flow from the nozzle to be directed away from a fuselage of the aircraft.

10. The apparatus of claim **8** wherein said plurality of core flow projections are arranged such that shorter nozzle projections are located upwardly relative to the longitudinal axis of the nozzle to cause the acoustic intensity vector of the gas flow from the nozzle to be directed upwardly relative to the longitudinal axis of the nozzle apparatus.

11. An aircraft having a fuselage and at least one jet engine having the nozzle apparatus of any one of claims **6** – **10**.

12. An aircraft system, comprising:

a turbofan engine exhaust nozzle that includes:

a fan flow duct having a first internal flow surface positioned to receive a fan flow;

a core flow duct having a second internal flow surface positioned to receive an engine core flow; and wherein at least one of the fan flow duct and the core flow duct has a varying flow area with a convergent section, a divergent section downstream of the convergent section, a throat between the convergent and divergent sections, and an exit aperture having a perimeter that includes multiple projections that are (a) non-moveable relative to the core flow duct or (b) moveable relative to the perimeter, the projections extending in an aft direction, with circumferentially adjacent projections spaced apart from each other by a gap, and wherein the gaps are positioned downstream of the throat; and

wherein a length of the projections in a first group decreases successively over at least three projections from one projection to the next along a first portion of the perimeter, and a length of projections in a second group decreasing successively over at least three projections from one projection to the next along a second portion of the perimeter, the second portion being mirrored relative to the first portion about an axially extending plane.

13. The system of claim **12** wherein the fan flow duct has the convergent section, the divergent section and the throat.

14. The system of claim **12** wherein the fan flow duct has the varying flow area with the convergent section, the divergent section downstream of the convergent section, the throat between the convergent and divergent sections, and wherein the core flow duct has the exit aperture with a perimeter that includes multiple projections extending in the aft direction, with an aft portion of individual neighboring projections spaced apart from each other by a gap.

15. The system of claim **14** wherein the projections of the fan flow duct vary in a first manner around the perimeter of the exit aperture of the fan flow duct, and wherein the projections of the core flow duct vary in a second manner around the perimeter of the exit aperture of the core flow duct, the second manner being different than the first manner.

16. The system of claim **14** wherein the projections of the fan flow duct are first projections at a first perimeter, and wherein the projections of the core flow duct are second projections at a second perimeter, and wherein:

the first projections decrease in length around the first perimeter from a **12**:**00** position at the first perimeter to a **6**:**00** position at the first perimeter; and

the second projections increase in length around the second perimeter from a **12**:**00** position at the second perimeter to a **6**:**00** position at the second perimeter.

17. The system of claim **12** wherein the projections have a length that varies in a monotonic manner around a portion of the perimeter.

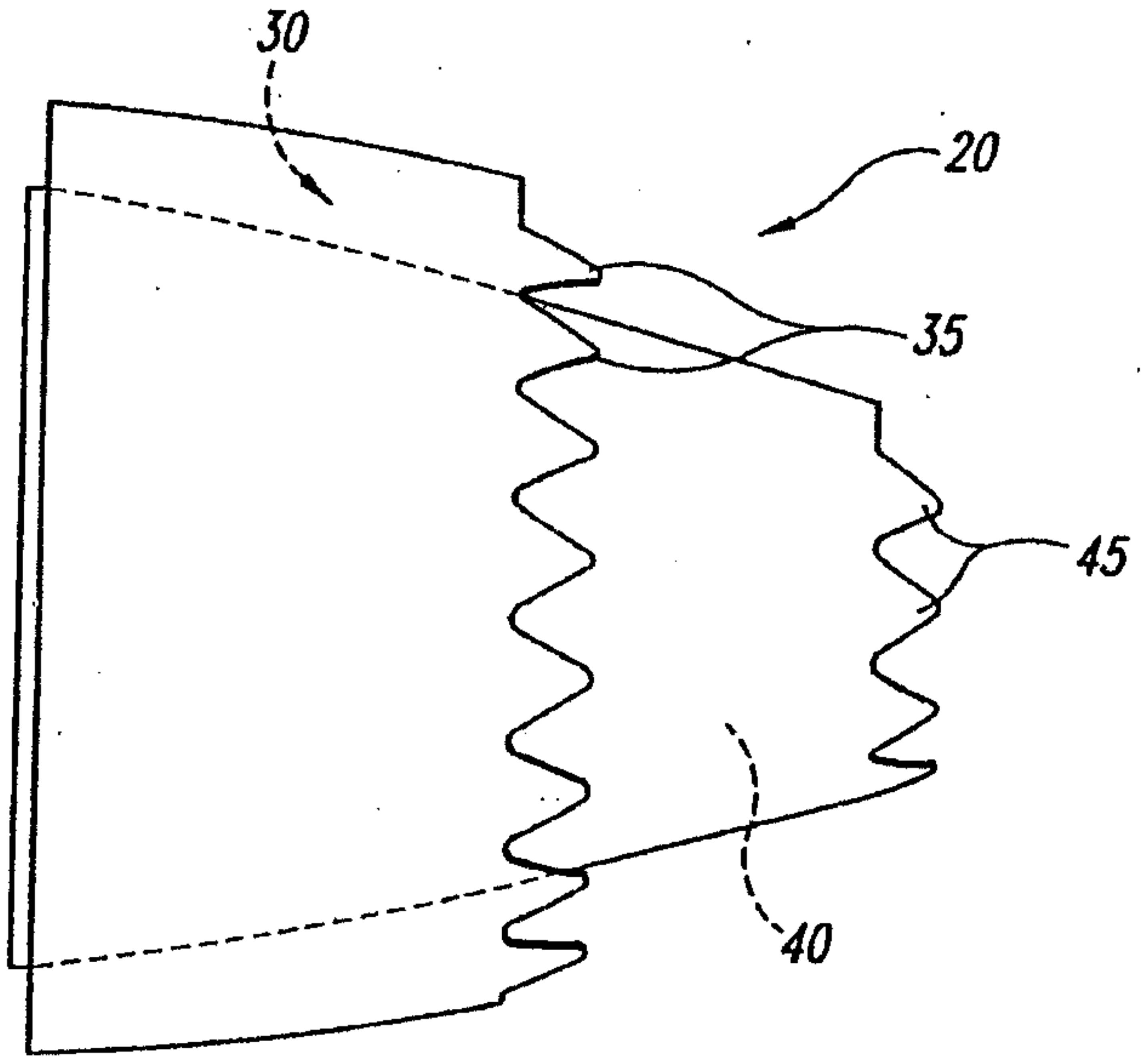

Fig. 1
(Prior Art)

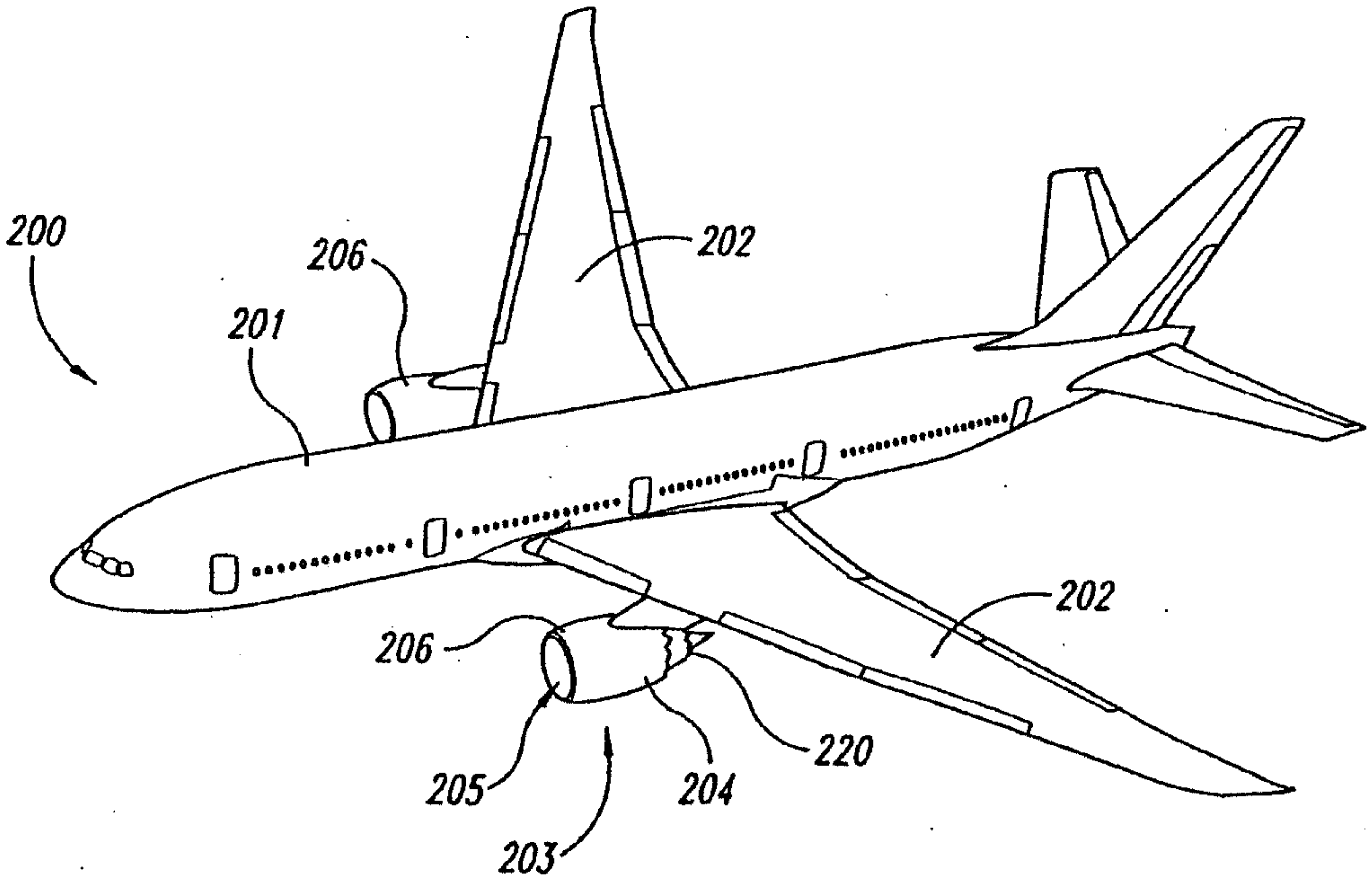

Fig. 2

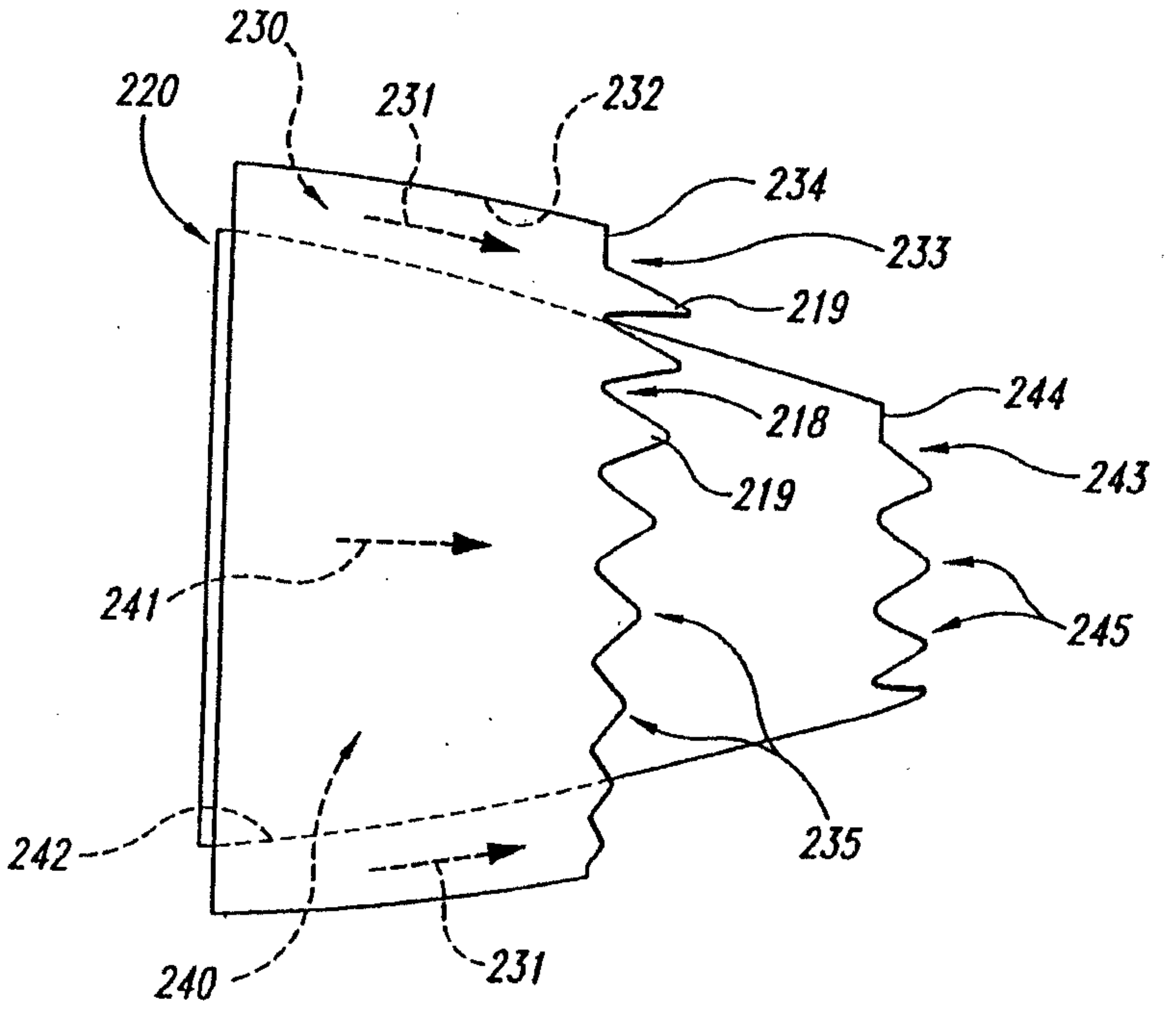
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Fig. 3

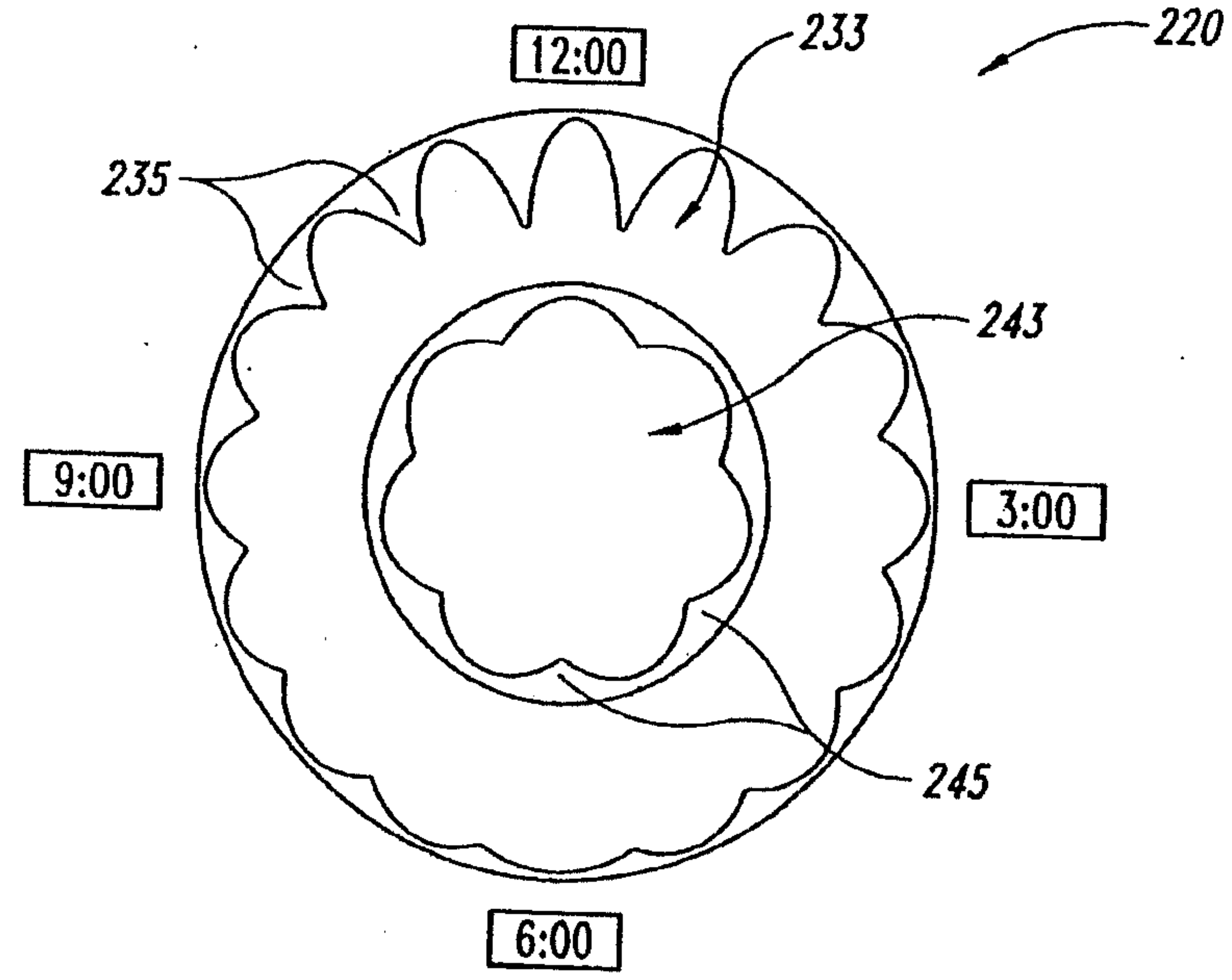
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Fig. 4

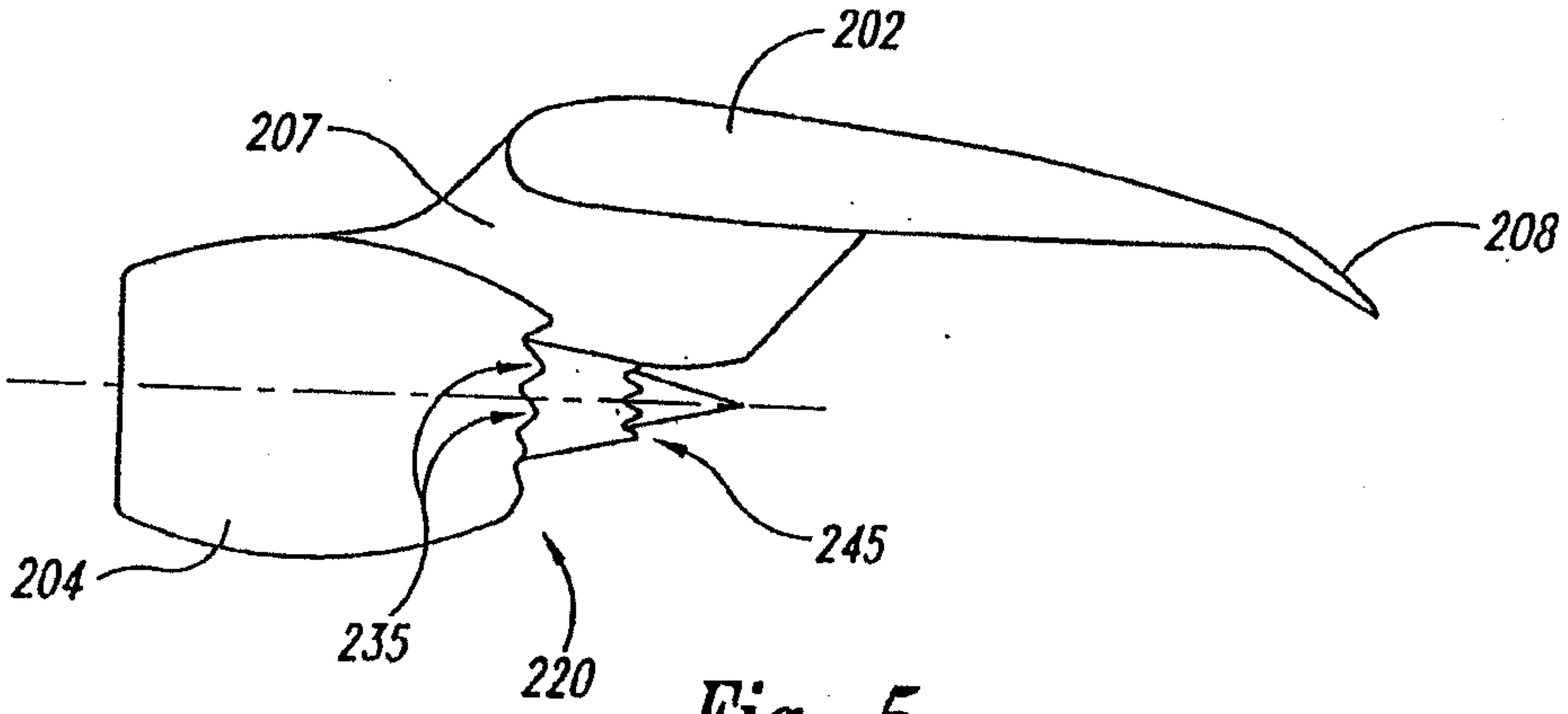


Fig. 5

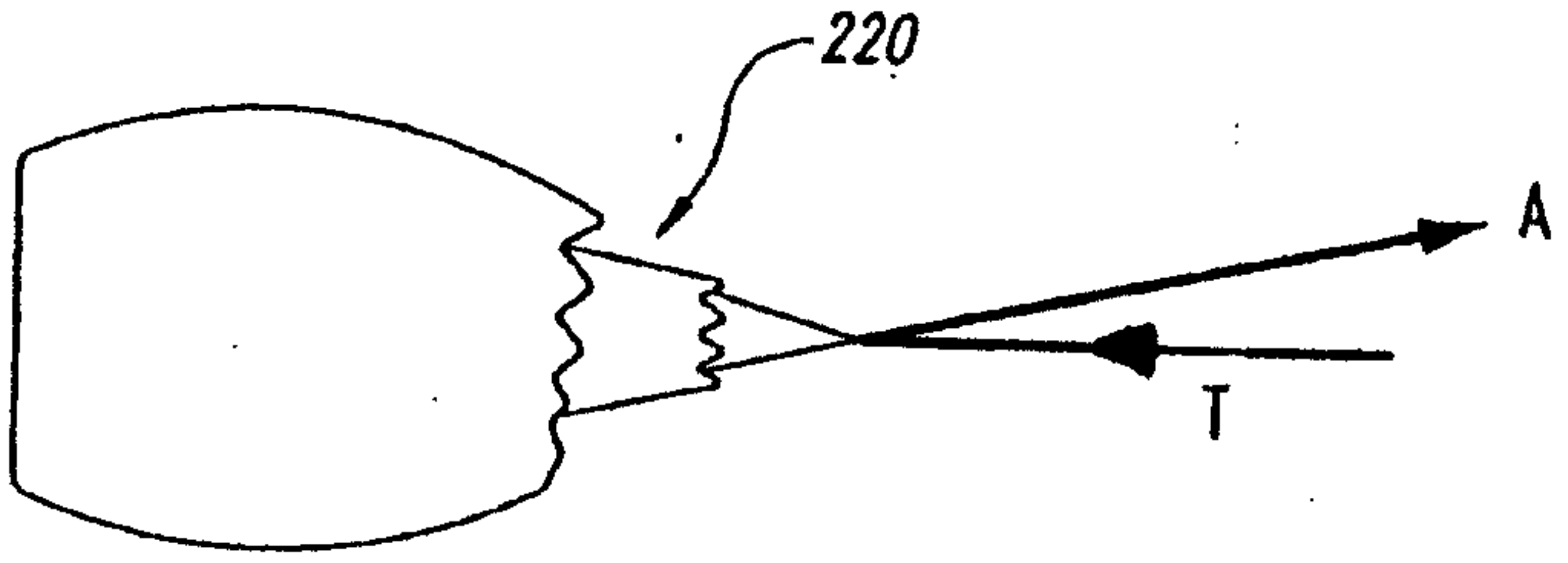


Fig. 6A

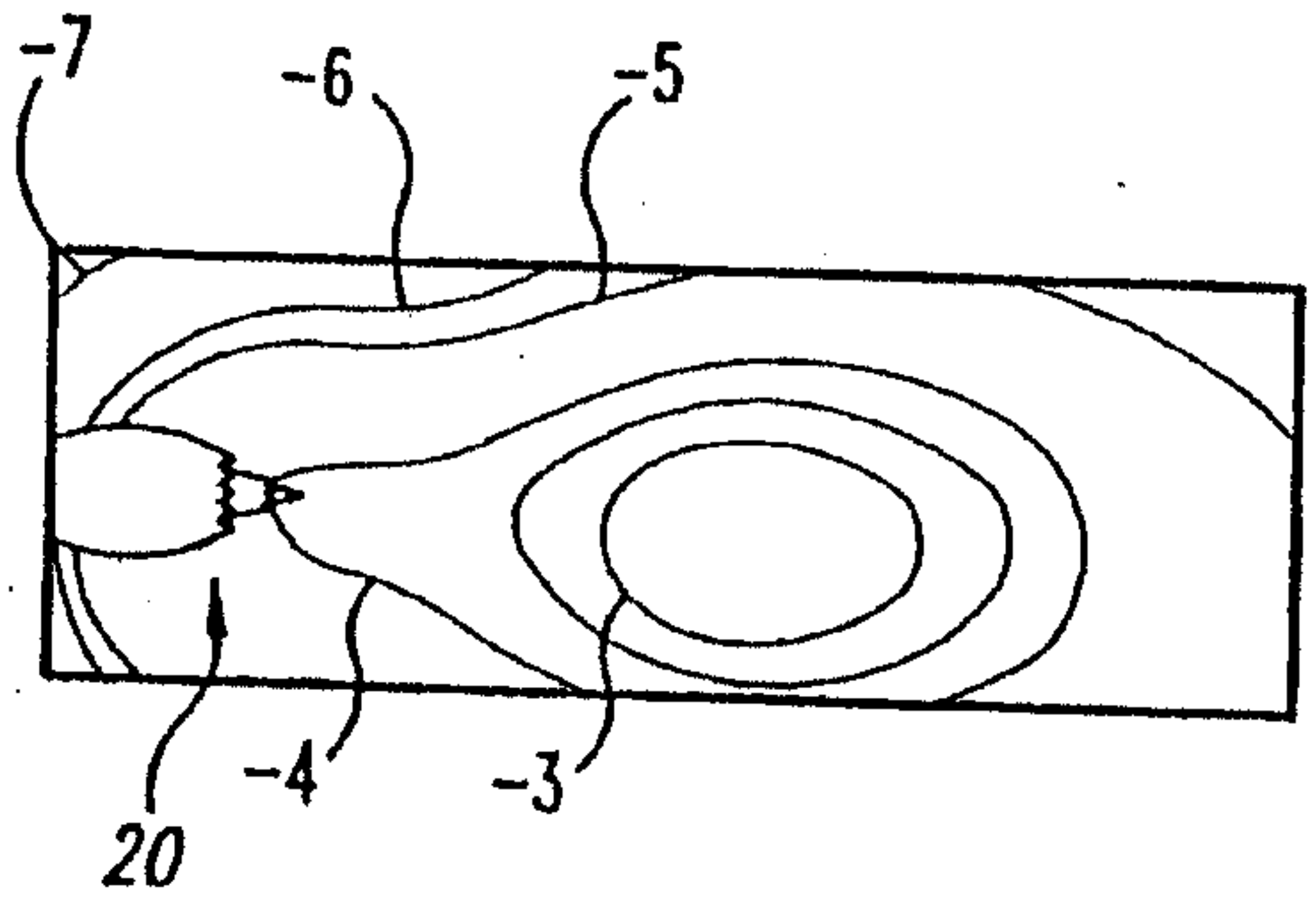


Fig. 6B

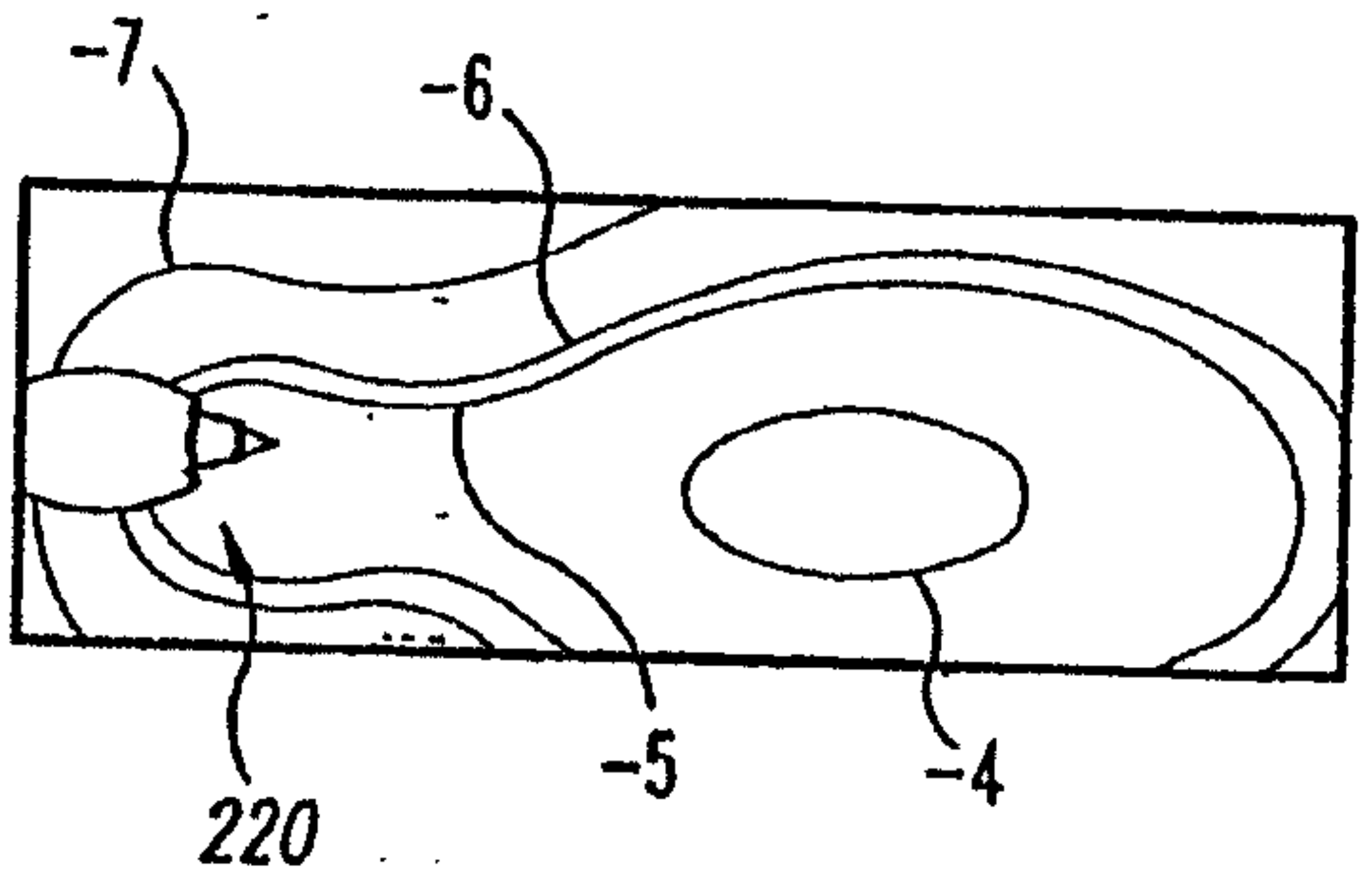


Fig. 6C

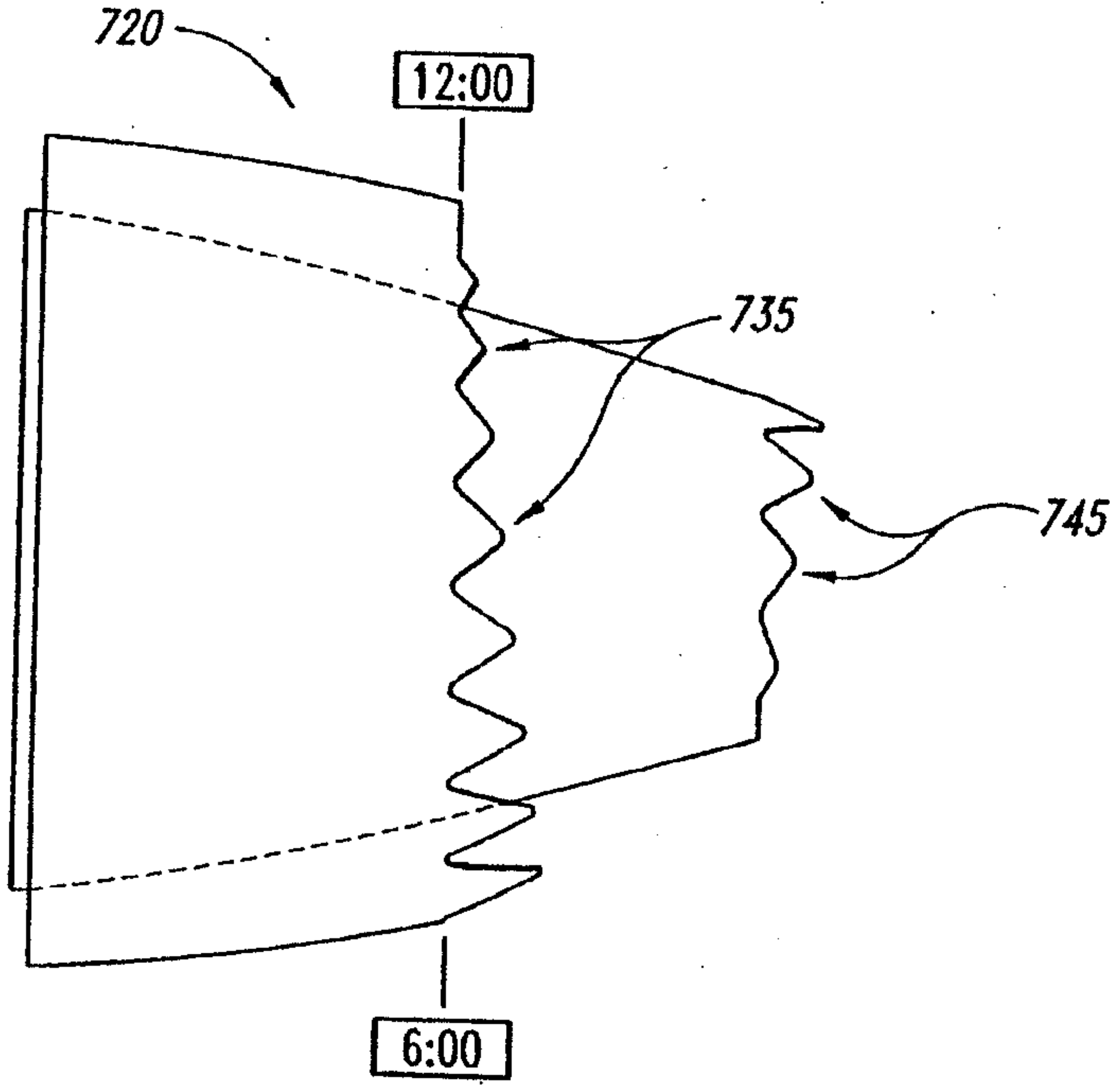


Fig. 7

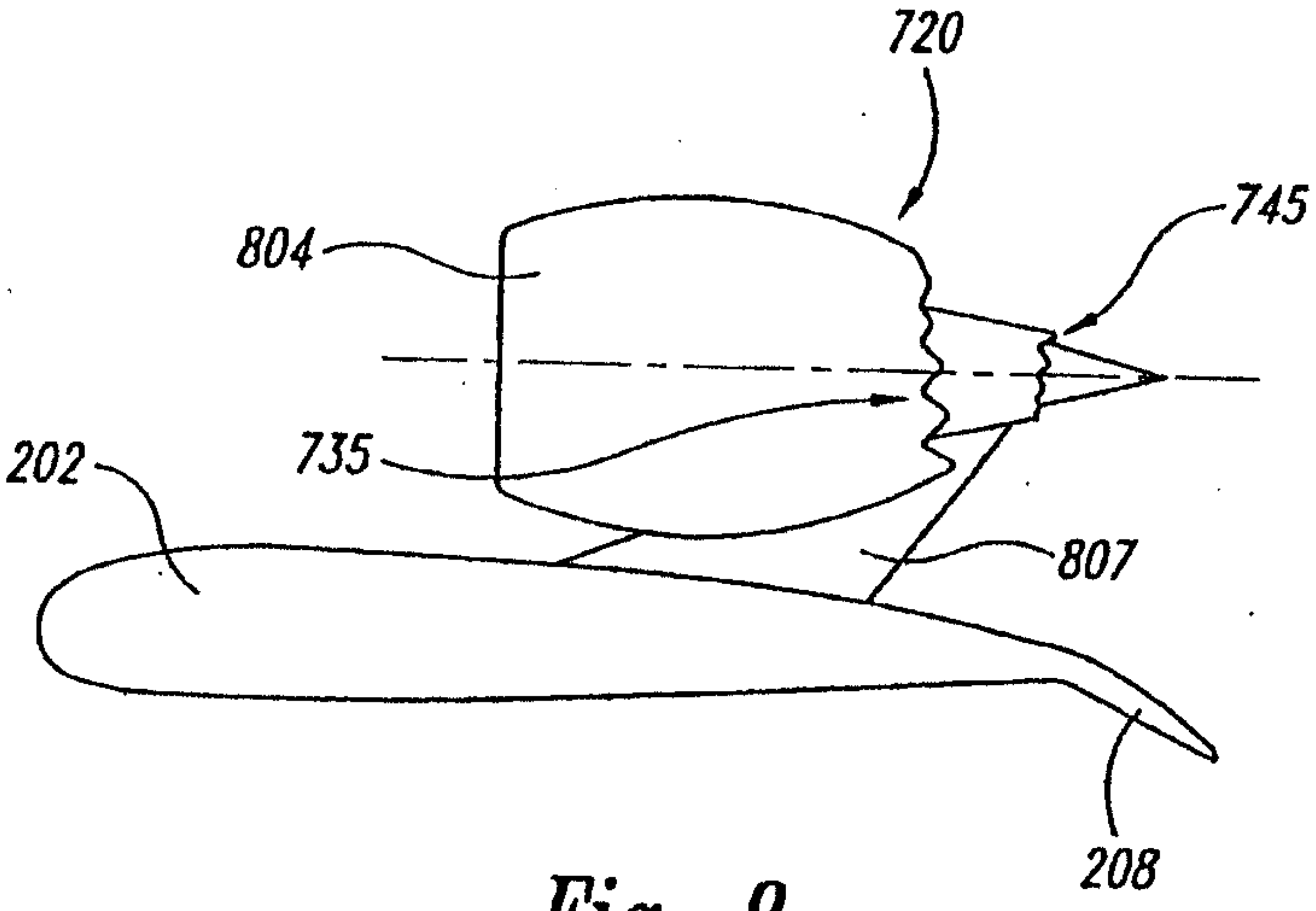


Fig. 8

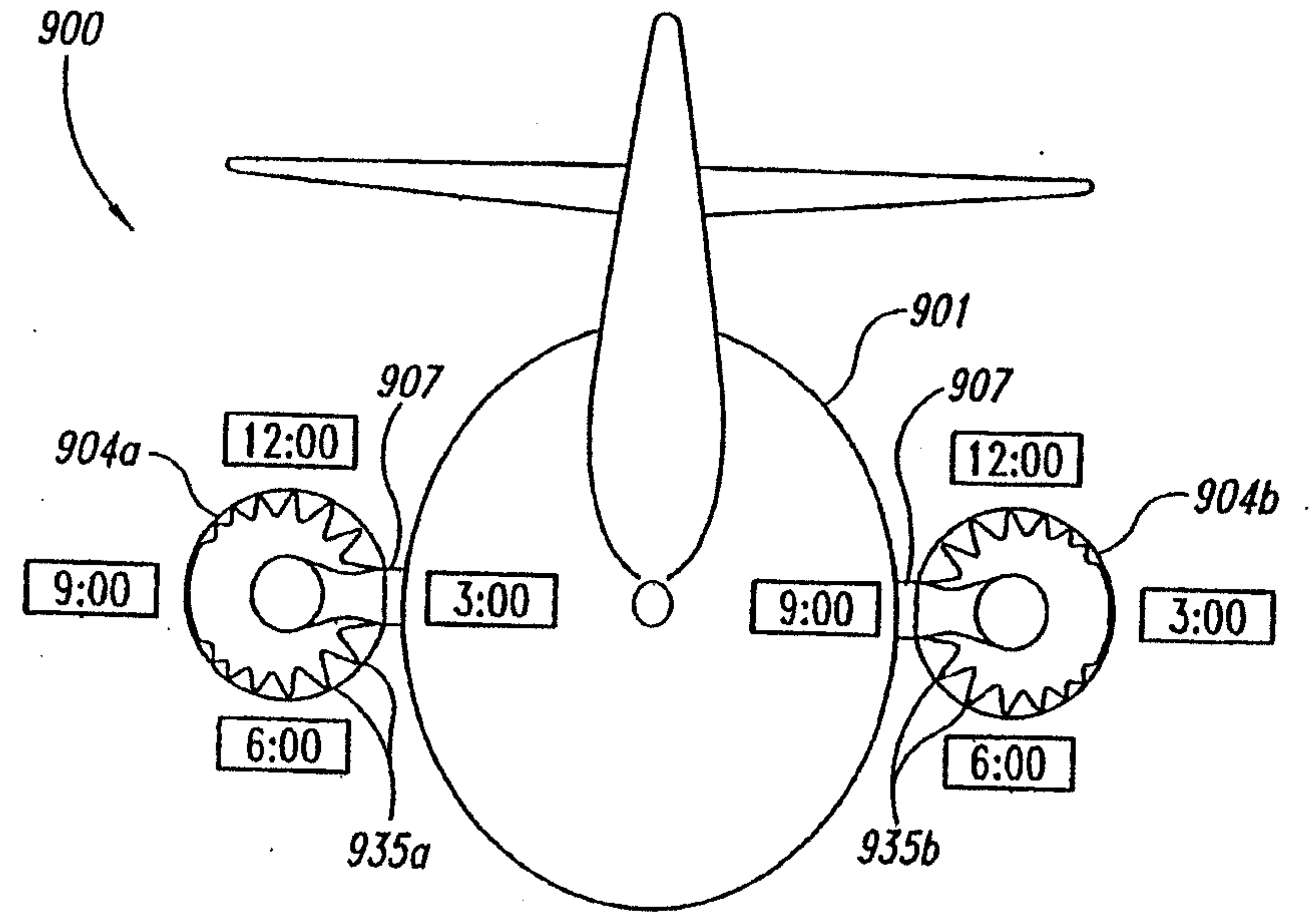


Fig. 9A

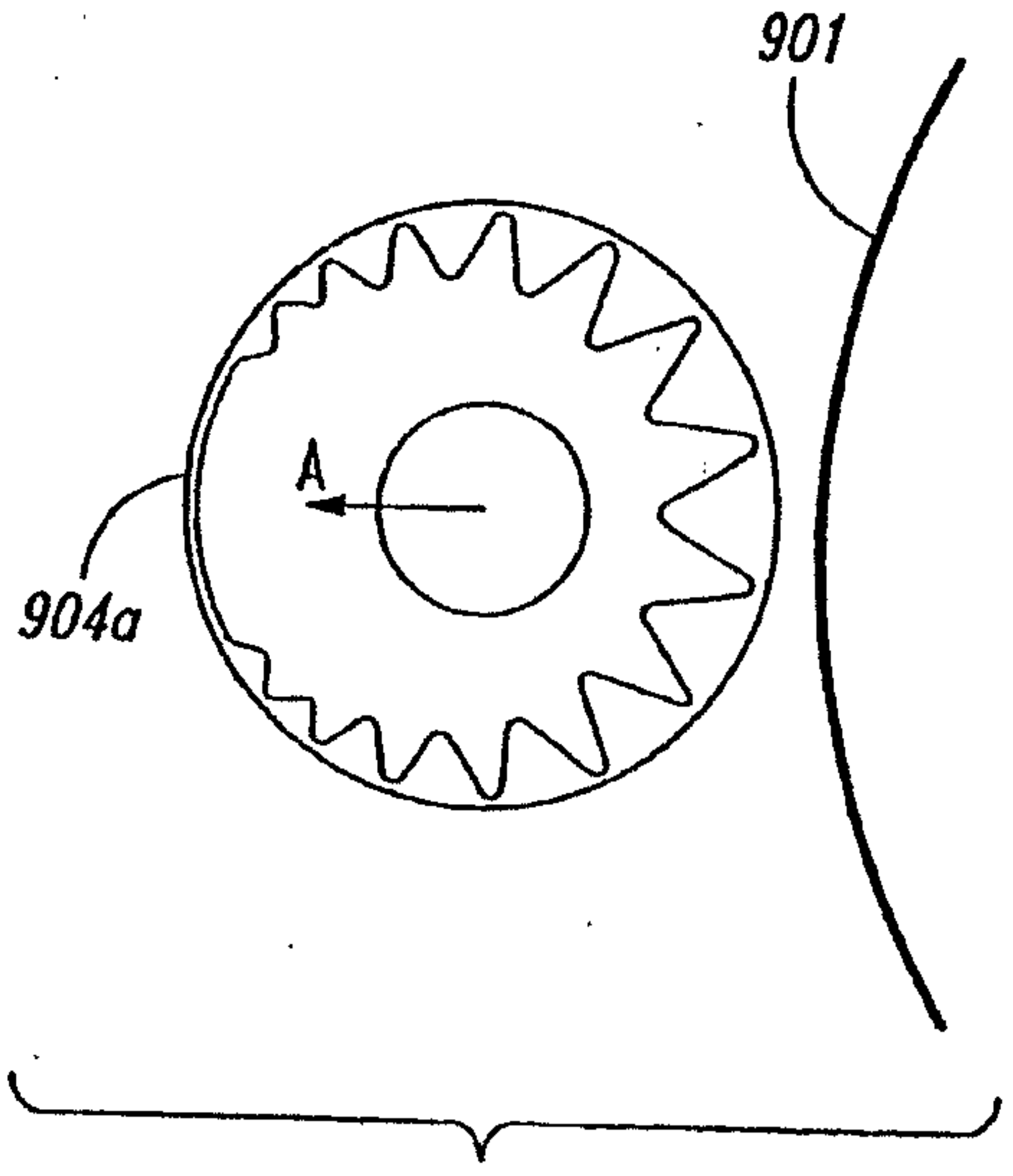


Fig. 9B

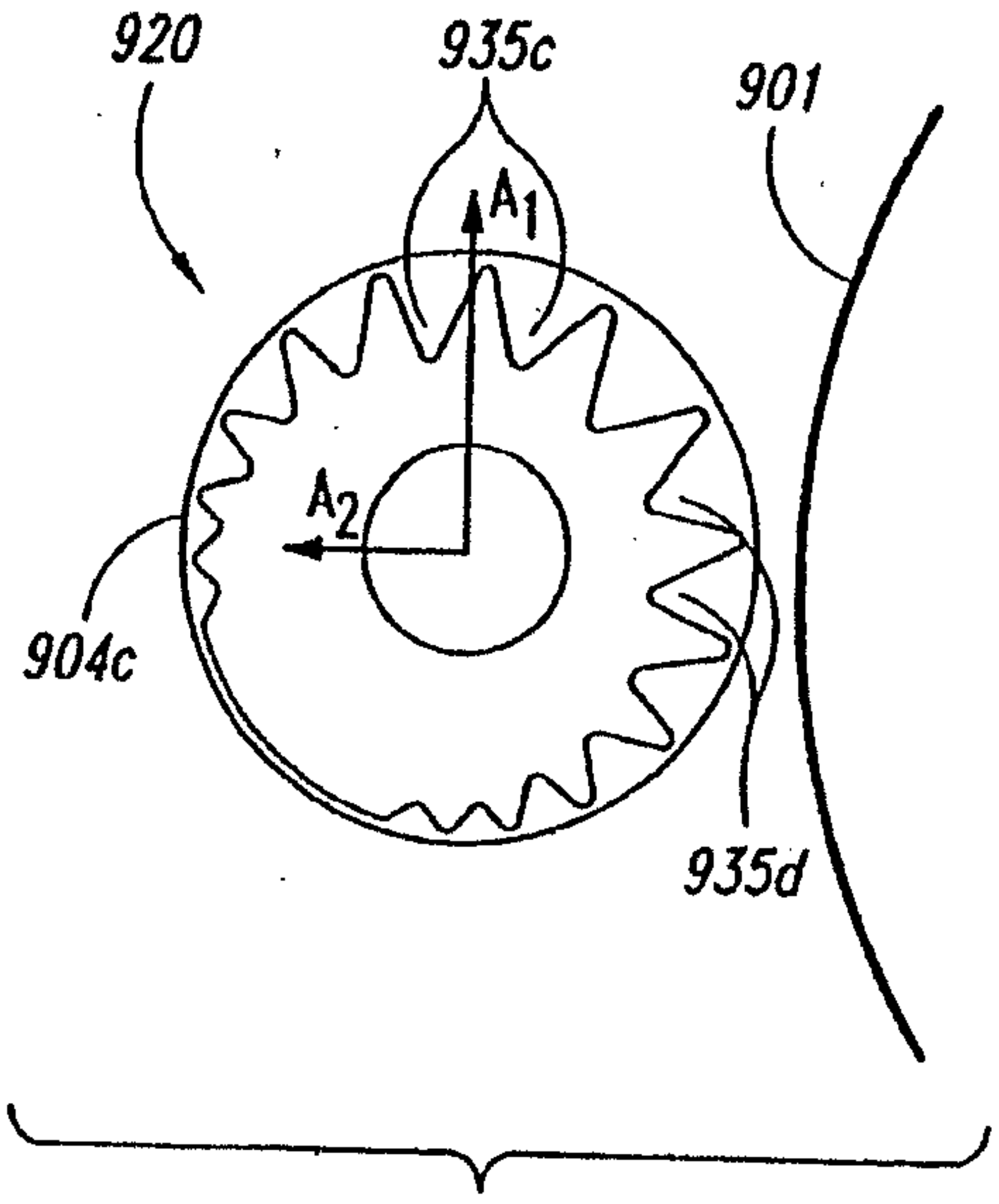


Fig. 9C

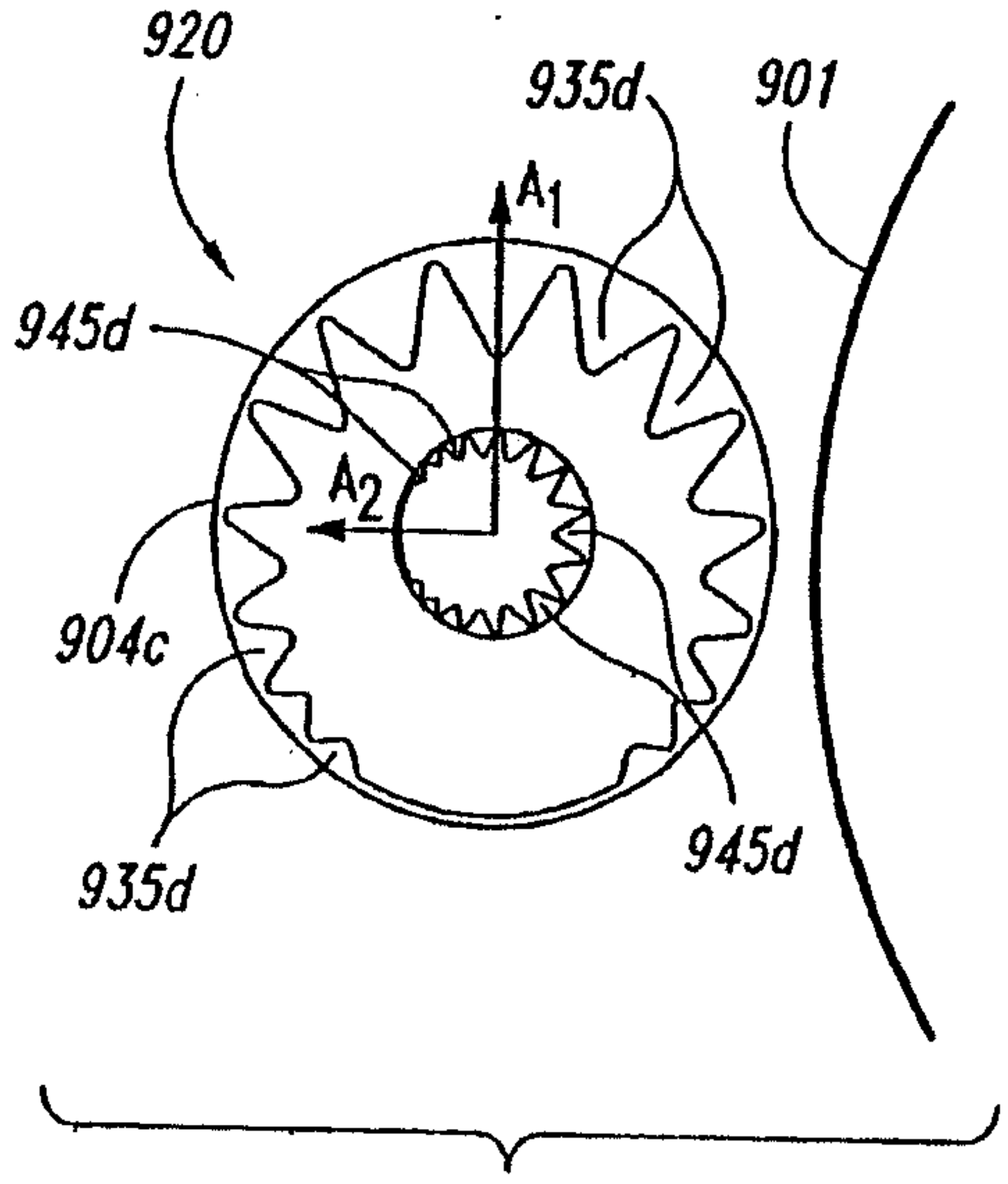


Fig. 9D

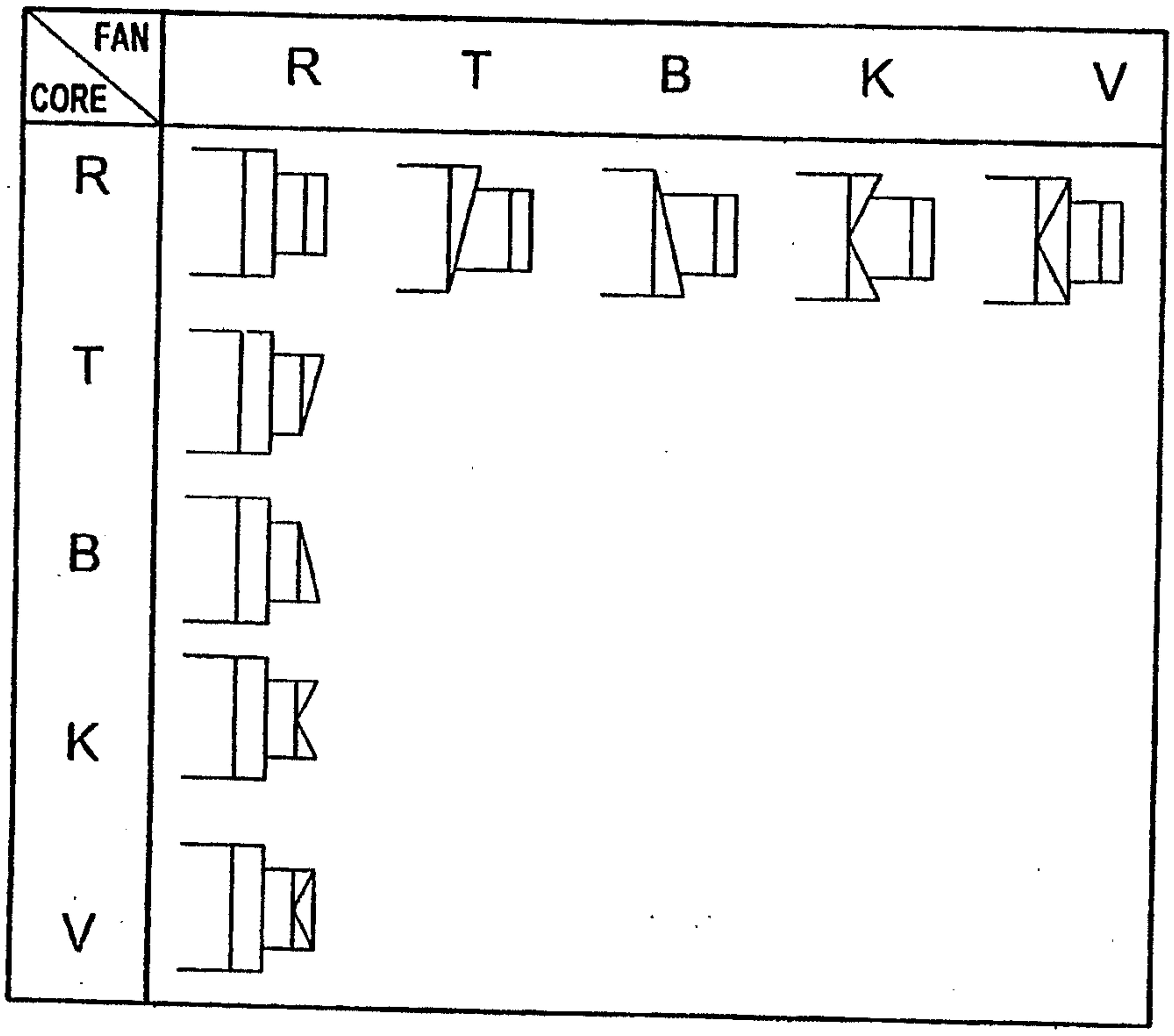


Fig. 10

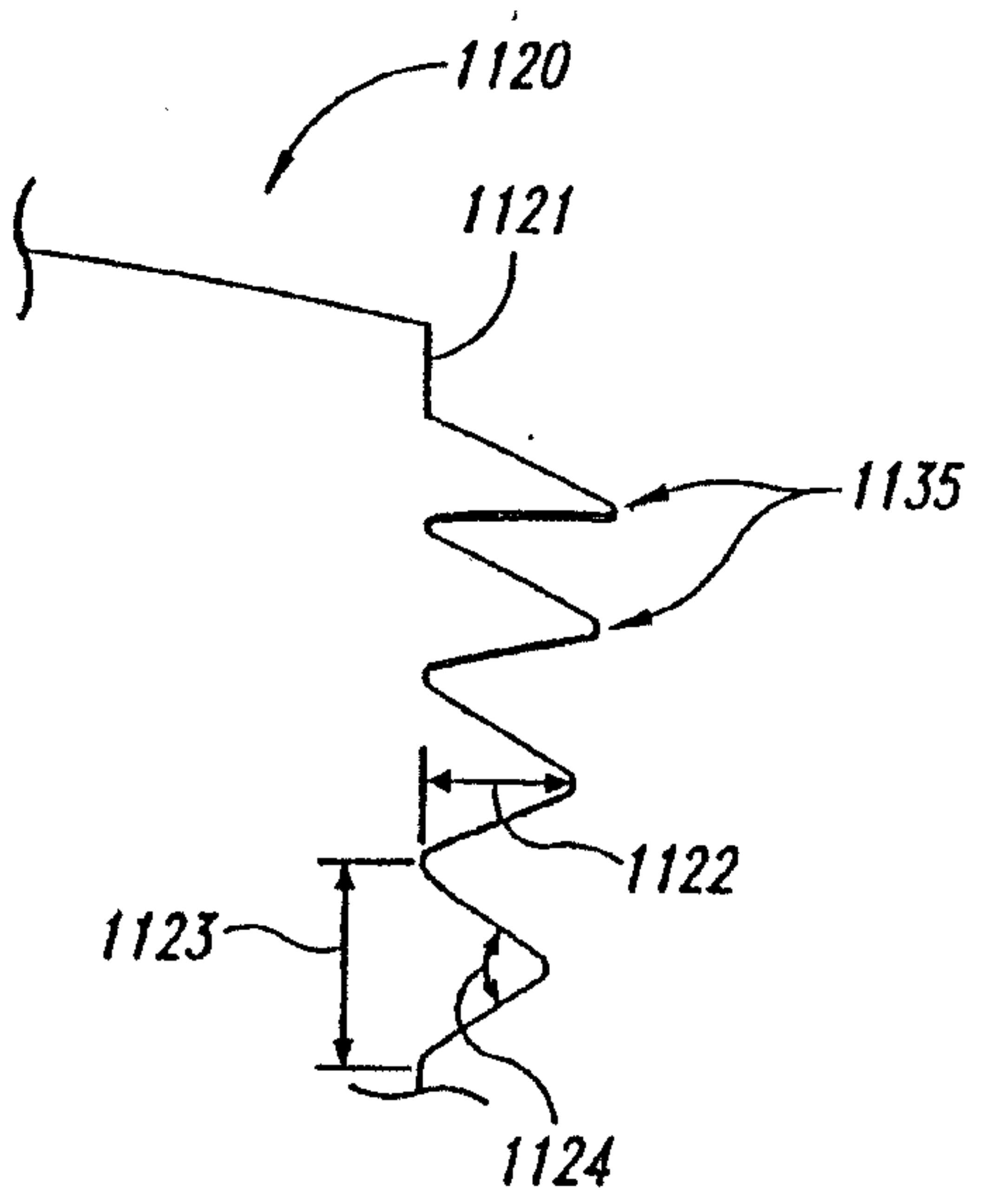


Fig. 11A

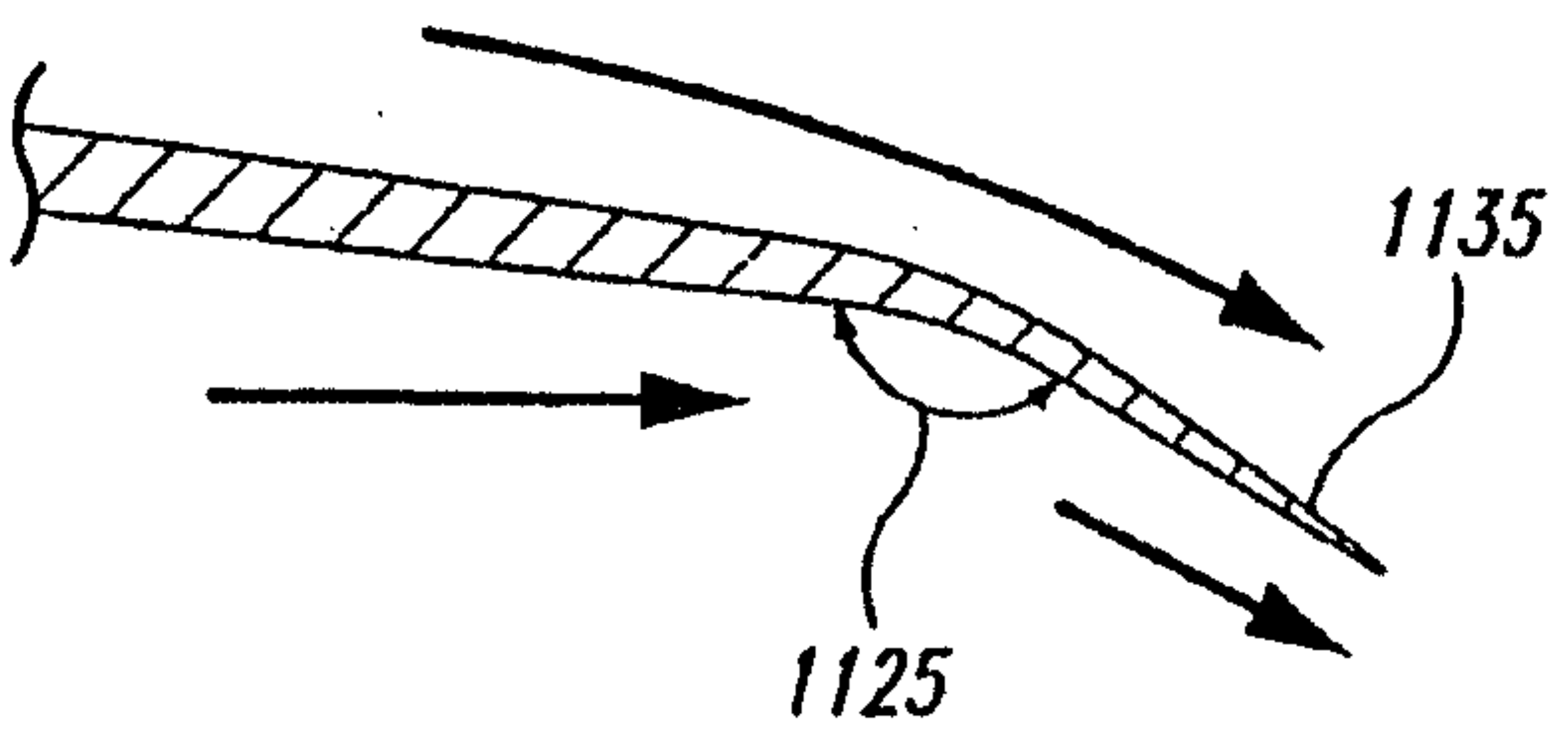


Fig. 11B

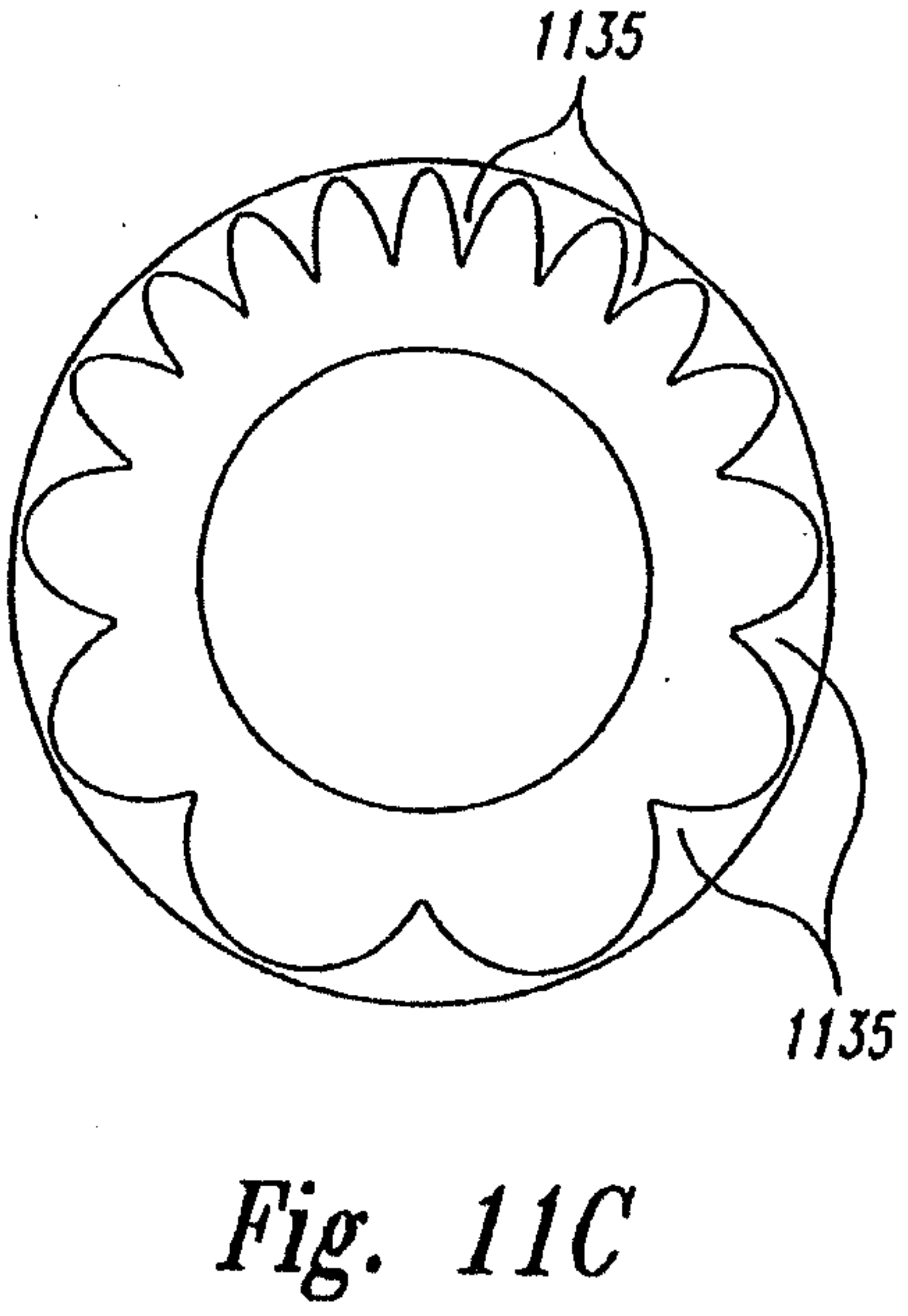


Fig. 11C

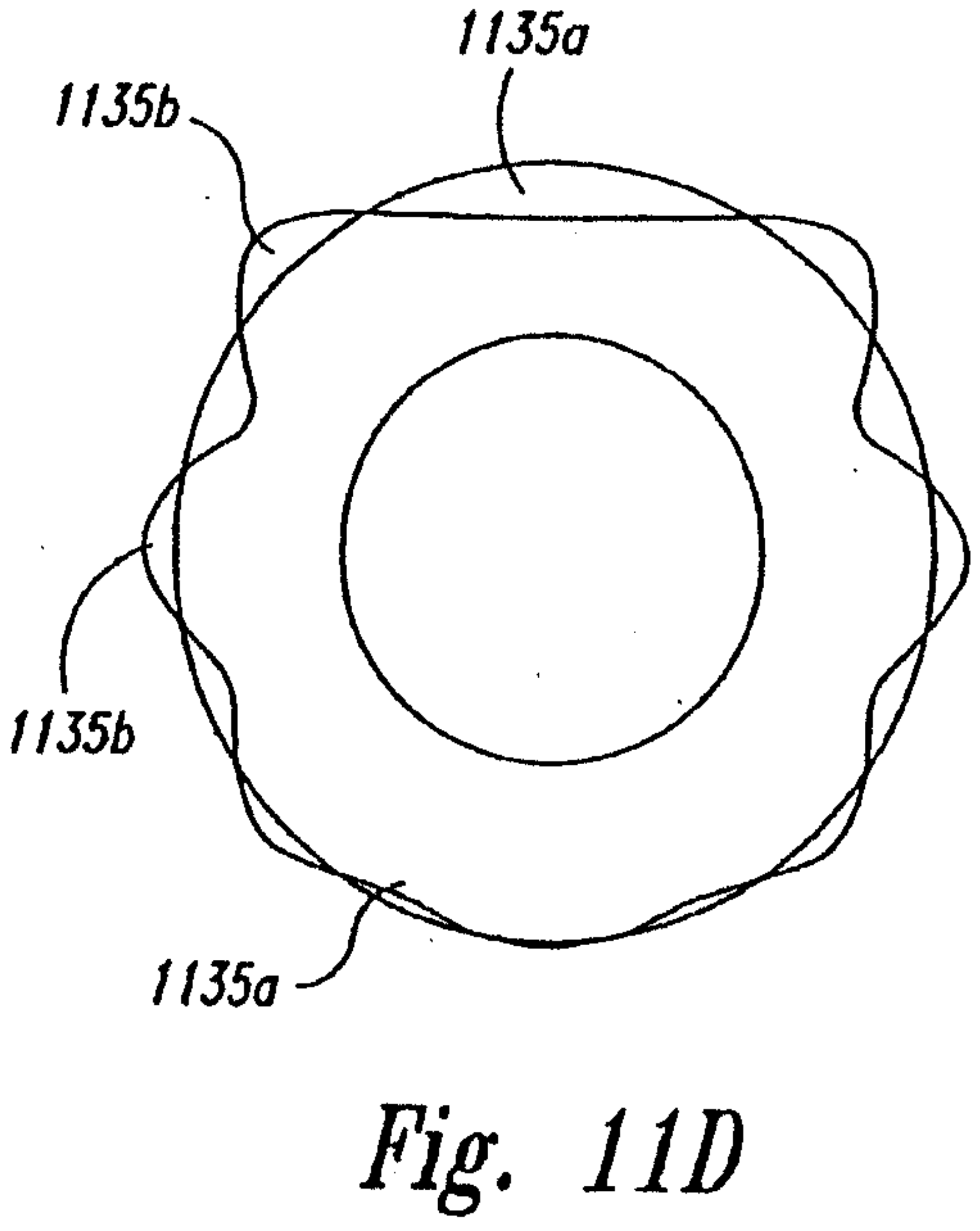
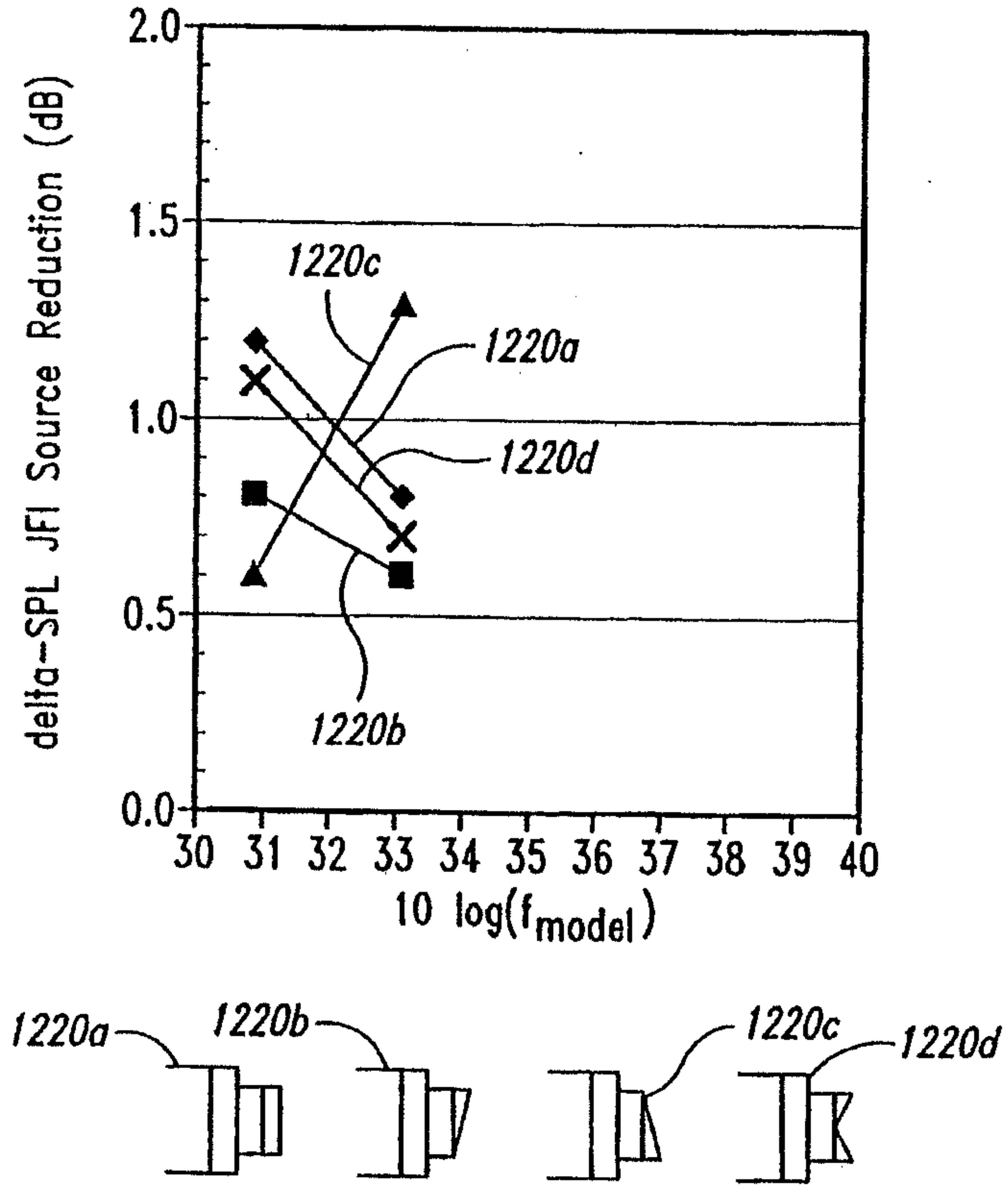


Fig. 11D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delta-SPL JFI Source Reduction (dB)
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Fig. 12

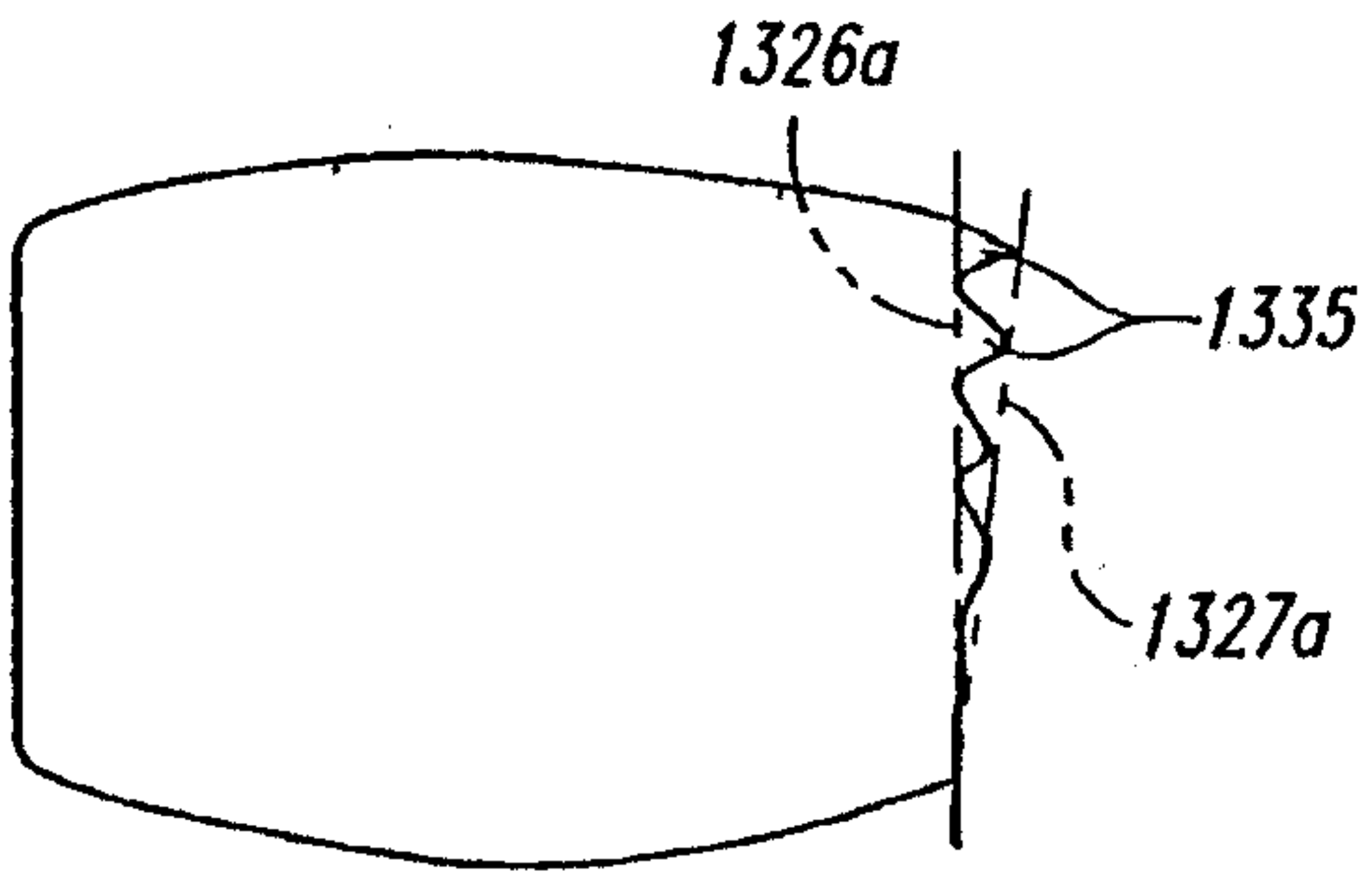


Fig. 13A

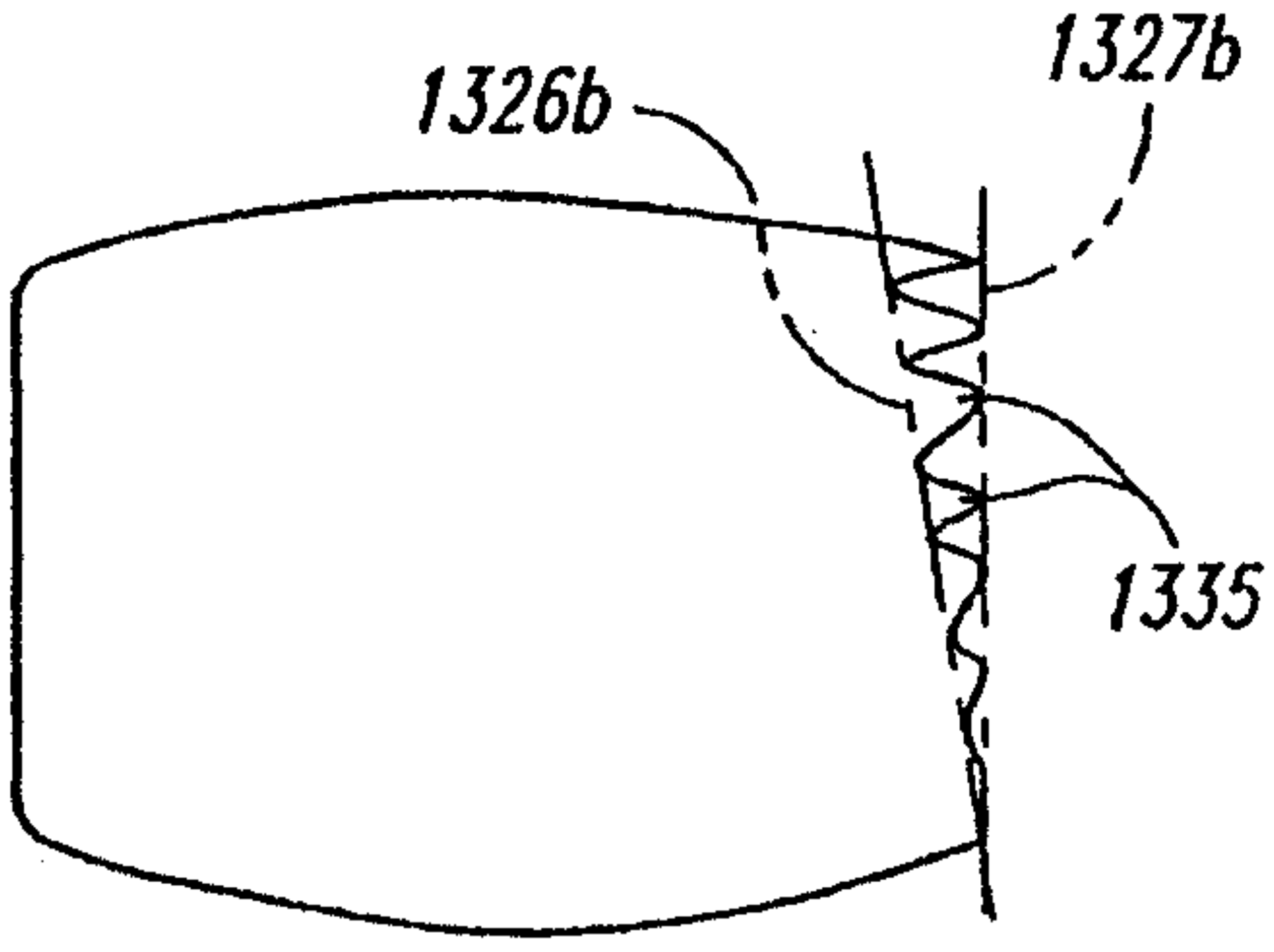


Fig. 13B

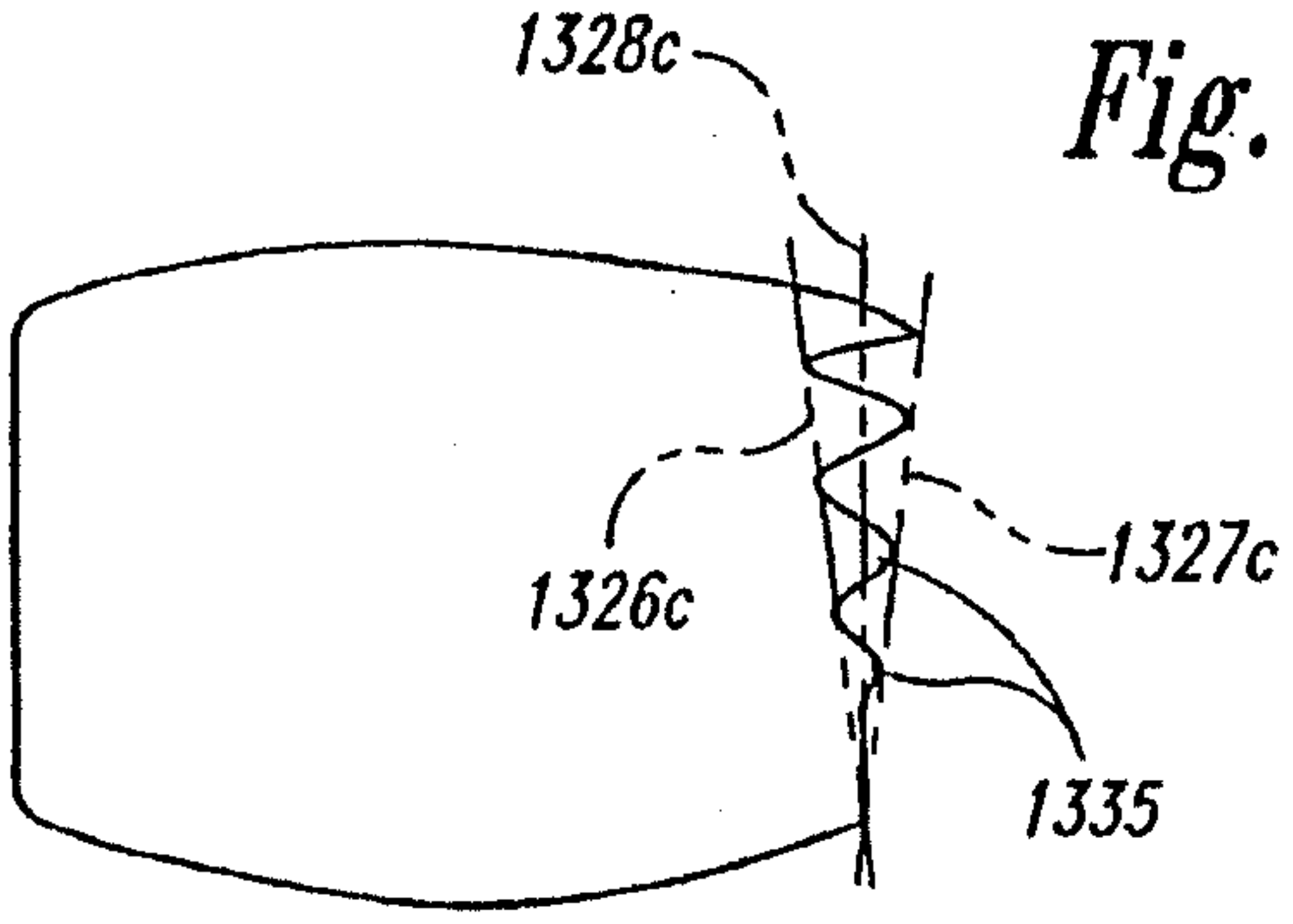


Fig. 13C

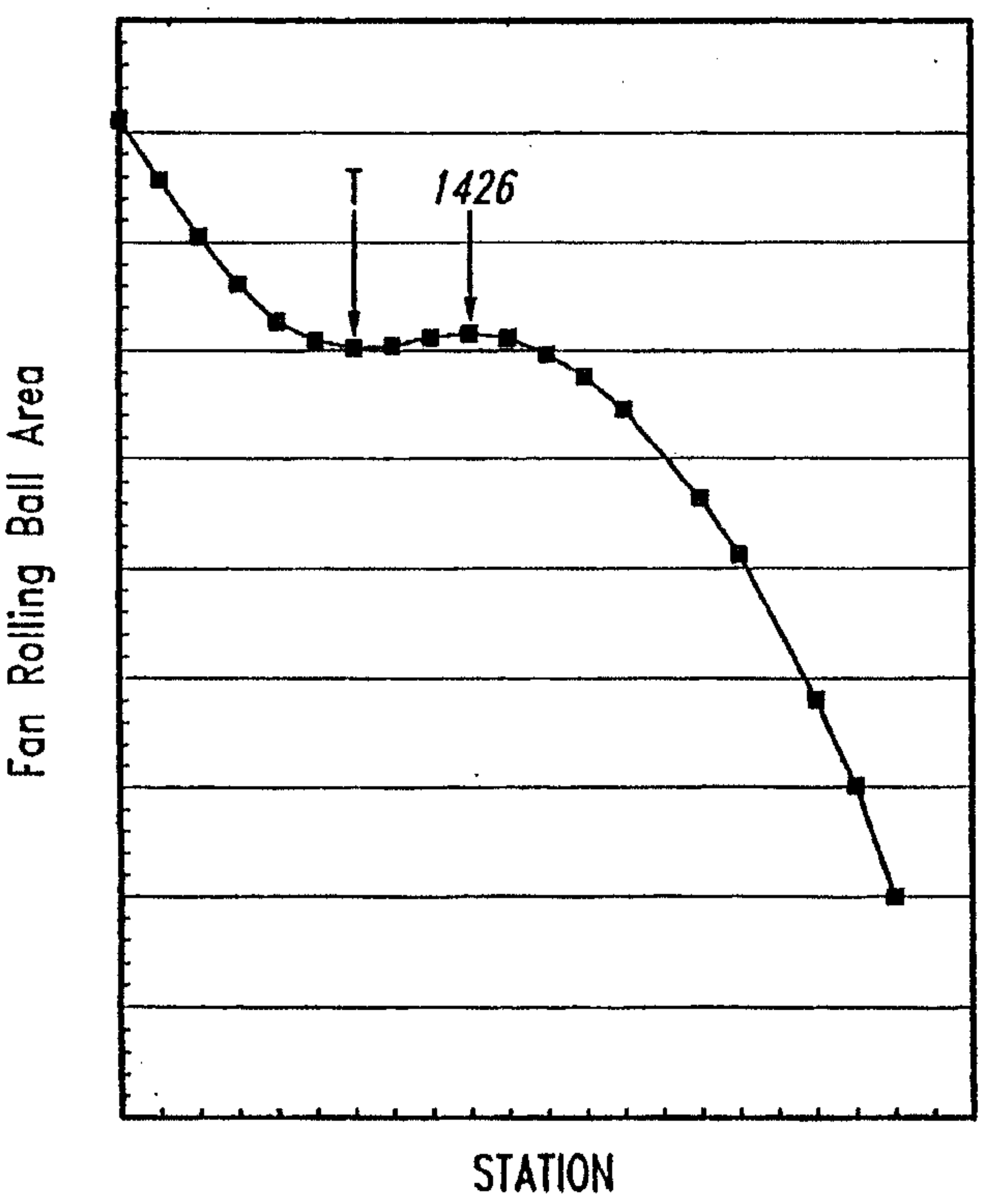
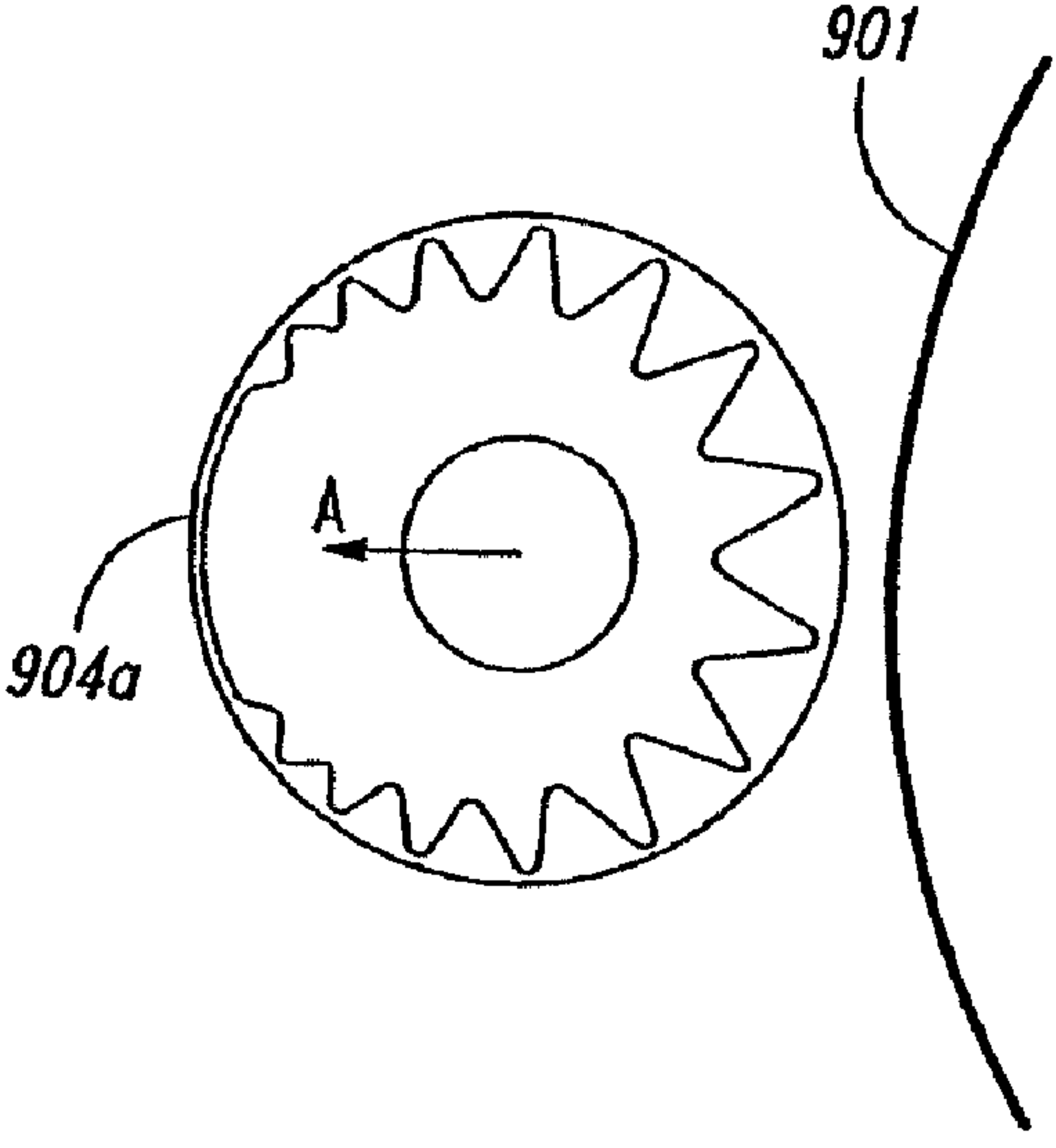


Fig. 14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